

# Buckling Limit State(Ship Rule)

Loading Condition       $\eta := 1.0$

## Material Information

$E := 206000$        $\nu := 0.3$        $f_y := 355$        $Pr := 0.6$

## Panel Information

$l := 3628$        $s := 1340$        $t := 19.0$        $C1 := 1.1$        $C2 := 1.2$   
 $f_L := 12.7$        $f_T := 70.6$        $f_{LT} := 103.0$

$$\left(\frac{f_L}{f_{cL}}\right)^2 + \left(\frac{f_T}{f_{cT}}\right)^2 + \left(\frac{f_{LT}}{f_{cLT}}\right)^2 \leq 1.0$$

Critical Buckling Stress :  $f_{cL}$

$$\left| \begin{array}{l} k_{ix} := 4 \cdot C1 \\ f_{EL} := k_{ix} \cdot \frac{\pi^2 \cdot E}{12 \cdot (1 - \nu^2)} \cdot \left(\frac{t}{s}\right)^2 \\ f_{cL} := \text{if } f_{EL} \leq Pr \cdot f_y \\ \quad f_{EL} \\ \text{else} \\ \quad f_y \cdot \left(1 - Pr \cdot (1 - Pr) \cdot \frac{f_y}{f_{EL}}\right) \end{array} \right|$$

$k_{ix} = 4.4$

$f_{EL} = 164.7$

$f_{cL} = 164.7$

Critical Buckling Stress :  $f_{cT}$

$$\left| \begin{array}{l} k_{iy} := C2 \cdot \left(1 + \left(\frac{s}{l}\right)^2\right)^2 \\ f_{ET} := k_{iy} \cdot \frac{\pi^2 \cdot E}{12 \cdot (1 - \nu^2)} \cdot \left(\frac{t}{s}\right)^2 \\ f_{cT} := \text{if } f_{ET} \leq Pr \cdot f_y \\ \quad f_{ET} \\ \text{else} \\ \quad f_y \cdot \left(1 - Pr \cdot (1 - Pr) \cdot \frac{f_y}{f_{ET}}\right) \end{array} \right|$$

$k_{iy} = 1.55$

$f_{ET} = 58.01$

$f_{cT} = 58.01$

Critical Buckling Stress :  $f_{cLT}$

$$\left| \begin{array}{l} f_t := \frac{1}{\sqrt{3}} \cdot f_y \\ k_s := \left(4.0 \cdot \left(\frac{s}{l}\right)^2 + 5.34\right) \cdot C1 \\ f_{ELT} := k_s \cdot \frac{\pi^2 \cdot E}{12 \cdot (1 - \nu^2)} \cdot \left(\frac{t}{s}\right)^2 \\ f_{cLT} := \text{if } f_{ELT} \leq Pr \cdot f_t \\ \quad f_{ELT} \\ \text{else} \\ \quad f_t \cdot \left(1 - Pr \cdot (1 - Pr) \cdot \frac{f_t}{f_{ELT}}\right) \end{array} \right|$$

$k_s = 6.474$

$f_{ELT} = 242.343$

$f_{cLT} = 163.357$

$$\left(\frac{f_L}{f_{cL}}\right)^2 + \left(\frac{f_T}{f_{cT}}\right)^2 + \left(\frac{f_{LT}}{f_{cLT}}\right)^2 = 1.885$$

# Ultimate Strength under Combined in-Plane Stresses

## Material Information

$$E := 206000 \quad \nu := 0.3 \quad f_y := 355 \quad Pr := 0.6 \quad S_m := 0.908$$

## Panel Information

$$l := 3628 \quad s := 1340 \quad t := 19.0 \quad C1 := 1.1 \quad C2 := 1.2$$

$$f_L := 12.7 \quad f_T := 70.6 \quad f_{LT} := 103.0$$

$$\left( \frac{f_L}{f_{uL}} \right)^2 + \left( \frac{f_T}{f_{uT}} \right)^2 - \eta \left( \frac{f_L}{f_{uL}} \right) \cdot \left( \frac{f_T}{f_{uT}} \right) + \left( \frac{f_{LT}}{f_{uLT}} \right)^2 \leq S_m$$

slenderness ratio  $\beta := \frac{s}{t} \cdot \sqrt{\frac{f_y}{E}} = 2.928$

coefficient to reflect interaction between longitudinal and transverse stresses  $\eta := 1.5 - \frac{\beta}{2} = 0.036$

$$C := \begin{cases} \frac{2.25}{\beta} - \frac{1.25}{\beta^2} & \text{if } \beta \geq 1.25 \\ 1.0 & \text{else} \end{cases} \quad b_{wLs} := C \quad b_{wTl} := C \cdot \frac{s}{l} + 0.115 \cdot \left( 1 - \frac{s}{l} \right) \cdot \left( 1 + \frac{1}{\beta^2} \right)^2$$

$$f_{uL} := f_y \cdot b_{wLs} \quad f_{uT} := f_y \cdot b_{wTl}$$

$$\alpha := \frac{1}{s}$$

$$f_{uLT} := f_{cLT} + 0.5 \cdot \frac{(f_y - \sqrt{3} \cdot f_{cLT})}{\sqrt{1 + \alpha + \alpha^2}}$$

$$f_{uL} = 221.052$$

$$f_{uT} = 113.75$$

$$f_{uLT} = 174.202$$

$$\left( \frac{f_L}{f_{uL}} \right)^2 + \left( \frac{f_{LT}}{f_{uLT}} \right)^2 = 0.353$$

$$\left( \frac{f_T}{f_{uT}} \right)^2 + \left( \frac{f_{LT}}{f_{uLT}} \right)^2 = 0.735$$

$$\left( \frac{f_L}{f_{uL}} \right)^2 + \left( \frac{f_T}{f_{uT}} \right)^2 - \eta \cdot \left( \frac{f_L}{f_{uL}} \right) \cdot \left( \frac{f_T}{f_{uT}} \right) + \left( \frac{f_{LT}}{f_{uLT}} \right)^2 = 0.737$$