

Computational Fluid Dynamics using the SU² Code

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1 Introduction

Before the invention of high speed computers, and GPU computing, fluid dynamics was a two discipline field – theoretical, and experimental. However, over the past few decades, a new discipline has emerged, computational fluid dynamics. Computational fluid dynamics has grown to an equal footing with the other two sub-fields, although, it will never be a replacement. Computational fluid dynamics is essentially a perfect wind tunnel; both give important information about the fluid flow around a shape such as an airplane, both allow parameters such as the Reynolds number, temperature, or flow speed to be changed, and both are an effective way to optimize the aerodynamics of an object. Where computational fluid dynamics pulls ahead of wind tunnels is its cost, speed, ease of use, and range of operation. Changing the Reynolds number, or temperature in a computational fluid dynamics experiment is as easy as typing in a new number; it's not quite this simple in a wind tunnel. Furthermore, computational fluid dynamics has the ability to provide a broader range of data, as well as simulate impractical situations to test in a wind tunnel, such as the extreme speeds and temperatures a sub-orbital spacecraft would experience. Computational fluid dynamics experiments can even be done from anywhere in the world, by anyone via remote access.

Computational fluid dynamics has a wide variety of applications even today. There are some things that only computational fluid dynamics simulations can predict, such as how molten iron will flow into a mold in a manufacturing plant. Computational fluid dynamics plays an important role in other areas as well including improving fuel efficiency of automobiles, optimizing pumps and turbines in dams, and even testing naval submarines and torpedoes. Computational fluid dynamics rivals other areas of physics and engineering as a vital area of research.

In this project we run basic computational fluid dynamics simulations for two dimensional flows. The Stanford University Unstructured (SU²) code was used to run both laminar and turbulent simulations. We first examine some background theory, and numerical simplifications; our first experiment looks at a classic airfoil. Then, a static cylinder using both Euler's equations as well as the Navier-Stokes equations was simulated. Finally, we experimented with turbulent flow past a static cylinder and observed vortex shedding and von Kármán vortex streets.

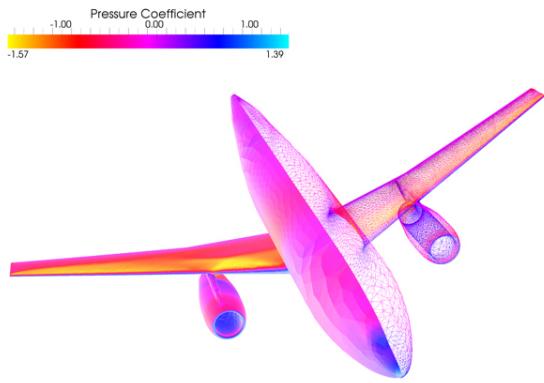


Figure 1: Computational fluid dynamics has a wide range of applications – passenger aircraft for instance [8].

2 Theory

2.1 Non-Turbulent Flow Equations

Non-viscous, ideal fluids obey two equations of motion collectively called Euler's equations. The first is the incompressibility condition

$$\nabla \cdot \mathbf{u} = 0, \quad (1)$$

which arises by assuming the volume of a fluid element, as well as its density are constant. The second equation,

$$\frac{\partial \mathbf{u}}{\partial t} + (\mathbf{u} \cdot \nabla) \mathbf{u} = -\frac{1}{\rho} \nabla p + \mathbf{g}, \quad (2)$$

can be derived from Newton's second law. The problem with Euler's equations is that they are only applicable for non-viscous fluids, which is not very practical. One of the subtler details is that the force experienced by a fluid element is always normal to its surface. However, due to the fluid elements interacting, tangential forces arise. Cauchy's equation,

$$\rho \left(\frac{\partial \mathbf{u}}{\partial t} + (\mathbf{u} \cdot \nabla) \mathbf{u} \right) = \nabla \cdot \mathbf{T} + \rho \mathbf{g}, \quad (3)$$

adds a stress tensor to handle these additional forces. If we chose $T_{ij} = -p\delta_{ij}$ we simply get (2), as expected, however, if we chose a more complex tensor with a deformation term, such as

$$T_{ij} = -p\delta_{ij} + \mu \left(\frac{\partial u_j}{\partial x_i} + \frac{\partial u_i}{\partial x_j} \right),$$

(3) reduces to

$$\begin{aligned} \rho \left(\frac{\partial \mathbf{u}}{\partial t} + (\mathbf{u} \cdot \nabla) \mathbf{u} \right) &= -\nabla p + \mu \nabla^2 \mathbf{u} + \rho \mathbf{g} \quad \text{or,} \\ \frac{\partial \mathbf{u}}{\partial t} + (\mathbf{u} \cdot \nabla) \mathbf{u} &= -\frac{1}{\rho} \nabla p + \nu \nabla^2 \mathbf{u} + \mathbf{g}, \end{aligned} \quad (4)$$

which along with the incompressibility condition, (1), are the Navier-Stokes equations. Notice that if $\nu = 0$, there is no interaction between fluid elements, and this reduces simply to Euler's equations, (2).

2.2 Reynolds-Averaged Navier-Stokes (RANS) Equation

The Navier-Stokes equations are notoriously difficult to solve since it is a second order partial differential equation, with three components, and four dependencies each. To make solving the Navier-Stokes equations easier, especially for turbulent flows, we can assume the flow is the superposition of the steady, and turbulent flows, and then take the time average. We can make the following variable substitutions:

$$\begin{aligned} \alpha_i &\equiv \bar{\alpha}_i + \alpha'_i, \\ \text{where } \bar{\alpha}_i &= \frac{1}{\tau} \int_0^\tau \alpha_i dt \end{aligned}$$

is the steady time average on some time scale τ , and α'_i is the turbulent perturbations. It follows from this definition that

$$\bar{\alpha}'_i = 0, \quad \frac{\partial \bar{\alpha}_i}{\partial t} = 0, \quad \bar{\alpha}_i = \bar{\alpha}_i, \quad \text{and} \quad \frac{\partial \bar{\alpha}_i}{\partial x_j} = \frac{\partial \bar{\alpha}_i}{\partial x_j}. \quad (5)$$

We can write (4) in the more suggestive component form:

$$\sum_j \frac{\partial u_i}{\partial t} + u_j \frac{\partial u_i}{\partial x_j} = \sum_j -\frac{1}{\rho} \frac{\partial p}{\partial x_i} + \nu \frac{\partial^2 u_i}{\partial x_j^2} + g_i,$$

then, making the appropriate substitutions, taking the time average, and using the properties from (5), we obtain

$$\sum_j \overline{u_j} \frac{\partial \overline{u_i}}{\partial x_j} + \overline{u'_j} \frac{\partial \overline{u'_i}}{\partial x_j} = \sum_j -\frac{1}{\rho} \frac{\partial \overline{p}}{\partial x_i} + \nu \frac{\partial^2 \overline{u_i}}{\partial x_j^2} + \overline{g_i}.$$

Rewriting this in vector form gives the Reynolds-Averaged Navier-Stokes equation,

$$(\bar{\mathbf{u}} \cdot \nabla) \bar{\mathbf{u}} + (\bar{\mathbf{u}}' \cdot \nabla) \bar{\mathbf{u}}' = -\frac{1}{\rho} \nabla \bar{p} + \nu \nabla^2 \bar{\mathbf{u}} + \bar{\mathbf{g}}. \quad (6)$$

2.3 Spalart-Allmaras (SA) Turbulent Model

While (6) makes fluid flow predictions easier, a model is still needed for ν . According to the Boussinesq hypothesis [4], an increase in the viscosity gives the effect of turbulence, and therefore $\nu = \bar{\nu} + \nu'$. One such way to model this altered viscosity and the turbulence, is the Spalart-Allmaras turbulence model. This model determines the turbulent viscosity using the following:

$$\nu' = \hat{\nu} f_{v1}; \quad f_{v1} = \frac{\chi^3}{\chi^3 + c_{v1}^3}; \quad \chi \equiv \frac{\hat{\nu}}{\nu}. \quad (7)$$

Assuming the flow is incompressible, and the diffusivity is constant, $\hat{\nu}$ is obtained by solving the convection-diffusion equation,

$$\frac{\partial \hat{\nu}}{\partial t} = D \nabla^2 \hat{\nu} - \bar{\mathbf{u}} \nabla \hat{\nu},$$

where D is the diffusivity.

3 Stanford University Unstructured (SU²) Code

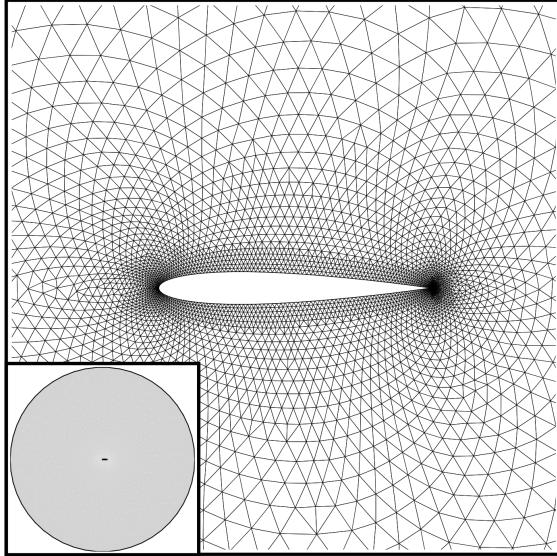


Figure 2: Unstructured mesh of an airfoil [7].

namics software are plagued with such as price, compatibility, and memory [1]. Some of the features of the SU² code include, but are not limited to:

The Stanford University Unstructured (SU²) code¹ is a relatively new, open source, constrained partial differential equation solver, optimized for computational fluid dynamics problems. It was initially developed by Francisco Palacios, Thomas Economou, and Juan Alonso from the Aerospace Design Laboratory of the Department of Aeronautics and Astronautics at Stanford University. SU² is written mostly in C++; and because it is open source, the capabilities have since been extended by many teams of developers from around the world, in addition to the team at Stanford University.

SU² was initially developed under the philosophy that it should remain open source, portable, and highly encapsulated while operating at a high performance standard. The code uses only widely available libraries and dependencies for C++ and Python; furthermore, it is written at a very high, abstract level to minimize disc space, and maximize performance. This philosophy was used to eliminate many of the problems other computational fluid dy-

¹<http://su2.stanford.edu/>

- Reynolds-Averaged Navier-Stokes equations (6)
- Spalart-Allmaras Turbulence model (7) ($c_{v1} = 7.1$ [1] [2])
- Menter Shear Stress Transport model (an alternative to the SA model)
- Compressible flows
- Wave equation, heat equation, and plasma equations
- Optimal shape design (i.e. Supersonic aircraft)
- Dynamic meshes

To conduct a simulation, two main files are required, the configuration file, and the mesh file. All of the parameters such as the fluid speed, Reynolds number, and governing equations are specified in the configuration file. The shape geometry of the problem is stored as an unstructured mesh – hence the name – in the mesh file, an example visualization is in Figure 2. The mesh is unstructured in the sense that there is no algebraic relation between nodes. Instead, the mesh is specified by listing the node coordinates, and the adjacent nodes it is connected to. The density of nodes is greatest at the surface since it is the most complex area, and any errors could quickly propagate through the solution.

Typically a particular node within the mesh will be surrounded by six triangular cells. Using the centroids of the cells, as well as the midpoints of the edges, we can construct a hexagonal region, Ω_i , which surrounds the i th node. Our partial differential equations can be discretized by

$$\int_{\Omega_i} \frac{\partial \mathbf{u}}{\partial t} d\Omega + R_i(\mathbf{u}) = 0,$$

where $R_i(\mathbf{u})$ is the residual. After the spacial components have been discretized, the governing equations can be numerically solved with a variety of methods and optimizations. Furthermore, due to the way the solution is discretized, the mesh can dynamically be updated; this is a key feature in the shape optimization process.

4 Scripting and Automation

Throughout this project Wolfram Mathematica 10.1.0 for Linux x86 was used to sift through the vast data files and extract the relevant information to produce the plots. The SU² code natively writes to the data files in the format used for Tecplot², a visualization and analysis tool for computational fluid dynamics. Since Tecplot was not used, a Wolfram function was needed to convert the data to a form Mathematica could use. Once the data was in a usable form, the `ListDensityPlot` function was used to create the images. The function to tidy the data along with the plotting function were combined into a Wolfram script such as Figure 3, which can be executed from the terminal. To fully automate the image generation, a shell script was written to iterate over each data file.

²<http://www.tecplot.com/>

```

1 #!/usr/local/bin/WolframScript -script
2
3 (*Change the current directory*)
4 SetDirectory["/home;brady/SU2/CFD/Results/Pitching_Airfoil_Turb"];
5 Print[ToString[$CommandLine[[4]]]]; (*To check the progress*)
6
7 CSV = "surface_flow_0" <> ToString[$CommandLine[[4]]] <> ".csv";
8 DAT = "flow_0" <> ToString[$CommandLine[[4]]] <> ".dat";
9 PNG = "Airfoil_Turb" <> ToString[$CommandLine[[4]]] <> ".png";
10
11 (*Set the plot limits, colour function, and the legend style*)
12 xyzlimits = {{-0.5, 2}, {-1, 1}, {0, 450}};
13 colfunc = ColorData["SunsetColors"][[xyzlimits[[3, 2]]]] &;
14 leg = BarLegend[{colfunc, xyzlimits[[3]]}, LegendLabel → "Velocity (m/s)",
15 LegendMarkerSize → 500];
16
17 (*Draw a gray ploygon using the points of the surface_flow.csv*)
18 shape = Graphics[{Gray, Polygon[Import[CSV][[2;;-1, {2, 3}]]]}];
19
20 (*Clean the data so it's in a usable form*)
21 (*Import the data file, and remove the preamble (three lines)*)
22 datafile = Import[DAT][[4 ;; -1]];
23
24 (*There is four seemingly random numbers per line for several lines at*)
25 (*the end, this ignores those lines*)
26 Do[If[Dimensions[datafile[[i]]][[1]] == 4,
27 {CleanData = datafile[[1 ;; i - 1]], Break[]}],
28 {i, 1, Dimensions[datafile][[1]]}];
29 (*Only the first 5 columns are needed: x, y, \[\[Rho], \[\[Rho]\]u, \[\[Rho]\]v*)
30 Data = CleanData[[All, 1 ;; 5]];
31
32 (*Declare and fill array for the velocity*)
33 velocity = {};
34 Do[AppendTo[velocity, {Data[[i, 1]], Data[[i, 2]],
35 Sqrt[(Data[[i, 4]]/Data[[i, 3]])^2 + (Data[[i, 5]]/Data[[i, 3]])^2]}],
36 {i, 1, Length[Data]}]
37
38 velplot = ListDensityPlot[velocity, ColorFunction → colfunc,
39 PlotRange → xyzlimits, AspectRatio → Automatic,
40 LabelStyle → {Black, FontSize → 18}, PlotLegends → leg,
41 ColorFunctionScaling → False, Frame → True,
42 FrameLabel → {"x", "y"}, PlotLabel → "Pitching Airfoil",
43 ImageSize → Full];
44
45 contplot = ListContourPlot[velocity, PlotRange → xyzlimits,
46 ContourShading → None, Contours → {200, 250, 300, 350}];
47
48 SetDirectory["/home;brady/SU2/CFD/TeX/Airfoil_Animation_Turb"];
49 Export[PNG, Show[velplot, contplot, shape]]

```

Figure 3: The Wolfram script used to generate Figure 4b. Syntax highlighting modified from [5].

5 Results

5.1 Airfoil

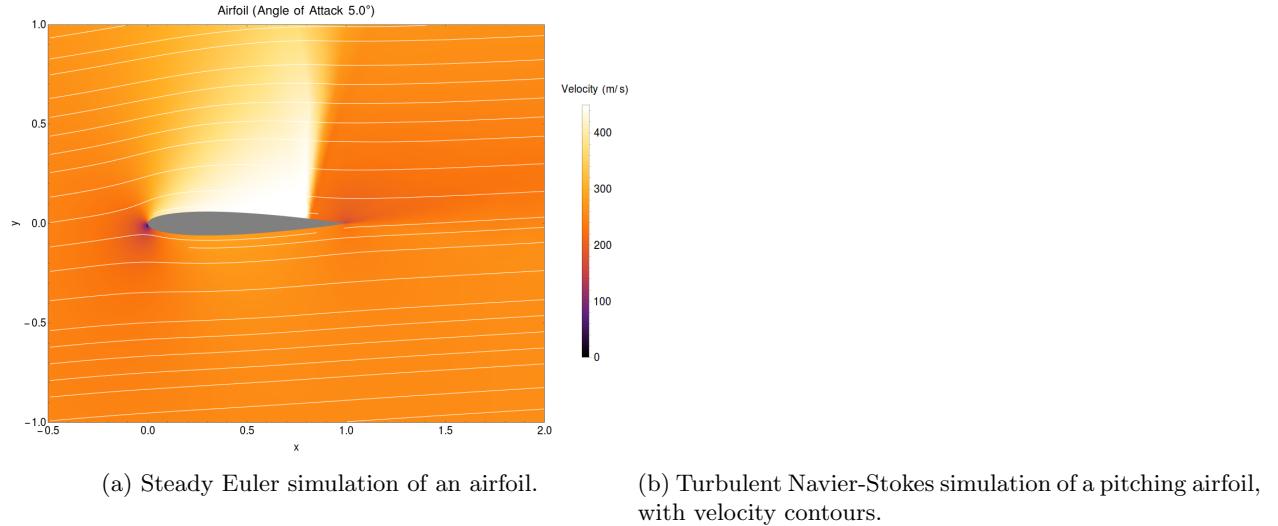


Figure 4: Results of the airfoil simulations.

A NACA 0012 airfoil is the first tutorial in the Quick Start section of the documentation [7]. This series of airfoils was developed by the National Advisory Committee for Aeronautics (NACA); for this particular case 0012 is the code for a symmetric airfoil with a thickness 12% of its length. As this is the first tutorial it walks the user through the process of configuring, and running SU² for the first time after it is installed – a natural place to start. In the first simulation Euler's equations were used to simulate an airfoil during takeoff. The full specifications of both simulations can be found in Table 1; this simulation took approximately 45 seconds to complete.

Figure 4a shows the streamlines, and velocity of the air around the airfoil from the data. It is clear from the data that due to the shape of the airfoil, the velocity of the air that flows over the wing of an aircraft is greatly increased compared to the air that flows under. A consequence of this increase in velocity is a decrease in pressure, which in turn provides lift for the aircraft.

The second simulation was of a pitching airfoil, the angle continuously varied from -4.0° to 4.0° – unlike the first simulation where the angle was a constant 5.0° . Furthermore, the Navier-Stokes equations with turbulence were used. Figure 4b shows the results of the simulation; using a viscous turbulent flow, such as real aircraft experience, there is undoubtedly a long turbulent wake off of the tail end of the airfoil. Another interesting property is that some of the velocity contours initially form in two distinct areas, and then come together as the airfoil flattens out.

5.2 Steady Flow Around a Static Cylinder

boundary layer

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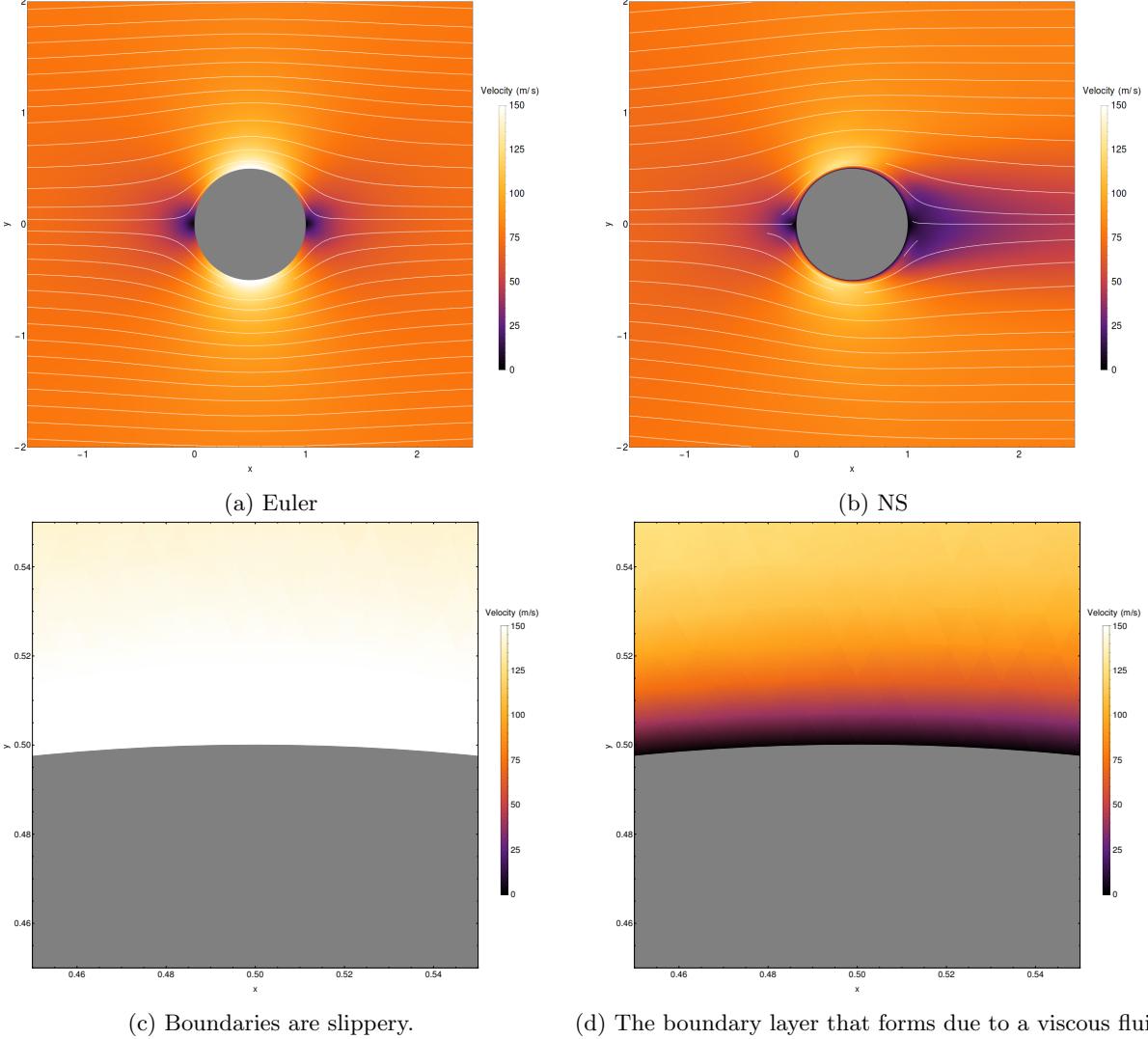


Figure 5: Static Cylinders

5.3 Turbulent Cylinder – Vortex Shedding

The equation (2) is interesting.

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Figure 6: Neat! The full video can be found at <https://youtu.be/1lBQN1EdXss>

malesuada eu, pulvinar at, mollis ac, nulla. Curabitur auctor semper nulla. Donec varius orci eget risus. Duis nibh mi, congue eu, accumsan eleifend, sagittis quis, diam. Duis eget orci sit amet orci dignissim rutrum.

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6 Conclusion

Fusce mauris. Vestibulum luctus nibh at lectus. Sed bibendum, nulla a faucibus semper, leo velit ultricies tellus, ac venenatis arcu wisi vel nisl. Vestibulum diam. Aliquam pellentesque, augue quis sagittis posuere, turpis lacus congue quam, in hendrerit risus eros eget felis. Maecenas eget erat in sapien mattis porttitor. Vestibulum porttitor. Nulla facilisi. Sed a turpis eu lacus commodo facilisis. Morbi fringilla, wisi in dignissim interdum, justo lectus sagittis dui, et vehicula libero dui cursus dui. Mauris tempor ligula sed lacus. Duis cursus enim ut augue. Cras ac magna. Cras nulla. Nulla egestas. Curabitur a leo. Quisque egestas wisi eget nunc. Nam feugiat lacus vel est. Curabitur consectetur.

Method	Euler	Navier-Stokes
Turbulence	None	None
Flow Speed	75 ms ⁻¹	75 ms ⁻¹
Reynolds Number	0	20000
Temperature	300.0 K	300.0 K
Pressure	1 atm	N/A
Iterations	250	250

Table 2: Parameters used in the non-turbulent flow simulations. Results shown in Figure 5.

Method	Navier-Stokes
Turbulence	Spalart-Allmaras
Flow Speed	75 ms ⁻¹
Reynolds Number	20000
Temperature	300.0 K
Time Step	0.0006 s
Iterations per Time Step	10
Total Iterations	2000
Total Time	1.200 s

Table 3: Parameters used in the turbulent flow simulations. Results shown in Figures 6, and 7.

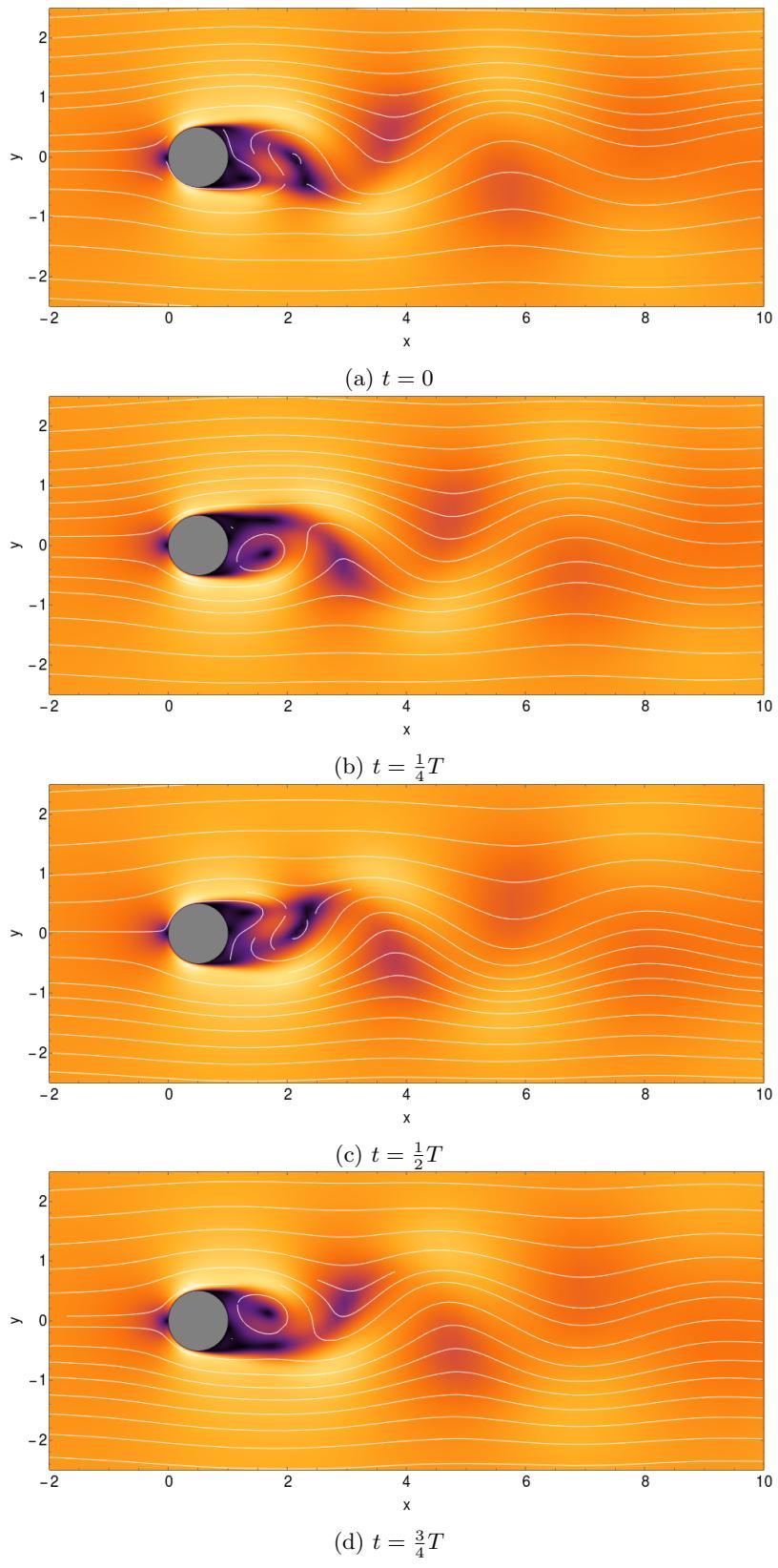


Figure 7: Vortex shedding!

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References

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