

Simulation of a Rocket Engine and Launch

Candidates: 15369 & 15401

Institute of Theoretical Astrophysics, University of Oslo

(Dated: December 5, 2022)

Using the Maxwell-Boltzmann distribution, a simulation of the gas inside of our rocket engine was created. This yielded in a rocket engine producing a thrust of $6 \cdot 10^5$ Newton using a total of $1.102 \cdot 10^{15}$ smaller engines and requiring 273.74 Kg/s of fuel in the form of Hydrogen. With these specifications, the rocket launch has been simulated. In the simulation, the rocket reached space at the coordinates $[-12'227, -710.2]$ km and a velocity of $[-13.3, -1-19]$ km/s. These results were satisfactory for our purposes, but did not include some elements such as air resistance, which may cause it to differ from reality.

I. INTRODUCTION

Humanity is the only known species to use and shape everything from elements to their entire planet to their benefit. Now, the next logical step is to look further and find out what there can be found beyond our home planet. There are different ways to get to, and explore space, but the most established way to do this is by using rockets.

Rocket engines, while highly advanced, still simply utilize Newton's laws of motion to create thrust. More specifically the third law, which states "For every action, there is an equal and opposite reaction". When wanting to propel our rocket engine as fast as possible upwards into space and being able to maneuver, we are therefore required to be able to expel the right amount of matter at the right time to create the force needed to propel the rocket as wanted.

The first step to creating such an engine is simulating both its inner workings and the engine as part of a rocket during a launch to determine engine-, rocket- and launch-parameters as well as gaining a better understanding of the complex system, which makes up a functional rocket. This will happen in a simulated solar system created by the ast2000tools package. The rocket in this paper will be using hot H_2 gas under high pressure which will be expelled out the end of the rocket engine to create a force. This is due to H_2 being an ideal gas, which simplifies calculations.

II. THEORY

To complete the calculations for the engine we are going to use a lot of statistics to simplify the behavior of the gas particles.

III. METHOD

As we are trying to simulate a rocket engine, using gas being expelled at high speed, it is important to be able to simulate and understand the gas. A gas is defined as "a fluid (such as air) that has neither independent shape

nor volume but tends to expand indefinitely" and can be comprised of one or multiple individual atoms, or in the case of Hydrogen gas molecules. Hence, if we want to simulate a gas, we need to be able to simulate individual molecules for themselves. To not make calculations excessively difficult we assume to have an ideal gas. This means it's particles can be looked at as point particles without any spatial extension and it's density and temperature being uniform throughout the gas. The particles will therefore be distributed at random positions. However, on the molecular scale some particles will have more energy than others, and since a molecules speed is tightly correlated to its energy, statistical physics need to be used to find their velocity. The most important function we will need is the gaussian probability distribution function, also called normal distribution function and given by

$$f(\mu, \sigma, x) = \frac{1}{\sqrt{2\pi}\sigma} \exp \left[-\frac{1}{2} \left(\frac{x - \mu}{\sigma} \right)^2 \right] \quad (1)$$

The mean value μ and standard deviation σ are parameters which help defining the distribution, and x is a free variable. The shape of the gaussian distribution is a very distinct bell curve. The position of the curve is given by the mean value μ , whereas the width of the curve is controlled by the standard deviation σ given by

$$\sigma = \sqrt{\frac{1}{N} \sum_{i=1}^N (x_i - \mu)^2}$$

This value can be very hard to visualise, as it is given by such an abstract formula. To make visualisation easier, the width of the curve can also be given by another unit called "Full width at half maximum" or in short FWHM. The definition of this unit is already in the name. The width of the curve at half of the maximum value. Since the maximum value of $f(\mu, \sigma; x)$ is attained at the mean value where $x = \mu$ and therefore equal to $f(\mu, \sigma; x = \mu)$, are we looking for values for which

$$f(\mu, \sigma; x_1) = \frac{1}{2} f(\mu, \sigma; \mu) \quad (2)$$

Half the width of the curve at half maximum will then be given by $W_{Half} = \sqrt{(\mu - x_1)^2}$. Due to the symmetry of the curve around $x = \mu$, the full width of the curve can be calculated by multiplying W_{Half} by 2. More specifically

$$FWHM = 2\sqrt{(\mu - x_1)^2}$$

Where x_1 satisfies 2. This unit for the shape of the curve is much easier to visualize for a curve with a given $FWHM$. Since both σ and $FWHM$ are controlling the width of the curve, they can be put in relation to one another. The relationship between σ and $FWHM$ is equal to

$$FWHM = 2\sqrt{2 \ln 2} \sigma$$

The mathematical derivation of this can be found in VII A. However, the gaussian probability distribution function alone is not sufficient to calculate the probability of a value as it only returns the probability density. To calculate the probability of the value x being in the interval $[a, b]$, we need to integrate the probability distribution function over the given interval.

$$P(a \leq x \leq b) = \int_a^b f(\mu, \sigma; x) dx$$

A handy rule of thumb for a value being inside an interval expressed by σ is

$$P(-1\sigma \leq x - \mu \leq 1\sigma) \approx 0.68$$

$$P(-2\sigma \leq x - \mu \leq 2\sigma) \approx 0.95$$

$$P(-3\sigma \leq x - \mu \leq 3\sigma) \approx 0.997$$

To simulate our gas later used in the rocket engine, we will use the Maxwell-Boltzmann probability distribution. This distribution is based on the gaussian probability distribution, and uses a particle's absolute temperature (T) and its mass (m) as arguments as well as the Boltzmann constant $k = 1.380649 \cdot 10^{-23} \text{ J/K}$.

When integrating the Maxwell-Boltzmann distribution over a given interval, it can be used to calculate the probability of a particle's speed being in the given interval.

$$P(v) = \int_a^b \left(\frac{m}{2\pi kT} \right)^{\frac{3}{2}} e^{-\frac{1}{2} \frac{mv^2}{kT}} 4\pi v^2 dv \quad (3)$$

$$P(v_x) = \int_a^b \sqrt{\frac{m}{2\pi kT}} e^{-\frac{1}{2} \frac{mv_x^2}{kT}} dv_x \quad (4)$$

The Maxwell-Boltzmann distribution for the absolute speed in three dimensions 3 and for one dimension 4.

It is important to not confuse these two distributions as one is only for one-dimensional velocities, whereas the

other is for the absolute velocity calculated from the velocity in all three spatial dimensions and will yield different results. Using $10^5 H_2$ molecules with a temperature of $T = 3000 \text{ K}$ each of the distributions can now be plotted to verify the formula and determine approximate values for both the absolute velocity and one-dimensional velocity. From here, the Maxwell-Boltzmann distribu-

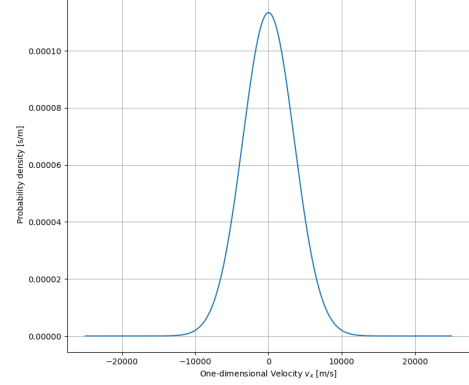


Figure 1. Maxwell-Boltzmann probability distribution for one-dimensional velocities of H_2 particles

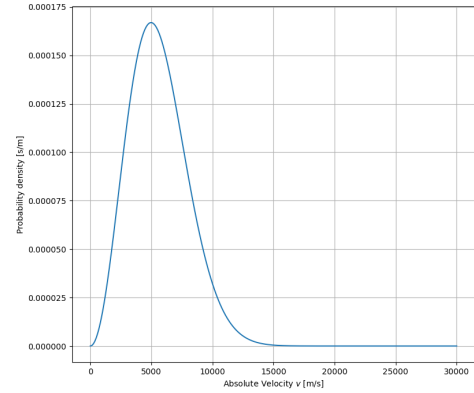


Figure 2. Maxwell-Boltzmann probability distribution for absolute velocity of H_2 particles

tion function can be used to derive some important formulae such as the average speed of a particle as well as the average kinetic energy of a particle in the gas. Note that these apply for ideal gases without intermolecular forces.

$$\langle v \rangle = \int_0^\infty v P(v) dv \quad (5)$$

$$\langle E \rangle = \frac{3}{2} kT \quad (6)$$

The calculations for 5 and 6 can be found in VIIB and can be useful as a reference to compare the exit speed of the particles in our rocket engine. Furthermore, we are able to derive the ideal gas law in section VIIC

$$P = nkT$$

With P being pressure, n being the number of particles, k the Boltzmann constant and T the temperature.

Using this information we are able to simulate the particles of a gas inside a box. To do this, a box is created with each side being of length L. The sides of the box in our simulation will be 10^{-6} meters. Since the box will have to act as a rocket engine, a circular nozzle has to be included. The nozzle in this simulation will simply be a hole in one of the walls with an area of $0.25L^2$. In the beginning the simulation will include 5000 particles with an absolute temperature of 3000 degrees Kelvin. Each particle will be assigned two three-dimensional arrays. One array to store the position, and one to store the velocity. Using the *random* python package and the seed "8", each particle will then be assigned a random position inside the box. Furthermore, an initial velocity is assigned to each particle. The initial velocity is normally distributed using the one-dimensional Maxwell-Boltzmann distribution.

The simulation is then run for a total time of $\tau = 10^{-9}$ seconds with 1000 timesteps of 10^{-12} seconds each using the Euler-Cromer method to advance each particle. In our simulation, we are neglecting the gravitational force, as well as assuming that the particles do not have any spatial dimension. This means, there are no particle-particle collisions. The only collisions happening are particle-wall collisions. We assume that these particle-wall collisions are fully elastic, which means the total kinetic energy is conserved. Hence, the angle of incidence must be equal to the angle of reflection. This can be achieved by keeping all velocities the same, but making the velocity along the axis perpendicular to the wall the particle is colliding with, negative.

As the simulation will be running, particles will escape through the nozzle and take their momentum with them. According to Newtons third law, this results in an equally large, but opposite force acting on the box (or rocket engine). An important thing to note is that, given the assumption that the temperature and pressure are equal in the box (or chamber) at all times, the number of particles inside the chamber needs to be constant. Hence, we will have to create a new particle for each particle escaping through the nozzle. Both the momentum and the number of escaped particles need to be accounted for to derive the thrust force our rocket engine produces as well as the fuel consumption. Our final rocket engine will be a superposition of many individual chambers. The exact number has to be determined imperically, according to the required thrust. Since the chambers are identical, the relation between number of chambers and the thrust force is linear. So, doubling the amount of chambers

means doubling the thrust.

To test the performance of our engine, a function to calculate the required fuel to achieve a speed boost Δv . The required fuel is calculated based on a given thrust force, the fuel consumption of our engine, the mass of our spacecraft and the speed boost Δv .

The chamber is now implemented into the spacecraft to simulate a launch from the home planet. During the launch we will be neglecting air resistance, so the only acting forces will be the thrust force from the rocket engine pointing forward along the axis of the rocket, and the gravitational force of the planet, pointing from the center of mass of the rocket to the center of the planet and given by

$$F = G \frac{Mm}{r^2}$$

Where G is the gravitational constant equal to $6.67430 \cdot 10^{-11}$, M is the mass of the planet, m is the mass of the spacecraft and r is the distance between the center of the planet and the spacecraft. The launch will happen in the reference frame of the center of the planet. When launching, the rocket will therefore have a tangential velocity due to the rotation of the planet. Since we are neglecting air resistance, this tangential velocity will be constant. The radial velocity will be initially zero, but increase due to the thrust force being greater than the gravitational force. Our simulation of the rocket launch will continue until the rocket is in outer space. According to our definition, this is when the rocket reaches escape velocity. Reaching escape velocity is defined as the kinetic energy being equal to the gravitational potential energy. This relation is given by the formula

$$v_e = \sqrt{\frac{2GM}{r}}$$

Where G is the universal gravitational constant, M is the mass of the planet and r is the distance between the spacecraft and the center of the planet. Our objective is to reach space with enough fuel left to continue on our space mission. We therefore need to take enough fuel with us to reach space, but not too much to not have any excessive weight. It is therefore not expedient to simply have as the highest thrust force possible and waste a lot of fuel.

After reaching space, our simulation will end and a few important things will have to be noted. These things are:

- The spacecraft's position when reaching space
- The spacecraft's velocity when reaching space
- The spacecrafts mass when reaching space (incl. fuel)
- The duration of the launch

After successfully launching, the next step will be to enter the solar system. This will mostly consist of changing our reference frame from the planets frame of reference, to the solar system's frame of reference with

the sun being in the center. Here distances are expressed in astronomical units (AU) which are roughly equal to the distance from the earth to the sun (approx. $1.495978707 \cdot 10^{11}$ meters), velocity expressed in AU per year and time is expressed in years. The solar system, generated by the ast2000tools python package has a star with a mass of 4.44081 solar masses and 8 planets, with our home planet being planet 0 with a semi-major axis of 9.00119 AU (Astronomical units).

IV. RESULTS

A. Engine Simulation

The engine simulation yielded a momentum of $5.444 \cdot 10^{-19} \text{ kg} \cdot \text{m/s}$ per box. Divided by the total time $\tau = 10^{-9}$ we get a force of $5.444 \cdot 10^{-10} \text{ N}$ per box. Using a total number of $1.102 \cdot 10^{15}$ boxes we get a thrust of $6 \cdot 10^5 \text{ N}$.

B. Rocket launch

The launch simulation was a success. Using $1.102 \cdot 10^{15}$ boxes, a thrust force of $6 \cdot 10^5$ was achieved. The total mass of the rocket including fuel was 1'001'100 Kg with the dry weight of the spacecraft being 1100 Kg. After simulating the launch, the rocket reached escape velocity, and therefore space after 16 minutes and 36 seconds. The final values for the spacecraft when reaching space were:

Position: x: -12'227 km, y: -710.2 km
 Velocity: v_x : -13.3 km/s, v_y : -1.19 km/s
 Final mass of spacecraft: 120.51 tons
 Remaining fuel: 119.41 tons

This gives us a remaining burn time of 7 minutes 16 seconds.

V. DISCUSSION

A. Engine simulation

The simulation of the rocket engine generally worked as expected. The results were partially flawed, as it sometimes would count a collision if the particle hit the wall opposite of the nozzle. To counter this we used an error factor of 0.5 as a means to quickly remove this error. Furthermore, the code ran slower than expected. We chose therefore to only simulate 10^4 particles instead of 10^5 , and then scaling up the thrust by a factor of 10. As we only counted the thrust of the particles going out of the nozzle along the z-axis,

we rely on the assumption that the law of big numbers to cancel out all momentum in the x and y direction.

B. Rocket launch simulation

The rocket simulation was able to accurately simulate the launch of the rocket from the surface of our planet. However, the results may not be extremely accurate compared to how the launch would be in reality. The most important thing to note, is that we chose to neglect air resistance. This can make a significant difference, as the forces created by air resistance are very high at these high speeds. Implementing this would slow down the rocket and may cause it to not reach space with the configuration we have used in this simulation. The rotation of the atmosphere and winds may also affect the trajectory of the rocket and lead to another final position and velocity.

Furthermore, the precision of the simulation was limited by the processing power of computers. Choosing smaller timesteps in the simulation would lead to more accurate results, but require more processing power and time.

The now chosen timestep size of 8.67×10^{-5} seconds is a compromise, which is manageable with our computing power, and gives a sufficient accuracy for our purposes.

VI. CONCLUSION

By creating a simulation of the gas inside our rocket engine, crucial parameters such as the thrust and mass flow rate were determined. This led to us being able to get a good grasp of the specifications of our engine, which are required to reach space. The engine consists of $1.102 \cdot 10^{15}$ boxes, which create a combined thrust of $6 \cdot 10^5 \text{ N}$. To create these amounts of thrust, the rocket engine had a fuel consumption of 273.74 Kg/s.

Using the specifications from the engine simulation, the rocket launch was simulated. After a launch duration of 16 minutes and 36 seconds, the rocket reached space at a position and velocity of

Position: x: -12'227 km, y: -710.2 km
 Velocity: v_x : -13.3 km/s, v_y : -1.19 km/s

At this point the spacecraft had a remaining fuel mass of 119.41 tons, which would result in 7 minutes and 16 seconds of remaining burn time.

The results were satisfactory for our purposes, however, we decided to neglect air resistance, which could make a significant impact on the final results. An actual launch would therefore probably differ from the results obtained by the simulation, and possibly result in not reaching space.

VII. APPENDIX: MATHEMATICAL DERIVATIONS

A. Full Width Half Maximum

Full width half maximum (FWHM) is a way to express the width of a curve by where $x = \frac{y_{\max}}{2}$.

$$P(x) = \frac{1}{\sigma\sqrt{2\pi}} e^{-\frac{(x-\mu)^2}{2\sigma^2}}$$

We know $P_{\max} = P(\mu)$ and therefore $P(x)_{\text{half max}} = \frac{1}{2}P(\mu)$. We solve for x .

$$\frac{1}{\sqrt{2\pi}\sigma} \exp\left[-\frac{1}{2}\left(\frac{x-\mu}{\sigma}\right)^2\right] = \frac{1}{2\sqrt{2\pi}\sigma}$$

$$\exp\left[-\frac{1}{2}\left(\frac{x-\mu}{\sigma}\right)^2\right] = \frac{1}{2}$$

$$-\frac{1}{2}\left(\frac{x-\mu}{\sigma}\right)^2 = \ln \frac{1}{2}$$

$$\left(\frac{x-\mu}{\sigma}\right)^2 = 2 \ln 2^{-1}$$

$$x - \mu = \pm\sqrt{2 \ln 2}\sigma$$

$$x = \pm\sqrt{2 \ln 2}\sigma - \mu$$

$$x_1 = \mu - \sqrt{2 \ln 2}\sigma \quad V \quad x_2 = \mu + \sqrt{2 \ln 2}\sigma$$

$$\text{FWHM} = x_2 - x_1$$

$$\mu + \sqrt{2 \ln 2}\sigma - \mu + \sqrt{2 \ln 2}\sigma$$

$$\text{FWHM} = 2\sqrt{2 \ln 2}\sigma$$

B. Mean Velocity of a Gas Particle

Maxwells-Boltzmann probability distribution

$$P(v) = \left(\frac{m}{2\pi kT}\right)^{\frac{3}{2}} e^{-\left(\frac{mv^2}{2kT}\right)} 4\pi v^2$$

$$\langle v \rangle = \int_0^\infty v P(v) dv$$

Solving the integral for all velocities from 0 to ∞

$$\langle v \rangle = \int_0^\infty v P(v) dv = \int_0^\infty v \left(\frac{m}{2\pi kT}\right)^{\frac{3}{2}} \exp\left[-\frac{1}{2}\frac{mv^2}{kT}\right] 4\pi v^2 dv$$

$$\langle v \rangle = \sqrt{\left(\frac{m}{2\pi kT}\right)} \int_0^\infty 4\pi v^3 \left(\frac{m}{2\pi kT}\right)^3 \exp\left[-\frac{1}{2}\frac{mv^2}{kT}\right] dx$$

C. Proof of the Ideal Gas Law

To derive $P = nkT$ we solve $\frac{1}{3} \int_0^\infty p v n(p) dp$. We start by substituting v with $\frac{p}{m}$.

$$P = \frac{1}{3} \int_0^\infty p \frac{p}{m} n(p) dp$$

Factoring out constants and adding what we know as the pressure integral

$$P = \frac{3}{3m} \int_0^\infty p^2 P(p) dp$$

$$P = \frac{n}{3m} \int_0^\infty p^2 (2\pi mkT)^{-\frac{3}{2}} \exp\left[-\frac{1}{2}\frac{p^2}{mkT}\right] 4\pi p^2 dp.$$