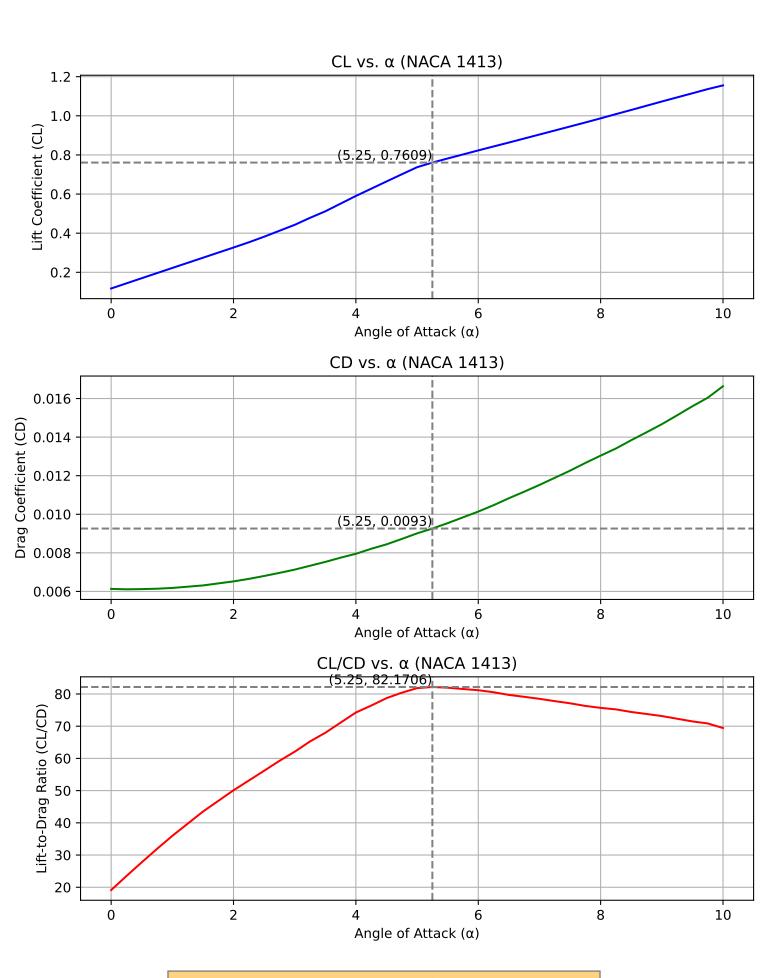
Airfoil Analysis Report

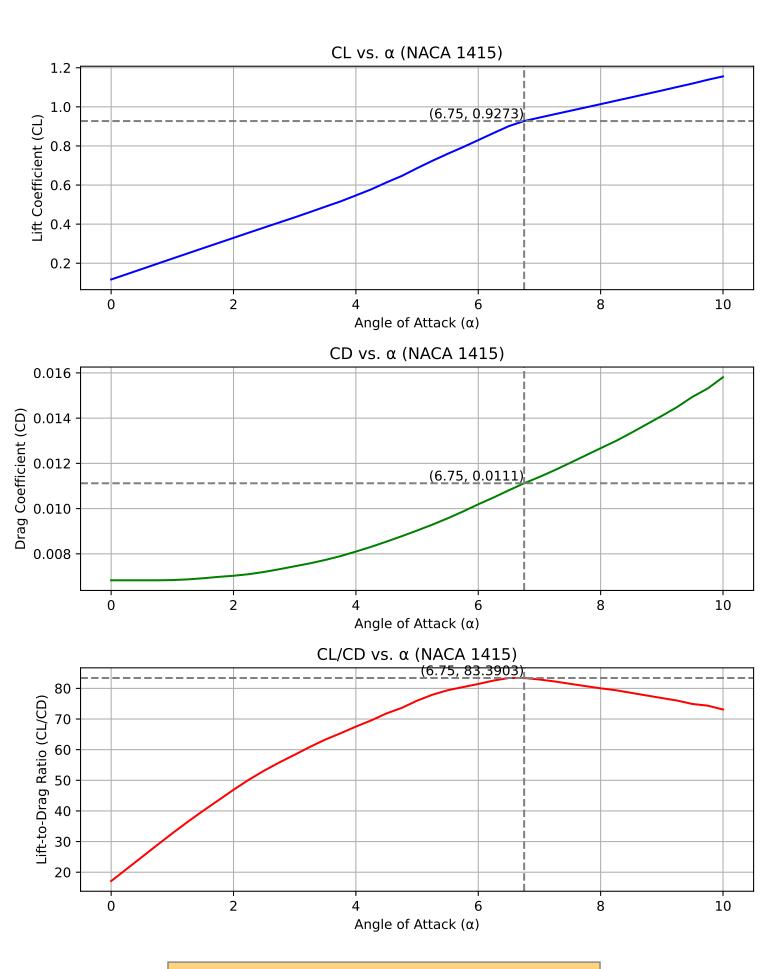
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NACA 1413, Re = 6.850e + 05



NACA 1415, Re = 6.850e + 05



NACA 1419, Re = 1.301e + 06

