Rotation Notation/Convention

$$\begin{Bmatrix} x \\ y \\ z \end{Bmatrix} = [\Lambda] \begin{Bmatrix} X \\ Y \\ Z \end{Bmatrix} = \begin{bmatrix} \hat{x}^T \\ \hat{y}^T \\ \hat{z}^T \end{bmatrix} \begin{Bmatrix} X \\ Y \\ Z \end{Bmatrix} \text{ (from global to local)}$$

or equivalently:

$$\begin{Bmatrix} X \\ Y \\ Z \end{Bmatrix} = [\Lambda]^T \begin{Bmatrix} x \\ y \\ z \end{Bmatrix} == \begin{bmatrix} \hat{x} & \hat{y} & \hat{z} \end{bmatrix} \begin{Bmatrix} x \\ y \\ z \end{Bmatrix}$$

where *X*/*Y*/*Z* are global coordinates, *x*/*y*/*z* are local coordinates, Λ is the DCM from global to local, and $\hat{x}\,/\,\hat{y}\,/\,\hat{z}$ are the unit vectors of the local coordinate system expressed in the global coordinate system.

$$\begin{Bmatrix} \theta_x \\ \theta_y \\ \theta_z \end{Bmatrix} = F^{EulerExtract}\left(\left[\Lambda\left(\theta_x, \theta_y, \theta_z\right)\right]\right)$$

where function $F^{EulerExtract}(\;)$ returns the 3 Euler angles of the *x-y-z* (1-2-3) rotation sequence used to form Λ (that is, first a rotation θ_x about the global *X* axis, followed by rotation θ_y about the *Y'* axis, followed by rotation θ_z about the *Z''* axis) defined as follows:

$$\Lambda\left(\theta_x, \theta_y, \theta_z\right) = \begin{bmatrix} COS(\theta_z) & SIN(\theta_z) & 0 \\ -SIN(\theta_z) & COS(\theta_z) & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} COS(\theta_y) & 0 & -SIN(\theta_y) \\ 0 & 1 & 0 \\ SIN(\theta_y) & 0 & COS(\theta_y) \end{bmatrix} \begin{bmatrix} 1 & 0 & 0 \\ 0 & COS(\theta_x) & SIN(\theta_x) \\ 0 & -SIN(\theta_x) & COS(\theta_x) \end{bmatrix}$$

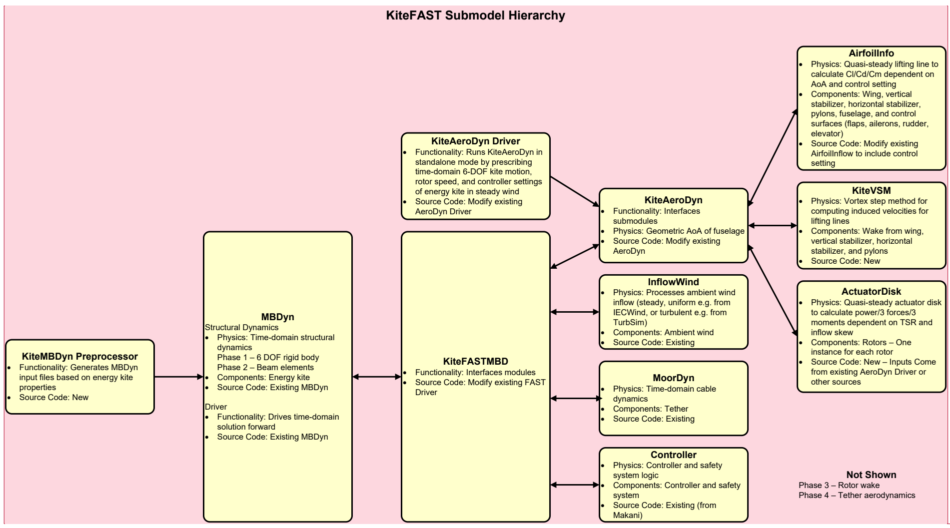
$$= \begin{bmatrix} COS(\theta_y)COS(\theta_z) & COS(\theta_x)SIN(\theta_z) + SIN(\theta_x)SIN(\theta_y)COS(\theta_z) & SIN(\theta_x)SIN(\theta_z) - COS(\theta_x)SIN(\theta_y)COS(\theta_z) \\ -COS(\theta_y)SIN(\theta_z) & COS(\theta_x)COS(\theta_z) - SIN(\theta_x)SIN(\theta_y)SIN(\theta_z) & SIN(\theta_x)COS(\theta_z) + COS(\theta_x)SIN(\theta_y)SIN(\theta_z) \\ SIN(\theta_y) & -SIN(\theta_x)COS(\theta_y) & COS(\theta_x)COS(\theta_y) \end{bmatrix}$$

Note the following simplifications:

$$\Lambda\left(0, \theta_y, \theta_z\right) = \begin{bmatrix} COS(\theta_y)COS(\theta_z) & SIN(\theta_z) & -SIN(\theta_y)COS(\theta_z) \\ -COS(\theta_y)SIN(\theta_z) & COS(\theta_z) & SIN(\theta_y)SIN(\theta_z) \\ SIN(\theta_y) & 0 & COS(\theta_y) \end{bmatrix}$$

$$\Lambda\left(\theta_x, 0, \theta_z\right) = \begin{bmatrix} COS(\theta_z) & COS(\theta_x)SIN(\theta_z) & SIN(\theta_x)SIN(\theta_z) \\ -SIN(\theta_z) & COS(\theta_x)COS(\theta_z) & SIN(\theta_x)COS(\theta_z) \\ 0 & -SIN(\theta_x) & COS(\theta_x) \end{bmatrix}$$

$$\Lambda\left(\theta_x, \theta_y, 0\right) = \begin{bmatrix} COS(\theta_y) & SIN(\theta_x)SIN(\theta_y) & -COS(\theta_x)SIN(\theta_y) \\ 0 & COS(\theta_x) & SIN(\theta_x) \\ SIN(\theta_y) & -SIN(\theta_x)COS(\theta_y) & COS(\theta_x)COS(\theta_y) \end{bmatrix}$$



Commented [JJ1]: We've split up KiteFASTMBD into KiteFASTMBD in C and KiteFASTMBD in Fortran. This plan is for KiteFASTMBD in Fortran.

KiteFASTMBD

Inputs	Outputs	States	Parameters
• ${}^{MBD}\vec{p}^{Wind}$ – Position of the ground station where the fixed wind measurement is taken (m) • ${}^{MBD}\vec{p}^{FusO}$ – Position (origin) of the fuselage (m) • ${}^{MBD}\Lambda^{FusO}$ – Rotation (absolute orientation) of the fuselage origin (-) • ${}^{MBD}\vec{v}^{FusO}$ – Translational velocity (absolute) of the fuselage origin (m/s) • ${}^{MBD}\vec{\omega}^{FusO}$ – Rotational velocity (absolute) of the fuselage origin (rad/s) • ${}^{MBD}\vec{a}^{FusO}$ – Translational acceleration (absolute) of the fuselage origin (m/s^2) • ${}^{MBD}\vec{\alpha}^{FusO}$ – Rotational acceleration (absolute) of the fuselage origin (rad/s^2) • ${}^{MBD}\vec{p}_j^{Fus}$ – Translational position (absolute) of the j^{th} node of the fuselage mesh (m)	• ${}^{MBD}\vec{F}_j^{Fus}$ – Aerodynamic applied concentrated forces at the j^{th} node of the fuselage mesh (N) • ${}^{MBD}\vec{M}_j^{Fus}$ – Aerodynamic applied concentrated moments at the j^{th} node of the fuselage mesh (N-m) • ${}^{MBD}\vec{F}_j^{SWn}$ – Aerodynamic and tether applied concentrated forces at the j^{th} node of the starboard wing mesh (N) • ${}^{MBD}\vec{M}_j^{SWn}$ – Aerodynamic applied and tether concentrated moments at the j^{th} node of the starboard wing mesh (N-m) • ${}^{MBD}\vec{F}_j^{PWn}$ – Aerodynamic and tether applied concentrated forces at the j^{th} node of the port wing	• $NewTime$ – Is this a new time step (in order to only call the controller once per step)? (flag) (other state) • ${}^{MD}x$ – MoorDyn continuous states (varied) • ${}^{KAD}z$ – KiteAeroDyn constraint states (varied)	• Δt – Time step (s) • N_{Flaps} – Number of flaps per wing (-) • N_{Pylons} – Number of pylons per wing (-) • $\Lambda^{FAST2Ctrl}$ – DCM conversion from the FAST ground system (X pointed nominally downwind; Z pointed vertically opposite gravity; Y transverse) to the ground system used by the controller (X pointed nominally upwind; Z pointed vertically downward, Y transverse) (-) • ${}^{MBD}\vec{g}$ – Gravity vector expressed in the global inertial-frame coordinate system (m/s^2) • ρ – Air density (kg/m^3) • $\vec{p}^{Anch}$ – Position of the ground station where the tether attaches (i.e. mooring line anchor) (m)

Commented [JJ2]: These are the data queried from the MBDyn model (through GetXPrev at t and GetXCur at t+dt) to be used within KiteFASTMBD.

Commented [JJ3]: These are the data sent to the MBDyn model from KiteFASTMBD.

Commented [JJ4]: Obvious parameters are not listed here.

Commented [JJ5]: I'm currently assuming that MBD is marching in time at 100 Hz (dt = 0.01 s), which is what the controller is running at. Changing the MBD time step will require a modification to this plan to ensure that the controller is called at 100 Hz.

- ${}^{MBD}\Lambda_j^{Fus}$ – Displaced rotation (absolute orientation) of the j th node of the fuselage mesh (-)
- ${}^{MBD}\vec{v}_j^{Fus}$ – Translational velocity (absolute) of the j th node of the fuselage mesh (m/s)
- ${}^{MBD}\vec{\omega}_j^{Fus}$ – Rotational velocity (absolute) of the j th node of the fuselage mesh (rad/s)
- ${}^{MBD}\vec{a}_j^{Fus}$ – Translational acceleration (absolute) of the j th node of the fuselage mesh (m/s^2)
- ${}^{MBD}\vec{FR}_j^{Fus}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the fuselage mesh (N)
- ${}^{MBD}\vec{MR}_j^{Fus}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the fuselage mesh (N-m)
- ${}^{MBD}\vec{p}^{SWnO}$ – Position (origin) of the starboard wing (m)
- ${}^{MBD}\vec{p}_j^{SWn}$ – Translational position (absolute) of the j th node of the starboard wing (m)
- ${}^{MBD}\Lambda_j^{SWn}$ – Displaced rotation (absolute orientation) of the j th node of the starboard wing mesh (-)
- ${}^{MBD}\vec{v}_j^{SWn}$ – Translational velocity (absolute) of the j th node of the starboard wing mesh (m/s)
- ${}^{MBD}\vec{\omega}_j^{SWn}$ – Rotational velocity (absolute) of the j th node of the starboard wing mesh (rad/s)

mesh (N)

- ${}^{MBD}\vec{M}_j^{PWn}$ – Aerodynamic and tether applied concentrated moments at the j th node of the port wing mesh (N-m)
- ${}^{MBD}\vec{F}_j^{VS}$ – Aerodynamic applied concentrated forces at the j th node of the vertical stabilizer mesh (N)
- ${}^{MBD}\vec{M}_j^{VS}$ – Aerodynamic applied concentrated moments at the j th node of the vertical stabilizer mesh (N-m)
- ${}^{MBD}\vec{F}_j^{SHS}$ – Aerodynamic applied concentrated forces at the j th node of the starboard horizontal stabilizer mesh (N)
- ${}^{MBD}\vec{M}_j^{SHS}$ – Aerodynamic applied concentrated moments at the j th node of the starboard horizontal stabilizer mesh (N-m)
- ${}^{MBD}\vec{F}_j^{PHS}$ – Aerodynamic applied concentrated forces at the j th node of the port horizontal stabilizer mesh (N)
- ${}^{MBD}\vec{M}_j^{PHS}$ – Aerodynamic applied concentrated moments at the j th node of the port horizontal stabilizer mesh (N-m)
- ${}^{MBD}\vec{F}_j^{SPy}\left[n_{Pylons}\right]$ – Aerodynamic applied concentrated forces at the j th node of the pylons on the starboard wing mesh (N)
- ${}^{MBD}\vec{M}_j^{SPy}\left[n_{Pylons}\right]$ –

- ${}^{MBD}m^{SPyRtr}\left[n_{Pylons}, n_2\right]$ – Mass of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh (kg)
- ${}^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons}, n_2\right]$ – Rotational inertia about the shaft axis of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh (kg·m^2)
- ${}^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons}, n_2\right]$ – Transverse inertia about the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh (kg·m^2)
- ${}^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons}, n_2\right]$ – Distance along the shaft from the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh to the center of mass of the rotor/drivetrain (positive along positive x) (m)
- ${}^{MBD}m^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Mass of the top and bottom rotors/drivetrains on the pylons on the port wing mesh (kg)
- ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Rotational inertia about the shaft axis of the top and bottom rotors/drivetrains on the pylons on the port wing mesh (kg·m^2)
- ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Transverse inertia about the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the port wing mesh (kg·m^2)

• $^{MBD}\vec{a}_j^{SWn}$ – Translational acceleration (absolute) of the j th node of the starboard wing mesh (m/s^2) • $^{MBD}\vec{FR}_j^{SWn}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the starboard wing mesh (N) • $^{MBD}\vec{MR}_j^{SWn}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the starboard wing mesh (N-m) • $^{MBD}\vec{p}^{PWnO}$ – Position (origin) of the port wing (m) • $^{MBD}\vec{p}_j^{PWn}$ – Translational position (absolute) of the j th node of the port wing mesh (m) • $^{MBD}\Lambda_j^{PWn}$ – Displaced rotation (absolute orientation) of the j th node of the port wing mesh (-) • $^{MBD}\vec{v}_j^{PWn}$ – Translational velocity (absolute) of the j th node of the port wing mesh (m/s) • $^{MBD}\vec{\omega}_j^{PWn}$ – Rotational velocity (absolute) of the j th node of the port wing mesh (rad/s) • $^{MBD}\vec{a}_j^{PWn}$ – Translational acceleration (absolute) of the j th node of the port wing mesh (m/s^2) • $^{MBD}\vec{FR}_j^{PWn}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the port wing mesh (N) • $^{MBD}\vec{MR}_j^{PWn}$ – Reaction moment (expressed in the	Aerodynamic applied concentrated moments at the j th node of pylons on the starboard wing mesh (N-m) • $^{MBD}\vec{F}_j^{PPy}\left[n_{Pylons}\right]$ – Aerodynamic applied concentrated forces at the j th node of the pylons on the port wing mesh (N) • $^{MBD}\vec{M}_j^{PPy}\left[n_{Pylons}\right]$ – Aerodynamic applied concentrated moments at the j th node of pylons on the port wing mesh (N-m) • $^{MBD}\vec{F}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction forces at the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (N) • $^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction moments at the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (N-m) • $^{MBD}\vec{F}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction forces at the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (N) • $^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction moments at the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (N-m)		• $^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Distance along the shaft from the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the port wing mesh to the center of mass of the rotor/drivetrain (positive along positive x) (m) • $^{MBD}\vec{p}_j^{FusR}$ – Reference position of the j th node of the fuselage mesh (m) • $^{MBD}\Lambda_j^{FusR}$ – Reference orientation of the j th node of the fuselage mesh (-) • $^{MBD}\vec{p}_j^{SWnR}$ – Reference position of the j th node of the starboard wing mesh (m) • $^{MBD}\Lambda_j^{SWnR}$ – Reference orientation of the j th node of the starboard wing mesh (-) • $^{MBD}\vec{p}_j^{PWnR}$ – Reference position of the j th node of the port wing mesh (m) • $^{MBD}\Lambda_j^{PWnR}$ – Reference orientation of the j th node of the port wing mesh (-) • $^{MBD}\vec{p}_j^{VSR}$ – Reference position of the j th node of the vertical stabilizer mesh (m) • $^{MBD}\Lambda_j^{VSR}$ – Reference orientation of the j th node of the vertical stabilizer mesh (-) • $^{MBD}\vec{p}_j^{SHSR}$ – Reference position of the j th node of the starboard horizontal stabilizer mesh (m) • $^{MBD}\Lambda_j^{SHSR}$ – Reference

Deleted: Aerodynamic applied c

Deleted: rotors

Deleted: Aerodynamic applied c

Deleted: plus generator torque

Deleted: rotors

Deleted: Aerodynamic applied c

Deleted: rotors

Deleted: Aerodynamic applied c

Deleted: plus generator torque

Deleted: rotors

local coordinate system) at the j^{th} Gauss point of the port wing mesh (N-m) • ${}^{MBD}\vec{p}^{VSO}$ – Position (origin) of the vertical stabilizer (m) • ${}^{MBD}\vec{p}_j^{VS}$ – Translational position (absolute) of the j^{th} node of the vertical stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{VS}$ – Displaced rotation (absolute orientation) of the j^{th} node of the vertical stabilizer mesh (-) • ${}^{MBD}\vec{v}_j^{VS}$ – Translational velocity (absolute) of the j^{th} node of the vertical stabilizer mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{VS}$ – Rotational velocity (absolute) of the j^{th} node of the vertical stabilizer mesh (rad/s) • ${}^{MBD}\vec{a}_j^{VS}$ – Translational acceleration (absolute) of the j^{th} node of the vertical stabilizer mesh (m/s^2) • ${}^{MBD}\vec{FR}_j^{VS}$ – Reaction force (expressed in the local coordinate system) at the j^{th} Gauss point of the vertical stabilizer mesh (N) • ${}^{MBD}\vec{MR}_j^{VS}$ – Reaction moment (expressed in the local coordinate system) at the j^{th} Gauss point of the vertical stabilizer mesh (N-m) • ${}^{MBD}\vec{p}^{SHSO}$ – Position (origin) of the starboard horizontal stabilizer (m) • ${}^{MBD}\vec{p}_j^{SHS}$ – Translational position (absolute) of the j^{th} node of the starboard horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{SHS}$ – Displaced rotation (absolute orientation)			orientation of the j^{th} node of the starboard horizontal stabilizer mesh (-) • ${}^{MBD}\vec{p}_j^{PHSR}$ – Reference position of the j^{th} node of the port horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{PHSR}$ – Reference orientation of the j^{th} node of the port horizontal stabilizer mesh (-) • ${}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right]$ – Reference position of the j^{th} node of the pylons on the starboard wing mesh (m) • ${}^{MBD}\Lambda_j^{SPyR}\left[n_{Pylons}\right]$ – Reference orientation of the j^{th} node of the pylons on the starboard wing mesh (-) • ${}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right]$ – Reference position of the j^{th} node of the pylons on the port wing mesh (m) • ${}^{MBD}\Lambda_j^{PPyR}\left[n_{Pylons}\right]$ – Reference orientation of the j^{th} node of the pylons on the port wing mesh (-) • ${}^{MBD}\vec{p}^{SPyRtrR}\left[n_{Pylons},n_2\right]$ – Reference positions (origins) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{SPyRtrR}\left[n_{Pylons},n_2\right]$ – Reference orientations of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (-) • ${}^{MBD}\vec{p}^{PPyRtrR}\left[n_{Pylons},n_2\right]$ – Reference positions

Deleted: rotors

Deleted: rotors

of the j th node of the starboard horizontal stabilizer mesh (-) • ${}^{MBD}\vec{v}_j^{SHS}$ – Translational velocity (absolute) of the j th node of the starboard horizontal stabilizer mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{SHS}$ – Rotational velocity (absolute) of the j th node of the starboard horizontal stabilizer mesh (rad/s) • ${}^{MBD}\vec{a}_j^{SHS}$ – Translational acceleration (absolute) of the j th node of the starboard horizontal stabilizer mesh (m/s^2) • ${}^{MBD}\vec{FR}_j^{SHS}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the starboard horizontal stabilizer mesh (N) • ${}^{MBD}\vec{MR}_j^{SHS}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the starboard horizontal stabilizer mesh (N-m) • ${}^{MBD}\vec{p}^{PHSO}$ – Position (origin) of the port horizontal stabilizer (m) • ${}^{MBD}\vec{p}_j^{PHS}$ – Translational position (absolute) of the j th node of the port horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{PHS}$ – Displaced rotation (absolute orientation) of the j th node of the port horizontal stabilizer mesh (-) • ${}^{MBD}\vec{v}_j^{PHS}$ – Translational velocity (absolute) of the j th node of the port horizontal stabilizer mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{PHS}$ – Rotational			(origins) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{PPyRtrR}\left[n_{Pylons}, n_2\right]$ – Reference orientations of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (-)

Deleted: rotors

Deleted: rotors

velocity (absolute) of the j^{th} node of the port horizontal stabilizer mesh (rad/s) • ${}^{MBD}\vec{a}_j^{PHS}$ – Translational acceleration (absolute) of the j^{th} node of the port horizontal stabilizer mesh (m/s^2) • ${}^{MBD}\vec{F}R_j^{PHS}$ – Reaction force (expressed in the local coordinate system) at the j^{th} Gauss point of the port horizontal stabilizer mesh (N) • ${}^{MBD}\vec{M}R_j^{PHS}$ – Reaction moment (expressed in the local coordinate system) at the j^{th} Gauss point of the port horizontal stabilizer mesh (N-m) • ${}^{MBD}\vec{p}^{SPyO}\left[n_{Pylons}\right]$ – Positions (origins) of pylons on the starboard wing (m) • ${}^{MBD}\vec{p}_j^{SPy}\left[n_{Pylons}\right]$ – Translational position (absolute) of the j^{th} node of the pylons on the starboard wing mesh (m) • ${}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons}\right]$ – Displaced rotation (absolute orientation) of the j^{th} node of the pylons on the starboard wing mesh (-) • ${}^{MBD}\vec{v}_j^{SPy}\left[n_{Pylons}\right]$ – Translational velocity (absolute) of the j^{th} node of the pylons on the starboard wing mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{SPy}\left[n_{Pylons}\right]$ – Rotational velocity (absolute) of the j^{th} node of the pylons on the starboard wing mesh (rad/s) • ${}^{MBD}\vec{a}_j^{SPy}\left[n_{Pylons}\right]$ – Translational acceleration			

(absolute) of the j^{th} node of the pylons on the starboard wing mesh (m/s^2) • $^{MBD}\vec{FR}_j^{SPy}\left[n_{Pylons}\right]$ – Reaction force (expressed in the local coordinate system) at the j^{th} Gauss point of the pylons on the starboard wing mesh (N) • $^{MBD}\vec{MR}_j^{SPy}\left[n_{Pylons}\right]$ – Reaction moment (expressed in the local coordinate system) at the j^{th} Gauss point of the pylons on the starboard wing mesh (N-m) • $^{MBD}\vec{p}^{PPyO}\left[n_{Pylons}\right]$ – Positions (origins) of pylons on the port wing (m) • $^{MBD}\vec{p}_j^{PPy}\left[n_{Pylons}\right]$ – Translational position (absolute) of the j^{th} node of the pylons on the port wing mesh (m) • $^{MBD}\Lambda_j^{PPy}\left[n_{Pylons}\right]$ – Displaced rotation (absolute orientation) of the j^{th} node of the pylons on the port wing mesh (-) • $^{MBD}\vec{v}_j^{PPy}\left[n_{Pylons}\right]$ – Translational velocity (absolute) of the j^{th} node of the pylons on the port wing mesh (m/s) • $^{MBD}\vec{\omega}_j^{PPy}\left[n_{Pylons}\right]$ – Rotational velocity (absolute) of the j^{th} node of the pylons on the port wing mesh (rad/s) • $^{MBD}\vec{a}_j^{PPy}\left[n_{Pylons}\right]$ – Translational acceleration (absolute) of the j^{th} node of the pylons on the port wing mesh (m/s^2) • $^{MBD}\vec{FR}_j^{PPy}\left[n_{Pylons}\right]$ – Reaction force (expressed in			

the local coordinate system) at the j^{th} Gauss point of the pylons on the port wing mesh (N) • ${}^{MBD}\vec{M}R_j^{PPy}\left[n_{Pylons}\right]$ – Reaction moment (expressed in the local coordinate system) at the j^{th} Gauss point of the pylons on the port wing mesh (N-m) • ${}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Translational position (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Displaced rotation (absolute orientation) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (-) • ${}^{MBD}\vec{v}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Translational velocity (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m/s) • ${}^{MBD}\vec{\omega}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational velocity (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (rad/s) • ${}^{MBD}\vec{a}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Translational acceleration (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m/s^2) • ${}^{MBD}\vec{\alpha}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational acceleration (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (rad/s^2)			

Deleted: rotors

Deleted: rotors

Deleted: rotors

• $^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Translational position (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m) • $^{MBD}\Lambda^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Displaced rotation (absolute orientation) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (-) • $^{MBD}\vec{v}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Translational velocity (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m/s) • $^{MBD}\vec{\omega}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational velocity (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (rad/s) • $^{MBD}\vec{a}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Translational acceleration (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m/s^2) • $^{MBD}\vec{\alpha}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational acceleration (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (rad/s^2)			

MiscVars: ^{Ctrl}y, ^{MD}y, ^{IfW}y, ^{KAD}y, $^{MD}x^{Copy}$, $^{KAD}z^{Copy}$

Mapping of Outputs to Inputs in KiteFASTMBD

Output depends on Input (Y/N)		Inputs				
		MBDyn	KiteAeroDyn	InflowWind	MoorDyn	Controller
Outputs	MBDyn		N	N	N	Y
	KiteAeroDyn	Y				Y
	InflowWind		Y			Y
	MoorDyn	Y				Y
	Controller	N	N			

Deleted: <#> $^{MBD}\Omega^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational speeds of the top and bottom rotors on the pylons on the starboard wing (rad/s)¶

Deleted: rotors

Deleted: rotors

Deleted: rotors

Deleted: ¶

Deleted: $^{MBD}\Omega^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational speeds of the top and bottom rotors on the pylons on the port wing (rad/s)

Commented [JJ6]: The outputs of each module (as calculated by their respective CalcOutput() routines) are stored as MiscVars in KiteFASTMBD.

Deleted: ?

Commented [JJ7]: This may technically not be true, but we can only call the Controller once anyway, so, we'll assume no.

Data Flow (stopping when reaching "N")

MBDyn Controller
KiteAeroDyn MBDyn Controller
Controller
InflowWind KiteAeroDyn MBDyn Controller
Controller
Controller
MoorDyn MBDyn Controller
Controller
Controller

Deleted: ?

Thus, no nonlinear solves are required
Order of calls: MBDyn, Controller, MoorDyn, InflowWind, KiteAeroDyn

Constructor

Commented [JJ8]: Does MBD need to know the aerodynamic loads (etc.) at t=0? If so, would that happen here?

This routine initializes KiteFASTMBD at $t=0$:

- Sets parameters
- Initializes states
- Calls module Init routines
- Opens the write output file
- Opens and writes the summary file

Query the MBDyn model to access the inputs at $t=0$.
Query the MBDyn model to access the names of the KiteAeroDyn, InflowWind, and MoorDyn primary input files

Commented [JJ9]: The names of the KiteAeroDyn input file etc., along with switches for enabling/disabling each module, must be queried from the MBDyn model. I haven't specifically included logic below to enable/disable modules, but this should implemented.

Set the parameters from inputs (Δt, N_{Flaps}, N_{Pylons}, ${}^{MBD}\vec{g}$, ${}^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons},n_2\right]$, and ${}^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]$). Trigger a fatal error if $N_{Flaps}\neq 3$ (to match the current controller interface), $N_{Pylons}\neq 2$ (to match the current controller interface), $\Delta t\neq 0.01$ (to match the controller time step), ${}^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]<0$, ${}^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons},n_2\right]<0$, ${}^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons},n_2\right]-{}^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]\left({}^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons},n_2\right]\right)^2<0$, ${}^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]<0$, ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons},n_2\right]<0$, or ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons},n_2\right]-{}^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]\left({}^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]\right)^2<0$. Note that:

Deleted: and

Commented [JJ10]: These must be queried from the MBDyn model.

Deleted: or

- The flap indices: $n_{Flaps}=\left\{1,2,\ldots,N_{Flaps}\right\}$
- The pylon indices: $n_{Pylons}=\left\{1,2,\ldots,N_{Pylons}\right\}$
- And: $n_2=\left\{1,2\right\}$

Set the DCM conversion parameter from the FAST ground system (X pointed nominally downwind; Z pointed vertically opposite gravity; Y transverse) to the ground system used by the controller (X pointed nominally upwind; Z pointed vertically downward, Y transverse):

$$\Lambda^{FAST2Ctrl} = \begin{bmatrix} -1 & 0 & 0 \\ 0 & 1 & 0 \\ 0 & 0 & -1 \end{bmatrix}$$

Set the reference positions (origins) needed as initialization inputs to KiteAeroDyn:

$$^{KAD}\vec{p}^{SWnOR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SWnO} - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{PWnOR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PWnO} - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{VSOR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{VSO} - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{SHSOR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SHSO} - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{PHSOR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PHSO} - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{SPyOR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SPyO}\left[n_{Pylons} \right] - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{PPyOR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PPyO}\left[n_{Pylons} \right] - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{SPyRtrR}\left[n_{Pylons}, n_2 \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons}, n_2 \right] - {}^{MBD}\vec{p}^{FusO} \right\}$$

$$^{KAD}\vec{p}^{PPyRtrR}\left[n_{Pylons}, n_2 \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons}, n_2 \right] - {}^{MBD}\vec{p}^{FusO} \right\}$$

Call KiteAeroDyn_Init()

Trigger a fatal error if $\left({}^{KAD}\Delta t \neq \Delta t \right)$

Set the air density for future reference: $\rho = {}^{KAD}\rho$

Determine the number of points where wind will be accessed within InflowWind by summing up the nodes on the AeroDyn input meshes, plus one for the fuselage origin and one for the ground station:

$$\begin{aligned} {}^{IfW}NumWindPoints = 2 & \\ & + {}^{KAD}NumFusNds \\ & + {}^{KAD}NumSWnNds \\ & + {}^{KAD}NumPWnNds \\ & + {}^{KAD}NumVSNds \\ & + {}^{KAD}NumSHSNds \\ & + {}^{KAD}NumPHSNds \\ & + {}^{KAD}NumPylNds\left(2N_{Pylons} \right) \\ & + 4N_{Pylons} \end{aligned}$$

Call InflowWind_Init()

Trigger a fatal error if $\left({}^{IfW}\Delta t \neq \Delta t \right)$

Set the initialization inputs to MoorDyn:

$$^{MD}g = \left\| ^{MBD}\vec{g} \right\|_2$$

$$^{MD}rhoW = \rho$$

$$^{MD}WtrDepth = 0$$

$$^{MD}PtfmInit = \begin{Bmatrix} ^{MBD}\vec{p}^{FusO} \\ ^{MBD}\Lambda^{FusO} \end{Bmatrix}$$

Call MoorDyn_Init()

Trigger a fatal error if $\left(^{MD}\Delta t \neq \Delta t \right)$

Trigger a fatal error if there is more than one anchor set in MoorDyn. Also, set the anchor position for future reference based on the initialization output from MoorDyn:

$\vec{p}^{Anch}$ = from MoorDyn initialization output

Call Controller_Init()

Trigger a fatal error if $\left(^{Ctrl}\Delta t \neq \Delta t \right)$

Set the reference positions and orientations of the line2 and point meshes from the inputs:

$$^{MBD}\vec{p}_j^{FusR} = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{Fus} - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumFusNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{FusR} = {}^{MBD}\Lambda_j^{Fus}\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumFusNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{SWnR} = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{SWn} - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumSWnNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{SWnR} = {}^{MBD}\Lambda_j^{SWn}\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumSWnNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{PWnR} = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{PWn} - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPWnNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{PWnR} = {}^{MBD}\Lambda_j^{PWn}\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPWnNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{VSR} = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{VS} - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumVSNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{VSR} = {}^{MBD}\Lambda_j^{VS}\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumVSNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{SHSR} = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{SHS} - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumSHSNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{SHSR} = {}^{MBD}\Lambda_j^{SHS}\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumSHSNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{PHSR} = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{PHS} - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPHSNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{PHSR} = {}^{MBD}\Lambda_j^{PHS}\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPHSNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{SPy}\left[n_{Pylons} \right] - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPylNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{SPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons} \right]\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPylNds\right\}\text{)}$$

$$^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda^{FusO}\left\{ ^{MBD}\vec{p}_j^{PPy}\left[n_{Pylons} \right] - {}^{MBD}\vec{p}^{FusO} \right\} \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPylNds\right\}\text{)}$$

$$^{MBD}\Lambda_j^{PPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons} \right]\left[^{MBD}\Lambda^{FusO} \right]^T \quad \text{(for } j = \left\{1,2,\ldots,{}^{MBD}NumPylNds\right\}\text{)}$$

Deleted: ^{MD}g = queried from the MBDyn model

Deleted: $^{MD}PtfmInit = \begin{Bmatrix} ^{MBD}\vec{p}^{FusO} \\ XXXXXXXXX \end{Bmatrix}$

Deleted: ¶

Commented [JJ12]: TBD. Are there any initialization inputs to the controller? Are there initialization output from the controller?

Right now I'm assuming that Controller_Init() returns the initial outputs.

Commented [JJ13]: Note: the motion meshes are line2 meshes (except for the rotors, which are point meshes), but the load meshes are point meshes.

$$
\begin{aligned}
&{}^{MBD}\vec{p}^{SPyRtrR}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons}, n_2\right] - {}^{MBD}\vec{p}^{FusO}\right\} \\
&{}^{MBD}\Lambda^{SPyRtrR}\left[n_{Pylons}, n_2\right] = I \\
&{}^{MBD}\vec{p}^{PPyRtrR}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons}, n_2\right] - {}^{MBD}\vec{p}^{FusO}\right\} \\
&{}^{MBD}\Lambda^{PPyRtrR}\left[n_{Pylons}, n_2\right] = I
\end{aligned}
$$

Set mesh-mappings between KiteFASTMBD-KiteAeroDyn and KiteFASTMBD-MoorDyn.

> **Commented [JJ14]:** The mesh-mapping routines can only handle one source and one destination mesh. To do this mapping, the MBDyn meshes for the starboard and port wings (SWn and PWn) have to be copied into a single mesh using a one-to-one transfer of reference positions, reference orientations, and fields (which I label as Wn in the mesh-mappings below).

Open the write Output File

Initial calculate output:

- Set inputs to MoorDyn at $t = 0$ from MBDyn (see below for solution at t)
- Call MoorDyn_CalcOutput()
- Set inputs to KiteAeroDyn from Controller at $t = 0$ (see below for solution at t)
- Set inputs to KiteAeroDyn from MBDyn at $t = 0$ (see below for solution at t)
- Set inputs to InflowWind at $t = 0$ based on the KiteAeroDyn inputs (see below for solution at t)
- Call InflowWind_CalcOutput()
- Set inputs to KiteAeroDyn from InflowWind at $t = 0$ (see below for solution at t)
- Call KiteAeroDyn_CalcOutput()
- Transfer outputs from KiteAeroDyn to MBDyn at $t = 0$ (see below for solution at t)
- Transfer outputs from MoorDyn to MBDyn at $t = 0$ (see below for solution at t)
- Model the rotor/drivetrain dynamics, including the effects from the Controller and KiteAeroDyn, and calculate the reaction loads on the pylons for transfer to MBDyn at $t = 0$ (see below for solution at t)

> **Deleted:** Transfer outputs from the Controller

> **Commented [JJ15]:** Do we need to do this here to calculate WriteOutputs at initialization and to set initial loads from MBDyn?

Initialize the states not previously initialized:

$NewTime = TRUE$

Open and write a summary file (if SumPrint = TRUE)

> **Commented [JJ16]:** SumPrint must be queried from the MBDyn model

> **Commented [JJ17]:** I'm only hand waving here because the implementation should be obvious (similar to other OpenFAST summary files)

KiteFASTMBD Summary File

Predictions were generated on DATE at TIME using KiteFASTMBD (VERSION, DATE)
compiled with
NWTC Subroutine Library (VERSION, DATE)
KiteAeroDyn (VERSION, DATE)
InflowWind (VERSION, DATE) for OpenFAST (VERSION DATE)
MoorDyn (VERSION, DATE)
Controller Wrapper (VERSION, DATE)
Controller (VERSION, DATE)
MBDyn (VERSION, DATE)

> **Commented [JJ18]:** The controller version will have to be an initialization output to write this.

Description from the MDyn input file: TITLE

> **Commented [JJ19]:** Probably not needed if TITLE is not easily accessible within the MBDyn user element.

Reference Points, MBDyn Finite-Element Nodes, and MBDyn Gauss Points

Component	Type	Number	Output Number	x	y	z
(-)	(-)	(-)	(-)	(m)	(m)	(m)
Fuselage	Reference point	-	-	0	0	0

Fuselage	Finite-element node	j	$\begin{cases} Fus\langle\beta\rangle & for\left(FusOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{FusR}$
Fuselage	Gauss point	j	$\begin{cases} Fus\langle\beta\rangle & for\left(FusOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{FusR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{FusR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{FusR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{FusR} & otherwise \end{cases}$
Starboard wing	Reference point	-	-	${}^{KAD}\vec{p}^{SWnOR}$
Starboard wing	Finite-element node	j	$\begin{cases} SWn\langle\beta\rangle & for\left(SWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{SWnR}$
Starboard wing	Gauss point	j	$\begin{cases} SWn\langle\beta\rangle & for\left(SWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SWnR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SWnR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SWnR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SWnR} & otherwise \end{cases}$
Port wing	Reference point	-	-	${}^{KAD}\vec{p}^{PWnOR}$
Port wing	Finite-element node	j	$\begin{cases} PWn\langle\beta\rangle & for\left(PWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{PWnR}$
Port wing	Gauss point	j	$\begin{cases} PWn\langle\beta\rangle & for\left(PWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PWnR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PWnR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PWnR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PWnR} & otherwise \end{cases}$
Vertical stabilizer	Reference point	-	-	${}^{KAD}\vec{p}^{VSOR}$

Vertical stabilizer	Finite-element node	j	$\begin{cases} VS\langle\beta\rangle & for\left(VSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{\,VSR}$
Vertical stabilizer	Gauss point	j	$\begin{cases} VS\langle\beta\rangle & for\left(VSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,VSR}+\left(\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,VSR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,VSR}+\left(1-\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,VSR} & otherwise \end{cases}$
Starboard horizontal stabilizer	Reference point	-	-	${}^{KAD}\vec{p}^{\,SHSOR}$
Starboard horizontal stabilizer	Finite-element node	j	$\begin{cases} SHS\langle\beta\rangle & for\left(SHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{\,SHSR}$
Starboard horizontal stabilizer	Gauss point	j	$\begin{cases} SHS\langle\beta\rangle & for\left(SHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,SHSR}+\left(\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,SHSR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,SHSR}+\left(1-\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,SHSR} & otherwise \end{cases}$
Port horizontal stabilizer	Reference point	-	-	${}^{KAD}\vec{p}^{\,PHSOR}$
Port horizontal stabilizer	Finite-element node	j	$\begin{cases} PHS\langle\beta\rangle & for\left(PHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{\,PHSR}$
Port horizontal stabilizer	Gauss point	j	$\begin{cases} PHS\langle\beta\rangle & for\left(PHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,PHSR}+\left(\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,PHSR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,PHSR}+\left(1-\dfrac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,PHSR} & otherwise \end{cases}$

Starboard pylon	n_{Pylons}	Reference point	-	-
$^{KAD}\vec{p}^{SPyOR}\left[n_{Pylons}\right]$				
Starboard pylon	n_{Pylons}	Finite-element node	j	$\begin{cases} SP\left\langle n_{Pylons}\right\rangle\left\langle\beta\right\rangle & for\left(PylOutNd\left[\beta\right]=j\right) \\ - & otherwise \end{cases}$
$^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right]$				
Starboard pylon	n_{Pylons}	Gauss point	j	$\begin{cases} SP\left\langle n_{Pylons}\right\rangle\left\langle\beta\right\rangle & for\left(PylOutNd\left[\beta\right]=j\right) \\ - & otherwise \end{cases}$
$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right]+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right] & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right]+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right] & otherwise \end{cases}$				
Port pylon	n_{Pylons}	Reference point	-	-
$^{KAD}\vec{p}^{PPyOR}\left[n_{Pylons}\right]$				
Port pylon	n_{Pylons}	Finite-element node	j	$\begin{cases} PP\left\langle n_{Pylons}\right\rangle\left\langle\beta\right\rangle & for\left(PylOutNd\left[\beta\right]=j\right) \\ - & otherwise \end{cases}$
$^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right]$				
Port pylon	n_{Pylons}	Gauss point	j	$\begin{cases} PP\left\langle n_{Pylons}\right\rangle\left\langle\beta\right\rangle & for\left(PylOutNd\left[\beta\right]=j\right) \\ - & otherwise \end{cases}$
$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right]+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right] & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right]+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right] & otherwise \end{cases}$				
Top rotor on starboard pylon	n_{Pylons}	Reference point	-	-
$^{KAD}\vec{p}^{SPyRtrR}\left[n_{Pylons},1\right]$				
Bottom rotor on starboard pylon	n_{Pylons}	Reference point	-	-
$^{KAD}\vec{p}^{SPyRtrR}\left[n_{Pylons},2\right]$				
Top rotor on port pylon	n_{Pylons}	Reference point	-	-
$^{KAD}\vec{p}^{PPyRtrR}\left[n_{Pylons},1\right]$				
Bottom rotor on port pylon	n_{Pylons}	Reference point	-	-
$^{KAD}\vec{p}^{PPyRtrR}\left[n_{Pylons},2\right]$				

Time Step (s): Δt

Commented [JJ20]: All time steps are the same. Is there a reason to allow for other time step of modules or output?

Deleted: Kite Mass, Center of Mass, and Inertia¶
¶

Requested Channels in KiteFASTMBD Output Files: NUMBER

Number	Name	Units	Generated by
0	Time	(s)	KiteFASTMBD
NUMBER	NAME	UNITS	(KiteFASTMBD, KiteAeroDyn, InflowWind, MoorDyn, or Controller Wrapper)

Deconstructor

This routine ends KiteFASTMBD:

- Calls module End routines
- Deallocates memory
- Closes the write output file

AssRes

This routine accesses inputs at both t (from GetXPrev) and $t+\Delta t$ (from GetXCur) for both the prediction and correction steps of each MBD time step, temporarily updates states from t to $t+\Delta t$, and calculates outputs at $t+\Delta t$:

- Calls module UpdateStates routines
- Calls module CalcOutput routines

Set the discrete-time counter:

$$n=\frac{t}{\Delta t}$$

Query the MBDyn model to access the inputs at both t (from GetXPrev) and $t+\Delta t$ (from GetXCur).

Calculate the translation displacements (relative) of the MBDyn input meshes at t and $t+\Delta t$:

$$^{MBD}\vec{u}_j^{Fus} = {}^{MBD}\vec{p}_j^{Fus} - {}^{MBD}\vec{p}_j^{FusR} \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumFusNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{SWn} = {}^{MBD}\vec{p}_j^{SWn} - {}^{MBD}\vec{p}_j^{SWnR} \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumSWnNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{PWn} = {}^{MBD}\vec{p}_j^{PWn} - {}^{MBD}\vec{p}_j^{PWnR} \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumPWnNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{VS} = {}^{MBD}\vec{p}_j^{VS} - {}^{MBD}\vec{p}_j^{VSR} \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumVSNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{SHS} = {}^{MBD}\vec{p}_j^{SHS} - {}^{MBD}\vec{p}_j^{SHSR} \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumSHSNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{PHS} = {}^{MBD}\vec{p}_j^{PHS} - {}^{MBD}\vec{p}_j^{PHSR} \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumPHSNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right] = {}^{MBD}\vec{p}_j^{SPy}\left[n_{Pylons}\right] - {}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right] \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumPylNds\}\text{)}$$

$$^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right] = {}^{MBD}\vec{p}_j^{PPy}\left[n_{Pylons}\right] - {}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right] \quad \text{(for } j=\{1,2,\ldots,{}^{MBD}NumPylNds\}\text{)}$$

$$^{MBD}\vec{u}^{SPyRtr}\left[n_{Pylons},n_2\right] = {}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons},n_2\right] - {}^{MBD}\vec{p}^{SPyRtrR}\left[n_{Pylons},n_2\right]$$

$$^{MBD}\vec{u}^{PPyRtr}\left[n_{Pylons},n_2\right] = {}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons},n_2\right] - {}^{MBD}\vec{p}^{PPyRtrR}\left[n_{Pylons},n_2\right]$$

Advance the controller only once per time step, obtaining the controller outputs at $t+\Delta t$:

IF $\left(NewTime\right)$ THEN

- Set inputs to Controller at t:

Commented [JJ21]: Technically, we only need the MBDyn inputs for MoorDyn (i.e. the wing deformation) and the controller at t.

$$^{Ctrl}dcm\_g2b = {}^{MBD}\Lambda^{FusO}\left[\Lambda^{FAST2Ctrl}\right]^T$$

$$^{Ctrl}pqr = {}^{MBD}\Lambda^{FusO}\ {}^{MBD}\vec{\omega}^{FusO}$$

$$^{Ctrl}acc\_norm = \left\| {}^{MBD}\vec{a}^{FusO} \right\|_2$$

$$^{Ctrl}Xg = \Lambda^{FAST2Ctrl}\left\{ {}^{MBD}\vec{p}^{FusO} - \vec{p}^{Anch} \right\}$$

$$^{Ctrl}Vg = \Lambda^{FAST2Ctrl}\ {}^{MBD}\vec{v}^{FusO}$$

$$^{Ctrl}Vb = {}^{MBD}\Lambda^{FusO}\ {}^{MBD}\vec{v}^{FusO}$$

$$^{Ctrl}Ag = \Lambda^{FAST2Ctrl}\ {}^{MBD}\vec{a}^{FusO}$$

$$^{Ctrl}Ab = {}^{MBD}\Lambda^{FusO}\ {}^{MBD}\vec{a}^{FusO}$$

$$^{Ctrl}rho = \rho$$

$$^{Ctrl}apparent\_wind = \Lambda^{FAST2Ctrl}\left\{ {}^{IfW}VelocityUVW(:,2) - {}^{MBD}\vec{v}^{FusO} \right\}$$

$$^{Ctrl}tether\_force\_b = {}^{MBD}\Lambda^{FusO}\left\{ \sum_{i=1}^{NFairs} {}^{MD}PtFairleadLoad\%Force(:,i) \right\}$$

$$^{Ctrl}wind\_g = \Lambda^{FAST2Ctrl}\ {}^{IfW}VelocityUVW(:,1)$$

$$^{Ctrl}aero\_torque^{SPyRtr}\left[n_{Pylons}, n_2\right] = \left\{ {}^{MBD}\hat{x}^{SPyRtr}\left[n_{Pylons}, n_2\right] \right\}^T\ {}^{KAD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right]$$

$$^{Ctrl}aero\_torque^{PPyRtr}\left[n_{Pylons}, n_2\right] = \left\{ {}^{MBD}\hat{x}^{PPyRtr}\left[n_{Pylons}, n_2\right] \right\}^T\ {}^{KAD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right]$$

- Call Controller_Step()

Ensure that we only call the controller once per time step:

$$NewTime = FALSE$$

END

Store a copy of the current states at t:

$$^{MD}x^{Copy} = {}^{MD}x$$

$$^{KAD}z^{Copy} = {}^{KAD}z$$

Set inputs to MoorDyn at both t and $t + \Delta t$ from MBDyn:

$$^{MD}PtFairleadDisplacement = M_u^{L2P}\left({}^{MBD}\vec{u}_j^{Wn},\ {}^{MBD}\Lambda_j^{Wn} \right)$$

Advance MoorDyn:
Call MoorDyn_UpdateStates()
Call MoorDyn_CalcOutput()

Set inputs to KiteAeroDyn from Controller at $t + \Delta t$:

$$^{KAD}Ctrl^{SFlp}\left[n_{Flaps}\right] = \begin{cases} {}^{Ctrl}kFlapA5 & for\left(n_{Flaps} = 1\right) \\ {}^{Ctrl}kFlapA7 & for\left(n_{Flaps} = 2\right) \\ {}^{Ctrl}kFlapA8 & for\left(n_{Flaps} = 3\right) \end{cases}$$

Commented [JJ22]: All filtered values (_f) are identical to the unfiltered values.

Commented [JJ23]: We are approximating this input to the controller as the vector sum of the fairlead tensions.

Deleted: $^{Ctrl}tether\_force\_b =$

Commented [JJ25]: These were added to the original controller inputs so that the controller could calculate the rotor/drivetrain acceleration and resulting generator speed and torque.

We should also ensure that the controller is using the same rotor/drivetrain rotational inertia.

We still need to confirm the sign of the rotor speeds, aerodynamic torques, and generator torques.

Commented [JJ26]: Use these MiscVars in the calls to MoorDyn and KiteAeroDyn UpdateStates and CalcOutput below rather than the actual states.

Commented [JJ27]: See earlier comment about mesh mapping with Wn above.

Commented [JJ28]: Input the time at t+dt in this call.

$$ {}^{KAD}Ctrl^{PFlp}\left[n_{Flaps}\right] = \begin{cases} {}^{Ctrl}kFlapA4 & for\left(n_{Flaps}=1\right) \\ {}^{Ctrl}kFlapA2 & for\left(n_{Flaps}=2\right) \\ {}^{Ctrl}kFlapA1 & for\left(n_{Flaps}=3\right) \end{cases} $$

$$ {}^{KAD}Ctrl^{Rudr}\left[n_2\right] = {}^{Ctrl}kFlapA10 $$

$$ {}^{KAD}Ctrl^{SElv}\left[n_2\right] = {}^{Ctrl}kFlapA9 $$

$$ {}^{KAD}Ctrl^{PElv}\left[n_2\right] = {}^{Ctrl}kFlapA9 $$

$$ {}^{KAD}\Omega^{SPyRtr}\left[n_{Pylons},n_2\right] = {}^{Ctrl}\Omega^{SPyRtr}\left[n_{Pylons},n_2\right] $$

$$ {}^{KAD}\Omega^{PPyRtr}\left[n_{Pylons},n_2\right] = {}^{Ctrl}\Omega^{PPyRtr}\left[n_{Pylons},n_2\right] $$

$$ {}^{KAD}\theta^{SPyRtr}\left[n_{Pylons},n_2\right] = 0 $$

$$ {}^{KAD}\theta^{PPyRtr}\left[n_{Pylons},n_2\right] = 0 $$

Set inputs to KiteAeroDyn from MBDyn at $t+\Delta t$ based on mesh-mapping:

$$ {}^{KAD}\vec{u}^{FusO} = {}^{MBD}\vec{p}^{FusO} $$

$$ {}^{KAD}\vec{u}_j^{Fus} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{Fus},\ {}^{MBD}\Lambda_j^{Fus}\right) $$

$$ {}^{KAD}\Lambda_j^{Fus} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{Fus}\right) $$

$$ {}^{KAD}\vec{v}_j^{Fus} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{Fus},\ {}^{MBD}\vec{u}_j^{Fus},\ {}^{MBD}\vec{v}_j^{Fus},\ {}^{MBD}\vec{\omega}_j^{Fus}\right) $$

$$ {}^{KAD}\vec{u}_j^{SWn} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{SWn},\ {}^{MBD}\Lambda_j^{SWn}\right) $$

$$ {}^{KAD}\Lambda_j^{SWn} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{SWn}\right) $$

$$ {}^{KAD}\vec{v}_j^{SWn} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{SWn},\ {}^{MBD}\vec{u}_j^{SWn},\ {}^{MBD}\vec{v}_j^{SWn},\ {}^{MBD}\vec{\omega}_j^{SWn}\right) $$

$$ {}^{KAD}\vec{u}_j^{PWn} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{PWn},\ {}^{MBD}\Lambda_j^{PWn}\right) $$

$$ {}^{KAD}\Lambda_j^{PWn} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{PWn}\right) $$

$$ {}^{KAD}\vec{v}_j^{PWn} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{PWn},\ {}^{MBD}\vec{u}_j^{PWn},\ {}^{MBD}\vec{v}_j^{PWn},\ {}^{MBD}\vec{\omega}_j^{PWn}\right) $$

$$ {}^{KAD}\vec{u}_j^{VS} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{VS},\ {}^{MBD}\Lambda_j^{VS}\right) $$

$$ {}^{KAD}\Lambda_j^{VS} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{VS}\right) $$

$$ {}^{KAD}\vec{v}_j^{VS} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{VS},\ {}^{MBD}\vec{u}_j^{VS},\ {}^{MBD}\vec{v}_j^{VS},\ {}^{MBD}\vec{\omega}_j^{VS}\right) $$

$$ {}^{KAD}\vec{u}_j^{SHS} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{SHS},\ {}^{MBD}\Lambda_j^{SHS}\right) $$

$$ {}^{KAD}\Lambda_j^{SHS} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{SHS}\right) $$

$$ {}^{KAD}\vec{v}_j^{SHS} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{SHS},\ {}^{MBD}\vec{u}_j^{SHS},\ {}^{MBD}\vec{v}_j^{SHS},\ {}^{MBD}\vec{\omega}_j^{SHS}\right) $$

$$ {}^{KAD}\vec{u}_j^{PHS} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{PHS},\ {}^{MBD}\Lambda_j^{PHS}\right) $$

$$ {}^{KAD}\Lambda_j^{PHS} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{PHS}\right) $$

Commented [JJ29]: Different controller documentation use kFlapRud in place of kFlapA10

Commented [JJ30]: Different controller documentation use kFlapEle in place of kFlapA9

Commented [JJ31]: These were added to the original controller outputs so that the controller could calculate the rotor/drivetrain acceleration and resulting generator speed and torque.

Deleted: ¶

Commented [JJ32]: The rotor-collective pitch angles are not currently commanded from the controller; assume zero for now.

Commented [JJ33]: You could use P2P mappings here, but there is no point, because the reference (0,0,0) is the same in both KiteAeroDyn and MBDyn.

$$ {}^{KAD}\vec{v}_j^{PHS} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{PHS}, {}^{MBD}\vec{u}_j^{PHS}, {}^{MBD}\vec{v}_j^{PHS}, {}^{MBD}\vec{\omega}_j^{PHS} \right) $$

$$ {}^{KAD}\vec{u}_j^{SPy}\left[n_{Pylons}\right] = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons}\right] \right) $$

$$ {}^{KAD}\Lambda_j^{SPy}\left[n_{Pylons}\right] = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons}\right] \right) $$

$$ {}^{KAD}\vec{v}_j^{SPy}\left[n_{Pylons}\right] = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{MBD}\vec{v}_j^{SPy}\left[n_{Pylons}\right], {}^{MBD}\vec{\omega}_j^{SPy}\left[n_{Pylons}\right] \right) $$

$$ {}^{KAD}\vec{u}_j^{PPy}\left[n_{Pylons}\right] = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons}\right] \right) $$

$$ {}^{KAD}\Lambda_j^{PPy}\left[n_{Pylons}\right] = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons}\right] \right) $$

$$ {}^{KAD}\vec{v}_j^{PPy}\left[n_{Pylons}\right] = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{MBD}\vec{v}_j^{PPy}\left[n_{Pylons}\right], {}^{MBD}\vec{\omega}_j^{PPy}\left[n_{Pylons}\right] \right) $$

$$ {}^{KAD}\vec{u}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{u}^{SPyRtr}\left[n_{Pylons}, n_2\right] $$

$$ {}^{KAD}\Lambda^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Lambda^{SPyRtr}\left[n_{Pylons}, n_2\right] $$

$$ {}^{KAD}\vec{v}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{v}^{SPyRtr}\left[n_{Pylons}, n_2\right] $$

$$ {}^{KAD}\vec{u}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{u}^{PPyRtr}\left[n_{Pylons}, n_2\right] $$

$$ {}^{KAD}\Lambda^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Lambda^{PPyRtr}\left[n_{Pylons}, n_2\right] $$

$$ {}^{KAD}\vec{v}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{v}^{PPyRtr}\left[n_{Pylons}, n_2\right] $$

Commented [JJ34]: You could use P2P mappings here, but there is no point, because the references are the same in both KiteAeroDyn and MBDyn.

Deleted: ${}^{KAD}\Omega^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Omega^{SPyRtr}\left[n_{Pyl\ldots}\right.$

¶

Commented [JJ35]: You could use P2P mappings here, but there is no point, because the references are the same in both KiteAeroDyn and MBDyn.

Deleted: ${}^{KAD}\Omega^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Omega^{PPyRtr}\left[n_{Pyl\ldots}\right.$

¶

Set inputs to InflowWind at $t + \Delta t$ based on the KiteAeroDyn inputs:

$$ {}^{IfW}PositionXYZ(:,1) = {}^{MBD}\vec{p}^{Wind} $$

$$ {}^{IfW}PositionXYZ(:,2) = {}^{MBD}\vec{p}^{FusO} $$

$$ {}^{IfW}PositionXYZ(:,j+2) = {}^{KAD\,In}\vec{p}_j^{FusR} + {}^{KAD}\vec{u}_j^{Fus} \quad \text{(for } j = \left\{1,2,\ldots,{}^{KAD}NumFusNds\right\}\text{)} $$

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ +{}^{KAD}NumFusNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{SWnR} + {}^{KAD}\vec{u}_j^{SWn} \quad \text{(for } j = \left\{1,2,\ldots,{}^{KAD}NumSWnNds\right\}\text{)} $$

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ +{}^{KAD}NumFusNds \\ +{}^{KAD}NumSWnNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{PWnR} + {}^{KAD}\vec{u}_j^{PWn} \quad \text{(for } j = \left\{1,2,\ldots,{}^{KAD}NumPWnNds\right\}\text{)} $$

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ +{}^{KAD}NumFusNds \\ +{}^{KAD}NumSWnNds \\ +{}^{KAD}NumPWnNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{VSR} + {}^{KAD}\vec{u}_j^{VS} \quad \text{(for } j = \left\{1,2,\ldots,{}^{KAD}NumVSNds\right\}\text{)} $$

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ +{}^{KAD}NumFusNds \\ +{}^{KAD}NumSWnNds \\ +{}^{KAD}NumPWnNds \\ +{}^{KAD}NumVSNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{SHSR} + {}^{KAD}\vec{u}_j^{SHS} \quad \text{(for } j = \left\{1,2,\ldots,{}^{KAD}NumSHSNds\right\}\text{)} $$

$$
{}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ \quad +{}^{KAD}NumFusNds \\ \quad +{}^{KAD}NumSWnNds \\ \quad +{}^{KAD}NumPWnNds \\ \quad +{}^{KAD}NumVSNds \\ \quad +{}^{KAD}NumSHSNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{\,PHSR} + {}^{KAD}\vec{u}_j^{\,PHS} \quad (\text{for } j=\left\{1,2,\ldots,{}^{KAD}NumPHSNds\right\})
$$

$$
{}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ \quad +{}^{KAD}NumFusNds \\ \quad +{}^{KAD}NumSWnNds \\ \quad +{}^{KAD}NumPWnNds \\ \quad +{}^{KAD}NumVSNds \\ \quad +{}^{KAD}NumSHSNds \\ \quad +{}^{KAD}NumPHSNds \\ \quad +{}^{KAD}NumPylNds\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{\,SPyR}\left[n_{Pylons}\right] + {}^{KAD}\vec{u}_j^{\,SPy}\left[n_{Pylons}\right] \quad (\text{for } j=\left\{1,2,\ldots,{}^{KAD}NumPylNds\right\})
$$

$$
{}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ \quad +{}^{KAD}NumFusNds \\ \quad +{}^{KAD}NumSWnNds \\ \quad +{}^{KAD}NumPWnNds \\ \quad +{}^{KAD}NumVSNds \\ \quad +{}^{KAD}NumSHSNds \\ \quad +{}^{KAD}NumPHSNds \\ \quad +{}^{KAD}NumPylNds\left(N_{Pylons}\right) \\ \quad +{}^{KAD}NumPylNds\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{\,PPyR}\left[n_{Pylons}\right] + {}^{KAD}\vec{u}_j^{\,PPy}\left[n_{Pylons}\right] \quad (\text{for } j=\left\{1,2,\ldots,{}^{KAD}NumPylNds\right\})
$$

$$
{}^{IfW}PositionXYZ\begin{pmatrix} :,n_2+2 \\ \quad +{}^{KAD}NumFusNds \\ \quad +{}^{KAD}NumSWnNds \\ \quad +{}^{KAD}NumPWnNds \\ \quad +{}^{KAD}NumVSNds \\ \quad +{}^{KAD}NumSHSNds \\ \quad +{}^{KAD}NumPHSNds \\ \quad +{}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ \quad +2\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD}\vec{p}_j^{\,SPyRtrR}\left[n_{Pylons},n_2\right] + {}^{KAD}\vec{u}^{\,SPyRtr}\left[n_{Pylons},n_2\right]
$$

$$
{}^{IfW}PositionXYZ\begin{pmatrix} :,n_2+2 \\ +\,{}^{KAD}NumFusNds \\ +\,{}^{KAD}NumSWnNds \\ +\,{}^{KAD}NumPWnNds \\ +\,{}^{KAD}NumVSNds \\ +\,{}^{KAD}NumSHSNds \\ +\,{}^{KAD}NumPHSNds \\ +\,{}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ +2\left(N_{Pylons}\right) \\ +2\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD}\vec{p}_j^{\,PPyRtrR}\left[n_{Pylons},n_2\right]+{}^{KAD}\vec{u}^{\,PPyRtr}\left[n_{Pylons},n_2\right]
$$

Call InflowWind_CalcOutput()

Commented [JJ36]: Input the time at t+dt in this call.

Set inputs to KiteAeroDyn from InflowWind at $t+\Delta t$:

$$
{}^{KAD}\vec{V}_j^{Fus} = {}^{IfW}VelocityUVW\left(:,j+2\right) \qquad \text{(for } j=\left\{1,2,\ldots,{}^{KAD}NumFusNds\right\}\text{)}
$$

$$
{}^{KAD}\vec{V}_j^{SWn} = {}^{IfW}VelocityUVW\begin{pmatrix} :,j+2 \\ +\,{}^{KAD}NumFusNds \end{pmatrix} \qquad \text{(for } j=\left\{1,2,\ldots,{}^{KAD}NumSWnNds\right\}\text{)}
$$

$$
{}^{KAD}\vec{V}_j^{PWn} = {}^{IfW}VelocityUVW\begin{pmatrix} :,j+2 \\ +\,{}^{KAD}NumFusNds \\ +\,{}^{KAD}NumSWnNds \end{pmatrix} \qquad \text{(for } j=\left\{1,2,\ldots,{}^{KAD}NumPWnNds\right\}\text{)}
$$

$$
{}^{KAD}\vec{V}_j^{VS} = {}^{IfW}VelocityUVW\begin{pmatrix} :,j+2 \\ +\,{}^{KAD}NumFusNds \\ +\,{}^{KAD}NumSWnNds \\ +\,{}^{KAD}NumPWnNds \end{pmatrix} \qquad \text{(for } j=\left\{1,2,\ldots,{}^{KAD}NumVSNds\right\}\text{)}
$$

$$
{}^{KAD}\vec{V}_j^{SHS} = {}^{IfW}VelocityUVW\begin{pmatrix} :,j+2 \\ +\,{}^{KAD}NumFusNds \\ +\,{}^{KAD}NumSWnNds \\ +\,{}^{KAD}NumPWnNds \\ +\,{}^{KAD}NumVSNds \end{pmatrix} \qquad \text{(for } j=\left\{1,2,\ldots,{}^{KAD}NumSHSNds\right\}\text{)}
$$

$$ {}^{KAD}\vec{V}_j^{PHS} = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \end{pmatrix} \qquad \text{(for } j = \left\{1, 2, \ldots, {}^{KAD}NumPHSNds\right\}\text{)} $$

$$ {}^{KAD}\vec{V}_j^{SPy}\left[n_{Pylons}\right] = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(n_{Pylons}-1\right) \end{pmatrix} \qquad \text{(for } j = \left\{1, 2, \ldots, {}^{KAD}NumPylNds\right\}\text{)} $$

$$ {}^{KAD}\vec{V}_j^{PPy}\left[n_{Pylons}\right] = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(N_{Pylons}\right) \\ \quad + {}^{KAD}NumPylNds\left(n_{Pylons}-1\right) \end{pmatrix} \qquad \text{(for } j = \left\{1, 2, \ldots, {}^{KAD}NumPylNds\right\}\text{)} $$

$$ {}^{KAD}\vec{V}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, n_2+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ \quad + 2\left(n_{Pylons}-1\right) \end{pmatrix} $$

$$
{}^{KAD}\vec{V}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{IfW}VelocityUVW\begin{pmatrix} :, n_2 + 2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \\ + {}^{KAD}NumVSNds \\ + {}^{KAD}NumSHSNds \\ + {}^{KAD}NumPHSNds \\ + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ + 2\left(N_{Pylons}\right) \\ + 2\left(n_{Pylons} - 1\right) \end{pmatrix}
$$

Copy KiteAeroDyn inputs at $t + \Delta t$ to t (for KiteAeroDyn_UpdateStates)

Advance KiteAeroDyn:
Call KiteAeroDyn_UpdateStates()
Call KiteAeroDyn_CalcOutput()

Commented [JJ37]: Input the time at t+dt in this call.

Transfer outputs from KiteAeroDyn to MBDyn at $t + \Delta t$:

$$
{}^{MBD}\vec{F}_j^{Fus} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{Fus}\right)
$$

$$
{}^{MBD}\vec{M}_j^{Fus} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{Fus}, {}^{KAD\,Out}\vec{u}_j^{Fus}, {}^{KAD}\vec{F}_j^{Fus}, {}^{KAD}\vec{M}_j^{Fus}\right)
$$

$$
{}^{MBD}\vec{F}_j^{SWn} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{SWn}\right)
$$

$$
{}^{MBD}\vec{M}_j^{SWn} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SWn}, {}^{KAD\,Out}\vec{u}_j^{SWn}, {}^{KAD}\vec{F}_j^{SWn}, {}^{KAD}\vec{M}_j^{SWn}\right)
$$

$$
{}^{MBD}\vec{F}_j^{PWn} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{PWn}\right)
$$

$$
{}^{MBD}\vec{M}_j^{PWn} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PWn}, {}^{KAD\,Out}\vec{u}_j^{PWn}, {}^{KAD}\vec{F}_j^{PWn}, {}^{KAD}\vec{M}_j^{PWn}\right)
$$

$$
{}^{MBD}\vec{F}_j^{VS} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{VS}\right)
$$

$$
{}^{MBD}\vec{M}_j^{VS} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{VS}, {}^{KAD\,Out}\vec{u}_j^{VS}, {}^{KAD}\vec{F}_j^{VS}, {}^{KAD}\vec{M}_j^{VS}\right)
$$

$$
{}^{MBD}\vec{F}_j^{SHS} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{SHS}\right)
$$

$$
{}^{MBD}\vec{M}_j^{SHS} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SHS}, {}^{KAD\,Out}\vec{u}_j^{SHS}, {}^{KAD}\vec{F}_j^{SHS}, {}^{KAD}\vec{M}_j^{SHS}\right)
$$

$$
{}^{MBD}\vec{F}_j^{PHS} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{PHS}\right)
$$

$$
{}^{MBD}\vec{M}_j^{PHS} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PHS}, {}^{KAD\,Out}\vec{u}_j^{PHS}, {}^{KAD}\vec{F}_j^{PHS}, {}^{KAD}\vec{M}_j^{PHS}\right)
$$

$$
{}^{MBD}\vec{F}_j^{SPy}\left[n_{Pylons}\right] = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{SPy}\left[n_{Pylons}\right]\right)
$$

$$
{}^{MBD}\vec{M}_j^{SPy}\left[n_{Pylons}\right] = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{KAD\,Out}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{KAD}\vec{F}_j^{SPy}\left[n_{Pylons}\right], {}^{KAD}\vec{M}_j^{SPy}\left[n_{Pylons}\right]\right)
$$

$${}^{MBD}\vec{F}_j^{PPy}\left[n_{Pylons}\right] = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{PPy}\left[n_{Pylons}\right]\right)$$

$${}^{MBD}\vec{M}_j^{PPy}\left[n_{Pylons}\right] = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{KADOut}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{KAD}\vec{F}_j^{PPy}\left[n_{Pylons}\right], {}^{KAD}\vec{M}_j^{PPy}\left[n_{Pylons}\right]\right)$$

Transfer outputs from MoorDyn to MBDyn at $t + \Delta t$:

$${}^{MBD}\vec{F}_j^{SWn} = {}^{MBD}\vec{F}_j^{SWn} + M_F^{P2P}\left({}^{MD}PtFairleadLoad\right)$$

$${}^{MBD}\vec{M}_j^{SWn} = {}^{MBD}\vec{M}_j^{SWn} + M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SWn}, {}^{MD}PtFairleadDisplacement, {}^{MD}PtFairleadLoad, \vec{0}\right)$$

$${}^{MBD}\vec{F}_j^{PWn} = {}^{MBD}\vec{F}_j^{PWn} + M_F^{P2P}\left({}^{MD}PtFairleadLoad\right)$$

$${}^{MBD}\vec{M}_j^{PWn} = {}^{MBD}\vec{M}_j^{PWn} + M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PWn}, {}^{MD}PtFairleadDisplacement, {}^{MD}PtFairleadLoad, \vec{0}\right)$$

Model the rotor/drivetrain dynamics, including the effects from the Controller and KiteAeroDyn, and calculate the reaction loads on the pylons for transfer to MBDyn at $t + \Delta t$:

Call Rotor(${}^{MBD}\Lambda^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{\omega}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{a}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{\alpha}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{Ctrl}\Omega^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{Ctrl}T^{GenSPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{KAD}\vec{F}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{KAD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{g}$, ${}^{MBD}m^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{F}^{SPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right]$)

Call Rotor(${}^{MBD}\Lambda^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{\omega}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{a}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{\alpha}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{Ctrl}\Omega^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{Ctrl}T^{GenPPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{KAD}\vec{F}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{KAD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{g}$, ${}^{MBD}m^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{F}^{PPyRtr}\left[n_{Pylons}, n_2\right]$, ${}^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right]$)

where:

$${}^{Ctrl}T^{GenSPyRtr}\left[n_{Pylons}, n_2\right] = \begin{cases} {}^{Ctrl}Motor\,7 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=1\right)\right) \\ {}^{Ctrl}Motor\,2 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=2\right)\right) \\ {}^{Ctrl}Motor\,8 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=1\right)\right) \\ {}^{Ctrl}Motor\,1 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=2\right)\right) \end{cases}$$

$${}^{Ctrl}T^{GenPPyRtr}\left[n_{Pylons}, n_2\right] = \begin{cases} {}^{Ctrl}Motor\,6 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=1\right)\right) \\ {}^{Ctrl}Motor\,3 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=2\right)\right) \\ {}^{Ctrl}Motor\,5 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=1\right)\right) \\ {}^{Ctrl}Motor\,4 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=2\right)\right) \end{cases}$$

Private SUBROUTINES

Deleted: ${}^{MBD}\vec{F}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{KAD}\vec{F}^{SPyRtr}\left[n_{Pylo\ldots}\right.$

${}^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{KAD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right]$

${}^{MBD}\vec{F}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{KAD}\vec{F}^{PPyRtr}\left[n_{Pylons}, n_2\right]$

${}^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{KAD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right]$

Commented [JJ38]: See earlier comment about mesh mapping with Wn above.

Deleted: Transfer outputs from

Deleted:

Commented [JJ39]: When this is updated, update the same line in the Constructor routine above.

Formatted: Lowered by 8 pt

Deleted:

${}^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons}, n_2\right]$

${}^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons}, n_2\right]$

${}^{MBD}\vec{M}_1^{SPy}\left[n_{Pylons}\right] = {}^{MBD}\vec{M}_1^{SPy}\left[n_{Pylons}\right] + {}^{MBD}\hat{x}^{SPyRtr\ldots}$

${}^{MBD}\vec{M}_{NumPylNds}^{SPy}\left[n_{Pylons}\right] = {}^{MBD}\vec{M}_{NumPylNds}^{SPy}\left[n_{Pylons}\right] +$

${}^{MBD}\vec{M}_1^{PPy}\left[n_{Pylons}\right] = {}^{MBD}\vec{M}_1^{PPy}\left[n_{Pylons}\right] + {}^{MBD}\hat{x}^{PPyR\ldots}$

${}^{MBD}\vec{M}_{NumPylNds}^{PPy}\left[n_{Pylons}\right] = {}^{MBD}\vec{M}_{NumPylNds}^{PPy}\left[n_{Pylons}\right] +$

Commented [JJ41]: This math assumes the top node of the pylon is node 1 and that the pylons are numbered from inboard to outboard.

Rotor (SUBROUTINE Rotor)

Implements the structural dynamics of a rotor/drivetrain analytically to calculate the reaction loads (forces and moments) applied on the nacelle, including the applied aerodynamic loads, rotor inertial loads, rotor gyroscopic loads, etc. The analytical formulation assumes that the rotor/drivetrain is a rigid body rotating about the local x-axis of the nacelle coordinate system and that the structure is axisymmetric about this axis (with no imbalances) such that the calculations do not depend on the azimuth angle of the rotor. That is, for a body-fixed (x,y,z) coordinate system in the rotor/drivetrain, it is assumed that:

$$ {}^{CM}y = {}^{CM}z = 0 $$

$$ I_{xy} = I_{yz} = I_{xz} = 0 $$

$$ I_{xx} = I^{Rot} $$

$$ I_{yy} = I_{zz} = I^{Tran} $$

Inputs	Outputs	States	Parameters
• Λ^{Nac} – Displaced rotation (absolute orientation) of the nacelle (-) • $\vec{\omega}^{Nac}$ – Rotational velocity (absolute) of the nacelle (rad/s) • $\vec{a}^{Nac}$ – Translational acceleration (absolute) of the nacelle at the rotor reference point (m/s^2) • $\vec{\alpha}^{Nac}$ – Rotational acceleration (absolute) of the nacelle (rad/s^2) • Ω^{Rtr} – Rotor speed about the shaft axis (relative to the nacelle) (rad/s) • T^{Gen} – electrical generator torque applied to the rotor/drivetrain about the shaft axis (N·m) • $\vec{F}^{Aero}$ – aerodynamic forces applied on the rotor at the rotor reference point expressed in the global inertial-frame coordinate system (N) • $\vec{M}^{Aero}$ – aerodynamic moments applied on the rotor about the	• $\vec{F}^{React}$ – reaction forces applied on the nacelle at the rotor reference point expressed in the global inertial-frame coordinate system (N) • $\vec{M}^{React}$ – reaction moments applied on the nacelle about the rotor reference point expressed in the global inertial-frame coordinate system (N·m)		

Commented [JJ42]: This is input in place of:

$\dot{\Omega}^{Rtr}$ – Rotor acceleration about the shaft axis (relative to the nacelle) (rad/s^2)

rotor reference point expressed in the global inertial-frame coordinate system (N·m) • $\vec{g}$ – gravity vector expressed in the global inertial-frame coordinate system (m/s^2) • m – rotor/drivetrain mass (kg) • I^{Rot} – rotor/drivetrain rotational inertia about the shaft axis (kg·m^2) • I^{Tran} – rotor/drivetrain transverse inertia about the rotor reference point (kg·m^2) • ${}^{CM}x$ – distance along the shaft from the rotor reference point to the center of mass of the rotor/drivetrain (positive along positive x) (m)			

Compute the inputs relative to the rotor/drivetrain CM and expressed in the local nacelle coordinate system:

$${}^{CM}\vec{r} = {}^{CM}x\hat{x}^{Nac}$$

$${}^{CM}I^{Tran} = I^{Tran} - m\,{}^{CM}x^2$$

$$\begin{Bmatrix} {}^{CM}F_x^{Aero} \\ {}^{CM}F_y^{Aero} \\ {}^{CM}F_z^{Aero} \end{Bmatrix} = \Lambda^{Nac}\vec{F}^{Aero}$$

$$\begin{Bmatrix} {}^{CM}M_x^{Aero} \\ {}^{CM}M_y^{Aero} \\ {}^{CM}M_z^{Aero} \end{Bmatrix} = \Lambda^{Nac}\left\{\vec{M}^{Aero} - {}^{CM}\vec{r}\times\vec{F}^{Aero}\right\}$$

$$\begin{Bmatrix} g_x \\ g_y \\ g_z \end{Bmatrix} = \Lambda^{Nac}\vec{g}$$

$$\vec{\omega}^{Rtr} = \vec{\omega}^{Nac} + \Omega^{Rtr}\hat{x}^{Nac}$$

$$\vec{\alpha}^{Rtr} = \vec{\alpha}^{Nac}$$

Commented [JJ43]: The equation implemented neglects the rotor acceleration about the shaft axis. The correct equation should be:

$$\vec{\alpha}^{Rtr} = \vec{\alpha}^{Nac} + \dot{\Omega}^{Rtr}\hat{x}^{Nac}$$

, but the rotor acceleration about the shaft axis is not needed because the generator torque is input instead.

$$\begin{Bmatrix} \omega_x^{Rtr} \\ \omega_y^{Rtr} \\ \omega_z^{Rtr} \end{Bmatrix} = \Lambda^{Nac} \vec{\omega}^{Rtr}$$

$$\begin{Bmatrix} {}^{CM}a_x^{Rtr} \\ {}^{CM}a_y^{Rtr} \\ {}^{CM}a_z^{Rtr} \end{Bmatrix} = \Lambda^{Nac} \left\{ \vec{a}^{Nac} + \vec{\alpha}^{Rtr} \times {}^{CM}\vec{r} + \vec{\omega}^{Rtr} \times \left\{ \vec{\omega}^{Rtr} \times {}^{CM}\vec{r} \right\} \right\}$$

$$\begin{Bmatrix} \alpha_x^{Rtr} \\ \alpha_y^{Rtr} \\ \alpha_z^{Rtr} \end{Bmatrix} = \Lambda^{Nac} \vec{\alpha}^{Rtr}$$

Compute the reaction loads applied to the rotor/drivetrain at the rotor/drivetrain CM and expressed in the local nacelle coordinate system:

$$\begin{Bmatrix} {}^{CM}F_x^{React} \\ {}^{CM}F_y^{React} \\ {}^{CM}F_z^{React} \end{Bmatrix} = \begin{Bmatrix} -{}^{CM}F_x^{Aero} - mg_x + m\,{}^{CM}a_x^{Rtr} \\ -{}^{CM}F_y^{Aero} - mg_y + m\,{}^{CM}a_y^{Rtr} \\ -{}^{CM}F_z^{Aero} - mg_z + m\,{}^{CM}a_z^{Rtr} \end{Bmatrix}$$

$$\begin{Bmatrix} {}^{CM}M_x^{React} \\ {}^{CM}M_y^{React} \\ {}^{CM}M_z^{React} \end{Bmatrix} = \begin{Bmatrix} T^{Gen} \\ -{}^{CM}M_y^{Aero} + I^{Rot}\alpha_y^{Rtr} + \left(I^{Rot} - {}^{CM}I^{Tran}\right)\omega_z^{Rtr}\omega_x^{Rtr} \\ -{}^{CM}M_z^{Aero} + I^{Rot}\alpha_z^{Rtr} - \left(I^{Rot} - {}^{CM}I^{Tran}\right)\omega_y^{Rtr}\omega_x^{Rtr} \end{Bmatrix}$$

Commented [JJ44]: The first equation should be:

$${}^{CM}M_x^{React} = -{}^{CM}M_x^{Aero} + I^{Rot}\alpha_x^{Rtr}$$

But this equals the equation implemented because the generator torque is input instead of the rotor acceleration about the shaft axis.

Compute the reaction loads applied to the nacelle (this is equal, but opposite to the reaction loads applied to the rotor/drivetrain) at the rotor/drivetrain reference point and expressed in the global inertial frame coordinate system:

$$\vec{F}^{React} = -\left[\Lambda^{Nac}\right]^T \begin{Bmatrix} {}^{CM}F_x^{React} \\ {}^{CM}F_y^{React} \\ {}^{CM}F_z^{React} \end{Bmatrix}$$

$$\vec{M}^{React} = -\left[\Lambda^{Nac}\right]^T \begin{Bmatrix} {}^{CM}M_x^{React} \\ {}^{CM}M_y^{React} \\ {}^{CM}M_z^{React} \end{Bmatrix} + {}^{CM}\vec{r} \times \vec{F}^{React}$$

AfterPredict

This routine updates the actual states based on the temporary states at the successful completion of time step $t + \Delta t$.

$$NewTime = TRUE$$
$${}^{MD}x = {}^{MD}x^{Copy}$$
$${}^{KAD}z = {}^{KAD}z^{Copy}$$

Output

This routine is called at initialization ($t = 0$) and at the successful completion of time step $t + \Delta t$ to write output data to a file.

Calculate the KiteFASTMBD write outputs and write them to the output file, together with the module-level write output data currently stored in MiscVars.

This is a list of all possible output parameters available within the KiteFASTMBD (not including the module-level outputs available from KiteAeroDyn, InflowWind, MoorDyn, and the Controller). The names are grouped by meaning, but can be ordered in the OUTPUTS section of the KiteMBDyn Preprocessor input file as you see fit.

Fusβ refers to output β on the fuselage, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***FusOutNd*** list. Setting β > ***NFusOuts*** yields invalid output.

SWnβ and PWnβ refer to output β on the starboard and port wings, respectively, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***SWnOutNd*** and ***PWnOutNd*** lists, respectively. Setting β > ***NSWnOuts*** and ***NPWnOuts***, respectively, yields invalid output.

VSβ refers to output β on the vertical stabilizer, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***VSOutNd*** list. Setting β > ***NVSOuts*** yields invalid output.

SHSβ and PHSβ refer to output β on the starboard and port horizontal stabilizers, respectively, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***SHSOutNd*** and ***PHSOutNd*** lists, respectively. Setting β > ***NSHSOuts*** and ***NPHSOuts***, respectively, yields invalid output.

SPα and PPα refer to pylon α on the starboard and port wings, respectively, where α is a one-digit number in the range [1,9]. SPαβ and PPαβ refer to output β on pylon α on the starboard and port wings, respectively, where α is a one-digit number in the range [1,9] and β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***PylOutNd*** list. Setting α > ***NumPylons*** or setting β > ***NPylOuts*** yields invalid output. If ***NumPylons*** > 9, only the first 9 pylons can be output.

For the fuselage, wings, vertical stabilizer, horizontal stabilizers, and pylons, the local structural coordinate system is used for output, where n is normal to the chord pointed toward the suction surface, c is along the chord pointed toward the trailing edge, and the spanwise (s) axis is directed into the airfoil following the right-hand rule i.e. $s = n \times c$.

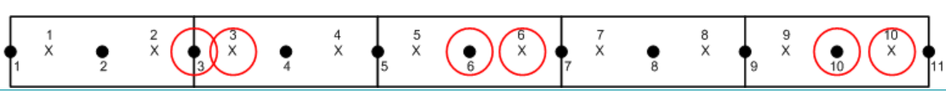


Figure: Example member with 5 finite elements, 11 nodes (•), and 10 Gauss points (x) (each finite element in MBDyn has 2 end nodes, 1 middle node, and 2 Gauss points). The red circles identify the finite-element nodes where motions are output and Gauss points where loads are output when ***NOuts*** = 3 and ***OutNd*** = 3, 6, 10.

Channel Name(s)	Unit(s)	Description
Fuselage		
FusβTDx, FusβTDy, FusβTDz, FusβRDx, FusβRDy, FusβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at Fusβ relative to the undeflected rigid-body position/orientation in the kite coordinate system;

Deleted: Write

Deleted: to the output file

Deleted: ¶

Commented [JJ45]: The new OUTPUT section of the KiteMBDyn Preprocessor input file should look something like this:

--- OUTPUT ---
True SumPrint Print summary data to <RootName>.sum? (flag)
"ES10.3E2" OutFmt Format used for text tabular output, excluding the time channel; resulting field should be 10 characters (string)
4 NFusOuts Number of fuselage outputs (-) [0 to 9]
2, 4, 6, 8 FusOutNd List of fuselage nodes/points whose values will be output (-) [1 to NFusOuts] [unused for NFusOuts=0]
4 NSWnOuts Number of starboard wing outputs (-) [0 to 9]
2, 4, 6, 8 SWnOutNd List of starboard wing nodes/points whose values will be output (-) [1 to NSWnOuts] [unused for NSWnOuts=0]
4 NPWnOuts Number of port wing outputs (-) [0 to 9]
2, 4, 6, 8 PWnOutNd List of port wing nodes/points whose values will be output (-) [1 to NPWnOuts] [unused for NPWnOuts=0]
2 NVSOuts Number of vertical stabilizer outputs (-) [0 to 9]
2, 4 VSOutNd List of vertical stabilizer nodes/points whose values will be output (-) [1 to NVSOuts] [unused for NVSOuts =0]
1 NSHSOuts Number of starboard horizontal stabilizer outputs (-) [0 to 9]
2 SHSOutNd List of starboard horizontal stabilizer nodes/points whose values will be output (-) [1 to NSHSOuts] [unused for NSHSOuts=0]
1 NPHSOuts Number of port horizontal stabilizer outputs (-) [0 to 9]
2 PHSOutNd List of port horizontal stabilizer nodes/points whose values will be output (-) [1 to NPHSOuts] [unused for NPHSOuts=0]
2 NPylOuts Number of pylon outputs (-) [0 to 9]
2, 4 PylOutNd List of pylon nodes/points whose values will be output (-) [1 to NPylOuts] [unused for NPylOuts=0]
OutList The next line(s) contains a list of output parameters. See OutListParameters.xlsx for a listing of available output channels (quoted string)
END of input file (the word "END" must appear in the first 3 columns of this last OutList line)

		the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
FusβRVn, FusβRVc, FusβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at Fusβ expressed in the local structural coordinate system
FusβTAn, FusβTAc, FusβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at Fusβ expressed in the local structural coordinate system (does not include gravity)
FusβFRn, FusβFRc, FusβFRs, FusβMRn, FusβMRc, FusβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at Fusβ expressed in the local structural coordinate system
Starboard (Right) Wing		
SWnβTDx, SWnβTDy, SWnβTDz, SWnβRDx, SWnβRDy, SWnβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at SWnβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
SWnβRVn, SWnβRVc, SWnβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at SWnβ expressed in the local structural coordinate system
SWnβTAn, SWnβTAc, SWnβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at SWnβ expressed in the local structural coordinate system (does not include gravity)
SWnβFRn, SWnβFRc, SWnβFRs, SWnβMRn, SWnβMRc, SWnβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at SWnβ expressed in the local structural coordinate system
Port (Left) Wing		
PWnβTDx, PWnβTDy, PWnβTDz, PWnβRDx, PWnβRDy, PWnβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at PWnβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
PWnβRVn, PWnβRVc, PWnβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at PWnβ expressed in the local structural coordinate system
PWnβTAn, PWnβTAc, PWnβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at PWnβ expressed in the local structural coordinate system (does not include gravity)
PWnβFRn, PWnβFRc, PWnβFRs, PWnβMRn, PWnβMRc, PWnβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at PWnβ expressed in the local structural coordinate system
Vertical Stabilizer		
VSβTDx, VSβTDy, VSβTDz, VSβRDx, VSβRDy, VSβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at VSβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
VSβRVn, VSβRVc, VSβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at VSβ expressed in the local structural coordinate system
VSβTAn, VSβTAc, VSβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at VSβ expressed in the local structural coordinate system (does not include gravity)
VSβFRn, VSβFRc, VSβFRs, VSβMRn, VSβMRc, VSβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at VSβ expressed in the local structural coordinate

		system
Starboard (Right) Horizontal Stabilizer		
SHSβTDx, SHSβTDy, SHSβTDz, SHSβRDx, SHSβRDy, SHSβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at SHSβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
SHSβRVn, SHSβRVc, SHSβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at SHSβ expressed in the local structural coordinate system
SHSβTAn, SHSβTAc, SHSβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at SHSβ expressed in the local structural coordinate system (does not include gravity)
SHSβFRn, SHSβFRc, SHSβFRs, SHSβMRn, SHSβMRc, SHSβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at SHSβ expressed in the local structural coordinate system
Port (Left) Horizontal Stabilizer		
PHSβTDx, PHSβTDy, PHSβTDz, PHSβRDx, PHSβRDy, PHSβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at PHSβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
PHSβRVn, PHSβRVc, PHSβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at PHSβ expressed in the local structural coordinate system
PHSβTAn, PHSβTAc, PHSβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at PHSβ expressed in the local structural coordinate system (does not include gravity)
PHSβFRn, PHSβFRc, PHSβFRs, PHSβMRn, PHSβMRc, PHSβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at PHSβ expressed in the local structural coordinate system
Pylons		
SPαβTDx, SPαβTDy, SPαβTDz, SPαβRDx, SPαβRDy, SPαβRDz, PPαβTDx, PPαβTDy, PPαβTDz, PPαβRDx, PPαβRDy, PPαβRDz	(m), (m), (m), (deg), (deg), (deg), (m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at SPαβ and PPαβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
SPαβRVn, SPαβRVc, SPαβRVs, PPαβRVn, PPαβRVc, PPαβRVs	(deg/s), (deg/s), (deg/s), (deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at SPαβ and PPαβ expressed in the local structural coordinate system
SPαβTAn, SPαβTAc, SPαβTAs, PPαβTAn, PPαβTAc, PPαβTAs	(m/s^2), (m/s^2), (m/s^2), (m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at SPαβ and PPαβ expressed in the local structural coordinate system (does not include gravity)
SPαβFRn, SPαβFRc, SPαβFRs, SPαβMRn, SPαβMRc, SPαβMRs, PPαβFRn, PPαβFRc, PPαβFRs, PPαβMRn, PPαβMRc, PPαβMRs	(N), (N), (N), (N·m), (N·m), (N·m), (N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at SPαβ and PPαβ expressed in the local structural coordinate system
Rotors		
SPαTRtSpd, SPαBRtSpd, PPαTRtSpd, PPαBRtSpd	(rad/s), (rad/s), (rad/s), (rad/s)	Rotor speed of the top (T) and bottom (B) rotor on SPα and PPα (relative to the nacelle)
SPαTRtAcc, SPαBRtAcc, PPαTRtAcc, PPαBRtAcc	(rad/s^2), (rad/s^2), (rad/s^2), (rad/s^2)	Rotor acceleration of the top (T) and bottom (B) rotor on SPα and PPα (relative to the nacelle)
Energy Kite		
KitePxi, KitePyi, KitePzi,	(m), (m), (m),	Translational position and rotational (angular)

Deleted: speed

KiteRoll, KitePitch, KiteYaw	(deg), (deg), (deg)	orientation of the energy kite fuselage reference point in the global inertial-frame coordinate system; the rotations are output as Euler angles in a X-Y'-Z'' (roll-pitch-yaw) rotation sequence
KiteTVx, KiteTVy, KiteTVz, KiteRVx, KiteRVy, KiteRVz	(m/s), (m/s), (m/s), (deg/s), (deg/s), (deg/s)	Absolute translational and rotational (angular) velocity of the energy kite fuselage reference point expressed in the kite coordinate system
KiteTAx, KiteTAy, KiteTAz, KiteRAx, KiteRAy, KiteRAz	(m/s^2), (m/s^2), (m/s^2), (deg/s^2), (deg/s^2), (deg/s^2)	Absolute translational and rotational (angular) acceleration of the energy kite fuselage reference point expressed in the kite coordinate system

These are calculated within KiteFASTMBD as follows:

Fuselage:

$$\begin{Bmatrix} Fus\beta TDx \\ Fus\beta TDy \\ Fus\beta TDz \\ Fus\beta RDx \\ Fus\beta RDy \\ Fus\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{\,Fus}_{FusOutNd[\beta]} - {}^{MBD}\vec{p}^{\,FusO} \right\} - {}^{MBD}\vec{p}^{\,FusR}_{FusOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{FusR}_{FusOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{Fus}_{FusOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} Fus\beta RVn \\ Fus\beta RVc \\ Fus\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{Fus}_{FusOutNd[\beta]} \; {}^{MBD}\vec{\omega}^{\,Fus}_{FusOutNd[\beta]}$$

$$\begin{Bmatrix} Fus\beta TAn \\ Fus\beta TAc \\ Fus\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{Fus}_{FusOutNd[\beta]} \; {}^{MBD}\vec{a}^{\,Fus}_{FusOutNd[\beta]}$$

$$\begin{Bmatrix} Fus\beta FRn \\ Fus\beta FRc \\ Fus\beta FRs \\ Fus\beta MRn \\ Fus\beta MRc \\ Fus\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{FR}^{Fus}_{FusOutNd[\beta]} \\ {}^{MBD}\vec{MR}^{Fus}_{FusOutNd[\beta]} \end{Bmatrix}$$

Starboard (Right) Wing:

$$\begin{Bmatrix} SWn\beta TDx \\ SWn\beta TDy \\ SWn\beta TDz \\ SWn\beta RDx \\ SWn\beta RDy \\ SWn\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{\,SWn}_{SWnOutNd[\beta]} - {}^{MBD}\vec{p}^{\,FusO} \right\} - {}^{MBD}\vec{p}^{\,SWnR}_{SWnOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{SWnR}_{SWnOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{SWn}_{SWnOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} SWn\beta RVn \\ SWn\beta RVc \\ SWn\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{SWn}_{SWnOutNd[\beta]} \; {}^{MBD}\vec{\omega}^{SWn}_{SWnOutNd[\beta]}$$

$$\begin{Bmatrix} SWn\beta TAn \\ SWn\beta TAc \\ SWn\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{SWn}_{SWnOutNd[\beta]} \; {}^{MBD}\vec{a}^{SWn}_{SWnOutNd[\beta]}$$

$$\begin{Bmatrix} SWn\beta FRn \\ SWn\beta FRc \\ SWn\beta FRs \\ SWn\beta MRn \\ SWn\beta MRc \\ SWn\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{SWn}_{SWnOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{SWn}_{SWnOutNd[\beta]} \end{Bmatrix}$$

Port (Left) Wing:

$$\begin{Bmatrix} PWn\beta TDx \\ PWn\beta TDy \\ PWn\beta TDz \\ PWn\beta RDx \\ PWn\beta RDy \\ PWn\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PWn}_{PWnOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{PWnR}_{PWnOutNd[\beta]} \\ \frac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{PWnR}_{PWnOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{PWn}_{PWnOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} PWn\beta RVn \\ PWn\beta RVc \\ PWn\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{PWn}_{PWnOutNd[\beta]} \; {}^{MBD}\vec{\omega}^{PWn}_{PWnOutNd[\beta]}$$

$$\begin{Bmatrix} PWn\beta TAn \\ PWn\beta TAc \\ PWn\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{PWn}_{PWnOutNd[\beta]} \; {}^{MBD}\vec{a}^{PWn}_{PWnOutNd[\beta]}$$

$$\begin{Bmatrix} PWn\beta FRn \\ PWn\beta FRc \\ PWn\beta FRs \\ PWn\beta MRn \\ PWn\beta MRc \\ PWn\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{PWn}_{PWnOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{PWn}_{PWnOutNd[\beta]} \end{Bmatrix}$$

Vertical Stabilizer:

$$\begin{Bmatrix} VS\beta TDx \\ VS\beta TDy \\ VS\beta TDz \\ VS\beta RDx \\ VS\beta RDy \\ VS\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{VS}_{VSOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{VSR}_{VSOutNd[\beta]} \\ \frac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{VSR}_{VSOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{VS}_{VSOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} VS\beta RVn \\ VS\beta RVc \\ VS\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{VS}_{VSOutNd[\beta]} \, {}^{MBD}\vec{\omega}^{VS}_{VSOutNd[\beta]}$$

$$\begin{Bmatrix} VS\beta TAn \\ VS\beta TAc \\ VS\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{VS}_{VSOutNd[\beta]} \, {}^{MBD}\vec{a}^{VS}_{VSOutNd[\beta]}$$

$$\begin{Bmatrix} VS\beta FRn \\ VS\beta FRc \\ VS\beta FRs \\ VS\beta MRn \\ VS\beta MRc \\ VS\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{FR}^{VS}_{VSOutNd[\beta]} \\ {}^{MBD}\vec{MR}^{VS}_{VSOutNd[\beta]} \end{Bmatrix}$$

Starboard (Right) Horizontal Stabilizer:

$$\begin{Bmatrix} SHS\beta TDx \\ SHS\beta TDy \\ SHS\beta TDz \\ SHS\beta RDx \\ SHS\beta RDy \\ SHS\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SHS}_{SHSOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{SHSR}_{SHSOutNd[\beta]} \\ \frac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{SHSR}_{SHSOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{SHS}_{SHSOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} SHS\beta RVn \\ SHS\beta RVc \\ SHS\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{SHS}_{SHSOutNd[\beta]} \, {}^{MBD}\vec{\omega}^{SHS}_{SHSOutNd[\beta]}$$

$$\begin{Bmatrix} SHS\beta TAn \\ SHS\beta TAc \\ SHS\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{SHS}_{SHSOutNd[\beta]} \, {}^{MBD}\vec{a}^{SHS}_{SHSOutNd[\beta]}$$

$$\left\{\begin{matrix} SHS\beta FRn \\ SHS\beta FRc \\ SHS\beta FRs \\ SHS\beta MRn \\ SHS\beta MRc \\ SHS\beta MRs \end{matrix}\right\} = \left\{\begin{matrix} {}^{MBD}\vec{F}R^{SHS}_{SHSOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{SHS}_{SHSOutNd[\beta]} \end{matrix}\right\}$$

Port (Left) Horizontal Stabilizer:

$$\left\{\begin{matrix} PHS\beta TDx \\ PHS\beta TDy \\ PHS\beta TDz \\ PHS\beta RDx \\ PHS\beta RDy \\ PHS\beta RDz \end{matrix}\right\} = \left\{\begin{matrix} {}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{PHS}_{PHSOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO}\right\} - {}^{MBD}\vec{p}^{PHSR}_{PHSOutNd[\beta]} \\ \frac{180}{\pi}F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO}\right]^T\left[{}^{MBD}\Lambda^{PHSR}_{PHSOutNd[\beta]}\right]^T {}^{MBD}\Lambda^{PHS}_{PHSOutNd[\beta]}\right) \end{matrix}\right\}$$

$$\left\{\begin{matrix} PHS\beta RVn \\ PHS\beta RVc \\ PHS\beta RVs \end{matrix}\right\} = \frac{180}{\pi}\,{}^{MBD}\Lambda^{PHS}_{PHSOutNd[\beta]}\;{}^{MBD}\vec{\omega}^{PHS}_{PHSOutNd[\beta]}$$

$$\left\{\begin{matrix} PHS\beta TAn \\ PHS\beta TAc \\ PHS\beta TAs \end{matrix}\right\} = {}^{MBD}\Lambda^{PHS}_{PHSOutNd[\beta]}\;{}^{MBD}\vec{a}^{PHS}_{PHSOutNd[\beta]}$$

$$\left\{\begin{matrix} PHS\beta FRn \\ PHS\beta FRc \\ PHS\beta FRs \\ PHS\beta MRn \\ PHS\beta MRc \\ PHS\beta MRs \end{matrix}\right\} = \left\{\begin{matrix} {}^{MBD}\vec{F}R^{PHS}_{PHSOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{PHS}_{PHSOutNd[\beta]} \end{matrix}\right\}$$

Pylons:

$$\begin{Bmatrix} SP\alpha\beta TDx \\ SP\alpha\beta TDy \\ SP\alpha\beta TDz \\ SP\alpha\beta RDx \\ SP\alpha\beta RDy \\ SP\alpha\beta RDz \\ PP\alpha\beta TDx \\ PP\alpha\beta TDy \\ PP\alpha\beta TDz \\ PP\alpha\beta RDx \\ PP\alpha\beta RDy \\ PP\alpha\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{\,SPy}_{PylOutNd[\beta]}[\alpha] - {}^{MBD}\vec{p}^{\,FusO} \right\} - {}^{MBD}\vec{p}^{\,SPyR}_{PylOutNd[\beta]}[\alpha] \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{SPyR}_{PylOutNd[\beta]}[\alpha] \right]^T {}^{MBD}\Lambda^{SPy}_{PylOutNd[\beta]}[\alpha] \right) \\ {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{\,PPy}_{PylOutNd[\beta]}[\alpha] - {}^{MBD}\vec{p}^{\,FusO} \right\} - {}^{MBD}\vec{p}^{\,PPyR}_{PylOutNd[\beta]}[\alpha] \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{PPyR}_{PylOutNd[\beta]}[\alpha] \right]^T {}^{MBD}\Lambda^{PPy}_{PylOutNd[\beta]}[\alpha] \right) \end{Bmatrix}$$

$$\begin{Bmatrix} SP\alpha\beta RVn \\ SP\alpha\beta RVc \\ SP\alpha\beta RVs \\ PP\alpha\beta RVn \\ PP\alpha\beta RVc \\ PP\alpha\beta RVs \end{Bmatrix} = \begin{Bmatrix} \dfrac{180}{\pi}\, {}^{MBD}\Lambda^{SPy}_{PylOutNd[\beta]}[\alpha]\, {}^{MBD}\vec{\omega}^{\,SPy}_{PylOutNd[\beta]}[\alpha] \\ \dfrac{180}{\pi}\, {}^{MBD}\Lambda^{PPy}_{PylOutNd[\beta]}[\alpha]\, {}^{MBD}\vec{\omega}^{\,PPy}_{PylOutNd[\beta]}[\alpha] \end{Bmatrix}$$

$$\begin{Bmatrix} SP\alpha\beta TAn \\ SP\alpha\beta TAc \\ SP\alpha\beta TAs \\ PP\alpha\beta TAn \\ PP\alpha\beta TAc \\ PP\alpha\beta TAs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{SPy}_{PylOutNd[\beta]}[\alpha]\, {}^{MBD}\vec{a}^{\,SPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\Lambda^{PPy}_{PylOutNd[\beta]}[\alpha]\, {}^{MBD}\vec{a}^{\,PPy}_{PylOutNd[\beta]}[\alpha] \end{Bmatrix}$$

$$\begin{Bmatrix} SP\alpha\beta FRn \\ SP\alpha\beta FRc \\ SP\alpha\beta FRs \\ SP\alpha\beta MRn \\ SP\alpha\beta MRc \\ SP\alpha\beta MRs \\ PP\alpha\beta FRn \\ PP\alpha\beta FRc \\ PP\alpha\beta FRs \\ PP\alpha\beta MRn \\ PP\alpha\beta MRc \\ PP\alpha\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{SPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\vec{M}R^{SPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\vec{F}R^{PPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\vec{M}R^{PPy}_{PylOutNd[\beta]}[\alpha] \end{Bmatrix}$$

Rotors

$$\begin{Bmatrix} SP\alpha TRtSpd \\ SP\alpha BRtSpd \\ PP\alpha TRtSpd \\ PP\alpha BRtSpd \end{Bmatrix} = \begin{Bmatrix} {}^{Ctrl}\Omega^{SPyRtr}[\alpha,1] \\ {}^{Ctrl}\Omega^{SPyRtr}[\alpha,2] \\ {}^{Ctrl}\Omega^{PPyRtr}[\alpha,1] \\ {}^{Ctrl}\Omega^{PPyRtr}[\alpha,2] \end{Bmatrix}$$

$$\begin{Bmatrix} SP\alpha TRtAcc \\ SP\alpha BRtAcc \\ PP\alpha TRtAcc \\ PP\alpha BRtAcc \end{Bmatrix} = \begin{Bmatrix} {}^{Ctrl}\alpha^{SPyRtr}[\alpha,1] \\ {}^{Ctrl}\alpha^{SPyRtr}[\alpha,2] \\ {}^{Ctrl}\alpha^{PPyRtr}[\alpha,1] \\ {}^{Ctrl}\alpha^{PPyRtr}[\alpha,2] \end{Bmatrix}$$

Energy Kite

$$\begin{Bmatrix} KitePxi \\ KitePyi \\ KitePzi \\ KiteRoll \\ KitePitch \\ KiteYaw \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{p}^{FusO} \\ \dfrac{180}{\pi} F^{EulerExtract}\left({}^{MBD}\Lambda^{FusO}\right) \end{Bmatrix}$$

$$\begin{Bmatrix} KiteTVx \\ KiteTVy \\ KiteTVz \\ KiteRVx \\ KiteRVy \\ KiteRVz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO\,MBD}\vec{v}^{FusO} \\ \dfrac{180}{\pi}\,{}^{MBD}\Lambda^{FusO\,MBD}\vec{\omega}^{FusO} \end{Bmatrix}$$

Deleted: $$\begin{Bmatrix} SP\alpha TRtSpd \\ SP\alpha BRtSpd \\ PP\alpha TRtSpd \\ PP\alpha BRtSpd \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Omega^{SPyRtr}[\alpha,1] \\ {}^{MBD}\Omega^{SPyRtr}[\alpha,2] \\ {}^{MBD}\Omega^{PPyRtr}[\alpha,1] \\ {}^{MBD}\Omega^{PPyRtr}[\alpha,2] \end{Bmatrix}$$

Commented [JJ46]: This can only be output if the controller outputs the rotor acceleration in addition to the generator torque. Does it? If not, this output should be removed.

Deleted: $$\begin{Bmatrix} SP\alpha TRtAcc \\ SP\alpha BRtAcc \\ PP\alpha TRtAcc \\ PP\alpha BRtAcc \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\alpha^{SPyRtr}[\alpha,1] \\ {}^{MBD}\alpha^{SPyRtr}[\alpha,2] \\ {}^{MBD}\alpha^{PPyRtr}[\alpha,1] \\ {}^{MBD}\alpha^{PPyRtr}[\alpha,2] \end{Bmatrix}$$

$$\begin{Bmatrix} KiteTAx \\ KiteTAy \\ KiteTAz \\ KiteRAx \\ KiteRAy \\ KiteRAz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO\,MBD}\vec{a}^{FusO} \\ \frac{180}{\pi}\,{}^{MBD}\Lambda^{FusO\,MBD}\vec{\alpha}^{FusO} \end{Bmatrix}$$