





Satellite Navigation

As discussed in module 6, pseudorange (i.e. code phase observations) can be expressed as

$$P_k^p = \rho_k^p + (\Delta t_k - \Delta t^p) \cdot c + \underbrace{I_k^p}_{\text{ionosphere delay}} + \underbrace{A_k^p}_{\text{troposphere delay}}$$
 (8.1)

where ρ_k^p is the true (geometrical) range between receiver k and satellite p. This range is equal to the Euclidian distance computed from the position of the satellite at transmission time t_p and the position of the receiver at the time t the signal was captured.

Dropping now the index k, i.e. focusing on a single receiver, we obtain

$$\rho^p(t,t^p) = \sqrt{[X^p(t^p) - x(t)]^2 + [Y^p(t^p) - y(t)]^2 + [Z^p(t^p) - z(t)]^2} \tag{8.2}$$

which implies that we have to deal with

- three position components of the receiver
- · the receiver clock offset
- the troposphere delay
- · the ionosphere delay

Besides the unknowns mentioned before we also need to know

- the satellite position at transmission time
- the satellite clock offset

This information is basically available from the broadcast ephemeris (see module 2 and others)

Example RINEX Nav-Format (see ftp://igs.org/pub/data/format/rinex210.txt):

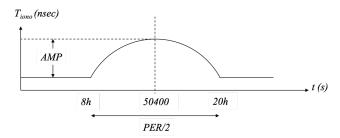
```
2
                   NAVIGATION DATA
                                                          RINEX VERSION / TYPE
CCRINEXN V1 6 0 HX CDDIS
                                       04-AIIG-18 17:31
                                                          PGM / RUN BY / DATE
IGS BROADCAST EPHEMERIS FILE
                                                          COMMENT
    0.4657D-08 0.1490D-07 -0.5960D-07 -0.5960D-07
                                                          ION ALPHA
   0.7782D+05 0.4915D+05 -0.6554D+05 -0.3277D+06
                                                          ION BETA
    0 186264514923D=08 0 888178419700D=14 61440
                                                     2013 DELTA-UTC: AO.A1.T.W
    18
                                                          LEAP SECONDS
                                                          END OF HEADER
 1 18 8 3 0 0 0.0-0.722697004676D-04-0.397903932026D-11 0.000000000000D+00
    0.66000000000D+02 0.84375000000D+00 0.461090634847D-08-0.107690595368D+01
    0.188127160072D-06 0.805073522497D-02 0.469572842121D-05 0.515366210747D+04
    0.43200000000D+06 0.203028321266D-06 0.240613246279D+01 0.223517417908D-07
    0.972082336160D+00 0.292250000000D+03 0.672681593740D+00-0.825677249915D-08
    0.288583449230D-09 0.10000000000D+01 0.20120000000D+04 0.0000000000D+00
    0.2000000000D+01 0.000000000D+00 0.558793544769D-08 0.6600000000D+02
    0.424818000000D+06 0.40000000000D+01 0.000000000D+00 0.000000000D+00
 2 18 8 3 0 0 0.0 0.395588576794D=04=0.112549969344D=10 0.000000000000D+00
    0.9100000000D+02 0.79062500000D+01 0.516057196975D-08-0.759203844230D+00
```

clock af₀, af₁, af₂ satellite orbit (Kepler elements)

(Simple) lonosphere corrections are also transmitted via the navigation message

```
NAVIGATION DATA
                                                     RINEX VERSION / TYPE
CCRINEXN V1.6.0 UX
                 CDDIS
                                   04-AUG-18 17:31
                                                     PGM / RUN BY / DATE
IGS BROADCAST EPHEMERIS FILE
                                                     COMMENT
   ION ALPHA
   0.7782D+05 0.4915D+05 -0.6554D+05 -0.3277D+06
                                                     ION BETA
   0.186264514923D-08 0.888178419700D-14
                                       61440
                                                 2013 DELTA-UTC: AO, A1, T, W
   18
                                                     LEAP SECONDS
```

The coefficients provide the so-called Klobuchar model which can be used to calculate a reasonable value for ionosphere delay T_{iono} in zenith direction.



We can relate T_{iono} to TEC by

$$T_{\text{iono}} \cdot c = 40.28 \frac{TEC}{f_1^2}$$
 (8.3)

where f_1 is the carrier frequency of GPS L1.

The actual computation follows a cookbook scheme

1. Calculate the earth-centered angle (elevation E in semicircles)

$$\psi = \frac{0.0137}{E + 0.11} - 0.022$$
 [semicircles]

2. Compute the latitude of the lonospheric Pierce Point (IPP)

$$\phi_i = \phi + \psi \cos A$$
 [semicircles]

If
$$\phi_i > 0.416$$
 then $\phi_i = 0.416$. If $\phi_i < -0.416$ then $\phi_i = -0.416$.

3. Compute the longitude of the IPP

$$\lambda_i = \lambda + \frac{\psi \sin A}{\cos \phi_i} \quad [\text{semicircles}]$$

4. Find the geomagnetic latitude of the IPP

$$\phi_m = \phi_i + 0.064 \cos(\lambda_i - 1.617)$$
 [semicircles]

5. Find the local time at the IPP

$$t = 43200\lambda_i + t_{\mathsf{GPS}} \quad [\mathsf{s}]$$

If t > 86400 s then subtract 86400 s. If t < 0 s then add 86400 s.

6. Compute the amplitude of ionospheric delay (use the broadcasted coeff.)

$$A_I = \sum_{n=0}^{3} \alpha_n \phi_n^m \quad [\mathbf{s}]$$

If $A_I < 0$ then $A_I = 0$.

7. Compute the period of ionospheric delay (use the broadcasted coeff.)

$$P_I = \sum_{n=0}^3 \beta_n \phi_m^n \quad \text{[s]}$$

If $P_I < 72000$ then $P_I = 72000$.

8. Compute the phase of ionospheric delay

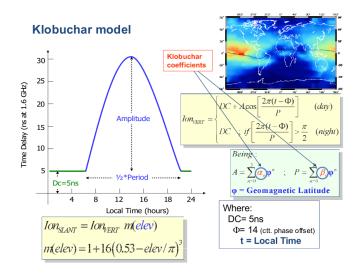
$$X_I = \frac{2\pi(t - 50400)}{P_I} \quad \text{[rad]}$$

9. Compute the slant factor (elevation E in semicircles).

$$F = 1.0 + 1.0(0.53 - E)^3$$

10. Compute the ionospheric time delay

$$T_{\mathsf{iono}} = \left\{ \begin{array}{l} \left[5 \cdot 10^{-9} + A_I \cdot (1 - \frac{X_I^2}{2} + \frac{X_I^4}{24}) \right] \cdot F \, ; \quad |X_I| \leq 1.57 \\ \\ 5 \cdot 10^{-9} \cdot F \, ; \quad |X_I| \geq 1.57 \end{array} \right.$$



The Saastamoinen Hydrostatic Delay Model

The zenith hydrostatic delay (ZHD) (see module 7) after Saastamoinen is given by:

$$ZHD = \frac{0.0022767 \cdot p}{1 - 0.00266 \cos(2\phi) - 0.00028 \cdot h}$$
 (8.4)

where ϕ is the ellipsoidal latitude, h is the height (in km) above the ellipsoid and p is the total pressure.

From where do we get surface pressure p?

- · barometer measurement at the station
- a very rough model, like the one from Berg (1948)

$$p(h) = 1013.25(1 - 0.0226 \cdot h)^{5.225}$$
 h in km

 from empirical models which provide us mean pressure for a certain location (latitude, longitude and height) on a given epoch (hour and day of the year), e.g. see https://link.springer.com/article/10.1007/s00190-007-0135-3

Assuming tropospheric and ionospheric delays are either taken care of or ignored, we can set up a non-linear equation system for N pseudorange observations collected at epoch t:

$$\begin{split} P^1 &= \sqrt{(X^1-x)^2 + (Y^1-y^2)^2 + (Z^1-z)^2} + c\tau - c\tau^1 + \epsilon_1 \\ P^2 &= \sqrt{(X^2-x)^2 + (Y^2-y^2)^2 + (Z^2-z)^2} + c\tau - c\tau^2 + \epsilon_2 \\ & \vdots \\ P^N &= \sqrt{(X^N-x)^2 + (Y^N-y^N)^2 + (Z^N-z)^2} + c\tau - c\tau^N + \epsilon_N \end{split} \tag{8.5}$$

This implies that we have four unknowns (three position components + the receiver clock).

Eq. (8.5) is non-linear in position coordinates (x,y,z) and the terms ϵ_k denote the (random) noise contribution to a pseudorange observation. Linearization of (8.2) of the pseudorange equations yields,

$$P^{k}(x, y, z, \tau) = \underbrace{P^{k}(x_{0}, y_{0}, z_{0}, \tau_{0})}_{P^{k} \text{ computed}} + \underbrace{(x - x_{0})}_{\Delta x} \underbrace{\frac{\partial P^{k}}{\partial x}}_{Dx} + \underbrace{(y - y_{0})}_{\Delta y} \underbrace{\frac{\partial P^{k}}{\partial y}}_{Dx} + \underbrace{(z - z_{0})}_{\Delta z} \underbrace{\frac{\partial P^{k}}{\partial z}}_{Dx} + \underbrace{(\tau - \tau_{0})}_{\Delta \tau} \underbrace{\frac{\partial P^{k}}{\partial z}}_{Dx}$$

$$(8.6)$$

In matrix notation

$$\Delta P^{k} = \begin{pmatrix} \frac{\partial P^{k}}{\partial x} & \frac{\partial P^{k}}{\partial y} & \frac{\partial P^{k}}{\partial z} & \frac{\partial P^{k}}{\partial \tau} \end{pmatrix} \cdot \begin{pmatrix} \frac{\Delta x}{\Delta y} \\ \frac{\Delta z}{\Delta \tau} \end{pmatrix} + \epsilon$$
 (8.8)

Thus, the equation array (8.5) can be written as

$$\Delta P = \underbrace{\left(\begin{array}{cccc} \frac{x_0 - x^1}{\rho^1} & \frac{y_0 - y^1}{\rho^1} & \frac{z_0 - z^1}{\rho^1} & c \\ \frac{x_0 - x^2}{\rho^2} & \frac{y_0 - y^2}{\rho^2} & \frac{z_0 - z^2}{\rho^2} & c \\ \vdots & \vdots & \vdots & \vdots \\ \frac{x_0 - x^N}{\rho^N} & \frac{y_0 - y^N}{\rho^N} & \frac{z_0 - z^N}{\rho^N} & c \end{array} \right)}_{\Delta x} \cdot \underbrace{\left(\begin{array}{c} \Delta x \\ \Delta y \\ \Delta z \\ \Delta \tau \end{array} \right)}_{\epsilon} + \underbrace{\left(\begin{array}{c} \epsilon_1 \\ \epsilon_2 \\ \vdots \\ \epsilon_N \end{array} \right)}_{\epsilon} \tag{8.9}$$

Eq. (8.9) can be used for Single Point Positioning (SPP) or referred to as code phase positioning), i.e.

- an epoch-wise least-squares adjustment, i.e. computing (x,y,z,t) independently if we have N>=4 GNSS pseudorange observations
- a Kalman filter where we model (x, y, z, t) as continuous stochastic processes which evolves over time and is updated by GNSS measurements

As for the ionosphere we can

- ignore its effect → our results will be biased
- significantly reduce its effect with empirical models
- remove its effect by dual-frequency observations (see last lesson)

As for the troposphere we can

- ignore its effect → our results will be slightly biased
- significantly reduce its effect with empirical models
- estimate its effect, by adding an additional unknown parameter that describes troposphere delay in zenith and which gets mapped down to the observation elevation (see last lecture)