Rotation Notation/Convention

$$\begin{Bmatrix} x \\ y \\ z \end{Bmatrix} = [\Lambda] \begin{Bmatrix} X \\ Y \\ Z \end{Bmatrix} = \begin{bmatrix} \hat{x}^T \\ \hat{y}^T \\ \hat{z}^T \end{bmatrix} \begin{Bmatrix} X \\ Y \\ Z \end{Bmatrix} \text{ (from global to local)}$$

or equivalently:

$$\begin{Bmatrix} X \\ Y \\ Z \end{Bmatrix} = [\Lambda]^T \begin{Bmatrix} x \\ y \\ z \end{Bmatrix} == \begin{bmatrix} \hat{x} & \hat{y} & \hat{z} \end{bmatrix} \begin{Bmatrix} x \\ y \\ z \end{Bmatrix}$$

where *X*/*Y*/*Z* are global coordinates, *x*/*y*/*z* are local coordinates, Λ is the DCM from global to local, and $\hat{x} / \hat{y} / \hat{z}$ are the unit vectors of the local coordinate system expressed in the global coordinate system.

$$\begin{Bmatrix} \theta_x \\ \theta_y \\ \theta_z \end{Bmatrix} = F^{EulerExtract}\left(\left[\Lambda\left(\theta_x, \theta_y, \theta_z\right)\right]\right)$$

where function $F^{EulerExtract}(\ )$ returns the 3 Euler angles of the *x-y-z* (1-2-3) rotation sequence used to form Λ (that is, first a rotation θ_x about the global *X* axis, followed by rotation θ_y about the *Y'* axis, followed by rotation θ_z about the *Z''* axis) defined as follows:

$$\Lambda\left(\theta_x,\theta_y,\theta_z\right) = \begin{bmatrix} COS(\theta_z) & SIN(\theta_z) & 0 \\ -SIN(\theta_z) & COS(\theta_z) & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} COS(\theta_y) & 0 & -SIN(\theta_y) \\ 0 & 1 & 0 \\ SIN(\theta_y) & 0 & COS(\theta_y) \end{bmatrix} \begin{bmatrix} 1 & 0 & 0 \\ 0 & COS(\theta_x) & SIN(\theta_x) \\ 0 & -SIN(\theta_x) & COS(\theta_x) \end{bmatrix}$$

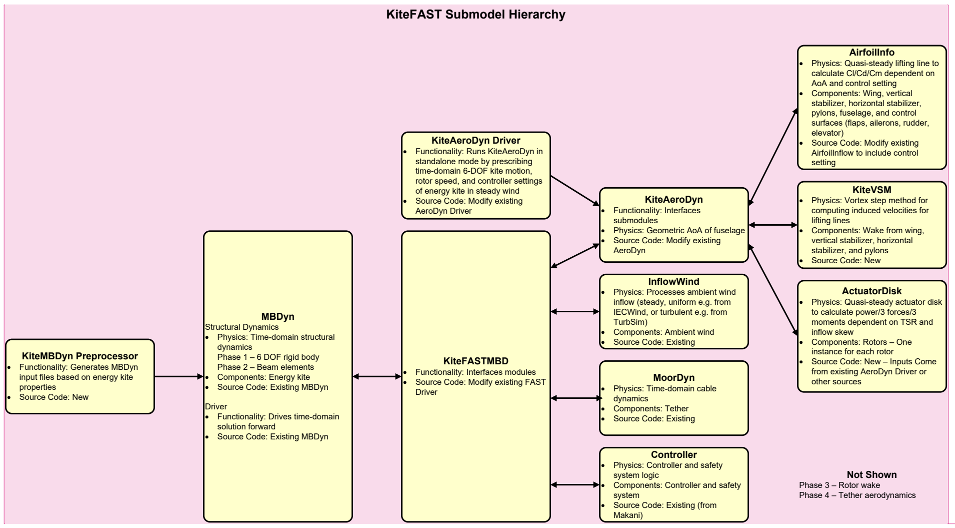
$$= \begin{bmatrix} COS(\theta_y)COS(\theta_z) & COS(\theta_x)SIN(\theta_z) + SIN(\theta_x)SIN(\theta_y)COS(\theta_z) & SIN(\theta_x)SIN(\theta_z) - COS(\theta_x)SIN(\theta_y)COS(\theta_z) \\ -COS(\theta_y)SIN(\theta_z) & COS(\theta_x)COS(\theta_z) - SIN(\theta_x)SIN(\theta_y)SIN(\theta_z) & SIN(\theta_x)COS(\theta_z) + COS(\theta_x)SIN(\theta_y)SIN(\theta_z) \\ SIN(\theta_y) & -SIN(\theta_x)COS(\theta_y) & COS(\theta_x)COS(\theta_y) \end{bmatrix}$$

Note the following simplifications:

$$\Lambda\left(0,\theta_y,\theta_z\right) = \begin{bmatrix} COS(\theta_y)COS(\theta_z) & SIN(\theta_z) & -SIN(\theta_y)COS(\theta_z) \\ -COS(\theta_y)SIN(\theta_z) & COS(\theta_z) & SIN(\theta_y)SIN(\theta_z) \\ SIN(\theta_y) & 0 & COS(\theta_y) \end{bmatrix}$$

$$\Lambda\left(\theta_x,0,\theta_z\right) = \begin{bmatrix} COS(\theta_z) & COS(\theta_x)SIN(\theta_z) & SIN(\theta_x)SIN(\theta_z) \\ -SIN(\theta_z) & COS(\theta_x)COS(\theta_z) & SIN(\theta_x)COS(\theta_z) \\ 0 & -SIN(\theta_x) & COS(\theta_x) \end{bmatrix}$$

$$\Lambda\left(\theta_x,\theta_y,0\right) = \begin{bmatrix} COS(\theta_y) & SIN(\theta_x)SIN(\theta_y) & -COS(\theta_x)SIN(\theta_y) \\ 0 & COS(\theta_x) & SIN(\theta_x) \\ SIN(\theta_y) & -SIN(\theta_x)COS(\theta_y) & COS(\theta_x)COS(\theta_y) \end{bmatrix}$$



KiteFASTMBD

Inputs	Outputs	States	Parameters
• ${}^{MBD}\vec{p}^{Wind}$ – Position of the ground station where the fixed wind measurement is taken (m) • ${}^{MBD}\vec{p}^{FusO}$ – Position (origin) of the fuselage (m) • ${}^{MBD}\Lambda^{FusO}$ – Rotation (absolute orientation) of the fuselage origin (-) • ${}^{MBD}\vec{v}^{FusO}$ – Translational velocity (absolute) of the fuselage origin (m/s) • ${}^{MBD}\vec{\omega}^{FusO}$ – Rotational velocity (absolute) of the fuselage origin (rad/s) • ${}^{MBD}\vec{a}^{FusO}$ – Translational acceleration (absolute) of the fuselage origin (m/s^2) • ${}^{MBD}\vec{\alpha}^{FusO}$ – Rotational acceleration (absolute) of the fuselage origin (rad/s^2) • ${}^{MBD}\vec{p}_j^{Fus}$ – Translational position (absolute) of the j th node of the fuselage mesh (m)	• ${}^{MBD}\vec{F}_j^{Fus}$ – Aerodynamic applied concentrated forces at the j th node of the fuselage mesh (N) • ${}^{MBD}\vec{M}_j^{Fus}$ – Aerodynamic applied concentrated moments at the j th node of the fuselage mesh (N-m) • ${}^{MBD}\vec{F}_j^{SWn}$ – Aerodynamic and tether applied concentrated forces at the j th node of the starboard wing mesh (N) • ${}^{MBD}\vec{M}_j^{SWn}$ – Aerodynamic applied and tether concentrated moments at the j th node of the starboard wing mesh (N-m) • ${}^{MBD}\vec{F}_j^{PWn}$ – Aerodynamic and tether applied concentrated forces at the j th node of the port wing	• ${}^{KAD}NewTime$ – Is this a new time step (in order to only call KiteAeroDyn once per step)? (flag) (other state) • ${}^{Ctrl}NewTime$ – Is this a new time step (in order to only call the controller once per step)? (flag) (other state) • ${}^{MBD}OtherStates$ – Inputs from MBDyn from the previous time step (stored as other states) • ${}^{MD}OtherStates$ – Inputs to MoorDyn from the previous time step (stored as other states) • ${}^{MD}x$ – MoorDyn continuous states (varied) • ${}^{KAD}z$ – KiteAeroDyn constraint states (varied) • ${}^{KAD}u(:)$ – Time history of KiteAeroDyn inputs (stored as other states)	• Δt – MBDyn time (s) • ${}^{KAD}\Delta t$ – KiteAeroD time step (s) • $InterpOrder$ – Interpolation order f input/output time his (-) {1=linear, 2=quadratic} • $N_{KAD/MBD}$ – Number of KiteAeroDyn time steps per MBDyn time step (-) • $N_{Ctrl/MBD}$ – Number of controller time steps MBDyn time step (- • N_{Flaps} – Number o per wing (-) • N_{Pylons} – Number c pylons per wing (-) • $\Lambda^{FAST2Ctrl}$ – DCM conversion from the FAST ground system (X pointed nominally downwind; Z pointed vertically opposite gravity; Y transverse) to

Commented [JJ1]: We've split up KiteFASTMBD into KiteFASTMBD in C and KiteFASTMBD in Fortran. This plan is for KiteFASTMBD in Fortran.

Commented [JJ2]: These are the data queried from the MBDyn model at t using GetXCur to be used within KiteFASTMBD. The outputs from MBDyn are inputs to KiteFASTMBD.

Commented [JJ3]: These are the data sent to the MBDyn model from KiteFASTMBD. The inputs to MBDyn are outputs from KiteFASTMBD.

Commented [JJ4]: Obvious parameters are not listed here.

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Deleted: <#> ${}^{KAD}OtherStates$ – Outputs from KiteAeroDyn from the previous time step (stored as other states)¶
<#> ${}^{IfW}OtherStates$ – Outputs from InflowWind from the previous time step (stored as other states)¶

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• $^{MBD}\Lambda_j^{Fus}$ – Displaced rotation (absolute orientation) of the j^{th} node of the fuselage mesh (-) • $^{MBD}\vec{v}_j^{Fus}$ – Translational velocity (absolute) of the j^{th} node of the fuselage mesh (m/s) • $^{MBD}\vec{\omega}_j^{Fus}$ – Rotational velocity (absolute) of the j^{th} node of the fuselage mesh (rad/s) • $^{MBD}\vec{a}_j^{Fus}$ – Translational acceleration (absolute) of the j^{th} node of the fuselage mesh (m/s^2) • $^{MBD}\vec{F}R_j^{Fus}$ – Reaction force (expressed in the local coordinate system) at the j^{th} Gauss point of the fuselage mesh (N) • $^{MBD}\vec{M}R_j^{Fus}$ – Reaction moment (expressed in the local coordinate system) at the j^{th} Gauss point of the fuselage mesh (N-m) • $^{MBD}\vec{p}^{SWnO}$ – Position (origin) of the starboard wing (m) • $^{MBD}\vec{p}_j^{SWn}$ – Translational position (absolute) of the j^{th} node of the starboard wing (m) • $^{MBD}\Lambda_j^{SWn}$ – Displaced rotation (absolute orientation) of the j^{th} node of the starboard wing mesh (-) • $^{MBD}\vec{v}_j^{SWn}$ – Translational velocity (absolute) of the j^{th} node of the starboard wing mesh (m/s) • $^{MBD}\vec{\omega}_j^{SWn}$ – Rotational velocity (absolute) of the	mesh (N) • $^{MBD}\vec{M}_j^{PWn}$ – Aerodynamic and tether applied concentrated moments at the j^{th} node of the port wing mesh (N-m) • $^{MBD}\vec{F}_j^{VS}$ – Aerodynamic applied concentrated forces at the j^{th} node of the vertical stabilizer mesh (N) • $^{MBD}\vec{M}_j^{VS}$ – Aerodynamic applied concentrated moments at the j^{th} node of the vertical stabilizer mesh (N-m) • $^{MBD}\vec{F}_j^{SHS}$ – Aerodynamic applied concentrated forces at the j^{th} node of the starboard horizontal stabilizer mesh (N) • $^{MBD}\vec{M}_j^{SHS}$ – Aerodynamic applied concentrated moments at the j^{th} node of the starboard horizontal stabilizer mesh (N-m) • $^{MBD}\vec{F}_j^{PHS}$ – Aerodynamic applied concentrated forces at the j^{th} node of the port horizontal stabilizer mesh (N) • $^{MBD}\vec{M}_j^{PHS}$ – Aerodynamic applied concentrated moments at the j^{th} node of the port horizontal stabilizer mesh (N-m) • $^{MBD}\vec{F}_j^{SPy}\left[n_{Pylons}\right]$ – Aerodynamic applied concentrated forces at the j^{th} node of the pylons on the starboard wing mesh (N) • $^{MBD}\vec{M}_j^{SPy}\left[n_{Pylons}\right]$ –	• $^{KAD}y(:)$ – Time history of KiteAeroDyn outputs (stored as other states) • $^{KAD}t(:)$ – Times associated with history of KiteAeroDyn inputs and outputs (stored as other states)	the ground system used by the controller (X pointed nominally upwind; Z pointed vertically downward, Y transverse) (-) • $^{MBD}\vec{g}$ – Gravity vector expressed in the global inertial-frame coordinate system (m/s^2) • ρ – Air density (kg/m^3) • $\vec{p}^{Anch}$ – Position of the ground station where the tether attaches (i.e. mooring line anchor) (m) • $^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Mass of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh (kg) • $^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational inertia about the shaft axis of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh (kg·m^2) • $^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Transverse inertia about the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh (kg·m^2) • $^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Distance along the shaft from the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the starboard wing mesh to the center of mass of the rotor/drivetrain (positive along positive x) (m) • $^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Mass of the top and bottom rotors/drivetrains on the pylons on the port wing mesh (kg)

j th node of the starboard wing mesh (rad/s) • ${}^{MBD}\vec{a}_j^{SWn}$ – Translational acceleration (absolute) of the j th node of the starboard wing mesh (m/s²) • ${}^{MBD}\vec{F}R_j^{SWn}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the starboard wing mesh (N) • ${}^{MBD}\vec{M}R_j^{SWn}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the starboard wing mesh (N-m) • ${}^{MBD}\vec{p}^{PWnO}$ – Position (origin) of the port wing (m) • ${}^{MBD}\vec{p}_j^{PWn}$ – Translational position (absolute) of the j th node of the port wing mesh (m) • ${}^{MBD}\Lambda_j^{PWn}$ – Displaced rotation (absolute orientation) of the j th node of the port wing mesh (-) • ${}^{MBD}\vec{v}_j^{PWn}$ – Translational velocity (absolute) of the j th node of the port wing mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{PWn}$ – Rotational velocity (absolute) of the j th node of the port wing mesh (rad/s) • ${}^{MBD}\vec{a}_j^{PWn}$ – Translational acceleration (absolute) of the j th node of the port wing mesh (m/s²) • ${}^{MBD}\vec{F}R_j^{PWn}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the port wing mesh (N)	Aerodynamic applied concentrated moments at the j th node of pylons on the starboard wing mesh (N-m) • ${}^{MBD}\vec{F}_j^{PPy}\left[n_{Pylons}\right]$ – Aerodynamic applied concentrated forces at the j th node of the pylons on the port wing mesh (N) • ${}^{MBD}\vec{M}_j^{PPy}\left[n_{Pylons}\right]$ – Aerodynamic applied concentrated moments at the j th node of pylons on the port wing mesh (N-m) • ${}^{MBD}\vec{F}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction forces at the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (N) • ${}^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction moments at the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (N-m) • ${}^{MBD}\vec{F}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction forces at the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (N) • ${}^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Concentrated reaction moments at the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (N-m)		• ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational inertia about the shaft axis of the top and bottom rotors/drivetrains on the pylons on the port wing mesh (kg·m²) • ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Transverse inertia about the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the port wing mesh (kg·m²) • ${}^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]$ – Distance along the shaft from the rotor reference point of the top and bottom rotors/drivetrains on the pylons on the port wing mesh to the center of mass of the rotor/drivetrain (positive along positive x) (m) • ${}^{MBD}\vec{p}_j^{FusR}$ – Reference position of the j th node of the fuselage mesh (m) • ${}^{MBD}\Lambda_j^{FusR}$ – Reference orientation of the j th node of the fuselage mesh (-) • ${}^{MBD}\vec{p}_j^{SWnR}$ – Reference position of the j th node of the starboard wing mesh (m) • ${}^{MBD}\Lambda_j^{SWnR}$ – Reference orientation of the j th node of the starboard wing mesh (-) • ${}^{MBD}\vec{p}_j^{PWnR}$ – Reference position of the j th node of the port wing mesh (m) • ${}^{MBD}\Lambda_j^{PWnR}$ – Reference orientation of the j th node of the port wing mesh (-)

• ${}^{MBD}\vec{M}R_j^{PWn}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the port wing mesh (N-m) • ${}^{MBD}\vec{p}^{VSO}$ – Position (origin) of the vertical stabilizer (m) • ${}^{MBD}\vec{p}_j^{VS}$ – Translational position (absolute) of the j th node of the vertical stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{VS}$ – Displaced rotation (absolute orientation) of the j th node of the vertical stabilizer mesh (-) • ${}^{MBD}\vec{v}_j^{VS}$ – Translational velocity (absolute) of the j th node of the vertical stabilizer mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{VS}$ – Rotational velocity (absolute) of the j th node of the vertical stabilizer mesh (rad/s) • ${}^{MBD}\vec{a}_j^{VS}$ – Translational acceleration (absolute) of the j th node of the vertical stabilizer mesh (m/s^2) • ${}^{MBD}\vec{F}R_j^{VS}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the vertical stabilizer mesh (N) • ${}^{MBD}\vec{M}R_j^{VS}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the vertical stabilizer mesh (N-m) • ${}^{MBD}\vec{p}^{SHSO}$ – Position (origin) of the starboard horizontal stabilizer (m) • ${}^{MBD}\vec{p}_j^{SHS}$ – Translational position (absolute) of the			• ${}^{MBD}\vec{p}_j^{VSR}$ – Reference position of the j th node of the vertical stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{VSR}$ – Reference orientation of the j th node of the vertical stabilizer mesh (-) • ${}^{MBD}\vec{p}_j^{SHSR}$ – Reference position of the j th node of the starboard horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{SHSR}$ – Reference orientation of the j th node of the starboard horizontal stabilizer mesh (-) • ${}^{MBD}\vec{p}_j^{PHSR}$ – Reference position of the j th node of the port horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{PHSR}$ – Reference orientation of the j th node of the port horizontal stabilizer mesh (-) • ${}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right]$ – Reference position of the j th node of the pylons on the starboard wing mesh (m) • ${}^{MBD}\Lambda_j^{SPyR}\left[n_{Pylons}\right]$ – Reference orientation of the j th node of the pylons on the starboard wing mesh (-) • ${}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right]$ – Reference position of the j th node of the pylons on the port wing mesh (m) • ${}^{MBD}\Lambda_j^{PPyR}\left[n_{Pylons}\right]$ – Reference orientation of the j th node of the pylons on the port wing mesh (-) • ${}^{MBD}\vec{p}^{SPyRtrR}\left[n_{Pylons},n_2\right]$

j th node of the starboard horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{SHS}$ – Displaced rotation (absolute orientation) of the j th node of the starboard horizontal stabilizer mesh (-) • ${}^{MBD}\vec{v}_j^{SHS}$ – Translational velocity (absolute) of the j th node of the starboard horizontal stabilizer mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{SHS}$ – Rotational velocity (absolute) of the j th node of the starboard horizontal stabilizer mesh (rad/s) • ${}^{MBD}\vec{a}_j^{SHS}$ – Translational acceleration (absolute) of the j th node of the starboard horizontal stabilizer mesh (m/s^2) • ${}^{MBD}\vec{F}R_j^{SHS}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the starboard horizontal stabilizer mesh (N) • ${}^{MBD}\vec{M}R_j^{SHS}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the starboard horizontal stabilizer mesh (N-m) • ${}^{MBD}\vec{p}^{PHSO}$ – Position (origin) of the port horizontal stabilizer (m) • ${}^{MBD}\vec{p}_j^{PHS}$ – Translational position (absolute) of the j th node of the port horizontal stabilizer mesh (m) • ${}^{MBD}\Lambda_j^{PHS}$ – Displaced rotation (absolute			– Reference positions (origins) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{SPyRtrR}\left[n_{Pylons}, n_2\right]$ – Reference orientations of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (-) • ${}^{MBD}\vec{p}^{PPyRtrR}\left[n_{Pylons}, n_2\right]$ – Reference positions (origins) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{PPyRtrR}\left[n_{Pylons}, n_2\right]$ – Reference orientations of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (-)

orientation) of the j th node of the port horizontal stabilizer mesh (-) • ${}^{MBD}\vec{v}_j^{PHS}$ – Translational velocity (absolute) of the j th node of the port horizontal stabilizer mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{PHS}$ – Rotational velocity (absolute) of the j th node of the port horizontal stabilizer mesh (rad/s) • ${}^{MBD}\vec{a}_j^{PHS}$ – Translational acceleration (absolute) of the j th node of the port horizontal stabilizer mesh (m/s^2) • ${}^{MBD}\vec{F}R_j^{PHS}$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the port horizontal stabilizer mesh (N) • ${}^{MBD}\vec{M}R_j^{PHS}$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the port horizontal stabilizer mesh (N-m) • ${}^{MBD}\vec{p}^{SPyO}\left[n_{Pylons}\right]$ – Positions (origins) of pylons on the starboard wing (m) • ${}^{MBD}\vec{p}_j^{SPy}\left[n_{Pylons}\right]$ – Translational position (absolute) of the j th node of the pylons on the starboard wing mesh (m) • ${}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons}\right]$ – Displaced rotation (absolute orientation) of the j th node of the pylons on the starboard wing mesh (-) • ${}^{MBD}\vec{v}_j^{SPy}\left[n_{Pylons}\right]$ – Translational velocity			

(absolute) of the j th node of the pylons on the starboard wing mesh (m/s) • ${}^{MBD}\vec{\omega}_j^{SPy}\left[n_{Pylons}\right]$ – Rotational velocity (absolute) of the j th node of the pylons on the starboard wing mesh (rad/s) • ${}^{MBD}\vec{a}_j^{SPy}\left[n_{Pylons}\right]$ – Translational acceleration (absolute) of the j th node of the pylons on the starboard wing mesh (m/s^2) • ${}^{MBD}\vec{F}R_j^{SPy}\left[n_{Pylons}\right]$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the pylons on the starboard wing mesh (N) • ${}^{MBD}\vec{M}R_j^{SPy}\left[n_{Pylons}\right]$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the pylons on the starboard wing mesh (N-m) • ${}^{MBD}\vec{p}^{PPyO}\left[n_{Pylons}\right]$ – Positions (origins) of pylons on the port wing (m) • ${}^{MBD}\vec{p}_j^{PPy}\left[n_{Pylons}\right]$ – Translational position (absolute) of the j th node of the pylons on the port wing mesh (m) • ${}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons}\right]$ – Displaced rotation (absolute orientation) of the j th node of the pylons on the port wing mesh (-) • ${}^{MBD}\vec{v}_j^{PPy}\left[n_{Pylons}\right]$ – Translational velocity (absolute) of the j th node of the pylons on the port wing mesh (m/s)			

• ${}^{MBD}\vec{\omega}_j^{PPy}\left[n_{Pylons}\right]$ – Rotational velocity (absolute) of the j th node of the pylons on the port wing mesh (rad/s) • ${}^{MBD}\vec{a}_j^{PPy}\left[n_{Pylons}\right]$ – Translational acceleration (absolute) of the j th node of the pylons on the port wing mesh (m/s^2) • ${}^{MBD}\vec{F}R_j^{PPy}\left[n_{Pylons}\right]$ – Reaction force (expressed in the local coordinate system) at the j th Gauss point of the pylons on the port wing mesh (N) • ${}^{MBD}\vec{M}R_j^{PPy}\left[n_{Pylons}\right]$ – Reaction moment (expressed in the local coordinate system) at the j th Gauss point of the pylons on the port wing mesh (N-m) • ${}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Translational position (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Displaced rotation (absolute orientation) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (-) • ${}^{MBD}\vec{v}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Translational velocity (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m/s) • ${}^{MBD}\vec{\omega}^{SPyRtr}\left[n_{Pylons},n_2\right]$ – Rotational velocity (absolute) of the top and			

bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (rad/s) • ${}^{MBD}\vec{a}^{SPyRtr}\left[n_{Pylons}, n_2\right]$ – Translational acceleration (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (m/s^2) • ${}^{MBD}\vec{\alpha}^{SPyRtr}\left[n_{Pylons}, n_2\right]$ – Rotational acceleration (absolute) of the top and bottom nacelles on the pylons on the starboard wing mesh at the rotor reference point (rad/s^2) • ${}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Translational position (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m) • ${}^{MBD}\Lambda^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Displaced rotation (absolute orientation) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (-) • ${}^{MBD}\vec{v}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Translational velocity (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m/s) ${}^{MBD}\vec{\omega}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Rotational velocity (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (rad/s) • ${}^{MBD}\vec{a}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Translational acceleration			

(absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (m/s^2) • ${}^{MBD}\vec{\alpha}^{PPyRtr}\left[n_{Pylons}, n_2\right]$ – Rotational acceleration (absolute) of the top and bottom nacelles on the pylons on the port wing mesh at the rotor reference point (rad/s^2)			

MiscVars: ${}^{Ctrl}y$, ${}^{MD}y$, ${}^{IfW}y$, ${}^{KAD}y$, ${}^{MBD}u$, ${}^{KAD}u$, ${}^{MD}u$ ${}^{MD}x^{Copy}$

Commented [JJ5]: The outputs of each module at time t (as calculated by their respective CalcOutput() routines) are stored as MiscVars in KiteFASTMBD.

Commented [JJ6]: The inputs from MBDyn and inputs to MoorDyn at time t and the extrapolated inputs to KiteAeroDyn at t+KAD^dt are stored as MiscVars in KiteFASTMBD.

Commented [JJ7]: The temporary states of MoorDyn are stored as MiscVars. In KiteFASTMBD.

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Mapping of Outputs to Inputs in KiteFASTMBD

Output depends on Input (Y/N)		Inputs				
		MBDyn	KiteAeroDyn	InflowWind	MoorDyn	Controller
Outputs	MBDyn		N	N	N	Y
	KiteAeroDyn	Y				Y
	InflowWind		Y			Y
	MoorDyn	Y				Y
	Controller	N	N			

Commented [JJ8]: This may technically not be true, but we can only call the Controller once anyway, so, we'll assume no.

Data Flow (stopping when reaching "N")

MBDyn Controller
KiteAeroDyn MBDyn Controller
Controller
InflowWind KiteAeroDyn MBDyn Controller
Controller
Controller
MoorDyn MBDyn Controller
Controller
Controller

Thus, no nonlinear solves are required
Order of calls: MBDyn, Controller, MoorDyn, InflowWind, KiteAeroDyn

Constructor

This routine initializes KiteFASTMBD at $t = 0$:

- Sets parameters
- Initializes states
- Calls module Init routines
- Opens the write output file
- Opens and writes the summary file

Commented [JJ9]: t=0 outputs are not set here, except for the Controller

Query the MBDyn model to access the inputs at $t = 0$.
Query the MBDyn model to access the names of the KiteAeroDyn, InflowWind, and MoorDyn primary input files

Commented [JJ10]: The names of the KiteAeroDyn input file etc., along with switches for enabling/disabling each module, must be queried from the MBDyn model. I haven't specifically included logic below to enable/disable modules, but this should implemented.

Set the parameters from inputs (Δt, $InterpOrder$, N_{Flaps}, N_{Pylons}, ${}^{MBD}\vec{g}$, ${}^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons},n_2\right]$, ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons},n_2\right]$, and ${}^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]$). Trigger a fatal error if ${}^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]<0$, ${}^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons},n_2\right]<0$, ${}^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons},n_2\right]-{}^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]\left({}^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons},n_2\right]\right)^2<0$, ${}^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]<0$, ${}^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons},n_2\right]<0$, or ${}^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons},n_2\right]-{}^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]\left({}^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]\right)^2<0$. Note that:

- The flap indices: $n_{Flaps}=\left\{1,2,\ldots,N_{Flaps}\right\}$
- The pylon indices: $n_{Pylons}=\left\{1,2,\ldots,N_{Pylons}\right\}$
- And: $n_2=\left\{1,2\right\}$

Set the DCM conversion parameter from the FAST ground system (X pointed nominally downwind; Z pointed vertically opposite gravity; Y transverse) to the ground system used by the controller (X pointed nominally upwind; Z pointed vertically downward, Y transverse):

$$\Lambda^{FAST2Ctrl}=\begin{bmatrix}-1 & 0 & 0\\ 0 & 1 & 0\\ 0 & 0 & -1\end{bmatrix}$$

Set the reference positions (origins) needed as initialization inputs to KiteAeroDyn:

$${}^{KAD}\vec{p}^{SWnOR}={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{SWnO}-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{PWnOR}={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{PWnO}-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{VSOR}={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{VSO}-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{SHSOR}={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{SHSO}-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{PHSOR}={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{PHSO}-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{SPyOR}\left[n_{Pylons}\right]={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{SPyO}\left[n_{Pylons}\right]-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{PPyOR}\left[n_{Pylons}\right]={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{PPyO}\left[n_{Pylons}\right]-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{SPyRtrR}\left[n_{Pylons},n_2\right]={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons},n_2\right]-{}^{MBD}\vec{p}^{FusO}\right\}$$
$${}^{KAD}\vec{p}^{PPyRtrR}\left[n_{Pylons},n_2\right]={}^{MBD}\Lambda^{FusO}\left\{{}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons},n_2\right]-{}^{MBD}\vec{p}^{FusO}\right\}$$

Call KiteAeroDyn_Init()

Calculate the number of KiteAeroDyn time steps per MBDyn time step: $N_{KAD/MBD}=NINT\left(\frac{{}^{KAD}\Delta t}{\Delta t}\right)$

Deleted:

Commented [JJ11]: These must be queried from the MBDyn model.

Trigger a fatal error if the KiteAeroDyn time step is not an integer multiple of the MBDyn time step i.e. if $N_{KAD/MBD}\Delta t - {}^{KAD}\Delta t \neq 0$

${}^{KAD}NewTime = TRUE$

Set the air density for future reference: $\rho = {}^{KAD}\rho$

Determine the number of points where wind will be accessed within InflowWind by summing up the nodes on the AeroDyn input meshes, plus one for the fuselage origin and one for the ground station:

$$\begin{aligned}{}^{IfW}NumWindPoints = 2 & \\ & + {}^{KAD}NumFusNds \\ & + {}^{KAD}NumSWnNds \\ & + {}^{KAD}NumPWnNds \\ & + {}^{KAD}NumVSNds \\ & + {}^{KAD}NumSHSNds \\ & + {}^{KAD}NumPHSNds \\ & + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ & + 4N_{Pylons}\end{aligned}$$

Call InflowWind_Init()

Set the initialization inputs to MoorDyn:

$${}^{MD}g = \left\| {}^{MBD}\vec{g} \right\|_2$$

$${}^{MD}rhoW = \rho$$

$${}^{MD}WtrDepth = 0$$

$${}^{MD}PtfmInit = \begin{Bmatrix} {}^{MBD}\vec{p}^{FusO} \\ {}^{MBD}\Lambda^{FusO} \end{Bmatrix}$$

Call MoorDyn_Init()

Trigger a fatal error if $\left({}^{MD}\Delta t \neq \Delta t\right)$

Trigger a fatal error if there is more than one anchor set in MoorDyn. Also, set the anchor position for future reference based on the initialization output from MoorDyn:

$\vec{p}^{Anch}$ = from MoorDyn initialization output

Call Controller_Init()

Calculate the number of controller time steps per MBDyn time step: $N_{Ctrl/MBD} = NINT\left(\frac{{}^{Ctrl}\Delta t}{\Delta t}\right)$

Trigger a fatal error if the controller time step is not an integer multiple of the MBDyn time step i.e. if $N_{Ctrl/MBD}\Delta t - {}^{Ctrl}\Delta t \neq 0$

${}^{Ctrl}NewTime = FALSE$

Deleted: Trigger a fatal error if $\left({}^{KAD}\Delta t \neq \Delta t\right)$¶

Deleted: Trigger a fatal error if $\left({}^{IfW}\Delta t \neq \Delta t\right)$¶

Commented [JJ12]: Note: the Controller_Init() call initializes the controller states and returns the initial controller outputs.

Commented [JJ13]: Note: the controller will trigger a fatal error if $N_{Flaps} \neq 3$ (to match the current controller interface), $N_{Pylons} \neq 2$ (to match the current controller interface)

Commented [JJ14]: If the controller takes larger steps than MBDyn, then we'll need to smooth the controller output to ensure that it is continuous (at least for the rotor velocity and acceleration). That is, the controller would have to be implemented like KiteAeroDyn.

Set the reference positions and orientations of the line2 and point meshes from the inputs:

$$ {}^{MBD}\vec{p}_j^{FusR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{Fus} - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumFusNds\}) $$

$$ {}^{MBD}\Lambda_j^{FusR} = {}^{MBD}\Lambda_j^{Fus}\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumFusNds\}) $$

$$ {}^{MBD}\vec{p}_j^{SWnR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{SWn} - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumSWnNds\}) $$

$$ {}^{MBD}\Lambda_j^{SWnR} = {}^{MBD}\Lambda_j^{SWn}\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumSWnNds\}) $$

$$ {}^{MBD}\vec{p}_j^{PWnR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{PWn} - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPWnNds\}) $$

$$ {}^{MBD}\Lambda_j^{PWnR} = {}^{MBD}\Lambda_j^{PWn}\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPWnNds\}) $$

$$ {}^{MBD}\vec{p}_j^{VSR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{VS} - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumVSNds\}) $$

$$ {}^{MBD}\Lambda_j^{VSR} = {}^{MBD}\Lambda_j^{VS}\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumVSNds\}) $$

$$ {}^{MBD}\vec{p}_j^{SHSR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{SHS} - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumSHSNds\}) $$

$$ {}^{MBD}\Lambda_j^{SHSR} = {}^{MBD}\Lambda_j^{SHS}\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumSHSNds\}) $$

$$ {}^{MBD}\vec{p}_j^{PHSR} = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{PHS} - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPHSNds\}) $$

$$ {}^{MBD}\Lambda_j^{PHSR} = {}^{MBD}\Lambda_j^{PHS}\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPHSNds\}) $$

$$ {}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{SPy}\left[n_{Pylons} \right] - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPylNds\}) $$

$$ {}^{MBD}\Lambda_j^{SPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons} \right]\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPylNds\}) $$

$$ {}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}_j^{PPy}\left[n_{Pylons} \right] - {}^{MBD}\vec{p}^{FusO} \right\} \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPylNds\}) $$

$$ {}^{MBD}\Lambda_j^{PPyR}\left[n_{Pylons} \right] = {}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons} \right]\left[{}^{MBD}\Lambda^{FusO} \right]^T \quad (\text{for } j = \{1,2,\ldots,{}^{MBD}NumPylNds\}) $$

$$ {}^{MBD}\vec{p}^{SPyRtrR}\left[n_{Pylons}, n_2 \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons}, n_2 \right] - {}^{MBD}\vec{p}^{FusO} \right\} $$

$$ {}^{MBD}\Lambda^{SPyRtrR}\left[n_{Pylons}, n_2 \right] = I $$

$$ {}^{MBD}\vec{p}^{PPyRtrR}\left[n_{Pylons}, n_2 \right] = {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons}, n_2 \right] - {}^{MBD}\vec{p}^{FusO} \right\} $$

$$ {}^{MBD}\Lambda^{PPyRtrR}\left[n_{Pylons}, n_2 \right] = I $$

Set mesh-mappings between KiteFASTMBD-KiteAeroDyn and KiteFASTMBD-MoorDyn.

Open the write Output File

Open and write a summary file (if SumPrint = TRUE)

KiteFASTMBD Summary File

Predictions were generated on DATE at TIME using KiteFASTMBD (VERSION, DATE)
compiled with
NWTC Subroutine Library (VERSION, DATE)
KiteAeroDyn (VERSION, DATE)
InflowWind (VERSION, DATE) for OpenFAST (VERSION DATE)
MoorDyn (VERSION, DATE)

Commented [JJ15]: Note: the motion meshes are line2 meshes (except for the rotors, which are point meshes), but the load meshes are point meshes.

Commented [JJ16]: The mesh-mapping routines can only handle one source and one destination mesh. To do this mapping, the MBDyn meshes for the starboard and port wings (SWn and PWn) have to be copied into a single mesh using a one-to-one transfer of reference positions, reference orientations, and fields (which I label as Wn in the mesh-mappings below).

Commented [JJ17]: SumPrint must be queried from the MBDyn model

Commented [JJ18]: I'm only hand waving here because the implementation should be obvious (similar to other OpenFAST summary files)

Deleted: Initialize the states not previously initialized:¶
$NewTime = TRUE$ ¶
¶

Commented [JJ19]: (VERSION,DATE) has been replaced with the a git hash

Controller Wrapper (VERSION, DATE)
Controller (VERSION, DATE)
MBDyn (VERSION, DATE)

Description from the MDyn input file: TITLE

Commented [JJ20]: Probably not needed if TITLE is not easily accessible within the MBDyn user element.

Time Step:

Deleted: (s)

Component	Time Step
(-)	(s)
MBDyn	Δt
KiteAeroDyn	${}^{KAD}\Delta t$
MoorDyn	Δt
Controller	${}^{Ctrl}\Delta t$

Deleted:

Deleted: ¶

Reference Points, MBDyn Finite-Element Nodes, and MBDyn Gauss Points

Component	Type	Number	Output Number	x	y	z
(-)	(-)	(-)	(-)	(m)	(m)	(m)
Fuselage	Reference point	-	-	0	0	0
Fuselage	Finite-element node	j	$\begin{cases} Fus\langle\beta\rangle & for\left(FusOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{\,FusR}$		
Fuselage	Gauss point	j	$\begin{cases} Fus\langle\beta\rangle & for\left(FusOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{\,FusR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,FusR} & for\left(Mod(j,2)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{\,FusR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,FusR} & otherwise \end{cases}$		
Starboard wing	Reference point	-	-	${}^{KAD}\vec{p}^{\,SWnOR}$		
Starboard wing	Finite-element node	j	$\begin{cases} SWn\langle\beta\rangle & for\left(SWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	${}^{MBD}\vec{p}_j^{\,SWnR}$		
Starboard wing	Gauss point	j	$\begin{cases} SWn\langle\beta\rangle & for\left(SWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$	$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{\,SWnR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,SWnR} & for\left(Mod(j,2)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{\,SWnR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{\,SWnR} & otherwise \end{cases}$		

Port wing | Reference point | - | -

$^{KAD}\vec{p}^{PWnOR}$

Port wing | Finite-element node | j | $\begin{cases} PWn\langle\beta\rangle & for\left(PWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$^{MBD}\vec{p}_{j}^{PWnR}$

Port wing | Gauss point | j | $\begin{cases} PWn\langle\beta\rangle & for\left(PWnOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{PWnR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j}^{PWnR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{PWnR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j}^{PWnR} & otherwise \end{cases}$$

Vertical stabilizer | Reference point | - | -

$^{KAD}\vec{p}^{VSOR}$

Vertical stabilizer | Finite-element node | j | $\begin{cases} VS\langle\beta\rangle & for\left(VSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$^{MBD}\vec{p}_{j}^{VSR}$

Vertical stabilizer | Gauss point | j | $\begin{cases} VS\langle\beta\rangle & for\left(VSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{VSR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j}^{VSR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{VSR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j}^{VSR} & otherwise \end{cases}$$

Starboard horizontal stabilizer | Reference point | - | -

$^{KAD}\vec{p}^{SHSOR}$

Starboard horizontal stabilizer | Finite-element node | j | $\begin{cases} SHS\langle\beta\rangle & for\left(SHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$^{MBD}\vec{p}_{j}^{SHSR}$

Starboard horizontal stabilizer | Gauss point | j | $\begin{cases} SHS\langle\beta\rangle & for\left(SHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{SHSR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j}^{SHSR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{SHSR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j}^{SHSR} & otherwise \end{cases}$$

Port horizontal stabilizer | Reference point | - | -

$^{KAD}\vec{p}^{PHSOR}$

Port horizontal stabilizer Finite-element node j $\begin{cases} PHS\langle\beta\rangle & for\left(PHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

${}^{MBD}\vec{p}_j^{PHSR}$

Port horizontal stabilizer Gauss point j $\begin{cases} PHS\langle\beta\rangle & for\left(PHSOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{PHSR}+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PHSR} & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{PHSR}+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PHSR} & otherwise \end{cases}$$

Starboard pylon n_{Pylons} Reference point - -

${}^{KAD}\vec{p}^{SPyOR}\left[n_{Pylons}\right]$

Starboard pylon n_{Pylons} Finite-element node j $\begin{cases} SP\langle n_{Pylons}\rangle\langle\beta\rangle & for\left(PylOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

${}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right]$

Starboard pylon n_{Pylons} Gauss point j $\begin{cases} SP\langle n_{Pylons}\rangle\langle\beta\rangle & for\left(PylOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{SPyR}\left[n_{Pylons}\right]+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right] & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{SPyR}\left[n_{Pylons}\right]+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{SPyR}\left[n_{Pylons}\right] & otherwise \end{cases}$$

Port pylon n_{Pylons} Reference point - -

${}^{KAD}\vec{p}^{PPyOR}\left[n_{Pylons}\right]$

Port pylon n_{Pylons} Finite-element node j $\begin{cases} PP\langle n_{Pylons}\rangle\langle\beta\rangle & for\left(PylOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

${}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right]$

Port pylon n_{Pylons} Gauss point j $\begin{cases} PP\langle n_{Pylons}\rangle\langle\beta\rangle & for\left(PylOutNd[\beta]=j\right) \\ - & otherwise \end{cases}$

$$\begin{cases} \left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{PPyR}\left[n_{Pylons}\right]+\left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right] & for\left(Mod\left(j,2\right)=1\right) \\ \left(\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_{j+1}^{PPyR}\left[n_{Pylons}\right]+\left(1-\frac{\sqrt{3}}{3}\right){}^{MBD}\vec{p}_j^{PPyR}\left[n_{Pylons}\right] & otherwise \end{cases}$$

Top rotor on starboard pylon n_{Pylons} Reference point - -

${}^{KAD}\vec{p}^{SPyRtrR}\left[n_{Pylons},1\right]$

Bottom rotor on starboard pylon n_{Pylons}	Reference point	-	-
${}^{KAD}\vec{p}^{\,SPyRtrR}\left[n_{Pylons},2\right]$			
Top rotor on port pylon n_{Pylons}	Reference point	-	-
${}^{KAD}\vec{p}^{\,PPyRtrR}\left[n_{Pylons},1\right]$			
Bottom rotor on port pylon n_{Pylons}	Reference point	-	-
${}^{KAD}\vec{p}^{\,PPyRtrR}\left[n_{Pylons},2\right]$			

Requested Channels in KiteFASTMBD Output Files: NUMBER

Number	Name	Units	Generated by
0	Time	(s)	KiteFASTMBD
NUMBER	NAME	UNITS	(KiteFASTMBD, KiteAeroDyn, InflowWind, MoorDyn, or Controller Wrapper)

Deconstructor

This routine ends KiteFASTMBD:

- Calls module End routines
- Deallocates memory
- Closes the write output file

AssRes

This routine accesses inputs at t (from GetXCur) (including $t=0$) for both the prediction and correction steps of each MBD time step, temporarily updates states from $t-\Delta t$ to t, and calculates outputs at t:

- Calls module UpdateStates and Controller_Step routines except at $t=0$
- Calls module CalcOutput routines

Set the discrete-time counter:

$$n=\frac{t}{\Delta t}-1$$

Query the MBDyn model to access the inputs at t (from GetXCur) i.e. ${}^{MBD}u$.

Calculate the translation displacements (relative) of the MBDyn input meshes at t:

$${}^{MBD}\vec{u}_j^{\,Fus}={}^{MBD}\vec{p}_j^{\,Fus}-{}^{MBD}\vec{p}_j^{\,FusR} \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumFusNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,SWn}={}^{MBD}\vec{p}_j^{\,SWn}-{}^{MBD}\vec{p}_j^{\,SWnR} \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumSWnNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,PWn}={}^{MBD}\vec{p}_j^{\,PWn}-{}^{MBD}\vec{p}_j^{\,PWnR} \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumPWnNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,VS}={}^{MBD}\vec{p}_j^{\,VS}-{}^{MBD}\vec{p}_j^{\,VSR} \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumVSNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,SHS}={}^{MBD}\vec{p}_j^{\,SHS}-{}^{MBD}\vec{p}_j^{\,SHSR} \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumSHSNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,PHS}={}^{MBD}\vec{p}_j^{\,PHS}-{}^{MBD}\vec{p}_j^{\,PHSR} \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumPHSNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,SPy}\left[n_{Pylons}\right]={}^{MBD}\vec{p}_j^{\,SPy}\left[n_{Pylons}\right]-{}^{MBD}\vec{p}_j^{\,SPyR}\left[n_{Pylons}\right] \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumPylNds\right\}\text{)}$$

$${}^{MBD}\vec{u}_j^{\,PPy}\left[n_{Pylons}\right]={}^{MBD}\vec{p}_j^{\,PPy}\left[n_{Pylons}\right]-{}^{MBD}\vec{p}_j^{\,PPyR}\left[n_{Pylons}\right] \quad \text{(for } j=\left\{1,2,\ldots,{}^{MBD}NumPylNds\right\}\text{)}$$

Commented [JJ21]: AssRes could access inputs at t-dt (from GetXPrev), but we save the previous inputs as OtherStates instead.

Commented [JJ22]: Note: the module UpdateStates and Controller_Step routines are not called at t=0 (except for KiteAeroDyn) because the states have already been initialized through the Init calls.

Deleted: <#>Note that AssRes has input argument $InitialTime=1$ at $t=0$ and $InitialTime=0$ at all other t¶

Commented [JJ23]: This is necessary because in OpenFAST, UpdateStates shifts from t to t+dt whereas AssRes shifts from t-dt to t.

$$ {}^{MBD}\vec{u}^{SPyRtr}\left[n_{Pylons},n_2\right] = {}^{MBD}\vec{p}^{SPyRtr}\left[n_{Pylons},n_2\right] - {}^{MBD}\vec{p}^{SPyRtrR}\left[n_{Pylons},n_2\right] $$
$$ {}^{MBD}\vec{u}^{PPyRtr}\left[n_{Pylons},n_2\right] = {}^{MBD}\vec{p}^{PPyRtr}\left[n_{Pylons},n_2\right] - {}^{MBD}\vec{p}^{PPyRtrR}\left[n_{Pylons},n_2\right] $$

Advance the controller only once per controller time step, updating the states to, and obtaining the controller outputs at, t:

IF $\left({}^{Ctrl}NewTime\right)$ THEN

First, calculate the InflowWind outputs at the ground station and fuselage using the most converged inputs from MBDyn (as data stored in ${}^{MBD}OtherStates$ from the previous step):

$$ {}^{IfW}PositionXYZ(:,1) = {}^{MBD}\vec{p}^{Wind} $$
$$ {}^{IfW}PositionXYZ(:,2) = {}^{MBD}\vec{p}^{FusO} $$

Call InflowWind_CalcOutput()

Set inputs to Controller using the most converged inputs from MBDyn and the outputs from KiteAeroDyn, InflowWind, and MoorDyn (as data stored in ${}^{MBD}OtherStates$, ${}^{KAD}y$, and ${}^{MD}y$ from the previous step):

$$ {}^{Ctrl}dcm\_g2b = {}^{MBD}\Lambda^{FusO}\left[\Lambda^{FAST2Ctrl}\right]^T $$
$$ {}^{Ctrl}pqr = {}^{MBD}\Lambda^{FusO}\,{}^{MBD}\vec{\omega}^{FusO} $$
$$ {}^{Ctrl}acc\_norm = \left\| {}^{MBD}\vec{a}^{FusO} \right\|_2 $$
$$ {}^{Ctrl}Xg = \Lambda^{FAST2Ctrl}\left\{ {}^{MBD}\vec{p}^{FusO} - \vec{p}^{Anch} \right\} $$
$$ {}^{Ctrl}Vg = \Lambda^{FAST2Ctrl}\,{}^{MBD}\vec{v}^{FusO} $$
$$ {}^{Ctrl}Vb = {}^{MBD}\Lambda^{FusO}\,{}^{MBD}\vec{v}^{FusO} $$
$$ {}^{Ctrl}Ag = \Lambda^{FAST2Ctrl}\,{}^{MBD}\vec{a}^{FusO} $$
$$ {}^{Ctrl}Ab = {}^{MBD}\Lambda^{FusO}\,{}^{MBD}\vec{a}^{FusO} $$
$$ {}^{Ctrl}rho = \rho $$
$$ {}^{Ctrl}apparent\_wind = \Lambda^{FAST2Ctrl}\left\{ {}^{IfW}VelocityUVW(:,2) - {}^{MBD}\vec{v}^{FusO} \right\} $$
$$ {}^{Ctrl}tether\_force\_b = {}^{MBD}\Lambda^{FusO}\left\{ \sum_{i=1}^{NFairs} {}^{MD}PtFairleadLoad\%Force(:,i) \right\} $$
$$ {}^{Ctrl}wind\_g = \Lambda^{FAST2Ctrl}\,{}^{IfW}VelocityUVW(:,1) $$
$$ {}^{Ctrl}aero\_torque^{SPyRtr}\left[n_{Pylons},n_2\right] = \left\{ {}^{MBD}\hat{x}^{SPyRtr}\left[n_{Pylons},n_2\right] \right\}^T \,{}^{KAD}\vec{M}^{SPyRtr}\left[n_{Pylons},n_2\right] $$
$$ {}^{Ctrl}aero\_torque^{PPyRtr}\left[n_{Pylons},n_2\right] = \left\{ {}^{MBD}\hat{x}^{PPyRtr}\left[n_{Pylons},n_2\right] \right\}^T \,{}^{KAD}\vec{M}^{PPyRtr}\left[n_{Pylons},n_2\right] $$

- Call Controller_Step()

Ensure that we only call the controller once per the controller time step:

$$ {}^{Ctrl}NewTime = FALSE $$

END

Deleted: $\left(\left(NewTime\right).AND.\left(InitialTime == 0\right)\right)$

Commented [JJ24]: One can call InflowWind_CalcOutput() with fewer than ${}^{IfW}NumWindPoints$.

Deleted: at $t-\Delta t$ using

Deleted: ${}^{IfW}OtherStates$,

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Commented [JJ25]: All filtered values (_f) are identical to the unfiltered values.

Commented [JJ26]: We are approximating this input to the controller as the vector sum of the fairlead tensions.

Commented [JJ27]: These were added to the original controller inputs so that the controller could calculate the rotor/drivetrain acceleration and resulting generator speed and torque.

We should also ensure that the controller is using the same rotor/drivetrain rotational inertia.

Deleted: $NewTime = FALSE$

Store a copy of the MoorDyn current states at $t-\Delta t$:

$$^{MD}x^{Copy} = {}^{MD}x$$

Set inputs to MoorDyn at t from MBDyn:

$$^{MD}PtFairleadDisplacement = M_u^{L2P}\left({}^{MBD}\vec{u}_j^{Wn}, {}^{MBD}\Lambda_j^{Wn}\right)$$

Advance MoorDyn:

IF $(t > 0)$ Call MoorDyn_UpdateStates()

Call MoorDyn_CalcOutput()

Advance KiteAeroDyn only once per KiteAeroDyn time step, interpolate the KiteAeroDyn outputs otherwise.

IF $\left({}^{KAD}NewTime\right)$ THEN

Shift the KiteAeroDyn input history:

IF $(t > 0)$

IF $(InterpOrder == 1)$ THEN

$$^{KAD}u(2) = {}^{KAD}u(1)$$

ELSEIF ! $(InterpOrder == 2)$

$$^{KAD}u(3) = {}^{KAD}u(2)$$

$$^{KAD}u(2) = {}^{KAD}u(1)$$

END IF

END IF

Set inputs to KiteAeroDyn—stored in $^{KAD}u(1)$—from Controller at t :

$$^{KAD}Ctrl^{SFlp}\left[n_{Flaps}\right] = \begin{cases} {}^{Ctrl}kFlapA5 & for\left(n_{Flaps} = 1\right) \\ {}^{Ctrl}kFlapA7 & for\left(n_{Flaps} = 2\right) \\ {}^{Ctrl}kFlapA8 & for\left(n_{Flaps} = 3\right) \end{cases}$$

$$^{KAD}Ctrl^{PFlp}\left[n_{Flaps}\right] = \begin{cases} {}^{Ctrl}kFlapA4 & for\left(n_{Flaps} = 1\right) \\ {}^{Ctrl}kFlapA2 & for\left(n_{Flaps} = 2\right) \\ {}^{Ctrl}kFlapA1 & for\left(n_{Flaps} = 3\right) \end{cases}$$

$$^{KAD}Ctrl^{Rudr}\left[n_2\right] = {}^{Ctrl}kFlapA10$$

$$^{KAD}Ctrl^{SElv}\left[n_2\right] = {}^{Ctrl}kFlapA9$$

$$^{KAD}Ctrl^{PElv}\left[n_2\right] = {}^{Ctrl}kFlapA9$$

$$^{KAD}\Omega^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{Ctrl}\Omega^{SPyRtr}\left[n_{Pylons}, n_2\right]$$

$$^{KAD}\Omega^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{Ctrl}\Omega^{PPyRtr}\left[n_{Pylons}, n_2\right]$$

Deleted: $^{KAD}z^{Copy} = {}^{KAD}z$ ¶

Commented [JJ28]: See earlier comment about mesh mapping with Wn above.

Commented [JJ29]: Input the time at t-dt in this call.

The input at t-dt comes from $^{MD}OtherStates$

Deleted: $(InitialTime == 0)$

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Commented [JJ30]: Different controller documentation use kFlapRud in place of kFlapA10

Commented [JJ31]: Different controller documentation use kFlapEle in place of kFlapA9

Commented [JJ32]: These were added to the original controller outputs so that the controller could calculate the rotor/drivetrain acceleration and resulting generator speed and torque.

$$^{KAD}\theta^{SPyRtr}\left[n_{Pylons},n_2\right]=0$$
$$^{KAD}\theta^{PPyRtr}\left[n_{Pylons},n_2\right]=0$$

Commented [JJ33]: The rotor-collective pitch angles are not currently commanded from the controller; assume zero for now.

Set inputs to KiteAeroDyn—stored in $^{KAD}u\left(1\right)$—from MBDyn at t based on mesh-mapping:

Deleted:

$$^{KAD}\vec{u}^{FusO} = {}^{MBD}\vec{p}^{FusO}$$

Commented [JJ34]: You could use P2P mappings here, but there is no point, because the reference (0,0,0) is the same in both KiteAeroDyn and MBDyn.

$$^{KAD}\vec{u}_j^{Fus} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{Fus},{}^{MBD}\Lambda_j^{Fus}\right)$$
$$^{KAD}\Lambda_j^{Fus} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{Fus}\right)$$
$$^{KAD}\vec{v}_j^{Fus} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{Fus},{}^{MBD}\vec{u}_j^{Fus},{}^{MBD}\vec{v}_j^{Fus},{}^{MBD}\vec{\omega}_j^{Fus}\right)$$
$$^{KAD}\vec{u}_j^{SWn} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{SWn},{}^{MBD}\Lambda_j^{SWn}\right)$$
$$^{KAD}\Lambda_j^{SWn} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{SWn}\right)$$
$$^{KAD}\vec{v}_j^{SWn} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{SWn},{}^{MBD}\vec{u}_j^{SWn},{}^{MBD}\vec{v}_j^{SWn},{}^{MBD}\vec{\omega}_j^{SWn}\right)$$
$$^{KAD}\vec{u}_j^{PWn} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{PWn},{}^{MBD}\Lambda_j^{PWn}\right)$$
$$^{KAD}\Lambda_j^{PWn} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{PWn}\right)$$
$$^{KAD}\vec{v}_j^{PWn} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{PWn},{}^{MBD}\vec{u}_j^{PWn},{}^{MBD}\vec{v}_j^{PWn},{}^{MBD}\vec{\omega}_j^{PWn}\right)$$
$$^{KAD}\vec{u}_j^{VS} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{VS},{}^{MBD}\Lambda_j^{VS}\right)$$
$$^{KAD}\Lambda_j^{VS} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{VS}\right)$$
$$^{KAD}\vec{v}_j^{VS} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{VS},{}^{MBD}\vec{u}_j^{VS},{}^{MBD}\vec{v}_j^{VS},{}^{MBD}\vec{\omega}_j^{VS}\right)$$
$$^{KAD}\vec{u}_j^{SHS} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{SHS},{}^{MBD}\Lambda_j^{SHS}\right)$$
$$^{KAD}\Lambda_j^{SHS} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{SHS}\right)$$
$$^{KAD}\vec{v}_j^{SHS} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{SHS},{}^{MBD}\vec{u}_j^{SHS},{}^{MBD}\vec{v}_j^{SHS},{}^{MBD}\vec{\omega}_j^{SHS}\right)$$
$$^{KAD}\vec{u}_j^{PHS} = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{PHS},{}^{MBD}\Lambda_j^{PHS}\right)$$
$$^{KAD}\Lambda_j^{PHS} = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{PHS}\right)$$
$$^{KAD}\vec{v}_j^{PHS} = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{PHS},{}^{MBD}\vec{u}_j^{PHS},{}^{MBD}\vec{v}_j^{PHS},{}^{MBD}\vec{\omega}_j^{PHS}\right)$$
$$^{KAD}\vec{u}_j^{SPy}\left[n_{Pylons}\right] = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right],{}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons}\right]\right)$$
$$^{KAD}\Lambda_j^{SPy}\left[n_{Pylons}\right] = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{SPy}\left[n_{Pylons}\right]\right)$$
$$^{KAD}\vec{v}_j^{SPy}\left[n_{Pylons}\right] = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{SPy}\left[n_{Pylons}\right],{}^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right],{}^{MBD}\vec{v}_j^{SPy}\left[n_{Pylons}\right],{}^{MBD}\vec{\omega}_j^{SPy}\left[n_{Pylons}\right]\right)$$
$$^{KAD}\vec{u}_j^{PPy}\left[n_{Pylons}\right] = M_u^{L2L}\left({}^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right],{}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons}\right]\right)$$
$$^{KAD}\Lambda_j^{PPy}\left[n_{Pylons}\right] = M_\Lambda^{L2L}\left({}^{MBD}\Lambda_j^{PPy}\left[n_{Pylons}\right]\right)$$
$$^{KAD}\vec{v}_j^{PPy}\left[n_{Pylons}\right] = M_v^{L2L}\left({}^{KAD}\vec{u}_j^{PPy}\left[n_{Pylons}\right],{}^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right],{}^{MBD}\vec{v}_j^{PPy}\left[n_{Pylons}\right],{}^{MBD}\vec{\omega}_j^{PPy}\left[n_{Pylons}\right]\right)$$

$$ {}^{KAD}\vec{u}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{u}^{SPyRtr}\left[n_{Pylons}, n_2\right] $$
$$ {}^{KAD}\Lambda^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Lambda^{SPyRtr}\left[n_{Pylons}, n_2\right] $$
$$ {}^{KAD}\vec{v}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{v}^{SPyRtr}\left[n_{Pylons}, n_2\right] $$
$$ {}^{KAD}\vec{u}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{u}^{PPyRtr}\left[n_{Pylons}, n_2\right] $$
$$ {}^{KAD}\Lambda^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\Lambda^{PPyRtr}\left[n_{Pylons}, n_2\right] $$
$$ {}^{KAD}\vec{v}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{MBD}\vec{v}^{PPyRtr}\left[n_{Pylons}, n_2\right] $$

Commented [JJ35]: You could use P2P mappings here, but there is no point, because the references are the same in both KiteAeroDyn and MBDyn.

Commented [JJ36]: You could use P2P mappings here, but there is no point, because the references are the same in both KiteAeroDyn and MBDyn.

Set inputs to InflowWind at t based on the KiteAeroDyn inputs—stored in ${}^{KAD}u\left(1\right)$:

$$ {}^{IfW}PositionXYZ\left(:,1\right) = {}^{MBD}\vec{p}^{Wind} $$
$$ {}^{IfW}PositionXYZ\left(:,2\right) = {}^{MBD}\vec{p}^{FusO} $$
$$ {}^{IfW}PositionXYZ\left(:,j+2\right) = {}^{KAD\,In}\vec{p}_j^{FusR} + {}^{KAD}\vec{u}_j^{Fus} $$ (for $j=\left\{1,2,\ldots,{}^{KAD}NumFusNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ +{}^{KAD}NumFusNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{SWnR} + {}^{KAD}\vec{u}_j^{SWn} $$ (for $j=\left\{1,2,\ldots,{}^{KAD}NumSWnNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ +{}^{KAD}NumFusNds \\ +{}^{KAD}NumSWnNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{PWnR} + {}^{KAD}\vec{u}_j^{PWn} $$ (for $j=\left\{1,2,\ldots,{}^{KAD}NumPWnNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ +{}^{KAD}NumFusNds \\ +{}^{KAD}NumSWnNds \\ +{}^{KAD}NumPWnNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{VSR} + {}^{KAD}\vec{u}_j^{VS} $$ (for $j=\left\{1,2,\ldots,{}^{KAD}NumVSNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :,j+2 \\ +{}^{KAD}NumFusNds \\ +{}^{KAD}NumSWnNds \\ +{}^{KAD}NumPWnNds \\ +{}^{KAD}NumVSNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{SHSR} + {}^{KAD}\vec{u}_j^{SHS} $$ (for $j=\left\{1,2,\ldots,{}^{KAD}NumSHSNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \\ + {}^{KAD}NumVSNds \\ + {}^{KAD}NumSHSNds \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{\,PHSR} + {}^{KAD}\vec{u}_j^{\,PHS} $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPHSNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \\ + {}^{KAD}NumVSNds \\ + {}^{KAD}NumSHSNds \\ + {}^{KAD}NumPHSNds \\ + {}^{KAD}NumPylNds\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{\,SPyR}\left[n_{Pylons}\right] + {}^{KAD}\vec{u}_j^{\,SPy}\left[n_{Pylons}\right] $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPylNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, j+2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \\ + {}^{KAD}NumVSNds \\ + {}^{KAD}NumSHSNds \\ + {}^{KAD}NumPHSNds \\ + {}^{KAD}NumPylNds\left(N_{Pylons}\right) \\ + {}^{KAD}NumPylNds\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD\,In}\vec{p}_j^{\,PPyR}\left[n_{Pylons}\right] + {}^{KAD}\vec{u}_j^{\,PPy}\left[n_{Pylons}\right] $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPylNds\right\}$)

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, n_2+2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \\ + {}^{KAD}NumVSNds \\ + {}^{KAD}NumSHSNds \\ + {}^{KAD}NumPHSNds \\ + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ + 2\left(n_{Pylons}-1\right) \end{pmatrix} = {}^{KAD}\vec{p}_j^{\,SPyRtrR}\left[n_{Pylons}, n_2\right] + {}^{KAD}\vec{u}^{\,SPyRtr}\left[n_{Pylons}, n_2\right] $$

$$ {}^{IfW}PositionXYZ\begin{pmatrix} :, n_2 + 2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \\ + {}^{KAD}NumVSNds \\ + {}^{KAD}NumSHSNds \\ + {}^{KAD}NumPHSNds \\ + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ + 2\left(N_{Pylons}\right) \\ + 2\left(n_{Pylons} - 1\right) \end{pmatrix} = {}^{KAD}\vec{p}_j^{\,PPyRtrR}\left[n_{Pylons}, n_2\right] + {}^{KAD}\vec{u}^{\,PPyRtr}\left[n_{Pylons}, n_2\right] $$

Call InflowWind_CalcOutput()

Commented [JJ37]: Input the time at t in this call.

Set inputs to KiteAeroDyn stored in ${}^{KAD}u\left(1\right)$ from InflowWind at t:

Deleted:

$$ {}^{KAD}\vec{V}_j^{\,Fus} = {}^{IfW}VelocityUVW\left(:, j+2\right) $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumFusNds\right\}$)

$$ {}^{KAD}\vec{V}_j^{\,SWn} = {}^{IfW}VelocityUVW\begin{pmatrix} :, j+2 \\ + {}^{KAD}NumFusNds \end{pmatrix} $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumSWnNds\right\}$)

$$ {}^{KAD}\vec{V}_j^{\,PWn} = {}^{IfW}VelocityUVW\begin{pmatrix} :, j+2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \end{pmatrix} $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPWnNds\right\}$)

$$ {}^{KAD}\vec{V}_j^{\,VS} = {}^{IfW}VelocityUVW\begin{pmatrix} :, j+2 \\ + {}^{KAD}NumFusNds \\ + {}^{KAD}NumSWnNds \\ + {}^{KAD}NumPWnNds \end{pmatrix} $$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumVSNds\right\}$)

$$^{KAD}\vec{V}_j^{SHS} = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \end{pmatrix}$$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumSHSNds\right\}$)

$$^{KAD}\vec{V}_j^{PHS} = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \end{pmatrix}$$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPHSNds\right\}$)

$$^{KAD}\vec{V}_j^{SPy}\left[n_{Pylons}\right] = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(n_{Pylons} - 1\right) \end{pmatrix}$$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPylNds\right\}$)

$$^{KAD}\vec{V}_j^{PPy}\left[n_{Pylons}\right] = {}^{IfW}VelocityUVW\begin{pmatrix} \vdots, j+2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(N_{Pylons}\right) \\ \quad + {}^{KAD}NumPylNds\left(n_{Pylons} - 1\right) \end{pmatrix}$$ (for $j = \left\{1, 2, \ldots, {}^{KAD}NumPylNds\right\}$)

$$ {}^{KAD}\vec{V}^{SPyRtr}\left[n_{Pylons}, n_2\right] = {}^{IfW}VelocityUVW\left(\begin{array}{l} :, n_2 + 2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ \quad + 2\left(n_{Pylons} - 1\right) \end{array} \right) $$

$$ {}^{KAD}\vec{V}^{PPyRtr}\left[n_{Pylons}, n_2\right] = {}^{IfW}VelocityUVW\left(\begin{array}{l} :, n_2 + 2 \\ \quad + {}^{KAD}NumFusNds \\ \quad + {}^{KAD}NumSWnNds \\ \quad + {}^{KAD}NumPWnNds \\ \quad + {}^{KAD}NumVSNds \\ \quad + {}^{KAD}NumSHSNds \\ \quad + {}^{KAD}NumPHSNds \\ \quad + {}^{KAD}NumPylNds\left(2N_{Pylons}\right) \\ \quad + 2\left(N_{Pylons}\right) \\ \quad + 2\left(n_{Pylons} - 1\right) \end{array} \right) $$

Initialize the KiteAeroDyn input history at $t = 0$:

IF $(t == 0)$ THEN

IF $(InterpOrder == 1)$ THEN

$$ {}^{KAD}u(2) = {}^{KAD}u(1) $$

$$ {}^{KAD}t(2) = -{}^{KAD}\Delta t $$

$$ {}^{KAD}t(1) = 0 $$

ELSEIF ! $(InterpOrder == 2)$

$$ {}^{KAD}u(3) = {}^{KAD}u(1) $$

$$ {}^{KAD}u(2) = {}^{KAD}u(1) $$

$^{KAD}t(3) = -2\,^{KAD}\Delta t$

$^{KAD}t(2) = {}^{KAD}\Delta t$

$^{KAD}t(1) = 0$

END IF

END IF

Advance KiteAeroDyn to $t + {}^{KAD}\Delta t$:

Call KiteAeroDyn_Input_ExtrapInterp($^{KAD}u(:)$, $^{KAD}t(:)$, ^{KAD}u , $t + {}^{KAD}\Delta t$)

Call KiteAeroDyn_UpdateStates()

Call KiteAeroDyn_CalcOutput()

Shift the KiteAeroDyn output history:

IF $(t > 0)$ THEN

IF $(InterpOrder == 1)$ THEN

$^{KAD}y(2) = {}^{KAD}y(1)$

$^{KAD}y(1) = {}^{KAD}y$

$^{KAD}t(2) = {}^{KAD}t(1)$

$^{KAD}t(1) = t + {}^{KAD}\Delta t$

ELSEIF ! $(InterpOrder == 2)$

$^{KAD}y(3) = {}^{KAD}y(2)$

$^{KAD}y(2) = {}^{KAD}y(1)$

$^{KAD}y(1) = {}^{KAD}y$

$^{KAD}t(3) = {}^{KAD}t(2)$

$^{KAD}t(2) = {}^{KAD}t(1)$

$^{KAD}t(1) = t + {}^{KAD}\Delta t$

END IF

ELSE ! $(t == 0)$

IF $(InterpOrder == 1)$ THEN

$^{KAD}y(2) = {}^{KAD}y$

$^{KAD}y(1) = {}^{KAD}y$

ELSEIF ! $(InterpOrder == 2)$

$^{KAD}y(3) = {}^{KAD}y$

$^{KAD}y(2) = {}^{KAD}y$

$^{KAD}y(1) = {}^{KAD}y$

Deleted: Copy KiteAeroDyn inputs at t to $t-\Delta t$ (for KiteAeroDyn_UpdateStates)¶
¶

Commented [JJ38]: Input the time at t in this call

Deleted: ¶

Deleted: IF $(InitialTime == 0)$

Commented [JJ39]: Input the time at t+KAD^dt in ths call.

END IF

END IF

Ensure that we only call KiteAeroDyn once per KiteAeroDyn time step:

$$^{KAD}NewTime = FALSE$$

END

Call KiteAeroDyn_Output_ExtrapInterp($^{KAD}y(:)$, $^{KAD}t(:)$, ^{KAD}y, t)

Model the rotor/drivetrain dynamics, including the effects from the Controller and KiteAeroDyn, and calculate the reaction loads on the pylons for transfer to MBDyn at t:

Call Rotor($^{MBD}\Lambda^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{\omega}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{a}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{\alpha}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{Ctrl}\Omega^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{Ctrl}T^{GenSPyRtr}\left[n_{Pylons},n_2\right]$, $^{KAD}\vec{F}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{KAD}\vec{M}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{g}$, $^{MBD}m^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}I_{Rot}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}I_{Tran}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}x_{CM}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{F}^{SPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{M}^{SPyRtr}\left[n_{Pylons},n_2\right]$)

Call Rotor($^{MBD}\Lambda^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{\omega}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{a}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{\alpha}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{Ctrl}\Omega^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{Ctrl}T^{GenPPyRtr}\left[n_{Pylons},n_2\right]$, $^{KAD}\vec{F}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{KAD}\vec{M}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{g}$, $^{MBD}m^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}I_{Rot}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}I_{Tran}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}x_{CM}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{F}^{PPyRtr}\left[n_{Pylons},n_2\right]$, $^{MBD}\vec{M}^{PPyRtr}\left[n_{Pylons},n_2\right]$)

where:

$$^{Ctrl}T^{GenSPyRtr}\left[n_{Pylons},n_2\right]=\begin{cases} ^{Ctrl}Motor\ 7 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=1\right)\right) \\ ^{Ctrl}Motor\ 2 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=2\right)\right) \\ ^{Ctrl}Motor\ 8 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=1\right)\right) \\ ^{Ctrl}Motor\ 1 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=2\right)\right) \end{cases}$$

$$^{Ctrl}T^{GenPPyRtr}\left[n_{Pylons},n_2\right]=\begin{cases} ^{Ctrl}Motor\ 6 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=1\right)\right) \\ ^{Ctrl}Motor\ 3 & for\left(\left(n_{Pylons}=1\right).AND.\left(n_2=2\right)\right) \\ ^{Ctrl}Motor\ 5 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=1\right)\right) \\ ^{Ctrl}Motor\ 4 & for\left(\left(n_{Pylons}=2\right).AND.\left(n_2=2\right)\right) \end{cases}$$

Commented [JJ40]: This math assumes the top node of the pylon is node 1 and that the pylons are numbered from inboard to outboard.

Commented [JJ41]: This math is now done in the C controller.

Transfer outputs from KiteAeroDyn to MBDyn at t:

$$^{MBD}\vec{F}_j^{Fus}=M_F^{P2P}\left(^{KAD}\vec{F}_j^{Fus}\right)$$

$$^{MBD}\vec{M}_j^{Fus}=M_M^{P2P}\left(^{MBD}\vec{u}_j^{Fus},\ ^{KADOut}\vec{u}_j^{Fus},\ ^{KAD}\vec{F}_j^{Fus},\ ^{KAD}\vec{M}_j^{Fus}\right)$$

$$^{MBD}\vec{F}_j^{SWn}=M_F^{P2P}\left(^{KAD}\vec{F}_j^{SWn}\right)$$

$$
\begin{aligned}
&{}^{MBD}\vec{M}_j^{SWn} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SWn}, {}^{KAD\,Out}\vec{u}_j^{SWn}, {}^{KAD}\vec{F}_j^{SWn}, {}^{KAD}\vec{M}_j^{SWn}\right)\\
&{}^{MBD}\vec{F}_j^{PWn} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{PWn}\right)\\
&{}^{MBD}\vec{M}_j^{PWn} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PWn}, {}^{KAD\,Out}\vec{u}_j^{PWn}, {}^{KAD}\vec{F}_j^{PWn}, {}^{KAD}\vec{M}_j^{PWn}\right)\\
&{}^{MBD}\vec{F}_j^{VS} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{VS}\right)\\
&{}^{MBD}\vec{M}_j^{VS} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{VS}, {}^{KAD\,Out}\vec{u}_j^{VS}, {}^{KAD}\vec{F}_j^{VS}, {}^{KAD}\vec{M}_j^{VS}\right)\\
&{}^{MBD}\vec{F}_j^{SHS} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{SHS}\right)\\
&{}^{MBD}\vec{M}_j^{SHS} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SHS}, {}^{KAD\,Out}\vec{u}_j^{SHS}, {}^{KAD}\vec{F}_j^{SHS}, {}^{KAD}\vec{M}_j^{SHS}\right)\\
&{}^{MBD}\vec{F}_j^{PHS} = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{PHS}\right)\\
&{}^{MBD}\vec{M}_j^{PHS} = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PHS}, {}^{KAD\,Out}\vec{u}_j^{PHS}, {}^{KAD}\vec{F}_j^{PHS}, {}^{KAD}\vec{M}_j^{PHS}\right)\\
&{}^{MBD}\vec{F}_j^{SPy}\left[n_{Pylons}\right] = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{SPy}\left[n_{Pylons}\right]\right)\\
&{}^{MBD}\vec{M}_j^{SPy}\left[n_{Pylons}\right] = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{KAD\,Out}\vec{u}_j^{SPy}\left[n_{Pylons}\right], {}^{KAD}\vec{F}_j^{SPy}\left[n_{Pylons}\right], {}^{KAD}\vec{M}_j^{SPy}\left[n_{Pylons}\right]\right)\\
&{}^{MBD}\vec{F}_j^{PPy}\left[n_{Pylons}\right] = M_F^{P2P}\left({}^{KAD}\vec{F}_j^{PPy}\left[n_{Pylons}\right]\right)\\
&{}^{MBD}\vec{M}_j^{PPy}\left[n_{Pylons}\right] = M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{KAD\,Out}\vec{u}_j^{PPy}\left[n_{Pylons}\right], {}^{KAD}\vec{F}_j^{PPy}\left[n_{Pylons}\right], {}^{KAD}\vec{M}_j^{PPy}\left[n_{Pylons}\right]\right)
\end{aligned}
$$

Transfer outputs from MoorDyn to MBDyn at t:

$$
\begin{aligned}
&{}^{MBD}\vec{F}_j^{SWn} = {}^{MBD}\vec{F}_j^{SWn} + M_F^{P2P}\left({}^{MD}PtFairleadLoad\right)\\
&{}^{MBD}\vec{M}_j^{SWn} = {}^{MBD}\vec{M}_j^{SWn} + M_M^{P2P}\left({}^{MBD}\vec{u}_j^{SWn}, {}^{MD}PtFairleadDisplacement, {}^{MD}PtFairleadLoad, \vec{0}\right)\\
&{}^{MBD}\vec{F}_j^{PWn} = {}^{MBD}\vec{F}_j^{PWn} + M_F^{P2P}\left({}^{MD}PtFairleadLoad\right)\\
&{}^{MBD}\vec{M}_j^{PWn} = {}^{MBD}\vec{M}_j^{PWn} + M_M^{P2P}\left({}^{MBD}\vec{u}_j^{PWn}, {}^{MD}PtFairleadDisplacement, {}^{MD}PtFairleadLoad, \vec{0}\right)
\end{aligned}
$$

Commented [JJ42]: See earlier comment about mesh mapping with Wn above.

Private SUBROUTINES

Rotor (SUBROUTINE Rotor)

Implements the structural dynamics of a rotor/drivetrain analytically to calculate the reaction loads (forces and moments) applied on the nacelle, including the applied aerodynamic loads, rotor inertial loads, rotor gyroscopic loads, etc. The analytical formulation assumes that the rotor/drivetrain is a rigid body rotating about the local x-axis of the nacelle coordinate system and that the structure is axisymmetric about this axis (with no imbalances) such that the calculations do not depend on the azimuth angle of the rotor. That is, for a body-fixed (x,y,z) coordinate system in the rotor/drivetrain, it is assumed that:

$$
\begin{aligned}
&{}^{CM}y = {}^{CM}z = 0\\
&I_{xy} = I_{yz} = I_{xz} = 0\\
&I_{xx} = I^{Rot}\\
&I_{yy} = I_{zz} = I^{Tran}
\end{aligned}
$$

Inputs	Outputs	States	Parameters
• Λ^{Nac} – Displaced rotation (absolute orientation) of the nacelle (-) • $\vec{\omega}^{Nac}$ – Rotational velocity (absolute) of the nacelle (rad/s) • $\vec{a}^{Nac}$ – Translational acceleration (absolute) of the nacelle at the rotor reference point (m/s^2) • $\vec{\alpha}^{Nac}$ – Rotational acceleration (absolute) of the nacelle (rad/s^2) • Ω^{Rtr} – Rotor speed about the shaft axis (relative to the nacelle) (rad/s) • T^{Gen} – electrical generator torque applied to the rotor/drivetrain about the shaft axis (N·m) • $\vec{F}^{Aero}$ – aerodynamic forces applied on the rotor at the rotor reference point expressed in the global inertial-frame coordinate system (N) • $\vec{M}^{Aero}$ – aerodynamic moments applied on the rotor about the rotor reference point expressed in the global inertial-frame coordinate system (N·m) • $\vec{g}$ – gravity vector expressed in the global inertial-frame coordinate system (m/s^2) • m – rotor/drivetrain mass (kg) • I^{Rot} – rotor/drivetrain rotational inertia about the shaft axis (kg·m^2) • I^{Tran} – rotor/drivetrain	• $\vec{F}^{React}$ – reaction forces applied on the nacelle at the rotor reference point expressed in the global inertial-frame coordinate system (N) • $\vec{M}^{React}$ – reaction moments applied on the nacelle about the rotor reference point expressed in the global inertial-frame coordinate system (N·m)		

Commented [JJ43]: This is input in place of:

$\dot{\Omega}^{Rtr}$ – Rotor acceleration about the shaft axis (relative to the nacelle) (rad/s^2)

transverse inertia about the rotor reference point (kg·m^2) • ${}^{CM}x$ – distance along the shaft from the rotor reference point to the center of mass of the rotor/drivetrain (positive along positive x) (m)			

Compute the inputs relative to the rotor/drivetrain CM and expressed in the local nacelle coordinate system:

$${}^{CM}\vec{r} = {}^{CM}x\hat{x}^{Nac}$$

$${}^{CM}I^{Tran} = I^{Tran} - m\,{}^{CM}x^2$$

$$\begin{Bmatrix} {}^{CM}F_x^{Aero} \\ {}^{CM}F_y^{Aero} \\ {}^{CM}F_z^{Aero} \end{Bmatrix} = \Lambda^{Nac}\vec{F}^{Aero}$$

$$\begin{Bmatrix} {}^{CM}M_x^{Aero} \\ {}^{CM}M_y^{Aero} \\ {}^{CM}M_z^{Aero} \end{Bmatrix} = \Lambda^{Nac}\left\{\vec{M}^{Aero} - {}^{CM}\vec{r}\times\vec{F}^{Aero}\right\}$$

$$\begin{Bmatrix} g_x \\ g_y \\ g_z \end{Bmatrix} = \Lambda^{Nac}\vec{g}$$

$$\vec{\omega}^{Rtr} = \vec{\omega}^{Nac} + \Omega^{Rtr}\hat{x}^{Nac}$$

$$\vec{\alpha}^{Rtr} = \vec{\alpha}^{Nac}$$

$$\begin{Bmatrix} \omega_x^{Rtr} \\ \omega_y^{Rtr} \\ \omega_z^{Rtr} \end{Bmatrix} = \Lambda^{Nac}\vec{\omega}^{Rtr}$$

$$\begin{Bmatrix} {}^{CM}a_x^{Rtr} \\ {}^{CM}a_y^{Rtr} \\ {}^{CM}a_z^{Rtr} \end{Bmatrix} = \Lambda^{Nac}\left\{\vec{a}^{Nac} + \vec{\alpha}^{Rtr}\times{}^{CM}\vec{r} + \vec{\omega}^{Rtr}\times\left\{\vec{\omega}^{Rtr}\times{}^{CM}\vec{r}\right\}\right\}$$

$$\begin{Bmatrix} \alpha_x^{Rtr} \\ \alpha_y^{Rtr} \\ \alpha_z^{Rtr} \end{Bmatrix} = \Lambda^{Nac}\vec{\alpha}^{Rtr}$$

Compute the reaction loads applied to the rotor/drivetrain at the rotor/drivetrain CM and expressed in the local nacelle coordinate system:

$$\begin{Bmatrix} {}^{CM}F_x^{React} \\ {}^{CM}F_y^{React} \\ {}^{CM}F_z^{React} \end{Bmatrix} = \begin{Bmatrix} -{}^{CM}F_x^{Aero} - mg_x + m\,{}^{CM}a_x^{Rtr} \\ -{}^{CM}F_y^{Aero} - mg_y + m\,{}^{CM}a_y^{Rtr} \\ -{}^{CM}F_z^{Aero} - mg_z + m\,{}^{CM}a_z^{Rtr} \end{Bmatrix}$$

Commented [JJ44]: The equation implemented neglects the rotor acceleration about the shaft axis. The correct equation should be:

$$\vec{\alpha}^{Rtr} = \vec{\alpha}^{Nac} + \dot{\Omega}^{Rtr}\hat{x}^{Nac}$$

, but the rotor acceleration about the shaft axis is not needed because the generator torque is input instead.

$$\begin{Bmatrix} {}^{CM}M_x^{React} \\ {}^{CM}M_y^{React} \\ {}^{CM}M_z^{React} \end{Bmatrix} = \begin{Bmatrix} T^{Gen} \\ -{}^{CM}M_y^{Aero} + I^{Rot}\alpha_y^{Rtr} + \left(I^{Rot} - {}^{CM}I^{Tran}\right)\omega_z^{Rtr}\omega_x^{Rtr} \\ -{}^{CM}M_z^{Aero} + I^{Rot}\alpha_z^{Rtr} - \left(I^{Rot} - {}^{CM}I^{Tran}\right)\omega_y^{Rtr}\omega_x^{Rtr} \end{Bmatrix}$$

Compute the reaction loads applied to the nacelle (this is equal, but opposite to the reaction loads applied to the rotor/drivetrain) at the rotor/drivetrain reference point and expressed in the global inertial frame coordinate system:

$$\vec{F}^{React} = -\left[\Lambda^{Nac}\right]^T \begin{Bmatrix} {}^{CM}F_x^{React} \\ {}^{CM}F_y^{React} \\ {}^{CM}F_z^{React} \end{Bmatrix}$$

$$\vec{M}^{React} = -\left[\Lambda^{Nac}\right]^T \begin{Bmatrix} {}^{CM}M_x^{React} \\ {}^{CM}M_y^{React} \\ {}^{CM}M_z^{React} \end{Bmatrix} + {}^{CM}\vec{r} \times \vec{F}^{React}$$

AfterPredict

This routine updates the actual states based on the temporary states at the successful completion of time step t (including $t = 0$). That said, time has already been updated to $t = t + \Delta t$ before this routine is called, so technically, this routine is first called at $t = \Delta t$.

IF $\left(MOD\left(NINT\left(\frac{t}{\Delta t}\right), N_{KAD/MBD}\right) == 0\right)$ THEN

$\quad {}^{KAD}NewTime = TRUE$

END

IF $\left(MOD\left(NINT\left(\frac{t}{\Delta t}\right), N_{Ctrl/MBD}\right) == 0\right)$ THEN

$\quad {}^{Ctrl}NewTime = TRUE$

$\quad {}^{MBD}OtherStates = {}^{MBD}u$

END

${}^{MD}OtherStates = {}^{MD}u$

${}^{MD}x = {}^{MD}x^{Copy}$

Output

This routine is called at the successful completion of time step t (including $t = 0$) to write output data to a file.

Calculate the KiteFASTMBD write outputs and write them to the output file, together with the module-level write output data currently stored in MiscVars.

This is a list of all possible output parameters available within the KiteFASTMBD (not including the module-level outputs available from KiteAeroDyn, InflowWind, MoorDyn, and the Controller). The names are grouped by meaning, but can be ordered in the OUTPUTS section of the KiteMBDyn Preprocessor input file as you see fit.

Commented [JJ45]: The first equation should be:

$${}^{CM}M_x^{React} = -{}^{CM}M_x^{Aero} + I^{Rot}\alpha_x^{Rtr}$$

But this equals the equation implemented because the generator torque is input instead of the rotor acceleration about the shaft axis.

Deleted: ¶

Deleted: ¶

Deleted: ${}^{KAD}OtherStates = {}^{KAD}y$ ¶
${}^{KAD}z = {}^{KAD}z^{Copy}$ ¶
${}^{IfW}OtherStates = {}^{IfW}y$ ¶

Formatted: Lowered by 3 pt

Deleted: ${}^{MD}OtherStates = \begin{Bmatrix} {}^{MD}u \\ {}^{MD}y \end{Bmatrix}$

Commented [JJ46]: The new OUTPUT section of the KiteMBDyn Preprocessor input file should look something like this:

```
--- OUTPUT ---
True        SumPrint     Print summary data to <RootName>.sum? (flag)
"ES10.3E2"  OutFmt       Format used for text tabular output, excluding the time channel; resulting field should be 10 characters (string)
4           NFusOuts     Number of fuselage outputs (-) [0 to 9]
2, 4, 6, 8  FusOutNd     List of fuselage nodes/points whose values will be output (-) [1 to NFusOuts] [unused for NFusOuts=0]
4           NSWnOuts     Number of starboard wing outputs (-) [0 to 9]
2, 4, 6, 8  SWnOutNd     List of starboard wing nodes/points whose values will be output (-) [1 to NSWnOuts] [unused for NSWnOuts=0]
4           NPWnOuts     Number of port wing outputs (-) [0 to 9]
2, 4, 6, 8  PWnOutNd     List of port wing nodes/points whose values will be output (-) [1 to NPWnOuts] [unused for NPWnOuts=0]
2           NVSOuts      Number of vertical stabilizer outputs (-) [0 to 9]
2, 4        VSOutNd      List of vertical stabilizer nodes/points whose values will be output (-) [1 to NVSOuts ] [unused for NVSOuts =0]
1           NSHSOuts     Number of starboard horizontal stabilizer outputs (-) [0 to 9]
2           SHSOutNd     List of starboard horizontal stabilizer nodes/points whose values will be output (-) [1 to NSHSOuts] [unused for NSHSOuts=0]
1           NPHSOuts     Number of port horizontal stabilizer outputs (-) [0 to 9]
2           PHSOutNd     List of port horizontal stabilizer nodes/points whose values will be output (-) [1 to NPHSOuts] [unused for NPHSOuts=0]
2           NPylOuts     Number of pylon outputs (-) [0 to 9]
2, 4        PylOutNd     List of pylon nodes/points whose values will be output (-) [1 to NPylOuts] [unused for NPylOuts=0]
            OutList      The next line(s) contains a list of output parameters.  See OutListParameters.xlsx for a listing of available output channels (quoted string)
END of input file (the word "END" must appear in the first 3 columns of this last OutList line)
```

Fusβ refers to output β on the fuselage, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***FusOutNd*** list. Setting β > ***NFusOuts*** yields invalid output.

SWnβ and PWnβ refer to output β on the starboard and port wings, respectively, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***SWnOutNd*** and ***PWnOutNd*** lists, respectively. Setting β > ***NSWnOuts*** and ***NPWnOuts***, respectively, yields invalid output.

VSβ refers to output β on the vertical stabilizer, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***VSOutNd*** list. Setting β > ***NVSOuts*** yields invalid output.

SHSβ and PHSβ refer to output β on the starboard and port horizontal stabilizers, respectively, where β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***SHSOutNd*** and ***PHSOutNd*** lists, respectively. Setting β > ***NSHSOuts*** and ***NPHSOuts***, respectively, yields invalid output.

SPα and PPα refer to pylon α on the starboard and port wings, respectively, where α is a one-digit number in the range [1,9]. SPαβ and PPαβ refer to output β on pylon α on the starboard and port wings, respectively, where α is a one-digit number in the range [1,9] and β is a one-digit number in the range [1,9] corresponding to the finite-element node for motions or Gauss point for loads identified by entry β in the ***PylOutNd*** list. Setting α > ***NumPylons*** or setting β > ***NPylOuts*** yields invalid output. If ***NumPylons*** > 9, only the first 9 pylons can be output.

For the fuselage, wings, vertical stabilizer, horizontal stabilizers, and pylons, the local structural coordinate system is used for output, where n is normal to the chord pointed toward the suction surface, c is along the chord pointed toward the trailing edge, and the spanwise (s) axis is directed into the airfoil following the right-hand rule i.e. $s = n \times c$.

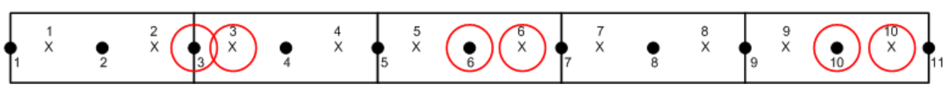

Figure: Example member with 5 finite elements, 11 nodes (•), and 10 Gauss points (x) (each finite element in MBDyn has 2 end nodes, 1 middle node, and 2 Gauss points). The red circles identify the finite-element nodes where motions are output and Gauss points where loads are output when ***NOuts*** = 3 and ***OutNd*** = 3, 6, 10.

Channel Name(s)	Unit(s)	Description
Fuselage		
FusβTDx, FusβTDy, FusβTDz, FusβRDx, FusβRDy, FusβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at Fusβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y’-z’’ (roll-pitch-yaw) rotation sequence
FusβRVn, FusβRVc, FusβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at Fusβ expressed in the local structural coordinate system
FusβTAn, FusβTAc, FusβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at Fusβ expressed in the local structural coordinate system (does not include gravity)
FusβFRn, FusβFRc, FusβFRs, FusβMRn, FusβMRc, FusβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at Fusβ expressed in the local structural coordinate system

Starboard (Right) Wing		
SWnβTDx, SWnβTDy, SWnβTDz, SWnβRDx, SWnβRDy, SWnβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at SWnβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
SWnβRVn, SWnβRVc, SWnβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at SWnβ expressed in the local structural coordinate system
SWnβTAn, SWnβTAc, SWnβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at SWnβ expressed in the local structural coordinate system (does not include gravity)
SWnβFRn, SWnβFRc, SWnβFRs, SWnβMRn, SWnβMRc, SWnβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at SWnβ expressed in the local structural coordinate system
Port (Left) Wing		
PWnβTDx, PWnβTDy, PWnβTDz, PWnβRDx, PWnβRDy, PWnβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at PWnβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
PWnβRVn, PWnβRVc, PWnβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at PWnβ expressed in the local structural coordinate system
PWnβTAn, PWnβTAc, PWnβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at PWnβ expressed in the local structural coordinate system (does not include gravity)
PWnβFRn, PWnβFRc, PWnβFRs, PWnβMRn, PWnβMRc, PWnβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at PWnβ expressed in the local structural coordinate system
Vertical Stabilizer		
VSβTDx, VSβTDy, VSβTDz, VSβRDx, VSβRDy, VSβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at VSβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
VSβRVn, VSβRVc, VSβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at VSβ expressed in the local structural coordinate system
VSβTAn, VSβTAc, VSβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at VSβ expressed in the local structural coordinate system (does not include gravity)
VSβFRn, VSβFRc, VSβFRs, VSβMRn, VSβMRc, VSβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at VSβ expressed in the local structural coordinate system
Starboard (Right) Horizontal Stabilizer		
SHSβTDx, SHSβTDy, SHSβTDz, SHSβRDx, SHSβRDy, SHSβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at SHSβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
SHSβRVn, SHSβRVc, SHSβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at SHSβ expressed in the local structural coordinate system
SHSβTAn, SHSβTAc, SHSβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at SHSβ

		expressed in the local structural coordinate system (does not include gravity)
SHSβFRn, SHSβFRc, SHSβFRs, SHSβMRn, SHSβMRc, SHSβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at SHSβ expressed in the local structural coordinate system
Port (Left) Horizontal Stabilizer		
PHSβTDx, PHSβTDy, PHSβTDz, PHSβRDx, PHSβRDy, PHSβRDz	(m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at PHSβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
PHSβRVn, PHSβRVc, PHSβRVs	(deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at PHSβ expressed in the local structural coordinate system
PHSβTAn, PHSβTAc, PHSβTAs	(m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at PHSβ expressed in the local structural coordinate system (does not include gravity)
PHSβFRn, PHSβFRc, PHSβFRs, PHSβMRn, PHSβMRc, PHSβMRs	(N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at PHSβ expressed in the local structural coordinate system
Pylons		
SPαβTDx, SPαβTDy, SPαβTDz, SPαβRDx, SPαβRDy, SPαβRDz, PPαβTDx, PPαβTDy, PPαβTDz, PPαβRDx, PPαβRDy, PPαβRDz	(m), (m), (m), (deg), (deg), (deg), (m), (m), (m), (deg), (deg), (deg)	Translational and rotational (angular) deflections at SPαβ and PPαβ relative to the undeflected rigid-body position/orientation in the kite coordinate system; the rotations are output as Euler angles in a x-y'-z'' (roll-pitch-yaw) rotation sequence
SPαβRVn, SPαβRVc, SPαβRVs, PPαβRVn, PPαβRVc, PPαβRVs	(deg/s), (deg/s), (deg/s), (deg/s), (deg/s), (deg/s)	Absolute rotational (angular) velocity at SPαβ and PPαβ expressed in the local structural coordinate system
SPαβTAn, SPαβTAc, SPαβTAs, PPαβTAn, PPαβTAc, PPαβTAs	(m/s^2), (m/s^2), (m/s^2), (m/s^2), (m/s^2), (m/s^2)	Absolute translational acceleration at SPαβ and PPαβ expressed in the local structural coordinate system (does not include gravity)
SPαβFRn, SPαβFRc, SPαβFRs, SPαβMRn, SPαβMRc, SPαβMRs, PPαβFRn, PPαβFRc, PPαβFRs, PPαβMRn, PPαβMRc, PPαβMRs	(N), (N), (N), (N·m), (N·m), (N·m), (N), (N), (N), (N·m), (N·m), (N·m)	Shear force and bending moment reaction loads at SPαβ and PPαβ expressed in the local structural coordinate system
Rotors		
SPαTRtSpd, SPαBRtSpd, PPαTRtSpd, PPαBRtSpd	(rad/s), (rad/s), (rad/s), (rad/s)	Rotor speed of the top (T) and bottom (B) rotor on SPα and PPα (relative to the nacelle)
SPαTRtAcc, SPαBRtAcc, PPαTRtAcc, PPαBRtAcc	(rad/s^2), (rad/s^2), (rad/s^2), (rad/s^2)	Rotor acceleration of the top (T) and bottom (B) rotor on SPα and PPα (relative to the nacelle)
Energy Kite		
KitePxi, KitePyi, KitePzi, KiteRoll, KitePitch, KiteYaw	(m), (m), (m), (deg), (deg), (deg)	Translational position and rotational (angular) orientation of the energy kite fuselage reference point in the global inertial-frame coordinate system; the rotations are output as Euler angles in a X-Y'-Z'' (roll-pitch-yaw) rotation sequence
KiteTVx, KiteTVy, KiteTVz, KiteRVx, KiteRVy, KiteRVz	(m/s), (m/s), (m/s), (deg/s), (deg/s), (deg/s)	Absolute translational and rotational (angular) velocity of the energy kite fuselage reference point expressed in the kite coordinate system
KiteTAx, KiteTAy, KiteTAz, KiteRAx, KiteRAy, KiteRAz	(m/s^2), (m/s^2), (m/s^2), (deg/s^2), (deg/s^2), (deg/s^2)	Absolute translational and rotational (angular) acceleration of the energy kite fuselage reference point expressed in the kite coordinate system

These are calculated within KiteFASTMBD as follows:

Fuselage:

$$\begin{Bmatrix} Fus\beta TDx \\ Fus\beta TDy \\ Fus\beta TDz \\ Fus\beta RDx \\ Fus\beta RDy \\ Fus\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO} \left\{ {}^{MBD}\vec{p}^{Fus}_{FusOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{FusR}_{FusOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{FusR}_{FusOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{Fus}_{FusOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} Fus\beta RVn \\ Fus\beta RVc \\ Fus\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{Fus}_{FusOutNd[\beta]} \, {}^{MBD}\vec{\omega}^{Fus}_{FusOutNd[\beta]}$$

$$\begin{Bmatrix} Fus\beta TAn \\ Fus\beta TAc \\ Fus\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{Fus}_{FusOutNd[\beta]} \, {}^{MBD}\vec{a}^{Fus}_{FusOutNd[\beta]}$$

$$\begin{Bmatrix} Fus\beta FRn \\ Fus\beta FRc \\ Fus\beta FRs \\ Fus\beta MRn \\ Fus\beta MRc \\ Fus\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{Fus}_{FusOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{Fus}_{FusOutNd[\beta]} \end{Bmatrix}$$

Starboard (Right) Wing:

$$\begin{Bmatrix} SWn\beta TDx \\ SWn\beta TDy \\ SWn\beta TDz \\ SWn\beta RDx \\ SWn\beta RDy \\ SWn\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO} \left\{ {}^{MBD}\vec{p}^{SWn}_{SWnOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{SWnR}_{SWnOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{SWnR}_{SWnOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{SWn}_{SWnOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} SWn\beta RVn \\ SWn\beta RVc \\ SWn\beta RVs \end{Bmatrix} = \frac{180}{\pi} {}^{MBD}\Lambda^{SWn}_{SWnOutNd[\beta]} \, {}^{MBD}\vec{\omega}^{SWn}_{SWnOutNd[\beta]}$$

$$\begin{Bmatrix} SWn\beta TAn \\ SWn\beta TAc \\ SWn\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{SWn}_{SWnOutNd[\beta]} \, {}^{MBD}\vec{a}^{SWn}_{SWnOutNd[\beta]}$$

$$\begin{Bmatrix} SWn\beta FRn \\ SWn\beta FRc \\ SWn\beta FRs \\ SWn\beta MRn \\ SWn\beta MRc \\ SWn\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{SWn}_{SWnOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{SWn}_{SWnOutNd[\beta]} \end{Bmatrix}$$

Port (Left) Wing:

$$\begin{Bmatrix} PWn\beta TDx \\ PWn\beta TDy \\ PWn\beta TDz \\ PWn\beta RDx \\ PWn\beta RDy \\ PWn\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PWn}_{PWnOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{PWnR}_{PWnOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{PWnR}_{PWnOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{PWn}_{PWnOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} PWn\beta RVn \\ PWn\beta RVc \\ PWn\beta RVs \end{Bmatrix} = \frac{180}{\pi}\, {}^{MBD}\Lambda^{PWn}_{PWnOutNd[\beta]}\, {}^{MBD}\vec{\omega}^{PWn}_{PWnOutNd[\beta]}$$

$$\begin{Bmatrix} PWn\beta TAn \\ PWn\beta TAc \\ PWn\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{PWn}_{PWnOutNd[\beta]}\, {}^{MBD}\vec{a}^{PWn}_{PWnOutNd[\beta]}$$

$$\begin{Bmatrix} PWn\beta FRn \\ PWn\beta FRc \\ PWn\beta FRs \\ PWn\beta MRn \\ PWn\beta MRc \\ PWn\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{PWn}_{PWnOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{PWn}_{PWnOutNd[\beta]} \end{Bmatrix}$$

Vertical Stabilizer:

$$\begin{Bmatrix} VS\beta TDx \\ VS\beta TDy \\ VS\beta TDz \\ VS\beta RDx \\ VS\beta RDy \\ VS\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{VS}_{VSOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{VSR}_{VSOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{VSR}_{VSOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{VS}_{VSOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} VS\beta RVn \\ VS\beta RVc \\ VS\beta RVs \end{Bmatrix} = \frac{180}{\pi}\, {}^{MBD}\Lambda^{VS}_{VSOutNd[\beta]}\, {}^{MBD}\vec{\omega}^{VS}_{VSOutNd[\beta]}$$

$$\begin{Bmatrix} VS\beta TAn \\ VS\beta TAc \\ VS\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{VS}_{VSOutNd[\beta]}\; {}^{MBD}\vec{a}^{VS}_{VSOutNd[\beta]}$$

$$\begin{Bmatrix} VS\beta FRn \\ VS\beta FRc \\ VS\beta FRs \\ VS\beta MRn \\ VS\beta MRc \\ VS\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{VS}_{VSOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{VS}_{VSOutNd[\beta]} \end{Bmatrix}$$

Starboard (Right) Horizontal Stabilizer:

$$\begin{Bmatrix} SHS\beta TDx \\ SHS\beta TDy \\ SHS\beta TDz \\ SHS\beta RDx \\ SHS\beta RDy \\ SHS\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{SHS}_{SHSOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{SHSR}_{SHSOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{SHSR}_{SHSOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{SHS}_{SHSOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} SHS\beta RVn \\ SHS\beta RVc \\ SHS\beta RVs \end{Bmatrix} = \frac{180}{\pi}\; {}^{MBD}\Lambda^{SHS}_{SHSOutNd[\beta]}\; {}^{MBD}\vec{\omega}^{SHS}_{SHSOutNd[\beta]}$$

$$\begin{Bmatrix} SHS\beta TAn \\ SHS\beta TAc \\ SHS\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{SHS}_{SHSOutNd[\beta]}\; {}^{MBD}\vec{a}^{SHS}_{SHSOutNd[\beta]}$$

$$\begin{Bmatrix} SHS\beta FRn \\ SHS\beta FRc \\ SHS\beta FRs \\ SHS\beta MRn \\ SHS\beta MRc \\ SHS\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{SHS}_{SHSOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{SHS}_{SHSOutNd[\beta]} \end{Bmatrix}$$

Port (Left) Horizontal Stabilizer:

$$\begin{Bmatrix} PHS\beta TDx \\ PHS\beta TDy \\ PHS\beta TDz \\ PHS\beta RDx \\ PHS\beta RDy \\ PHS\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO}\left\{ {}^{MBD}\vec{p}^{PHS}_{PHSOutNd[\beta]} - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{PHSR}_{PHSOutNd[\beta]} \\ \dfrac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{PHSR}_{PHSOutNd[\beta]} \right]^T {}^{MBD}\Lambda^{PHS}_{PHSOutNd[\beta]} \right) \end{Bmatrix}$$

$$\begin{Bmatrix} PHS\beta RVn \\ PHS\beta RVc \\ PHS\beta RVs \end{Bmatrix} = \frac{180}{\pi} \, {}^{MBD}\Lambda^{PHS}_{PHSOutNd[\beta]} \, {}^{MBD}\vec{\omega}^{PHS}_{PHSOutNd[\beta]}$$

$$\begin{Bmatrix} PHS\beta TAn \\ PHS\beta TAc \\ PHS\beta TAs \end{Bmatrix} = {}^{MBD}\Lambda^{PHS}_{PHSOutNd[\beta]} \, {}^{MBD}\vec{a}^{PHS}_{PHSOutNd[\beta]}$$

$$\begin{Bmatrix} PHS\beta FRn \\ PHS\beta FRc \\ PHS\beta FRs \\ PHS\beta MRn \\ PHS\beta MRc \\ PHS\beta MRs \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\vec{F}R^{PHS}_{PHSOutNd[\beta]} \\ {}^{MBD}\vec{M}R^{PHS}_{PHSOutNd[\beta]} \end{Bmatrix}$$

Pylons:

$$\begin{Bmatrix} SP\alpha\beta TDx \\ SP\alpha\beta TDy \\ SP\alpha\beta TDz \\ SP\alpha\beta RDx \\ SP\alpha\beta RDy \\ SP\alpha\beta RDz \\ PP\alpha\beta TDx \\ PP\alpha\beta TDy \\ PP\alpha\beta TDz \\ PP\alpha\beta RDx \\ PP\alpha\beta RDy \\ PP\alpha\beta RDz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO} \left\{ {}^{MBD}\vec{p}^{SPy}_{PylOutNd[\beta]}[\alpha] - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{SPyR}_{PylOutNd[\beta]}[\alpha] \\ \frac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{SPyR}_{PylOutNd[\beta]}[\alpha] \right]^T {}^{MBD}\Lambda^{SPy}_{PylOutNd[\beta]}[\alpha] \right) \\ {}^{MBD}\Lambda^{FusO} \left\{ {}^{MBD}\vec{p}^{PPy}_{PylOutNd[\beta]}[\alpha] - {}^{MBD}\vec{p}^{FusO} \right\} - {}^{MBD}\vec{p}^{PPyR}_{PylOutNd[\beta]}[\alpha] \\ \frac{180}{\pi} F^{EulerExtract}\left(\left[{}^{MBD}\Lambda^{FusO} \right]^T \left[{}^{MBD}\Lambda^{PPyR}_{PylOutNd[\beta]}[\alpha] \right]^T {}^{MBD}\Lambda^{PPy}_{PylOutNd[\beta]}[\alpha] \right) \end{Bmatrix}$$

$$\begin{Bmatrix} SP\alpha\beta RVn \\ SP\alpha\beta RVc \\ SP\alpha\beta RVs \\ PP\alpha\beta RVn \\ PP\alpha\beta RVc \\ PP\alpha\beta RVs \end{Bmatrix} = \begin{Bmatrix} \frac{180}{\pi} \, {}^{MBD}\Lambda^{SPy}_{PylOutNd[\beta]}[\alpha] \, {}^{MBD}\vec{\omega}^{SPy}_{PylOutNd[\beta]}[\alpha] \\ \frac{180}{\pi} \, {}^{MBD}\Lambda^{PPy}_{PylOutNd[\beta]}[\alpha] \, {}^{MBD}\vec{\omega}^{PPy}_{PylOutNd[\beta]}[\alpha] \end{Bmatrix}$$

$$\left\{\begin{matrix} SP\alpha\beta TAn \\ SP\alpha\beta TAc \\ SP\alpha\beta TAs \\ PP\alpha\beta TAn \\ PP\alpha\beta TAc \\ PP\alpha\beta TAs \end{matrix}\right\} = \left\{\begin{matrix} {}^{MBD}\Lambda^{SPy}_{PylOutNd[\beta]}[\alpha] & {}^{MBD}\vec{a}^{SPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\Lambda^{PPy}_{PylOutNd[\beta]}[\alpha] & {}^{MBD}\vec{a}^{PPy}_{PylOutNd[\beta]}[\alpha] \end{matrix}\right\}$$

$$\left\{\begin{matrix} SP\alpha\beta FRn \\ SP\alpha\beta FRc \\ SP\alpha\beta FRs \\ SP\alpha\beta MRn \\ SP\alpha\beta MRc \\ SP\alpha\beta MRs \\ PP\alpha\beta FRn \\ PP\alpha\beta FRc \\ PP\alpha\beta FRs \\ PP\alpha\beta MRn \\ PP\alpha\beta MRc \\ PP\alpha\beta MRs \end{matrix}\right\} = \left\{\begin{matrix} {}^{MBD}\vec{F}R^{SPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\vec{M}R^{SPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\vec{F}R^{PPy}_{PylOutNd[\beta]}[\alpha] \\ {}^{MBD}\vec{M}R^{PPy}_{PylOutNd[\beta]}[\alpha] \end{matrix}\right\}$$

Rotors

$$\left\{\begin{matrix} SP\alpha TRtSpd \\ SP\alpha BRtSpd \\ PP\alpha TRtSpd \\ PP\alpha BRtSpd \end{matrix}\right\} = \left\{\begin{matrix} {}^{Ctrl}\Omega^{SPyRtr}[\alpha,1] \\ {}^{Ctrl}\Omega^{SPyRtr}[\alpha,2] \\ {}^{Ctrl}\Omega^{PPyRtr}[\alpha,1] \\ {}^{Ctrl}\Omega^{PPyRtr}[\alpha,2] \end{matrix}\right\}$$

$$\left\{\begin{matrix} SP\alpha TRtAcc \\ SP\alpha BRtAcc \\ PP\alpha TRtAcc \\ PP\alpha BRtAcc \end{matrix}\right\} = \left\{\begin{matrix} {}^{Ctrl}\alpha^{SPyRtr}[\alpha,1] \\ {}^{Ctrl}\alpha^{SPyRtr}[\alpha,2] \\ {}^{Ctrl}\alpha^{PPyRtr}[\alpha,1] \\ {}^{Ctrl}\alpha^{PPyRtr}[\alpha,2] \end{matrix}\right\}$$

Energy Kite

$$\left\{\begin{matrix} KitePxi \\ KitePyi \\ KitePzi \\ KiteRoll \\ KitePitch \\ KiteYaw \end{matrix}\right\} = \left\{\begin{matrix} {}^{MBD}\vec{p}^{FusO} \\ \dfrac{180}{\pi} F^{EulerExtract}\left({}^{MBD}\Lambda^{FusO}\right) \end{matrix}\right\}$$

$$\begin{Bmatrix} KiteTVx \\ KiteTVy \\ KiteTVz \\ KiteRVx \\ KiteRVy \\ KiteRVz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO\,MBD}\vec{v}^{FusO} \\ \dfrac{180}{\pi}\,{}^{MBD}\Lambda^{FusO\,MBD}\vec{\omega}^{FusO} \end{Bmatrix}$$

$$\begin{Bmatrix} KiteTAx \\ KiteTAy \\ KiteTAz \\ KiteRAx \\ KiteRAy \\ KiteRAz \end{Bmatrix} = \begin{Bmatrix} {}^{MBD}\Lambda^{FusO\,MBD}\vec{a}^{FusO} \\ \dfrac{180}{\pi}\,{}^{MBD}\Lambda^{FusO\,MBD}\vec{\alpha}^{FusO} \end{Bmatrix}$$