

CFL3D provides multiple options for solving CFD problems. A variety of 0-equation, 1-equation, and 2-equation turbulence models are available. The code has static or dynamic mesh capabilities. If the grid has multiple zones, there are several choices for communication between the zones which can be used independently or in conjunction with one another. For convergence acceleration, multigrid and mesh sequencing are available.

The test cases described in this chapter provide a sampling of CFL3D's capabilities. After studying the test cases, the user will hopefully be able to choose the best strategy for his or her particular applications. For information on how to obtain the files needed for the test cases see "Acquiring the Code and Example Files" on page 7.

Several two-dimensional test cases discussed in this chapter involve airfoils and flat plates. The use of a single block is exemplified with a RAE 2822 airfoil case. A NACA 0012 airfoil case is used as an example for both grid patching and grid overlapping. Also included is a multielement airfoil case, involving grid overlapping. The flat plate examples include a turbulent flat plate case and a vibrating flat plate case which illustrates the dynamic mesh capabilities of CFL3D. Also included are a multistream nozzle case and a rotor-stator case.

One three-dimensional example is for an axisymmetric bump. By taking advantage of periodicity, it is solved on a grid with only two planes in the circumferential direction. Three of the three-dimensional examples are for wing topologies. A single block case is set up for an F-5 wing. A case solving for the viscous flow over the Onera M-6 wing is also set up using a single block. A delta wing case with laminar flow is also available. Keep in mind that, in order to have cases that are "quick" to run, the three-dimensional grids used in some of these examples are relatively coarse compared to what one *should* use to adequately resolve the flow.

Note: you may see slight differences in your results, due to errors in CFL3D that have been corrected since the plots in this chapter were generated.

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### *9.1 Two-dimensional Test Cases*

CFL3D solves for the primitive variables at the *cell centers* of a grid. Therefore, for two-dimensional cases, *two* grid planes are needed for one plane of cell-center points to exist. The "2-d direction" is the *i* direction designated by setting **idim** = 2 (and **i2d** = 1). Typically, after a 2-d grid is generated, it is simply duplicated such that identical planes

exist at  $i = 1$  and  $i = 2$  with a constant value in the third direction. For example, if  $x$  is the third direction,  $x$  is typically set to 0.0 at  $i = 1$  and  $x$  might equal 1.0 or  $-1.0$  at  $i = 2$ . When setting up the third direction by duplicating the grid plane, keep in mind that the right-hand rule *must* be satisfied. See “The Right-Hand Rule” on page 67. Also note that, while this step *is* doubling the number of grid points, only *one* plane of data is actually computed. Therefore, the number of points in *one* plane of the grid should be used when estimating the time required to run the code.

### 9.1.1 RAE 2822 Airfoil

This test case solves for the viscous flow over the RAE 2822 airfoil at  $\alpha = 2.72^\circ$  with  $M_\infty = 0.75$ . These are corrected conditions from Case 10 of Cook et al.<sup>15</sup> The grid consists of a single zone with 24929 points in one plane. Menter’s  $k - \omega$  SST model is used to solve the turbulent flow with a Reynolds number of 6.2 million. The memory requirement for this example is 5.7 million words. A typical timing for this case is 382 CPU seconds on a CRAY YMP (NASA LaRC’s Sabre as of June 1996). A close-up of the grid near the airfoil is shown in Figure 9-1.

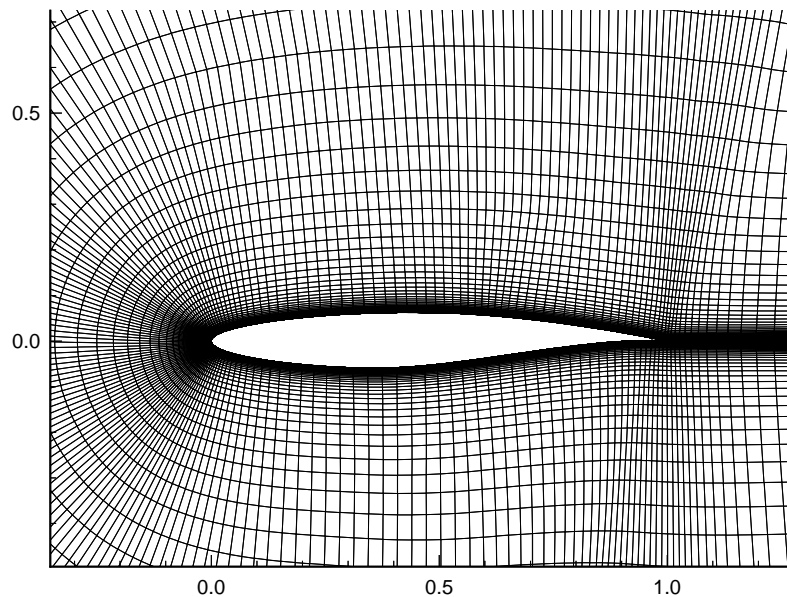


Figure 9-1. Single zone RAE 2822 airfoil grid.

Besides the CFL3D code, the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
rae10.inp	input for CFL3D
rae10.grd	formatted single plane grid
grid2dto3d.f	converter for creating 2 grid planes

The steps for running this case on the YMP are as follows:

### Step 1

Compile the grid converter code:

```
cft77 grid2dto3d.f
```

### Step 2

Link the grid converter object file:

```
segldr -o grid2dto3d grid2dto3d.o
```

### Step 3

Run the grid converter program (the binary file rae10.bin will be output):

```
grid2dto3d
```

In answer to the questions, type:

```
rae10.grd
rae10.bin
2
0
```

### Step 4

Use the makefile to compile, link, and create the executable for the precfl3d code (be sure precfl.h is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 5

Run the precfl3d code (the cflx.h files will be output):

```
precfl3d < rae10.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

### Step 7

Run the CFL3D code:

```
cfl3d < rael0.inp
```

The input file for this case is:

```
FILES:
rael0.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
RAE case 10, with SST model
  XMACH      ALPHA      BETA  REUE,MIL    TINF,DR      IALPH      IHSTRY
    0.7500    02.720      0.0    6.2000    460.0        0          0
    SREF      CREF      BREF      XMC        YMC        ZMC
  1.00000    1.00000    1.0000    0.00000    0.00        0.00
    DT        IREST      IFLAGTS      FMAX      IUNST      CFLTAU
  -5.0000    0          000    05.0000    0          10.0
  NGRID      NPLOT3D      NPRINT      NWREST      ICHK      I2D      NTSTEP      ITA
    1          1          1          6100      0          1          0001      1
    NCG        IEM      IADVANCE      IFORCE      IVISC(I)    IVISC(J)    IVISC(K)
    2          0          0          1          0          0          7
    IDIM      JDIM      KDIM
    2          257      97
  ILAMLO      ILAMHI      JLAMLO      JLAMHI      KLAMLO      KLAMHI
    1          2          88      159        1          97
  INEWG      IGRIDC      IS          JS          KS          IE          JE          KE
    0          0          0          0          0          0          0          0
  IDIAG(I)    IDIAG(J)    IDIAG(K)    IFLIM(I)    IFLIM(J)    IFLIM(K)
    1          1          1          3          3          3
  IFDS(I)     IFDS(J)     IFDS(K)     RKAP0(I)    RKAP0(J)    RKAP0(K)
    1          1          1          0.3333     0.3333     0.3333
  GRID        NBCIO      NBCIDIM      NBCJO      NBCJDIM      NBCKO      NBCKDIM      IOVRLP
    1          1          1          1          1          3          1          0
I0:  GRID      SEGMENT      BCTYPE      JSTA      JEND      KSTA      KEND      NDATA
    1          1          1001      0          0          0          0          0
IDIM: GRID      SEGMENT      BCTYPE      JSTA      JEND      KSTA      KEND      NDATA
    1          1          1002      0          0          0          0          0
J0:  GRID      SEGMENT      BCTYPE      ISTA      IEND      KSTA      KEND      NDATA
    1          1          1002      0          0          0          0          0
JDIM: GRID      SEGMENT      BCTYPE      ISTA      IEND      KSTA      KEND      NDATA
    1          1          1002      0          0          0          0          0
K0:  GRID      SEGMENT      BCTYPE      ISTA      IEND      JSTA      JEND      NDATA
    1          1          0          0          0          1          41          0
    1          2          2004      0          0          41          217          2
      TWTYPE      CQ
        0.          0.
    1          3          0          0          0          217          257          0
KDIM: GRID      SEGMENT      BCTYPE      ISTA      IEND      JSTA      JEND      NDATA
    1          1          1003      0          0          0          0          0
  MSEQ      MGFLAG      ICONSF      MTT      NGAM
    1          1          0          0          02
  ISSC EPSSSR(1) EPSSSR(2) EPSSSR(3)      ISSR EPSSSR(1) EPSSSR(2) EPSSSR(3)
    0          0.3      0.3      0.3          0          0.3      0.3      0.3
  NCYC      MGLEVG      NEMGL      NITFO
```

```

      500      03      00      000
      MIT1     MIT2     MIT3     MIT4     MIT5     MIT6     MIT7     MIT8
      01      01      01      01      01      1       1       1
1-1 BLOCKING DATA:
  NBLI
  1
NUMBER  GRID    :   ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  1      1      :   1     1     1     2     41    1     1     2
NUMBER  GRID    :   ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  1      1      :   1    257    1     2    217    1     1     2
PATCH SURFACE DATA:
  NINTER
  0
PLOT3D OUTPUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
  1      0      1     01     1     01    999    1     1    999    1
MOVIE
  0
PRINT OUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
  1      0      1     01     1     41    217    1     1     1     1
CONTROL SURFACE:
  NCS
  0
  GRID ISTART  IEND  JSTART  JEND  KSTART  KEND  IWALL  INORM

```

After running this test case a result such as that shown in Figure 9-2 should be obtained. In the figure, surface pressure coefficients are plotted along with experimental data for this case. The computational surface pressures can be obtained from file `cf13d.prout`. Experimental surface pressure coefficients from Cook et. al<sup>15</sup> are included with this test case for comparison purposes. The file is called `rae10.cpexp`. The residual plots shown in Figure 9-3 should also be duplicated. These convergence histories can be found in `cf13d.res`.

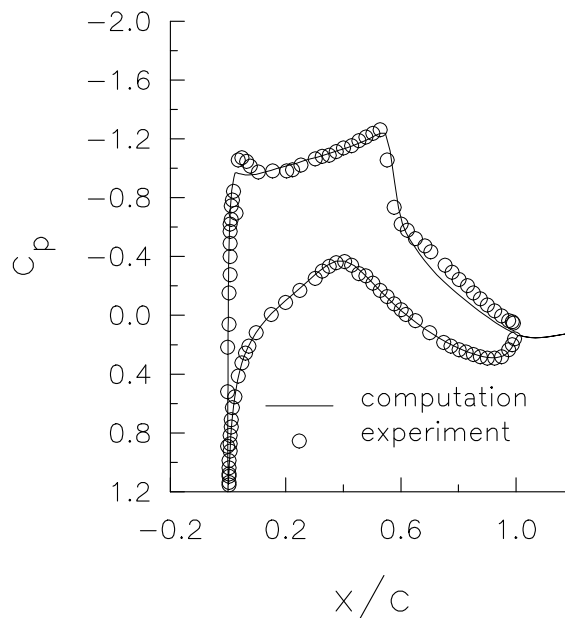
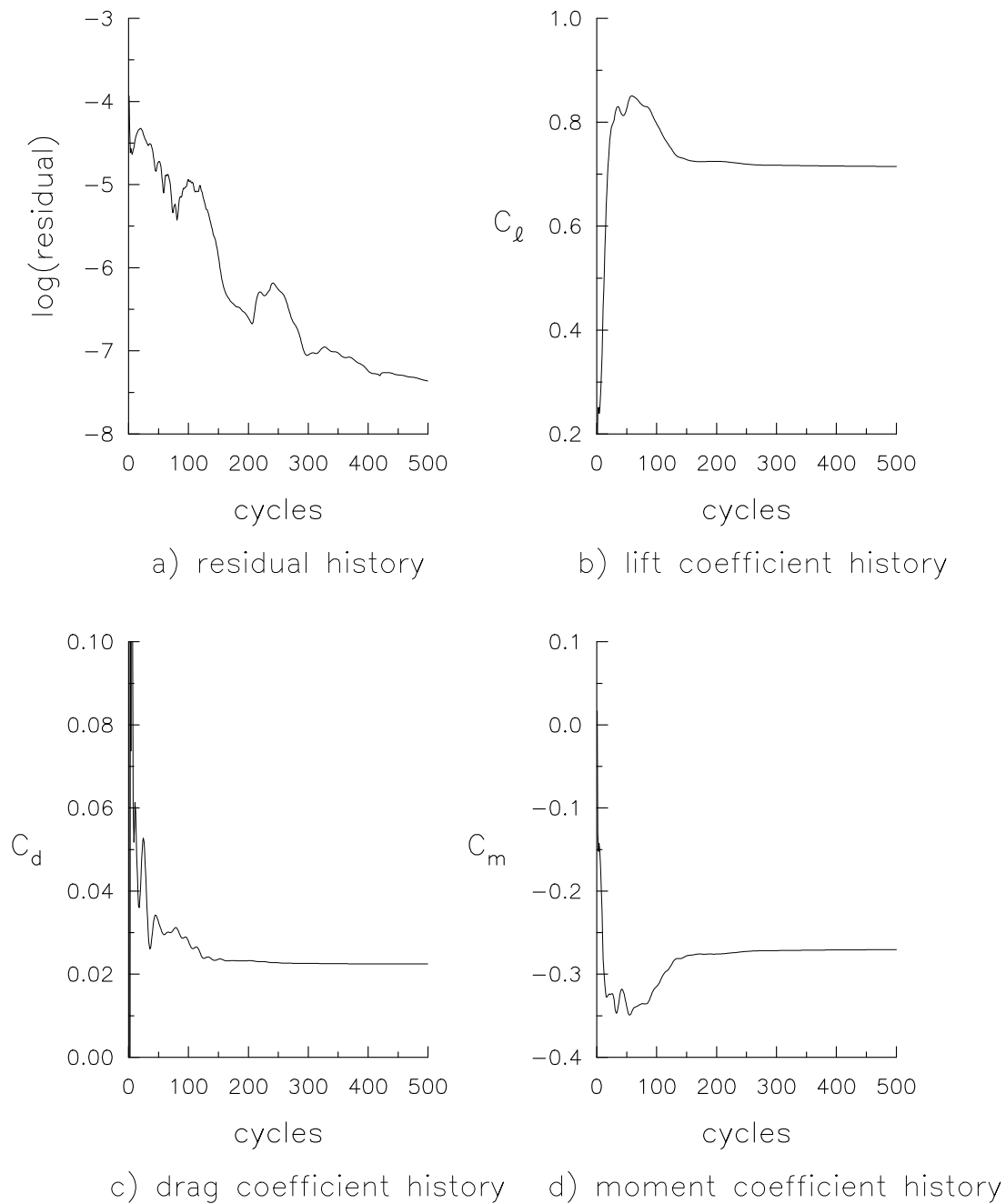


Figure 9-2. Surface pressure coefficients for RAE 2822 airfoil;

$$\alpha = 2.72^\circ, M_\infty = 0.75, Re_{\bar{L}_R} = 6.2 \times 10^6.$$



**Figure 9-3.** Residual and coefficient histories for RAE 2822 airfoil case;

$$\alpha = 2.72^\circ, M_\infty = 0.75, Re_{\tilde{L}_R} = 6.2 \times 10^6$$

### 9.1.2 NACA 0012 Airfoil with Overlapped Grids

This test case solves for the inviscid flow over the NACA 0012 airfoil at  $\alpha = 5^\circ$  with  $M_\infty = 0.2$ . The grid has a total of 4850 points on two grid zones which communicate with one another through overlapped grid stencils. Therefore, the MaGGiE code is used in addition to CFL3D. The memory requirement for this case is 1.8 million words. A typical timing for this case is 43 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of June 1996). A close-up of the grid near the airfoil is shown in Figure 9-4.

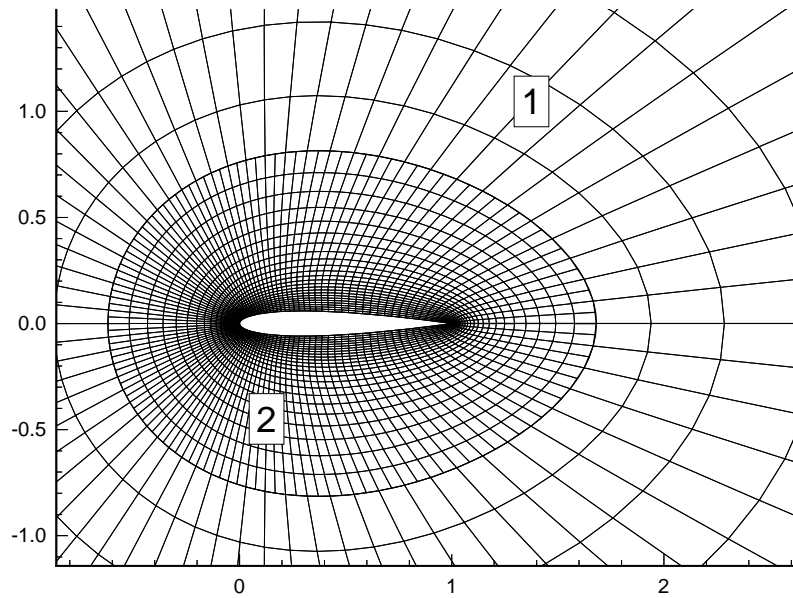


Figure 9-4. Two-zone overlapped grid system for NACA 0012 airfoil.

Besides the CFL3D and MaGGiE codes the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
0012x.inp	input for CFL3D
0012x.fmt	formatted grid in PLOT3D format
fmttobin_p3d.f	grid converter
mag1.h	parameters for MaGGiE makefile
maggie.inp	input for MaGGiE

The steps for running this case on the YMP are as follows:

### Step 1

Compile the grid converter code:

```
cft77 fmttobin_p3d.f
```

### Step 2

Link the grid converter object file:

```
sgldr -o fmttobin_p3d fmttobin_p3d.o
```

### Step 3

Run the grid converter program (the binary file 0012x.bin will be output):

```
fmttobin_p3d
```

### Step 4

Use the makefile to compile, link, and create the executable for the MaGGiE code (be sure mag1.h is in the current directory):

```
make -f makemaggie_cray
```

### Step 5

Run the MaGGiE code (the file ovrlp.bin will be output):

```
maggie < maggie.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the precfl3d code (be sure precfl.h is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 7

Run the precfl3d code (the cflx.h files will be output):

```
precfl3d < 0012x.inp
```

### Step 8

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

### Step 9

Run the CFL3D code:



```
cfl3d < 0012x.inp
```

The input file for this case is:

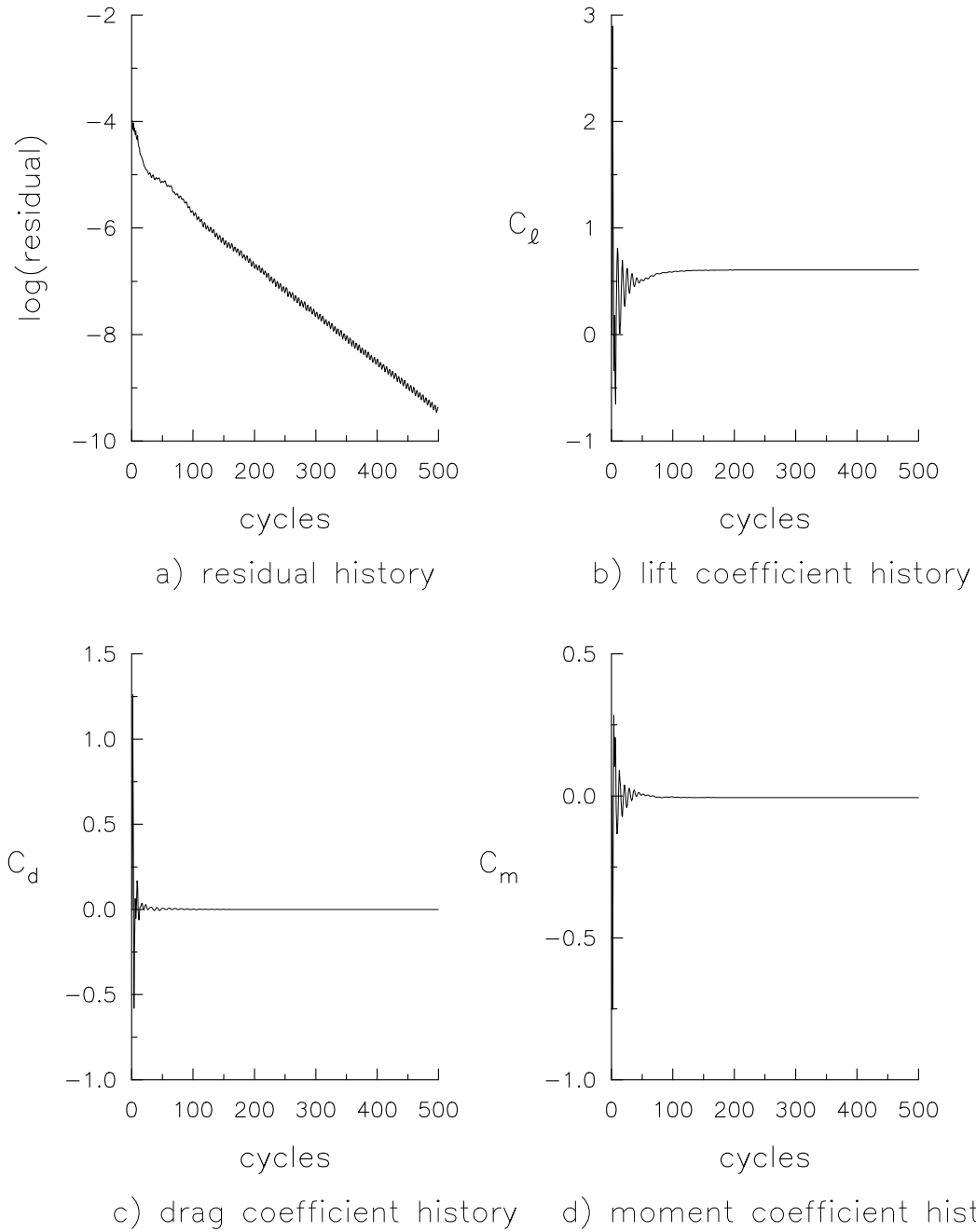
```
I/O FILES
0012x.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
2-block 0012 airfoil as simple chimera test
  XMACH ALPHA BETA REUE,MIL TINF,DR IALPH IHSTRY
    .200 5.000 0.0 0.0 520.0 1 0
  SREF CREF BREF XMC YMC ZMC
1.00000 1.00000 1.0000 0.25000 0.00 0.00
  DT IREST IFLAGTS FMAX IUNST CFLTAU
 -5.00 0 000 1.00 0 10.
 NGRID NPLOT3D NPRINT NWREST ICHK I2D NTSTEP ITA
   -2 2 0 100 0 1 1 1
  NCG IEM IADVANCE IFORCE IVISC(I) IVISC(J) IVISC(K)
    2 0 0 0 0 0 0 0
    2 0 0 1 0 0 0 0
  IDIM JDIM KDIM
    002 65 25
    002 129 25
  ILAMLO ILAMHI JLAMLO JLAMHI KLAMLO KLAMHI
    00 00 000 000 0 0000
    00 00 000 000 0 0000
  INEWG IGRIDC IS JS KS IE JE KE
    0 0 0 0 0 0 0 0
    0 0 0 0 0 0 0 0
  IDIAG(I) IDIAG(J) IDIAG(K) IFLIM(I) IFLIM(J) IFLIM(K)
    1 1 1 0 0 0
    1 1 1 0 0 0
  IFDS(I) IFDS(J) IFDS(K) RKAP0(I) RKAP0(J) RKAP0(K)
    1 1 1 .3333 .3333 .3333
    1 1 1 .3333 .3333 .3333
  GRID NBCIO NBCIDIM NBCJO NBCJDIM NBCKO NBCKDIM IOVRLP
    1 1 1 1 1 1 1 1
    2 1 1 1 1 1 1 1
IO:  GRID SEGMENT BCTYPE JSTA JEND KSTA KEND NDATA
    1 1 1002 0 0 0 0 0
    2 1 1002 0 0 0 0 0
IDIM: GRID SEGMENT BCTYPE JSTA JEND KSTA KEND NDATA
    1 1 1002 0 0 0 0 0
    2 1 1002 0 0 0 0 0
JO:  GRID SEGMENT BCTYPE ISTA IEND KSTA KEND NDATA
    1 1 0 0 0 0 0 0
    2 1 0 0 0 0 0 0
JDIM: GRID SEGMENT BCTYPE ISTA IEND KSTA KEND NDATA
    1 1 0 0 0 0 0 0
    2 1 0 0 0 0 0 0
KO:  GRID SEGMENT BCTYPE ISTA IEND JSTA JEND NDATA
    1 1 0 0 0 0 0 0
    2 1 1005 0 0 0 0 0
KDIM: GRID SEGMENT BCTYPE ISTA IEND JSTA JEND NDATA
    1 1 1003 0 0 0 0 0
    2 1 0 0 0 0 0 0
  MSEQ MGFLAG ICONSF MTT NGAM
    1 1 1 0 01
  ISSC EPSSC(1) EPSSC(2) EPSSC(3) ISSR EPSSR(1) EPSSR(2) EPSSR(3)
    0 .3 .3 .3 0 .3 .3 .3
```

```

      NCYC      MGLEVG      NEMGL      NITFO
      500        03        00        000
      MIT1      MIT2      MIT3      MIT4      MIT5
      01        01        01        01        01
1-1 BLOCKING DATA:
  NBLI
  2
NUMBER  GRID      :      ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  1      1      :      1      1      1      2      1      25      1      3
  2      2      :      1      1      1      2      1      25      1      3
NUMBER  GRID      :      ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  1      1      :      1      65      1      2      65      25      1      3
  2      2      :      1     129      1      2     129      25      1      3
PATCH SURFACE DATA:
  NINTER
  0
PLOT3D OUTPUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
  1      0      1      001    1      01     999    1      1     999    1
  2      0      1      001    1      01     999    1      1     999    1
MOVIE
  0
PRINT OUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
CONTROL SURFACE:
NCS
  0
  GRID ISTART  IEND  JSTART  JEND  KSTART  KEND  IWALL  INORM

```

The residual and coefficient histories for this case are plotted in Figure 9-5.



**Figure 9-5.** NACA 0012 with overlapped grids residual and coefficient histories;  
 $\alpha = 5^\circ$ ,  $M_\infty = 0.2$ .

### 9.1.3 NACA 0012 Airfoil with Patched Grids

This test case solves for the inviscid flow over the NACA 0012 airfoil at  $\alpha = 1.25^\circ$  with  $M_\infty = 0.8$ . The grid has a total of 4949 points in seven zones which communicate with one another utilizing the patching option. Therefore, the ronnie code is used in addition to CFL3D. An advantage of using patched grids is that finer grids can be placed in high gradient regions while relatively coarser grids can be placed elsewhere thus reducing the CPU time and memory needed. In this case, the finest grids are located in the regions where the upper and lower shocks are expected to occur in order to better resolve these flow phenomena. The memory requirement for this example is 1.9 million words. A typical timing is 87 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of June 1996). A close-up of the grid near the airfoil is shown in Figure 9-6. In the figure, the grids are labelled one through seven and this is the grid order in which the information is set up in the input file.

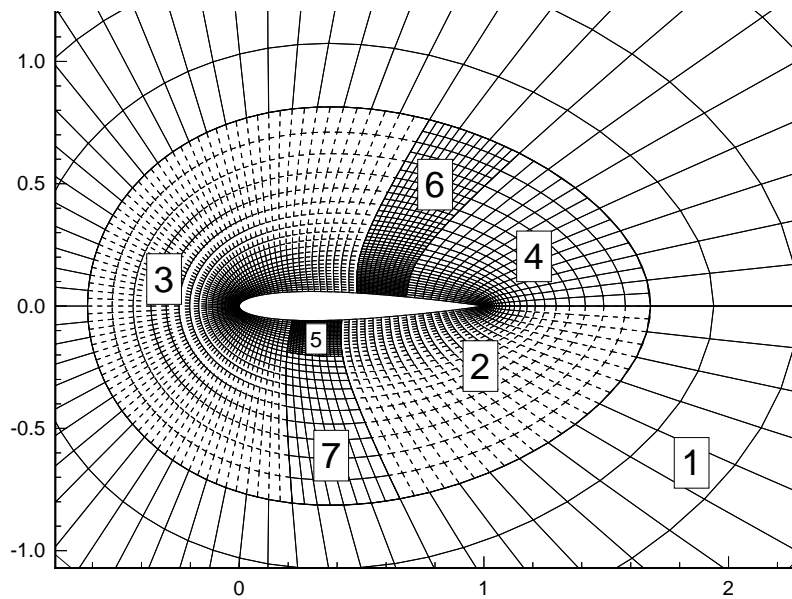


Figure 9-6. Seven-zone patched grid system for NACA 0012 airfoil.

Besides the CFL3D and ronnie codes the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
0012.inp	input for CFL3D
0012.fmt	formatted grid
fmttobin.f	grid converter

<u>File</u>	<u>Description</u>
ron1.h	parameters for ronnie makefile
ronnie.inp	input for ronnie

The steps for running this case on the YMP are as follows:

### Step 1

Compile the grid converter code:

```
cft77 fmttobin.f
```

### Step 2

Link the grid converter object file:

```
segldr -o fmttobin fmttobin.o
```

### Step 3

Run the grid converter program (the binary file 0012.bin will be output):

```
fmttobin
```

### Step 4

Use the makefile to compile, link, and create the executable for the ronnie code (be sure ron1.h is in the current directory):

```
make -f makeronnie_cray
```

### Step 5

Run the ronnie code (the file patch.bin\_0012 will be output):

```
ronnie < ronnie.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the precf13d code (be sure precf1.h is in the current directory):

```
make -f makeprecf13d_cray
```

### Step 7

Run the precf13d code (the cflx.h files will be output):

```
precf13d < 0012.inp
```

Step 8

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

Step 9

Run the CFL3D code:

```
cfl3d < 0012.inp
```

The input file for this case is:

```
I/O FILES
0012.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin_0012
restart.bin
input for 7 block patched 0012 grids - iopt = 1 in assemble.f
  XMACH      ALPHA      BETA  REUE,MIL    TINF,DR    IALPH      IHSTRY
    0.80      1.25      0.0    0.000    122.0      0          0
  SREF        CREF        BREF        XMC        YMC        ZMC
1.00000      1.00000      1.0000      0.25000      0.00      0.00
  DT          IREST      IFLAGTS      FMAX      IUNST      CFLTAU
 -5.00        0          000      1.00      0          10.0
  NGRID      NPLOT3D      NPRINT      NWREST      ICHK      I2D      NTSTEP      ITA
    7          7          0      100      0          1          1          1
  NCG          IEM      IADVANCE      IFORCE      IVISC(I)      IVISC(J)      IVISC(K)
    1          0          0          0          0          0          0
    1          0          0          1          0          0          0
    1          0          0          1          0          0          0
    1          0          0          1          0          0          0
    1          0          0          1          0          0          0
    1          0          0          1          0          0          0
    1          0          0          0          0          0          0
  IDIM      JDIM      KDIM
    2          65      13
    2          23      25
    2          79      25
    2          15      25
    2          17      25
    2          13      49
    2          9       13
  ILAMLO      ILAMHI      JLAMLO      JLAMHI      KLAMLO      KLAMHI
    00          00      000      000      0      0000
    00          00      000      000      0      0000
    00          00      000      000      0      0000
    00          00      000      000      0      0000
    00          00      000      000      0      0000
    00          00      000      000      0      0000
    00          00      000      000      0      0000
  INEWG      IGRIDC      IS      JS      KS      IE      JE      KE
    0          0          0          0          0          0          0          0
    0          0          0          0          0          0          0          0
    0          0          0          0          0          0          0          0
    0          0          0          0          0          0          0          0
    0          0          0          0          0          0          0          0
```

0	0	0	0	0	0	0	0	
0	0	0	0	0	0	0	0	
IDIAG(I)	IDIAG(J)	IDIAG(K)	IFLIM(I)	IFLIM(J)	IFLIM(K)			
0	0	0	3	3	3			
0	0	0	3	3	3			
0	0	0	3	3	3			
0	0	0	3	3	3			
0	0	0	3	3	3			
0	0	0	3	3	3			
0	0	0	3	3	3			
IFDS(I)	IFDS(J)	IFDS(K)	RKAP0(I)	RKAP0(J)	RKAP0(K)			
0	0	0	.3333	.3333	.3333			
0	0	0	.3333	.3333	.3333			
0	0	0	.3333	.3333	.3333			
0	0	0	.3333	.3333	.3333			
0	0	0	.3333	.3333	.3333			
0	0	0	.3333	.3333	.3333			
0	0	0	.3333	.3333	.3333			
GRID	NBCI0	NBCIDIM	NBCJ0	NBCJDIM	NBCK0	NBCKDIM	IOVRLP	
1	1	1	1	1	1	1	0	
2	1	1	1	1	1	1	0	
3	1	1	1	1	1	1	0	
4	1	1	1	1	1	1	0	
5	1	1	1	1	1	1	0	
6	1	1	1	1	1	1	0	
7	1	1	1	1	1	1	0	
I0:	GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA
1	1	1	1002	0	0	0	0	0
2	1	1	1002	0	0	0	0	0
3	1	1	1002	0	0	0	0	0
4	1	1	1002	0	0	0	0	0
5	1	1	1002	0	0	0	0	0
6	1	1	1002	0	0	0	0	0
7	1	1	1002	0	0	0	0	0
IDIM:	GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA
1	1	1	1002	0	0	0	0	0
2	1	1	1002	0	0	0	0	0
3	1	1	1002	0	0	0	0	0
4	1	1	1002	0	0	0	0	0
5	1	1	1002	0	0	0	0	0
6	1	1	1002	0	0	0	0	0
7	1	1	1002	0	0	0	0	0
J0:	GRID	SEGMENT	BCTYPE	ISTA	IEND	KSTA	KEND	NDATA
1	1	1	0	0	0	0	0	0
2	1	1	0	0	0	0	0	0
3	1	1	0	0	0	0	0	0
4	1	1	0	0	0	0	0	0
5	1	1	0	0	0	0	0	0
6	1	1	0	0	0	0	0	0
7	1	1	0	0	0	0	0	0
JDIM:	GRID	SEGMENT	BCTYPE	ISTA	IEND	KSTA	KEND	NDATA
1	1	1	0	0	0	0	0	0
2	1	1	0	0	0	0	0	0
3	1	1	0	0	0	0	0	0
4	1	1	0	0	0	0	0	0
5	1	1	0	0	0	0	0	0
6	1	1	0	0	0	0	0	0
7	1	1	0	0	0	0	0	0
K0:	GRID	SEGMENT	BCTYPE	ISTA	IEND	JSTA	JEND	NDATA
1	1	1	0	0	0	0	0	0
2	1	1	1005	0	0	0	0	0
3	1	1	1005	0	0	0	0	0
4	1	1	1005	0	0	0	0	0
5	1	1	1005	0	0	0	0	0
6	1	1	1005	0	0	0	0	0
7	1	1	0	0	0	0	0	0
KDIM:	GRID	SEGMENT	BCTYPE	ISTA	IEND	JSTA	JEND	NDATA
1	1	1	1003	0	0	0	0	0
2	1	1	0	0	0	0	0	0
3	1	1	0	0	0	0	0	0
4	1	1	0	0	0	0	0	0
5	1	1	0	0	0	0	0	0

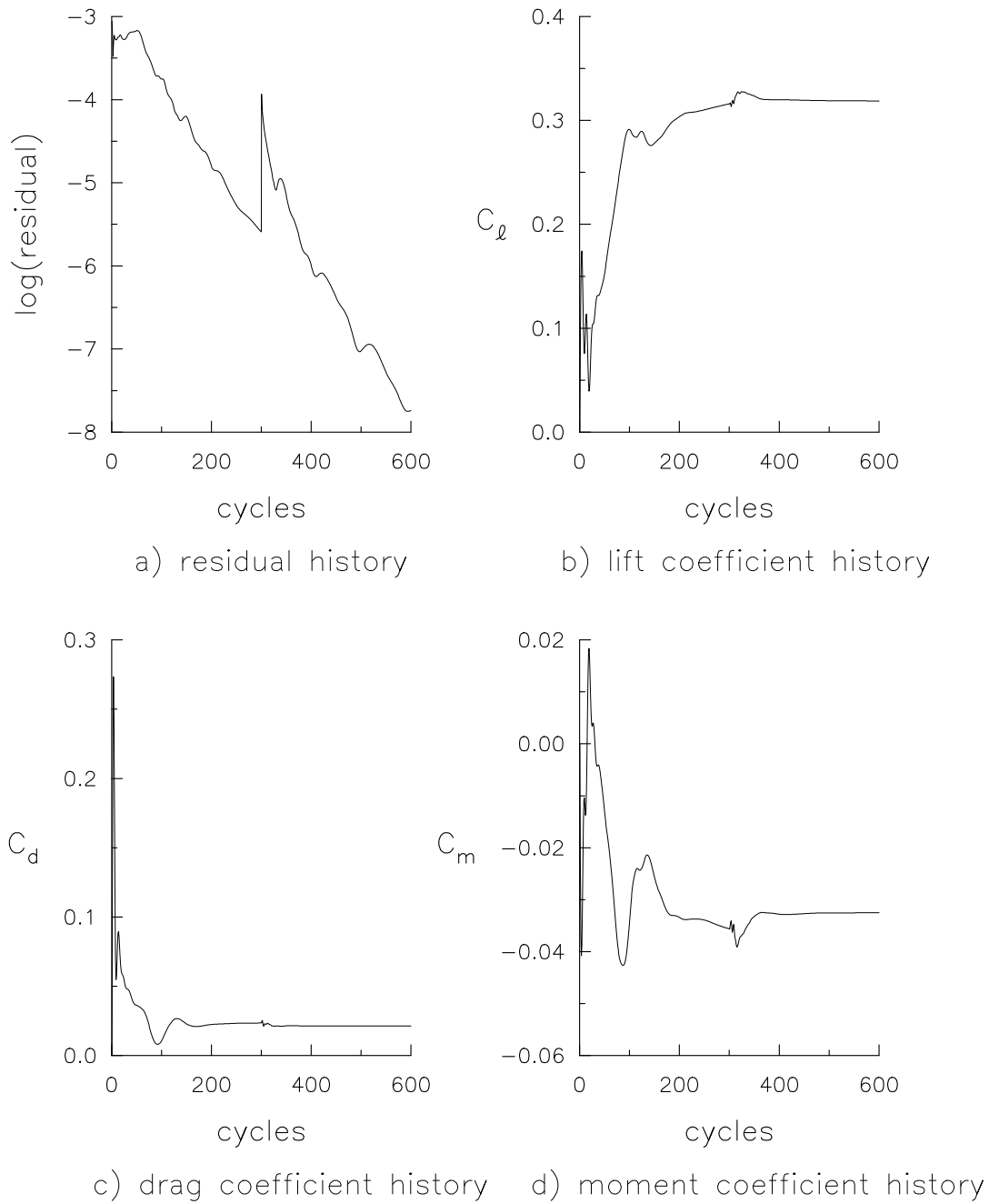
```

        6          1          0          0          0          0          0          0
        7          1          0          0          0          0          0          0
MSEQ      MGFLAG      ICONSF      MTT      NGAM
        2          1          1          0          02
ISSC      EPSSC(1)    EPSSC(2)    EPSSC(3)    ISSR      EPSSR(1)    EPSSR(2)    EPSSR(3)
        0          .3          .3          .3          0          .3          .3          .3
NCYC      MGLEVG      NEMGL      NITFO
0300          01          00          000
0300          02          00          000
MIT1      MIT2      MIT3      MIT4      MIT5
        01          01          01          01          01
        01          01          01          01          01
1-1 BLOCKING DATA:
NBLI
1
NUMBER    GRID      :      ISTA      JSTA      KSTA      IEND      JEND      KEND      ISVA1      ISVA2
1          1          :          1          1          1          2          1          13          1          3
NUMBER    GRID      :      ISTA      JSTA      KSTA      IEND      JEND      KEND      ISVA1      ISVA2
1          1          :          1          65          1          2          65          13          1          3
PATCH SURFACE DATA:
NINTER
-1
PLOT3D OUTPUT:
GRID IPTYPE ISTART      IEND      IINC JSTART      JEND      JINC KSTART      KEND      KINC
1          0          1      001          1          01      999          1          1      999          1
2          0          1      001          1          01      999          1          1      999          1
3          0          1      001          1          01      999          1          1      999          1
4          0          1      001          1          01      999          1          1      999          1
5          0          1      001          1          01      999          1          1      999          1
6          0          1      001          1          01      999          1          1      999          1
7          0          1      001          1          01      999          1          1      999          1
MOVIE
0
PRINT OUT:
GRID IPTYPE ISTART      IEND      IINC JSTART      JEND      JINC KSTART      KEND      KINC
CONTROL SURFACE:
NCS
0
GRID ISTART      IEND      JSTART      JEND      KSTART      KEND      IWALL      INORM

```

The residual and force coefficient history plots for this case is shown in Figure 9-7. The sharp spike in the residual history plot depicts the iteration at which the grid levels changed for mesh sequencing. These convergence histories can be found in `cfl3d.res`.

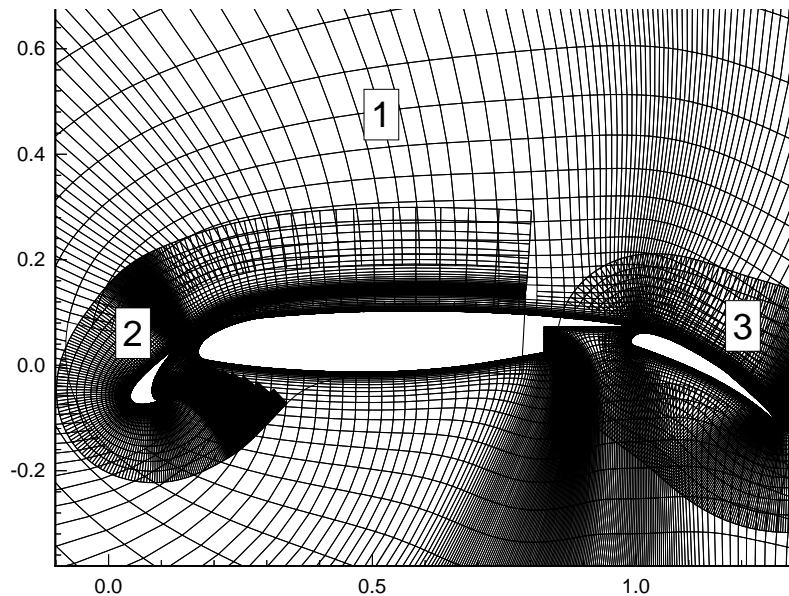




**Figure 9-7.** Seven-zone NACA 0012 case residual and coefficient histories;  
 $\alpha = 1.25^\circ$ ,  $M_\infty = 0.8$ .

### 9.1.4 Multielement Airfoil with Overlapped Grids

This test case solves for the viscous, turbulent flow over a three-element airfoil with  $M_\infty = 0.2$ ,  $\alpha = 8.109^\circ$ , and a Reynolds number of 9 million. The Spalart-Allmaras turbulence model is used. The grid, with a total of 59051 points, consists of three zones, one for each element. The grid zones communicate with one another utilizing the grid overlapping option. Therefore, the MaGGiE code is used in addition to CFL3D. The memory requirement for this case is 9.5 million words. A typical timing is 2849 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of June 1996). A close-up of the grid near the airfoil is shown in Figure 9-8. In the figure, the grids are labelled one through three and this is the grid order in which the information is set up in the input file.



**Figure 9-8.** Three-zone overlapped grid system for a three-element airfoil.

Besides the CFL3D and MaGGiE codes the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
multi.inp	input for CFL3D
grid.fmt	formatted grid
fmttobin.f	grid converter
mag1.h	parameters for MaGGiE makefile
mag.inp_multi	input for MaGGiE

The steps for running this case on the YMP are as follows:

Step 1

Compile the grid converter code:

```
cft77 fmttobin.f
```

Step 2

Link the grid converter object file:

```
segldr -o fmttobin fmttobin.o
```

Step 3

Run the grid converter program (the binary file `multi.bin` will be output):

```
fmttobin
```

Step 4

Use the makefile to compile, link, and create the executable for the MaGGiE code (be sure `mag1.h` is in the current directory):

```
make -f makemaggie_cray
```

Step 5

Run the MaGGiE code (the file `ovrlp.bin` will be output):

```
maggie < mag.inp_multi
```

Step 6

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

Step 7

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < multi.inp
```

Step 8

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

Step 9

Run the CFL3D code:

```
cfl3d < multi.inp
```

The input file for this case is:

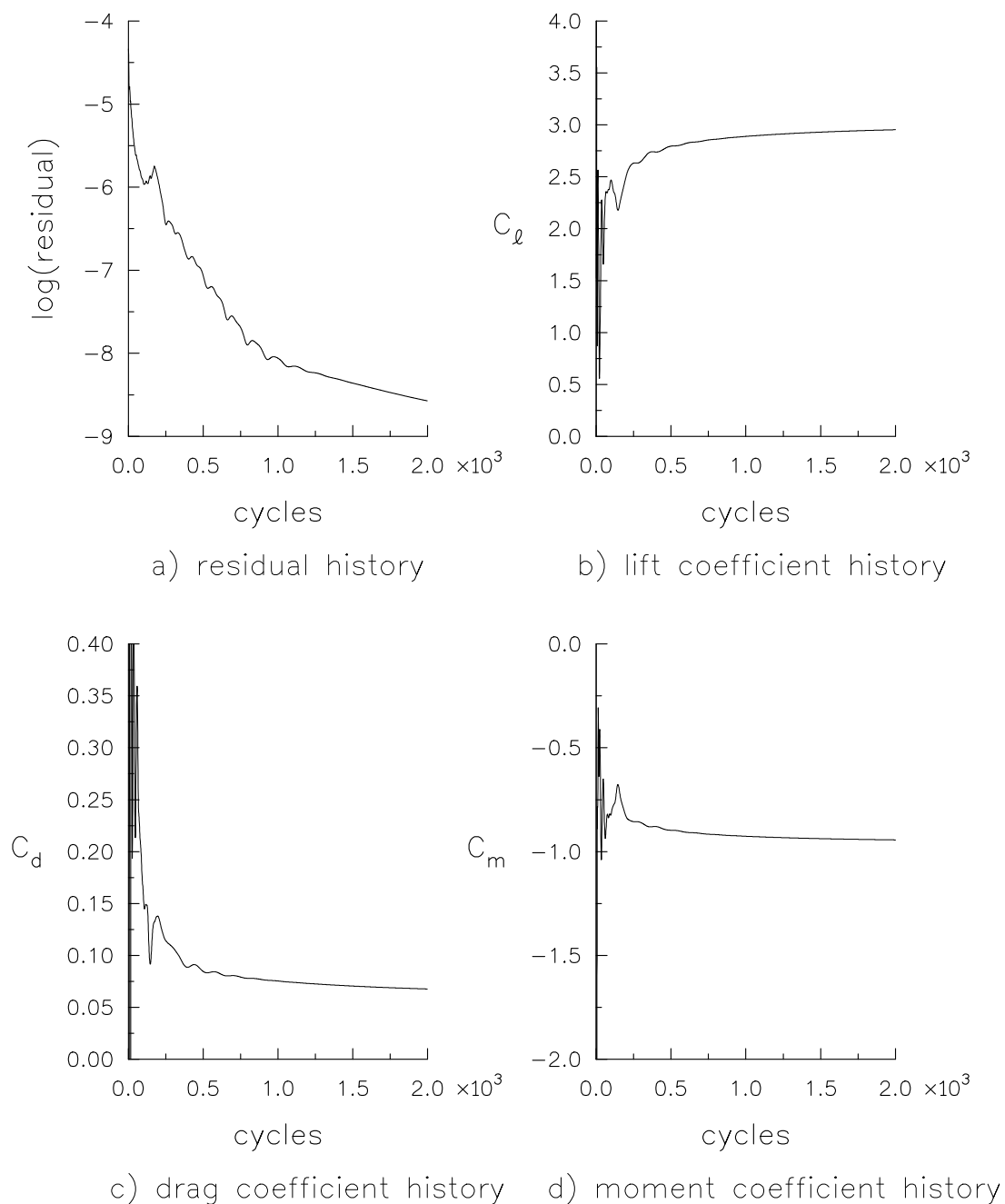
```
I/O FILES
multi.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
3 element airfoil - chimera-type grids - Spalart-Allmaras turb model
  XMACH      ALPHA      BETA  REUE,MIL  TINF,DR    IALPH      IHSTRY
    .2000      8.109      0.0      9.0      520.0      0          0
  SREF        CREF        BREF        XMC        YMC        ZMC
1.00000      1.00000      1.0000      0.25000      0.00      0.00
  DT          IREST      IFLAGTS      FMAX      IUNST      CFLTAU
 -5.00      0          000      1.00      0          10.0
 NGRID        NPLOT3D      NPRINT      NWREST      ICHK      I2D      NTSTEP      ITA
    3          3          0          100      0          1          1          1
  NCG          IEM      IADVANCE      IFORCE      IVISC(I)      IVISC(J)      IVISC(K)
    2          0          0          001      0          0          5
    2          0          0          001      0          0          5
    2          0          0          001      0          0          5
  IDIM          JDIM          KDIM
    002          361          65
    002          369          57
    002          297          49
  ILAMLO      ILAMHI      JLAMLO      JLAMHI      KLAMLO      KLAMHI
    00          00          000      000      0          0000
    00          00          000      000      0          0000
    00          00          000      000      0          0000
  INEWG      IGRIDC      IS          JS          KS          IE          JE          KE
    0          0          0          0          0          0          0          0
    0          0          0          0          0          0          0          0
    0          0          0          0          0          0          0          0
  IDIAG(I)      IDIAG(J)      IDIAG(K)      IFLIM(I)      IFLIM(J)      IFLIM(K)
    1          1          1          3          3          3
    1          1          1          3          3          3
    1          1          1          3          3          3
  IFDS(I)      IFDS(J)      IFDS(K)      RKAP0(I)      RKAP0(J)      RKAP0(K)
    1          1          1      .3333      .3333      .3333
    1          1          1      .3333      .3333      .3333
    1          1          1      .3333      .3333      .3333
  GRID          NBCI0      NBCIDIM      NBCJ0      NBCJDIM      NBCK0      NBCKDIM      IOVRLP
    1          1          1          1          1          3          1          1
    2          1          1          1          1          3          1          1
    3          1          1          1          1          3          1          1
I0:  GRID      SEGMENT      BCTYPE      JSTA      JEND      KSTA      KEND      NDATA
    1          1          1002      0          0          0          0          0
    2          1          1002      0          0          0          0          0
    3          1          1002      0          0          0          0          0
IDIM: GRID      SEGMENT      BCTYPE      JSTA      JEND      KSTA      KEND      NDATA
    1          1          1002      0          0          0          0          0
    2          1          1002      0          0          0          0          0
    3          1          1002      0          0          0          0          0
J0:  GRID      SEGMENT      BCTYPE      ISTA      IEND      KSTA      KEND      NDATA
    1          1          1003      0          0          0          0          0
```

```

      2      1      0      0      0      0      0      0
      3      1      0      0      0      0      0      0
JDIM: GRID SEGMENT BCTYPE ISTA IEND KSTA KEND NDATA
      1      1      1003      0      0      0      0      0
      2      1      0      0      0      0      0      0
      3      1      0      0      0      0      0      0
K0:  GRID SEGMENT BCTYPE ISTA IEND JSTA JEND NDATA
      1      1      0      0      0      1      73      0
      1      2      2004      0      0      73      289      2
      TWTYPE CQ
      0.      0.
      1      3      0      0      0      289      361      0
      2      1      0      0      0      1      65      0
      2      2      2004      0      0      65      305      2
      TWTYPE CQ
      0.      0.
      2      3      0      0      0      305      369      0
      3      1      0      0      0      1      49      0
      3      2      2004      0      0      49      249      2
      TWTYPE CQ
      0.      0.
      3      3      0      0      0      249      297      0
KDIM: GRID SEGMENT BCTYPE ISTA IEND JSTA JEND NDATA
      1      1      1003      0      0      0      0      0
      2      1      0      0      0      0      0      0
      3      1      0      0      0      0      0      0
MSEQ MGFLAG ICONSF MTT NGAM
      1      1      1      0      01
ISSC EPSSC(1) EPSSC(2) EPSSC(3) ISSR EPSSR(1) EPSSR(2) EPSSR(3)
      0      .3      .3      .3      0      .3      .3      .3
NCYC MGLEVG NEMGL NITFO
      2000      03      00      000
MIT1 MIT2 MIT3 MIT4 MIT5
      01      01      01      01      01
1-1 BLOCKING DATA:
NBLI
      3
NUMBER GRID : ISTA JSTA KSTA IEND JEND KEND ISVA1 ISVA2
      1      1      1      1      1      2      73      1      1      2
      2      2      1      1      1      2      65      1      1      2
      3      3      1      1      1      2      49      1      1      2
NUMBER GRID : ISTA JSTA KSTA IEND JEND KEND ISVA1 ISVA2
      1      1      1      361      1      2      289      1      1      2
      2      2      1      369      1      2      305      1      1      2
      3      3      1      297      1      2      249      1      1      2
PATCH SURFACE DATA:
NINTER
      0
PLOT3D OUTPUT:
GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
      1      1      1      001      1      01      999      1      1      999      1
      2      1      1      001      1      01      999      1      1      999      1
      3      1      1      001      1      01      999      1      1      999      1
MOVIE
      0
PRINT OUT:
GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
CONTROL SURFACE:
NCS
      0
GRID ISTART IEND JSTART JEND KSTART KEND IWALL INORM

```

After running this test case, the residual and force coefficient convergence histories should look like those in Figure 9-9. These convergence histories can be found in file `cfl3d.res`. Note the unusually high number of multigrid cycles required to converge this case. While quite large, this is the behavior typically seen (with CFL3D) for multielement airfoil cases, even when one-to-one blocking is employed.



**Figure 9-9.** Convergence histories for three-element airfoil case;

$$\alpha = 8.109^\circ, Re_{\tilde{L}_R} = 9 \times 10^6.$$

### 9.1.5 Flat Plate

The viscous, turbulent flow with a Reynolds number of 6 million over a flat plate is solved in this test case. The grid consists of a single grid zone with 6305 points. Menter's  $k - \omega$  SST turbulence model is utilized in this example. The memory requirement is 2.3 million words. A typical timing for this case is 157 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of June 1996). The entire flat plate grid is illustrated in Figure 9-10.

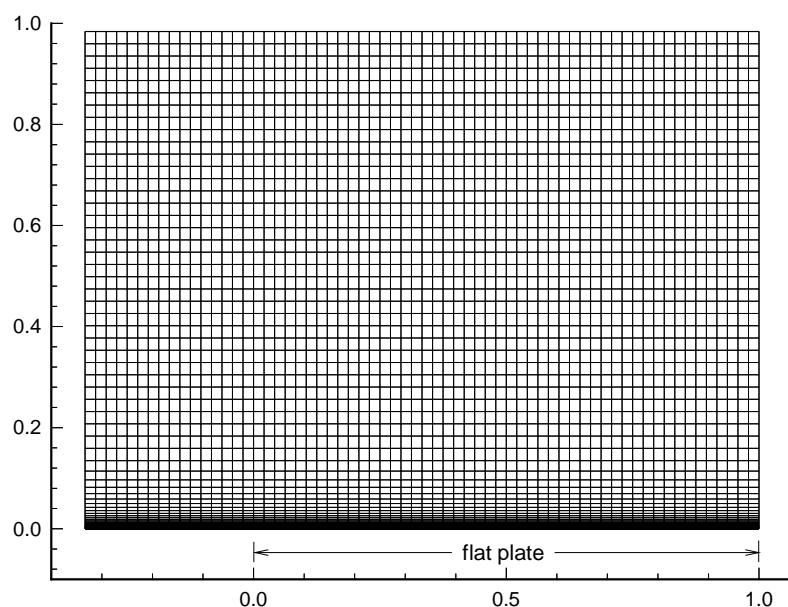


Figure 9-10. Single zone flat plate grid.

Besides the CFL3D code the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
grdflat5.inp	input for CFL3D
grdflat5.grd	formatted grid
grid2dto3d.f	grid converter

The steps for running this case on the YMP are as follows:

#### Step 1

Compile the grid converter code:

```
cft77 grid2dto3d.f
```

### Step 2

Link the grid converter object file:

```
segldr -o grid2dto3d grid2dto3d.o
```

### Step 3

Run the grid converter program (the binary file `grdflat5.bin` will be output):

```
grid2dto3d
```

In answer to the questions, type:

```
grdflat5.grd
grdflat5.bin
2
0
```

### Step 4

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 5

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < grdflat5.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the `CFL3D` code:

```
make -f makecfl3d_cray
```

### Step 7

Run the `CFL3D` code:

```
cfl3d < grdflat5.inp
```

The input file for this case is:

```
I/O FILES
grdflat5.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
```



```

cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
turbulent flat plate (plate from j=17-65, prior to 17 is symmetry)
  XMACH      ALPHA      BETA  REUE,MIL    TINF,DR    IALPH      IHSTRY
  0.2000     00.000     0.0   06.000     460.0       0          0
  SREF       CREF       BREF       XMC       YMC       ZMC
  1.00000    1.00000    1.0000  0.00000    0.00       0.00
  DT         IREST      IFLAGTS  FMAX      IUNST      CFLTAU
  -5.000     1         000     05.0000    0          10.
  NGRID      NPLOT3D    NPRINT    NWREST     ICHK       I2D        NTSTEP      ITA
  1           1         2         1200       0          1          1          1
  NCG        IEM        IADVANCE  IFORCE     IVISC(I)   IVISC(J)   IVISC(K)
  2           0         0         001         0          0          12
  IDIM       JDIM       KDIM
  02         65        97
  ILAMLO     ILAMHI     JLAMLO     JLAMHI     KLAMLO     KLAMHI
  1           2         1         17          1         97
  INEWG      IGRIDC     IS         JS         KS         IE         JE         KE
  0           0         0         0         0         0         0         0
  IDIAG(I)   IDIAG(J)   IDIAG(K)   IFLIM(I)   IFLIM(J)   IFLIM(K)
  1           1         1         0         0         0
  IFDS(I)    IFDS(J)    IFDS(K)    RKAP0(I)   RKAP0(J)   RKAP0(K)
  1           1         1         0.3333     0.3333     0.3333
  GRID       NBCI0      NBCIDIM    NBCJ0      NBCJDIM    NBCK0      NBCKDIM      IOVRLP
  1           1         1         1         1         2         1         0
  I0:  GRID  SEGMENT    BCTYPE    JSTA      JEND      KSTA      KEND      NDATA
  1           1         1001       0         0         0         0         0
  IDIM: GRID  SEGMENT    BCTYPE    JSTA      JEND      KSTA      KEND      NDATA
  1           1         1002       0         0         0         0         0
  J0:  GRID  SEGMENT    BCTYPE    ISTA      IEND      KSTA      KEND      NDATA
  1           1         1008       0         0         0         0         0
  JDIM: GRID  SEGMENT    BCTYPE    ISTA      IEND      KSTA      KEND      NDATA
  1           1         1002       0         0         0         0         0
  K0:  GRID  SEGMENT    BCTYPE    ISTA      IEND      JSTA      JEND      NDATA
  1           1         1001       0         0         1         17         0
  1           2         2004       0         0         17         65         2
  TWTYPE     CQ
  0.          0.
  KDIM: GRID  SEGMENT    BCTYPE    ISTA      IEND      JSTA      JEND      NDATA
  1           1         1003       0         0         0         0         0
  MSEQ       MGFLAG     ICONSF     MTT       NGAM
  1           1         0         0         02
  ISSC EPSSSC(1) EPSSSC(2) EPSSSC(3) ISSR EPSSSR(1) EPSSSR(2) EPSSSR(3)
  0         0.3       0.3       0.3       0         0.3       0.3       0.3
  NCYC       MGLEVG     NEMGL     NITFO
  0500       03         00         000
  MIT1       MIT2       MIT3       MIT4       MIT5       MIT6       MIT7       MIT8
  01         01         01         01         01         1         1         1
  1-1 BLOCKING DATA:
  NBLI
  0
  NUMBER  GRID  :  ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  NUMBER  GRID  :  ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  PATCH SURFACE DATA:
  NINTER
  0
  PLOT3D OUTPUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
  1         0         0         0         0         0         0         0         0         0
  IMOVIE
  0
  PRINT OUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
  1         0         0         0         0         0         0         1         1         1
  1         0         0         0         0         49        49         1         0         0
  CONTROL SURFACE:
  NCS
  0
  GRID ISTART  IEND  JSTART  JEND  KSTART  KEND  IWALL  INORM

```

After this test case is run, the residual history, found in file `cfl3d.res`, should look like that plotted in Figure 9-11. In Figure 9-12, values of  $u^+$  versus  $y^+$  are plotted at two cross sections of the flat plate and compared with theoretical values. The  $u^+$  and  $y^+$  values were extracted from the PLOT3D grid and q files using a postprocessor currently not available for general use.

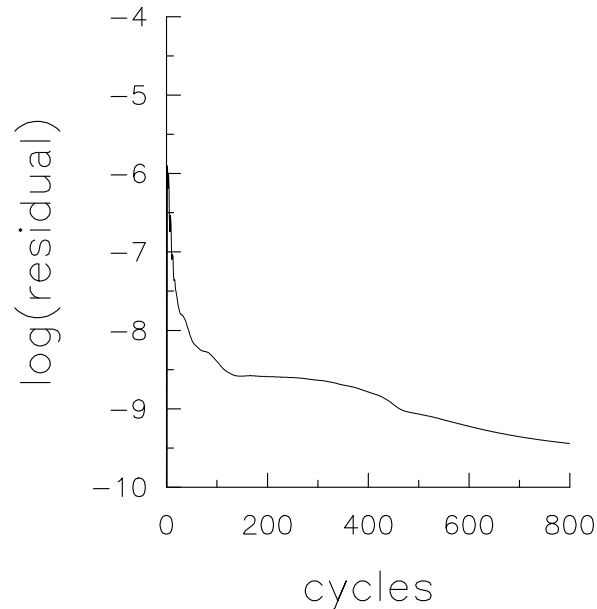


Figure 9-11. Residual history for single grid flat plate case.

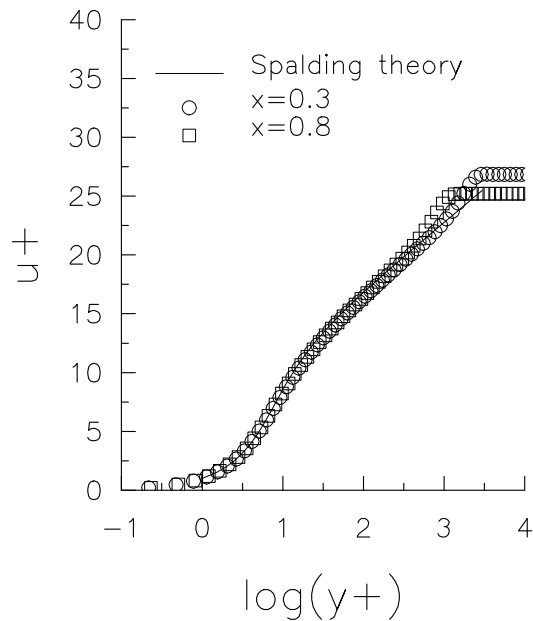


Figure 9-12. Flat plate calculation compared with experiment;  $Re_{\tilde{L}_R} = 6 \times 10^6$ .

This flat plate case has been studied with *all* the turbulence models currently available in CFL3D and a summary of the timings is tabulated below:

Turbulence Model	<b>ivisc</b>	Approximate CPU seconds per cycle	Percent increase over Baldwin-Lomax per cycle
Baldwin-Lomax	2	0.153	0
Baldwin-Lomax with Degani-Schiff	3	0.153	0
Baldwin-Barth	4	0.177	16
Spalart-Allmaras	5	0.182	19
Wilcox $k - \omega$	6	0.186	22
SST $k - \omega$	7	0.196	28
$k - \omega$ EASM Gatski-Speziale (Linear)	8	0.218	43
$k - \varepsilon$ EASM Girimaji (Linear)	9	0.236	54
$k - \varepsilon$ (Abid version)	10	0.197	29
$k - \varepsilon$ EASM Gatski-Speziale (Nonlinear)	11	0.236	54
$k - \omega$ EASM Gatski-Speziale (Nonlinear)	12	0.236	54
$k - \varepsilon$ EASM Girimaji (Nonlinear)	13	0.254	66

This case requires between 800 to 1800 cycles to converge, depending on the turbulence model employed and the convergence criterion chosen. Generally,  $k - \varepsilon$  models take longer than  $k - \omega$  models and two-equation models tend to take longer than one-equation models. (See Appendix H.) Memory requirements also depend on which turbulence model is being used, varying from 2.2 to 2.4 million words.

### 9.1.6 Vibrating Flat Plates

This test case solves for the unsteady, time-accurate inviscid flow through an “infinite” row of vibrating flat plates. The plates, located from  $x = 0.0$  to  $x = 1.0$ , are vibrating up and down with a sinusoidal motion. The maximum displacement is  $h = 0.001$  and the nominal distance between the plates is 1.0. The reduced frequency, defined by  $k_r = \tilde{\omega} \tilde{c} / 2|\tilde{\mathbf{V}}|_\infty$ , where  $\omega$  is in radians/second, is 4.0. (Note that the input to CFL3D defines  $\tilde{\omega}$  in cycles/second; therefore, **rfreq** = 0.63662.) The plate vibration generates acoustic waves which propagate both downstream and upstream. The solution invokes periodicity at both the upper and lower boundaries (except between  $x = 0.0$  and  $x = 1.0$ ), thus mimicking an infinite row of flat plates.

This example also employs a second block downstream of  $x = 2.0$ . This block is a “sliding block” that is used to test the effect of a sliding block interface (such as might be used in a rotor-stator computation) on the transmission of acoustic waves. Currently, it is set to translate “up” with speed  $w/a_\infty = 1/\pi = 0.31831$ .

This test problem is set up to run for 961 time steps, using three multigrid sub-iterations per time step. The time step is  $\pi/320 = 0.0098175$ . At this time step, 160 time steps yield one complete cycle of plate oscillation, so 961 steps yield six complete cycles of plate oscillations. The code, taking advantage of the periodicity of the solution, “resets” the sliding zone (zone 2) whenever its motion exceeds **dzmax** = 1.0. At the end of time step 961, zone 2 is again physically aligned with zone 1. If the number of steps taken is such that **ntstep**-1 is not evenly divisible by 160, then zone 2 will appear displaced some distance “up” from zone 1 when looking at the solution. While this is not a problem since the solution is periodic, it is easier to visualize the whole flow field when the two zones are aligned.

The grid consists of two grid zones with a total of 5502 points. The memory requirement for this case is 2.1 million words. A typical timing for six cycles of plate oscillation is 422 CPU seconds on a CRAY YMP (NASA LaRC’s Sabre as of July 1996).

Besides the CFL3D code the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
vibrate.inp	input for CFL3D
cartesian.f	grid generator

The steps for running this case on the YMP are as follows:

### Step 1

Compile the grid generator code:

```
cft77 cartesian.f
```

### Step 2

Link the grid converter object file:

```
segldr -o cartesian cartesian.o
```

### Step 3

Run the grid generator program (the binary file `grid.bin` will be output):

```
cartesian
```

In answer to the questions, type

```

2
-6,2,0,1
.05,.05
0
2,7,0,1
.05,.05
0

```

#### Step 4

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

#### Step 5

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < vibrate.inp
```

#### Step 6

Use the makefile to compile, link, and create the executable for the `CFL3D` code:

```
make -f makecfl3d_cray
```

#### Step 7

Run the `CFL3D` code:

```
cfl3d < vibrate.inp
```

The input file for this case is:

```

I/O FILES
grid.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
Flat plate vibrating cascade with sliding interface downstream
  XMACH      ALPHA      BETA  REUE,MIL    TINF,DR    IALPH      IHSTRY
    .500      0.00      0.0    1.07      520.0      0          0
  SREF      CREF      BREF      XMC      YMC      ZMC
1.00000    1.00000    1.0000    0.25000    0.00      0.00
    DT      IREST    IFLAGTS      FMAX    IUNST    CFLTAU
.0098175      0      000      1.00      1      5.

```

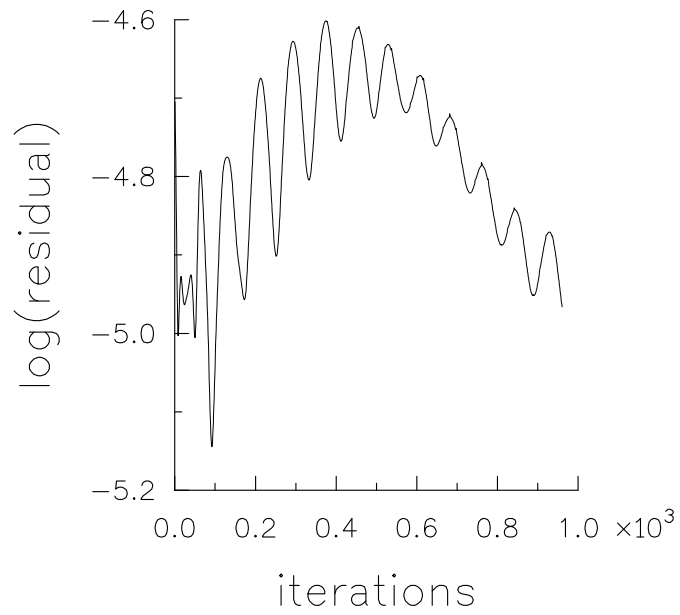
NGRID	NPLOT3D	NPRINT	NWREST	ICLK	I2D	NTSTEP	ITA			
2	2	2	2000	0	1	961	-2			
NCG	IEM	IADVANCE	IFORCE	IVISC(I)	IVISC(J)	IVISC(K)				
2	0	0	001	0	0	0				
2	0	0	001	0	0	0				
IDIM	JDIM	KDIM								
2	161	21								
2	101	21								
ILAMLO	ILAMHI	JLAMLO	JLAMHI	KLAMLO	KLAMHI					
00	00	000	000	0	0000					
00	00	000	000	0	0000					
INEWG	IGRIDC	IS	JS	KS	IE	JE	KE			
0	0	0	0	0	0	0	0			
0	0	0	0	0	0	0	0			
IDIAG(I)	IDIAG(J)	IDIAG(K)	IFLIM(I)	IFLIM(J)	IFLIM(K)					
1	1	1	0	0	0					
1	1	1	0	0	0					
IFDS(I)	IFDS(J)	IFDS(K)	RKAP0(I)	RKAP0(J)	RKAP0(K)					
1	1	1	.3333	.3333	.3333					
1	1	1	.3333	.3333	.3333					
GRID	NBCIO	NBCIDIM	NBCJO	NBCJDIM	NBCKO	NBCKDIM	IOVRLP			
1	1	1	1	1	3	3	0			
2	1	1	1	1	1	1	0			
I0: GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA			
1	1	1001	0	0	0	0	0			
2	1	1001	0	0	0	0	0			
IDIM: GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA			
1	1	1002	0	0	0	0	0			
2	1	1002	0	0	0	0	0			
J0: GRID	SEGMENT	BCTYPE	ISTA	IEND	KSTA	KEND	NDATA			
1	1	1003	0	0	0	0	0			
2	1	0	0	0	0	0	0			
JDIM: GRID	SEGMENT	BCTYPE	ISTA	IEND	KSTA	KEND	NDATA			
1	1	0	0	0	0	0	0			
2	1	1003	0	0	0	0	0			
K0: GRID	SEGMENT	BCTYPE	ISTA	IEND	JSTA	JEND	NDATA			
1	1	2005	1	2	1	121	4			
	NBLP	DTHTX	DTHTY	DTHTZ						
	1	0.0	0.	0.						
1	2	1005	1	2	121	141	0			
1	3	2005	1	2	141	161	4			
	NBLP	DTHTX	DTHTY	DTHTZ						
	1	0.0	0.	0.						
2	1	2005	0	0	0	0	4			
	NBLP	DTHTX	DTHTY	DTHTZ						
	2	0.0	0.	0.						
KDIM: GRID	SEGMENT	BCTYPE	ISTA	IEND	JSTA	JEND	NDATA			
1	1	2005	1	2	1	121	4			
	NBLP	DTHTX	DTHTY	DTHTZ						
	1	0.0	0.	0.						
1	2	1005	1	2	121	141	0			
1	3	2005	1	2	141	161	4			
	NBLP	DTHTX	DTHTY	DTHTZ						
	1	0.0	0.	0.						
2	1	2005	0	0	0	0	4			
	NBLP	DTHTX	DTHTY	DTHTZ						
	2	0.0	0.	0.						
MSEQ	MGFLAG	ICONSF	MTT	NGAM						
1	1	1	0	02						
ISSC	EPSSC(1)	EPSSC(2)	EPSSC(3)	ISSR	EPSSR(1)	EPSSR(2)	EPSSR(3)			
0	.3	.3	.3	0	.3	.3	.3			
NCYC	MGLEVG	NEMGL	NITFO							
3	03	00	000							
MIT1	MIT2	MIT3	MIT4	MIT5						
01	01	01	01	01						
1-1 BLOCKING DATA:										
NBLI										
0										
NUMBER	GRID	:	ISTA	JSTA	KSTA	IEND	JEND	KEND	ISVA1	ISVA2
NUMBER	GRID	:	ISTA	JSTA	KSTA	IEND	JEND	KEND	ISVA1	ISVA2
PATCH SURFACE DATA:										
NINTER										

```

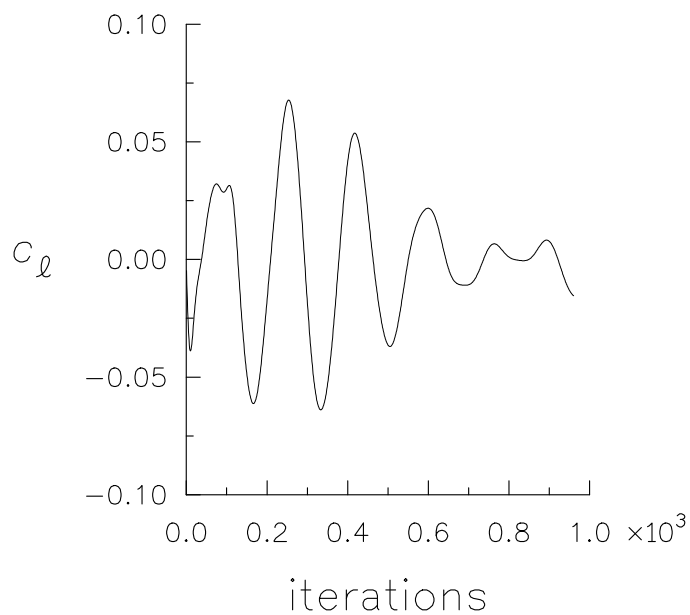
0
PLOT3D OUTPUT:
  GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
    1      0      1   999    1    01   999    1    1   999    1
    2      0      1   999    1    01   999    1    1   999    1
MOVIE
0
PRINT OUT:
  GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
    1      1      1     2     1   999   999    1    1   999    1
    2      1      1     2     1     1     1    1    1   999    1
CONTROL SURFACE:
NCS
0
  GRID ISTART IEND JSTART JEND KSTART KEND IWALL INORM
MOVING GRID DATA - TRANSLATION
NTRANS
2
  LREF
0.5
  GRID ITRANS RFREQ XMAG YMAG ZMAG
    1      2 .31831  0.  0.  0.00100
    2      1  0.  0.  0.  0.31831
  GRID DXMAX DYMAX DZMAX
    1      0.  0.  0.000
    2      0.  0.  1.000
MOVING GRID DATA - ROTATION
NROTAT
0
  LREF
  GRID IROTAT RFREQ THXMAG THYMAG THZMAG XORIG YORIG ZORIG
  GRID THXMAX THYMAX THZMAX
DYNAMIC PATCH INPUT DATA
NINTER
2
INT IFIT LIMIT ITMAX MCXIE MCETA C-0 IORPH ITOSS
  1  -1     5    20     0     0     0     0     1
  2  -1     5    20     0     0     0     0     1
INT TO XIE1 XIE2 ETA1 ETA2 NFB
  1  122    0     0     0     0     2
    FROM XIE1 XIE2 ETA1 ETA2 FACTJ FACTK
      221    0     0     0     0     0.  0.
        DX DY DZ DTHETX DTHETY DTHETZ
        0.  0.  0.  0.  0.  0.
    FROM XIE1 XIE2 ETA1 ETA2 FACTJ FACTK
      221    0     0     0     0     0.  0.
        DX DY DZ DTHETX DTHETY DTHETZ
        0.  0.  0.  0.  0.  0.
INT TO XIE1 XIE2 ETA1 ETA2 NFB
  2  221    0     0     0     0     2
    FROM XIE1 XIE2 ETA1 ETA2 FACTJ FACTK
      122    0     0     0     0     0.  0.
        DX DY DZ DTHETX DTHETY DTHETZ
        0.  0.  0.  0.  0.  0.
    FROM XIE1 XIE2 ETA1 ETA2 FACTJ FACTK
      122    0     0     0     0     0.  0.
        DX DY DZ DTHETX DTHETY DTHETZ
        0.  0.  +1.  0.  0.  0.

```

The resulting residual and lift coefficient histories for this case are shown in Figure 9-13 and Figure 9-14, respectively. The oscillatory nature of the flow is clearly evident. Figure 9-15 shows a profile of the flow as defined by pressure contours.

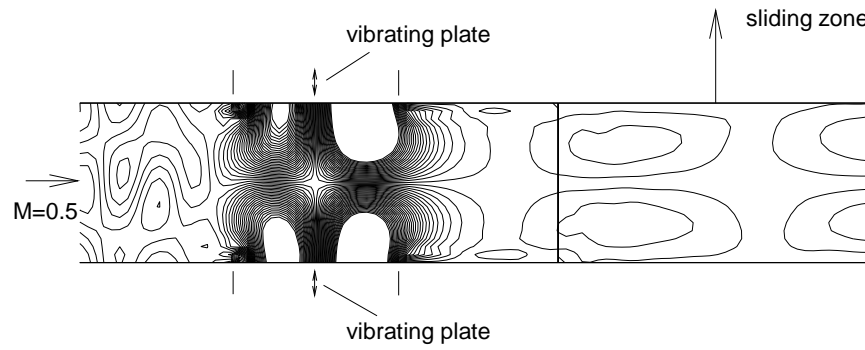


**Figure 9-13.** Residual history for inviscid flow through vibrating flat plates.



**Figure 9-14.** Lift coefficient history for inviscid flow through vibrating flat plates;  
 $M_\infty = 0.5$ .





**Figure 9-15.** Pressure contours for inviscid flow through vibrating flat plates;  $M_{\infty} = 0.5$ .

### 9.1.7 Multistream Nozzle

This case simulates, in two dimensions, the flow through a converging/diverging nozzle with multiple streams. The case is meant to model the exhaust from an engine (with a hot core and cooler outer flow modeled as a “top hat” temperature profile) entering an s-shaped converging/diverging nozzle. Two additional streams are injected downstream of the throat to provide additional cooling of the exhaust.

The grid consists of thirteen patched zones with a total of 15897 points in one plane. The memory requirement for this example is 3.2 million words. A typical timing for this case is 1550 CPU seconds on a CRAY YMP (NASA LaRC’s Sabre as of October 1996). A cross-section of the grid is shown in Figure 9-16.

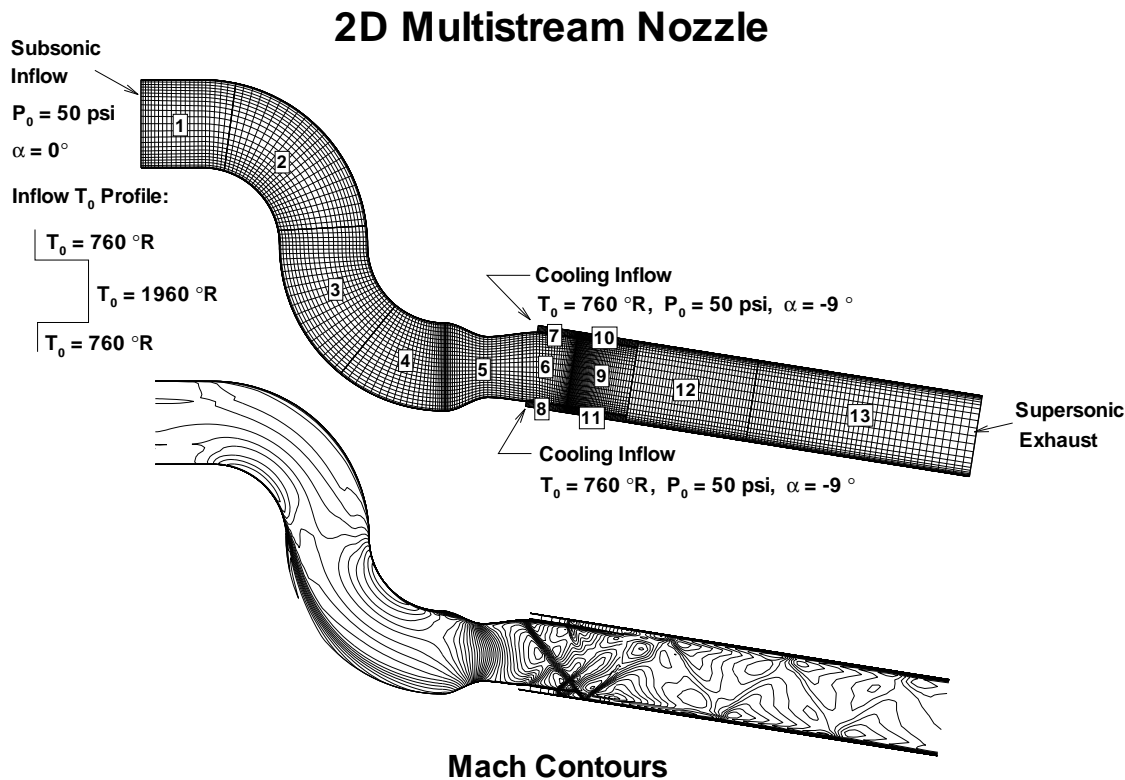


Figure 9-16. Cross-section of multistream nozzle zonal patching and Mach contours.

Boundary conditions with user-defined input, including constant data supplied via the main input deck and variable (point-to-point) data supplied via an auxiliary boundary condition data file are exemplified. Control surfaces are used to monitor mass-flow convergence (here, two control surfaces are used to measure the “difference” between mass in

and mass out). In addition, the case illustrates how one can define reference conditions for CFL3D in the case of a purely internal flow, as discussed below.

#### 9.1.7.1 Nondimensionalization

The conditions provided for this case are that the total pressure of the primary inflow is 50 psi, with a total temperature of 1960°R in the core and 760°R in the outer region. The flow enters the s-duct with zero angle. The secondary cooling flows also have a total pressure of 50 psi and a uniform total temperature of 760°R. The cooling flow enters at –9 degrees relative to horizontal. The exhaust from the nozzle system is supersonic. The throat height is 1 foot. The primary inflow height is 1.449 feet; the core flow region (i.e. where the total temperature is 1960°R) spans the central 0.769 feet.

Although inflow stagnation conditions would be a natural reference state for this problem, the associated Mach number is zero. Since the viscous terms are scaled with the reference Mach number, another reference state is needed. A second natural reference state for nozzle flows is the sonic point. From isentropic relations, the sonic conditions can be obtained once the inflow total conditions are known. However, for this problem there are two total conditions owing to the “top hat” temperature profile. Thus, to have just one reference state, the total temperature of the inflow is area-averaged and the resulting average total temperature, together with the given total pressure, is used to determine the sonic conditions at the throat.

In what follows, stagnation conditions are denoted by 0, sonic conditions by \*, and dimensional quantities by ~. First, determine the average inflow total temperature:

$$\tilde{T}_0 = \frac{0.769}{1.449(1960)} + \frac{(1.449 - 0.769)}{1.449(760)} = 1397^\circ\text{R} \quad (9-1)$$

Next, determine the stagnation density and speed of sound:

$$\tilde{\rho}_0 = \frac{\tilde{p}_0}{\tilde{R}\tilde{T}_0} = \frac{50(144)}{1716(1397)} = 0.00300 \text{ slugs/feet}^3 \quad (9-2)$$

$$\tilde{a}_0 = \sqrt{\gamma\tilde{R}\tilde{T}_0} = \sqrt{1.4(1716)(1397)} = 1832 \text{ feet/second} \quad (9-3)$$

where  $\tilde{R} = 1716 \text{ feet}^2/(\text{second}^2 \cdot ^\circ\text{R})$ . From the isentropic relations,

$$\frac{p}{p_0} = 0.528, \frac{T}{T_0} = 0.833, \frac{\rho}{\rho_0} = 0.634, \frac{a}{a_0} = 0.913 \quad (9-4)$$

Thus, the desired reference pressure, temperature, speed of sound and density are

$$\begin{aligned}
 \tilde{p} &= 50(0.528) = 26.4 \text{ psi} \\
 \tilde{T} &= 1397(0.833) = 1164^\circ\text{R} \\
 \tilde{\rho} &= 0.003(0.634) = 0.0019 \text{ slug/feet}^3 \\
 \tilde{a} &= 1832(0.913) = 1673 \text{ feet/second}
 \end{aligned}
 \tag{9-5}$$

Assuming a molecular viscosity coefficient of  $3.7 \times 10^{-7}$  slugs/(feet-seconds) for a temperature of  $520^\circ\text{R}$ , then the power law  $\mu_2/\mu_1 = (T_2/T_1)^{0.76}$  gives

$$\tilde{\mu}^* = 3.7 \times 10^{-7} \left( \frac{1164}{520} \right)^{0.76} = 6.82 \times 10^{-7} \text{ slugs/(feet-seconds)}
 \tag{9-6}$$

The reference Reynolds number based on throat height and the reference sonic values is

$$Re^* = \frac{\tilde{\rho}^* \tilde{u}^* (\text{throat height})}{\tilde{\mu}^*} = 0.0019(1673) \frac{1}{6.82 \times 10^{-7}} = 4.66 \times 10^6
 \tag{9-7}$$

In the grid, the throat height is 12 inches, so the input parameter **reue** is

$$\mathbf{reue} = \frac{Re^* \times 10^{-6}}{12} = \frac{4.66}{12} = 0.388
 \tag{9-8}$$

Finally, the nondimensional input values for boundary condition type 2003 are determined from the reference sonic conditions (note that in CFL3D parlance, in this problem the \* conditions are the “infinity” conditions):

$$\begin{aligned}
 p_t &= 50 \text{ psi} \rightarrow \frac{p_t}{p^*} = \frac{p_t}{p_\infty} = \frac{50}{26.4} = 1.894 \\
 T_t &= 760^\circ\text{R} \rightarrow \frac{T_t}{T^*} = \frac{T_t}{T_\infty} = \frac{760}{1164} = 0.653 \\
 T_t &= 1960^\circ\text{R} \rightarrow \frac{T_t}{T^*} = \frac{T_t}{T_\infty} = \frac{1960}{1164} = 1.684
 \end{aligned}
 \tag{9-9}$$

Boundary condition type 2003 also needs an estimate of the local Mach number. For the primary inflow, the inlet height (area) to throat height (area) is 1.449/1. The isentropic relations give a corresponding Mach number of approximately 0.45. The local Mach number for the cooling inflow cannot be determined a priori; 1.0 is used in the boundary condition. The computations give the cooling inflow Mach number as approximately 0.85; 1.0 is deemed as a sufficiently close estimate since the solution does not change perceptibly if 0.85 is used instead of 1.0.

The auxiliary boundary condition data file provided for this example (`inflow.data`) contains the data for the primary inflow:

$$\begin{aligned} M &= 0.45 \\ \frac{p_t}{p_\infty} &= 1.894 \\ \frac{T_t}{T_\infty} &= \frac{0.653}{1.684} \end{aligned} \tag{9-10}$$

in top hat distribution,  $\alpha = 0$ ,  $\beta = 0$ . The cooling inflow:

$$\begin{aligned} M &= 1.0 \\ \frac{p_t}{p_\infty} &= 1.894 \\ \frac{T_t}{T_\infty} &= 0.653 \\ \alpha &= -9 \\ \beta &= 0 \end{aligned} \tag{9-11}$$

is specified explicitly in the main data file.

### 9.1.7.2 Running CFL3D

Besides the CFL3D and ronnie codes the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
<code>multistream.inp</code>	input for CFL3D
<code>grid_multistream.fmt</code>	formatted grid
<code>formtobin.f</code>	grid converter
<code>inflow.data</code>	auxiliary boundary condition data
<code>ronl.h</code>	parameters for ronnie makefile
<code>ronnie.inp</code>	input for ronnie

The steps for running this case on the YMP are as follows:

#### Step 1

Compile the grid converter code:

```
cft77 formtobin.f
```

### Step 2

Link the grid converter object file:

```
segldr -o formtobin formtobin.o
```

### Step 3

Run the grid generator program (the binary file `grid_multistream.bin` will be output):

```
formtobin
```

### Step 4

Use the makefile to compile, link, and create the executable for the ronnie code (be sure `ron1.h` is in the current directory):

```
make -f makeronnie_cray
```

### Step 5

Run the ronnie code (the file `patch_multistream.bin` will be output):

```
ronnie < ronnie.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 7

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < multistream.inp
```

### Step 8

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

### Step 9

Run the CFL3D code (be sure the `inflow.data` file is available and correct for this case):

```
cfl3d < multistream.inp
```

The input file for this case is:

```
I/O FILES
grid_multistream.p3d
plot3dg.bin
```

```

plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch_multistream.bin
restart.bin
Multistream Nozzle (sonic conditions as reference state)
  XMACH      ALPHA      BETA  REUE,MIL  TINF,DR      IALPH      IHIST
    1.000      0.00      0.0    0.388    1163.0        1        1
  SREF      CREF      BREF      XMC      YMC      ZMC
1.00000  1.00000  1.0000  0.25000  0.00      0.00
  DT      IREST  IFLAGTS      FMAX      IUNST  CFL_TAU
-1.0000  0      000      1.0      +1      5.
  NGRID  NPLOT3D  NPRINT  NWREST  ICHK      I2D      NTSTEP      ITA
   -13    13      0      500      0      1      2      -2
  NCG      IEM  IADVANCE  IFORCE  IVISC(I)  IVISC(J)  IVISC(K)
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
    1      0      0      000      0      0      +7
  IDIM      JDIM      KDIM
    2      23      41
    2      23      41
    2      23      41
    2      23      41
    2      25      41
    2      25      41
    2      17      21
    2      17      21
    2      49      41
    2      49      21
    2      49      21
    2      41      61
    2      49      57
  ILAMLO  ILAMHI  JLAMLO  JLAMHI  KLAMLO  KLAMHI
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
    00      00      000      000      0      0000
  INEWG  IGRIDC  IS      JS      KS      IE      JE      KE
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0
    0      0      0      0      0      0      0      0

```

	0	0	0	0	0	0	0	0
	0	0	0	0	0	0	0	0
	0	0	0	0	0	0	0	0
	0	0	0	0	0	0	0	0
IDIAG(I)	IDIAG(J)	IDIAG(K)	IFLIM(I)	IFLIM(J)	IFLIM(K)			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
1	1	1	3	3	3			
IFDS(I)	IFDS(J)	IFDS(K)	RKAP0(I)	RKAP0(J)	RKAP0(K)			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
1	1	1	.3333	.3333	.3333			
GRID	NBCI0	NBCIDIM	NBCJ0	NBCJDIM	NBCK0	NBCKDIM	IOVRLP	
1	1	1	1	1	1	1	0	
2	1	1	1	1	1	1	0	
3	1	1	1	1	1	1	0	
4	1	1	1	1	1	1	0	
5	1	1	1	1	1	1	0	
6	1	1	1	1	1	1	0	
7	1	1	1	1	1	1	0	
8	1	1	1	1	1	1	0	
9	1	1	1	1	1	1	0	
10	1	1	1	1	1	1	0	
11	1	1	1	1	1	1	0	
12	1	1	1	1	1	1	0	
13	1	1	1	1	1	1	0	
I0:	GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA
1	1	1	1002	0	0	0	0	0
2	1	1	1002	0	0	0	0	0
3	1	1	1002	0	0	0	0	0
4	1	1	1002	0	0	0	0	0
5	1	1	1002	0	0	0	0	0
6	1	1	1002	0	0	0	0	0
7	1	1	1002	0	0	0	0	0
8	1	1	1002	0	0	0	0	0
9	1	1	1002	0	0	0	0	0
10	1	1	1002	0	0	0	0	0
11	1	1	1002	0	0	0	0	0
12	1	1	1002	0	0	0	0	0
13	1	1	1002	0	0	0	0	0
IDIM:	GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA
1	1	1	1002	0	0	0	0	0
2	1	1	1002	0	0	0	0	0
3	1	1	1002	0	0	0	0	0
4	1	1	1002	0	0	0	0	0
5	1	1	1002	0	0	0	0	0
6	1	1	1002	0	0	0	0	0
7	1	1	1002	0	0	0	0	0
8	1	1	1002	0	0	0	0	0
9	1	1	1002	0	0	0	0	0
10	1	1	1002	0	0	0	0	0



	12		1	1002	0	0	0	0	0
	13		1	1002	0	0	0	0	0
J0:	GRID	SEGMENT		BCTYPE	ISTA	IEND	KSTA	KEND	NDATA
	1		1	2003	0	0	0	0	-5
Mach	Pt/Pinf	Tt/Tinf		alpha	beta				
inflow.data									
	2		1	0	0	0	0	0	0
	3		1	0	0	0	0	0	0
	4		1	0	0	0	0	0	0
	5		1	0	0	0	0	0	0
	6		1	0	0	0	0	0	0
	7		1	2003	0	0	0	0	5
Mach	Pt/Pinf	Tt/Tinf		alpha	beta				
1.00	1.894	0.653		-9.	0.				
	8		1	2003	0	0	0	0	5
Mach	Pt/Pinf	Tt/Tinf		alpha	beta				
1.00	1.894	0.653		-9.	0.				
	9		1	0	0	0	0	0	0
	10		1	0	0	0	0	0	0
	11		1	0	0	0	0	0	0
	12		1	0	0	0	0	0	0
	13		1	0	0	0	0	0	0
JDIM:	GRID	SEGMENT		BCTYPE	ISTA	IEND	KSTA	KEND	NDATA
	1		1	0	0	0	0	0	0
	2		1	0	0	0	0	0	0
	3		1	0	0	0	0	0	0
	4		1	0	0	0	0	0	0
	5		1	0	0	0	0	0	0
	6		1	0	0	0	0	0	0
	7		1	0	0	0	0	0	0
	8		1	0	0	0	0	0	0
	9		1	0	0	0	0	0	0
	10		1	0	0	0	0	0	0
	11		1	0	0	0	0	0	0
	12		1	0	0	0	0	0	0
	13		1	1002	0	0	0	0	0
K0:	GRID	SEGMENT		BCTYPE	ISTA	IEND	JSTA	JEND	NDATA
	1		1	2004	0	0	0	0	2
Tw	Cq								
0.									
	2		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	3		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	4		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	5		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	6		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	7		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	8		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	9		1	0	0	0	0	0	0
	10		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	11		1	0	0	0	0	0	0
	12		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								
	13		1	2004	0	0	0	0	2
Tw	Cq								
0.	0.								

KDIM:	GRID	SEGMENT	BCTYPE	ISTA	IEND	JSTA	JEND	NDATA				
	1	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	2	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	3	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	4	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	5	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	6	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	7	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	8	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	9	1	0	0	0	0	0	0				
	10	1	0	0	0	0	0	0				
	11	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	12	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	13	1	2004	0	0	0	0	2				
Tw	Cq											
0.	0.											
	MSEQ	MGFLAG	ICONSF	MTT	NGAM							
	1	1	1	0	02							
	ISSC	EPSSC(1)	EPSSC(2)	EPSSC(3)	ISSR	EPSSR(1)	EPSSR(2)	EPSSR(3)				
	0	.3	.3	.3	0	.3	.3	.3				
	NCYC	MGLEVG	NEMGL	NITFO								
	3000	02	00	0								
	MIT1	MIT2	MIT3	MIT4	MIT5							
	01	01	01	01	01							
	1-1 BLOCKING DATA:											
	NBLI											
	0											
	NUMBER	GRID	:	ISTA	JSTA	KSTA	IEND	JEND	KEND	ISVA1	ISVA2	
	NUMBER	GRID	:	ISTA	JSTA	KSTA	IEND	JEND	KEND	ISVA1	ISVA2	
	PATCH SURFACE DATA:											
	NINTER											
	-1											
	PLOT3D OUTPUT:											
	grid	iptyp		ista	iend	iinc	jsta	jend	jinc	ksta	kend	kinc
	1	0		0	0	0	0	0	0	0	0	0
	2	0		0	0	0	0	0	0	0	0	0
	3	0		0	0	0	0	0	0	0	0	0
	4	0		0	0	0	0	0	0	0	0	0
	5	0		0	0	0	0	0	0	0	0	0
	6	0		0	0	0	0	0	0	0	0	0
	7	0		0	0	0	0	0	0	0	0	0
	8	0		0	0	0	0	0	0	0	0	0
	9	0		0	0	0	0	0	0	0	0	0
	10	0		0	0	0	0	0	0	0	0	0
	11	0		0	0	0	0	0	0	0	0	0
	12	0		0	0	0	0	0	0	0	0	0
	13	0		0	0	0	0	0	0	0	0	0
	MOVIE											
	0											
	PRINT OUT:											
	GRID	IPATYPE	ISTART	IEND	IINC	JSTART	JEND	JINC	KSTART	KEND	KINC	
	CONTROL SURFACES:											

```

NCS
4
GRID  ISTA  IEND  JSTA  JEND  KSTA  KEND  IWALL  INORM
1      0      0      1      1      0      0      0      -1
7      0      0      1      1      0      0      0      -1
8      0      0      1      1      0      0      0      -1
13     0      0     49     49      0      0      0      1

```

The inflow.data file is:

```

auxiliary bc data, j-face of block 1, multistream nozzle
40, 2*1
5
40*0.4499999999999993, 40*1.893999999999998, 16*0.6530000000000001,
8*1.6850000000000004, 16*0.6530000000000001, 40*0., 40*0.

```

After running this test case, the residual history and mass flow convergence history shown in Figure 9-17 results. Also, a plot of Mach contours should have the flow features of those plotted in Figure 9-16.

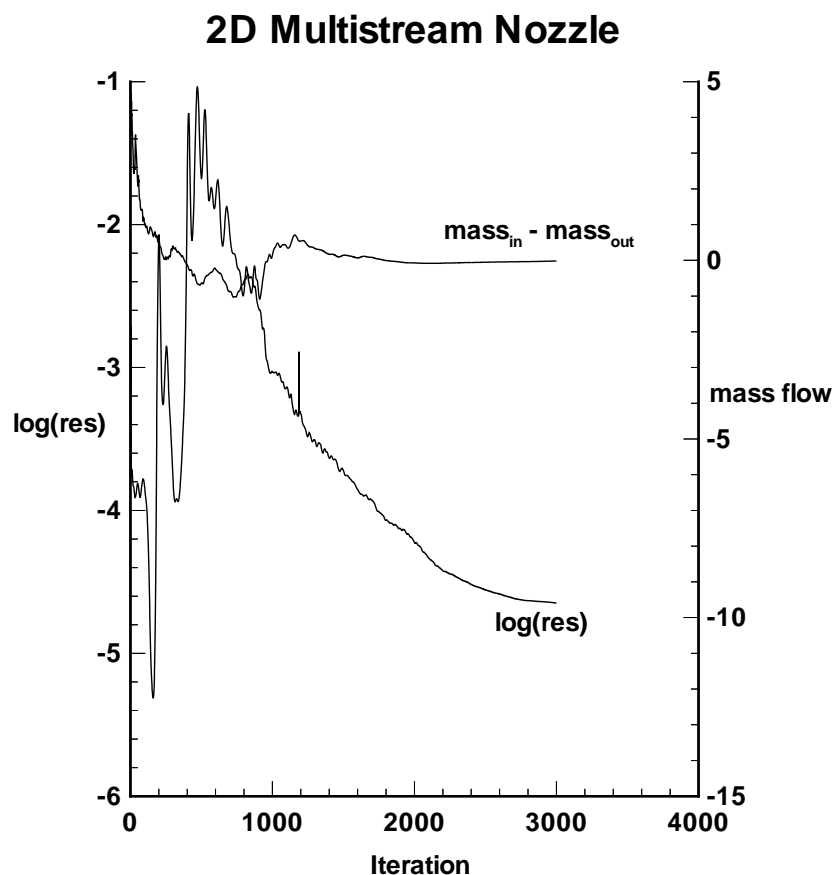


Figure 9-17. Multistream nozzle case residual and mass flow convergence history.

### 9.1.8 Rotor Stator

This case simulates, in two dimensions, the unsteady flow through a single stage turbine in which the ratio of stator to rotor blades is 3:4. The axial gap between the two blades is 50% of the blade chord. The case exercises a number of capabilities of CFL3D including unsteady flow, moving (translating) zones, dynamic patching between zones in relative motion, grid overlapping, and boundary conditions with user-defined input.

The original grid for this case was provided by D. J. Dorney<sup>17</sup> of Western Michigan University, although the grid given out for the test case contains only half the number of points of the original grid. The grid consists of fourteen zones with a total of 18374 points in one plane. A close-up of the grid near the airfoil is shown in Figure 9-18. The grid zones communicate with one another through both patching and overlapping. At a time step of 1.0, it takes 270 time steps for the eight rotor zones (containing four blades) to completely traverse the six stator zones (containing three blades). The rotor zones are reset after each complete traverse. The input file is set for 1500 time steps (using five multigrid sub-iterations per time step), which is sufficient to establish a time-periodic solution. The memory requirement for this example is 4.0 million words. A typical timing for this case (1500 time steps) is 4205 CPU seconds on a Cray YMP (NASA LaRC's Sabre as of October 1996). On a DEC Alpha workstation, the timing is 18303 CPU seconds, using single precision (as of June 1996).

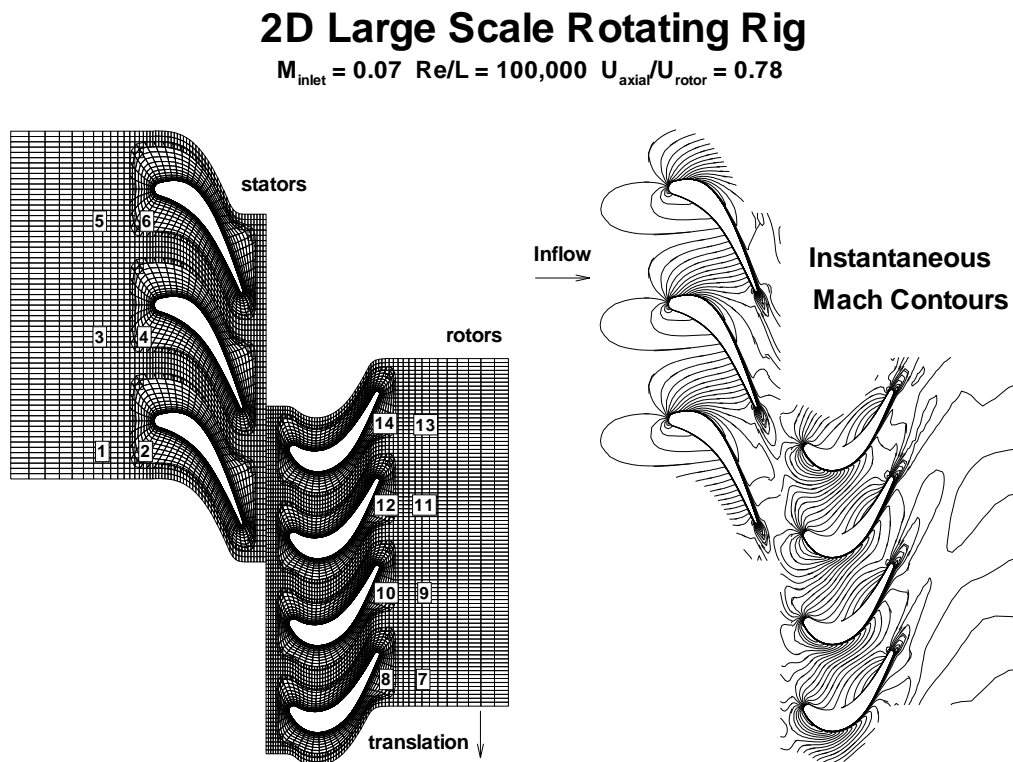


Figure 9-18. Fourteen zone rotor-stator grid system and Mach contours.

### 9.1.8.1 Experimental Conditions

The experimental blade count was 22 stator blades and 28 rotor blades. In order to run an exact simulation, a minimum of 11 stator blades and 14 rotor blades would be required (Dring et.al<sup>18</sup>). To reduce the problem size in the computation, the ratio of stator to rotor blades was reduced to 3:4 (equivalent to 21:28), and the stators were scaled by a factor of 22/21 to maintain the same pitch-to-chord ratio as in the experiment. The experimental set-up was a three-dimensional configuration; the corresponding 2-d simulation was set up from conditions at the mid-span radius of 27 inches, for a rotor speed of 410 rpm, with a nominal axial velocity of 75 feet/second. The inlet Mach number in the experiment was approximately 0.07, and the Reynolds number/inch was approximately 100,000.

### 9.1.8.2 Input Setup

Inlet conditions are used as the reference conditions, so **xmach** = 0.07. The grid is full scale, with dimensions in inches. Therefore, **reue** = 0.1. The inlet temperature was assumed to be 60°F, so **tin** = 520°R.

Boundary condition type 2003 is used to specify total pressure and total temperature at the inlet. From isentropic flow relations or tables, for an inlet flow Mach number of 0.07,

$$\begin{aligned}
 M_{inlet} &= 0.07 \\
 \frac{P_{t,inlet}}{P_{\infty}} &= 1.0035 \\
 \frac{T_{t,inlet}}{T_{\infty}} &= 1.0010
 \end{aligned}
 \tag{9-12}$$

Also,  $\alpha = \beta = 0$  (purely axial flow is assumed).

Boundary condition type 2002 is used to specify an exit pressure. Dorney gives a ratio of static pressure to inlet total pressure = 0.963 at the rotor trailing edge plane. Assuming this value to hold at the exit as well gives

$$\frac{P_{exit}}{P_{\infty}} = \frac{P_{exit}}{P_{t,inlet}} \frac{P_{t,inlet}}{P_{\infty}} = 0.963 \times 1.0035 = 0.967
 \tag{9-13}$$

Note that the inflow Mach number used in boundary condition type 2003 is an estimate; if the exit pressure were not set correctly, the computed inflow Mach number would not be close to the specified inflow value (when a time-periodic state is reached or at convergence in steady state). By specifying control surfaces at the inflow plane, the user is able to verify after the computation is complete that the average inflow Mach number is approximately 0.071; this was deemed to be close enough to the desired value. If desired,

the exit pressure could be adjusted (raised in this case) and the solution re-run until a new time-periodic solution (and a new inlet Mach number) is established.

It should be noted that the input grid is in PLOT3D format, with  $y$  as the “up” direction (**ialph** = 1;  $z$  is the spanwise, 2-d direction). However, the grid motion parameters *must* be set as if  $z$  is the up direction. Recall that if the input grid has  $y$  as the up direction, CFL3D will internally swap  $y$  and  $z$  so that the code always computes on a grid in which  $z$  is up. (See the caution in “LT35 - Translational Information and Velocities” on page 44.)

Given the rotor speed and mid-span radius, the translational velocity for a 2-d simulation corresponding to the mid-span radius is

$$\tilde{w}_{trans} = \omega r = (410/60 \times 2\pi)(27/12) = 96.6 \text{ feet/second} \quad (9-14)$$

This gives

$$\frac{\tilde{u}_{axial}}{\tilde{w}_{trans}} = \frac{75}{96.6} = 0.78 \quad (9-15)$$

The input value **wtrans** is  $\tilde{w}_{trans}/\tilde{a}_{\infty}$ , so with the reference Mach number 0.07, **wtrans** =  $0.07/0.78 = (-) 0.0897$  (the negative gives a downward rotor motion).

In order to be able to run an arbitrarily long simulation, the grid resetting option was employed. The top-to-bottom length of the grid is 24.23514 inches and the rotor and stator zones start out in alignment, so **dzmax** = 24.23514. Thus the rotor zones are reset whenever the displacement exceeds 24.23514 inches.

### 9.1.8.3 Running CFL3D

Besides the CFL3D code, the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
lsrr.inp	input for CFL3D
lsrr_coarse.p3d_fmt	formatted single plane grid
fmttobin_p3d.f	converter for creating 2 grid planes
mag1.h	parameters for MaGGiE makefile
maggie.inp	input for MaGGiE

The steps for running this case on the DEC are as follows:

### Step 1

Compile the grid converter code:

```
cft77 fmttobin_p3d.f
```

### Step 2

Link the grid converter object file:

```
segldr -o fmttobin_p3d fmttobin_p3d.o
```

### Step 3

Run the grid converter program (the binary file `lsrr_coarse.p3d` will be output):

```
fmttobin_p3d
```

### Step 4

Use the makefile to compile, link, and create the executable for the MaGGiE code (be sure `mag1.h` is in the current directory):

```
make -f makemaggie_cray
```

### Step 5

Run the MaGGiE code (the file `ovrlp.bin` will be output):

```
maggie < maggie.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 7

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < lsrr.inp
```

### Step 8

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

### Step 9

Run the CFL3D code:

```
cf13d < lsrr.inp
```

The input file for this case is:

[illegible]



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	10	1	1002	0	0	0	0	0
	11	1	1002	0	0	0	0	0
	12	1	1002	0	0	0	0	0
	13	1	1002	0	0	0	0	0
	14	1	1002	0	0	0	0	0
IDIM:	GRID	SEGMENT	BCTYPE	JSTA	JEND	KSTA	KEND	NDATA
	1	1	1002	0	0	0	0	0
	2	1	1002	0	0	0	0	0
	3	1	1002	0	0	0	0	0
	4	1	1002	0	0	0	0	0
	5	1	1002	0	0	0	0	0
	6	1	1002	0	0	0	0	0
	7	1	1002	0	0	0	0	0
	8	1	1002	0	0	0	0	0
	9	1	1002	0	0	0	0	0
	10	1	1002	0	0	0	0	0
	11	1	1002	0	0	0	0	0
	12	1	1002	0	0	0	0	0
	13	1	1002	0	0	0	0	0
	14	1	1002	0	0	0	0	0
J0:	GRID	SEGMENT	BCTYPE	ISTA	IEND	KSTA	KEND	NDATA
	1	1	2003	0	0	0	0	5
Mach	Pt/Pinf	Tt/Tinf	alpha	beta				
0.07	1.0035	1.0010	0.	0.				
	2	1	0	0	0	0	0	0
	3	1	2003	0	0	0	0	5
Mach	Pt/Pinf	Tt/Tinf	alpha	beta				
0.07	1.0035	1.0010	0.	0.				
	4	1	0	0	0	0	0	0
	5	1	2003	0	0	0	0	5
Mach	Pt/Pinf	Tt/Tinf	alpha	beta				
0.07	1.0035	1.0010	0.	0.				
	6	1	0	0	0	0	0	0
	7	1	0	0	0	0	0	0
	8	1	0	0	0	0	0	0
	9	1	0	0	0	0	0	0
	10	1	0	0	0	0	0	0
	11	1	0	0	0	0	0	0
	12	1	0	0	0	0	0	0
	13	1	0	0	0	0	0	0
	14	1	0	0	0	0	0	0
JDIM:	GRID	SEGMENT	BCTYPE	ISTA	IEND	KSTA	KEND	NDATA
	1	1	0	0	0	0	0	0
	2	1	0	0	0	0	0	0
	3	1	0	0	0	0	0	0
	4	1	0	0	0	0	0	0
	5	1	0	0	0	0	0	0
	6	1	0	0	0	0	0	0
	7	1	2002	0	0	0	0	1
pexit/pinf								
0.967								
	8	1	0	0	0	0	0	0
	9	1	2002	0	0	0	0	1
pexit/pinf								
0.967								
	10	1	0	0	0	0	0	0
	11	1	2002	0	0	0	0	1
pexit/pinf								
0.967								
	12	1	0	0	0	0	0	0
	13	1	2002	0	0	0	0	1
pexit/pinf								
0.967								
	14	1	0	0	0	0	0	0
K0:	GRID	SEGMENT	BCTYPE	ISTA	IEND	JSTA	JEND	NDATA
	1	1	0	0	0	0	0	0
	2	1	2004	0	0	0	0	2
Tw/Tinf	C_q							
0.	0.							
	3	1	0	0	0	0	0	0
	4	1	2004	0	0	0	0	2
Tw/Tinf	C_q							

```

0.      0.
      5      1      0      0      0      0      0      0
      6      1      2004      0      0      0      0      2
Tw/Tinf C_q
0.      7      1      0      0      0      0      0      0
      8      1      2004      0      0      0      0      2
Tw/Tinf C_q
0.      9      1      0      0      0      0      0      0
      10     1      2004      0      0      0      0      2
Tw/Tinf C_q
0.     11      1      0      0      0      0      0      0
      12     1      2004      0      0      0      0      2
Tw/Tinf C_q
0.     13      1      0      0      0      0      0      0
      14     1      2004      0      0      0      0      2
Tw/Tinf C_q
0.     14     1
KDIM: GRID SEGMENT BCTYPE ISTA IEND JSTA JEND NDATA
      1      1      0      0      0      0      0      0
      2      1      0      0      0      0      0      0
      3      1      0      0      0      0      0      0
      4      1      0      0      0      0      0      0
      5      1      0      0      0      0      0      0
      6      1      0      0      0      0      0      0
      7      1      0      0      0      0      0      0
      8      1      0      0      0      0      0      0
      9      1      0      0      0      0      0      0
     10      1      0      0      0      0      0      0
     11      1      0      0      0      0      0      0
     12      1      0      0      0      0      0      0
     13      1      0      0      0      0      0      0
     14      1      0      0      0      0      0      0
MSEQ MGFLAG ICONSF MTT NGAM
  1      1      1      0      01
ISSC EPSSC(1) EPSSC(2) EPSSC(3) ISSR EPSSR(1) EPSSR(2) EPSSR(3)
  0      .3      .3      .3      0      .3      .3      .3
NCYC MGLEVG NEMGL NITFO
  5      02      00      000
MIT1 MIT2 MIT3 MIT4 MIT5
  01      01      01      01      01
1-1 BLOCKING DATA:
NBLI
  14
NUMBER GRID : ISTA JSTA KSTA IEND JEND KEND ISVA1 ISVA2
  1      2      1      1      1      2      1      21      1      3
  2      4      1      1      1      2      1      21      1      3
  3      6      1      1      1      2      1      21      1      3
  4      8      1      1      1      2      1      21      1      3
  5     10      1      1      1      2      1      21      1      3
  6     12      1      1      1      2      1      21      1      3
  7     14      1      1      1      2      1      21      1      3
  8      1      1      1      1      2     55      1      1      2
  9      7      1      1      1      2     61      1      1      2
 10      1      1      1     23      2     55     23      1      2
 11      3      1      1     23      2     55     23      1      2
 12      7      1      1     23      2     61     23      1      2
 13      9      1      1     23      2     61     23      1      2
 14     11      1      1     23      2     61     23      1      2
NUMBER GRID : ISTA JSTA KSTA IEND JEND KEND ISVA1 ISVA2
  1      2      1     61      1      2     61     21      1      3
  2      4      1     61      1      2     61     21      1      3
  3      6      1     61      1      2     61     21      1      3
  4      8      1     61      1      2     61     21      1      3
  5     10      1     61      1      2     61     21      1      3
  6     12      1     61      1      2     61     21      1      3
  7     14      1     61      1      2     61     21      1      3
  8      5      1      1     23      2     55     23      1      2
  9     13      1      1     23      2     61     23      1      2

```

10	3	1	1	1	2	55	1	1	2
11	5	1	1	1	2	55	1	1	2
12	9	1	1	1	2	61	1	1	2
13	11	1	1	1	2	61	1	1	2
14	13	1	1	1	2	61	1	1	2

PATCH SURFACE DATA:  
NINTER  
0

PLOT3D OUTPUT:

BLOCK	IPTYPE	ISTART	IEND	IINC	JSTART	JEND	JINC	KSTART	KEND	KINC
1	0	1	001	1	01	999	1	1	999	1
2	0	1	001	1	01	999	1	1	999	1
3	0	1	001	1	01	999	1	1	999	1
4	0	1	001	1	01	999	1	1	999	1
5	0	1	001	1	01	999	1	1	999	1
6	0	1	001	1	01	999	1	1	999	1
7	0	1	001	1	01	999	1	1	999	1
8	0	1	001	1	01	999	1	1	999	1
9	0	1	001	1	01	999	1	1	999	1
10	0	1	001	1	01	999	1	1	999	1
11	0	1	001	1	01	999	1	1	999	1
12	0	1	001	1	01	999	1	1	999	1
13	0	1	001	1	01	999	1	1	999	1
14	0	1	001	1	01	999	1	1	999	1

MOVIE  
0

PRINT OUT:

BLOCK	IPTYPE	ISTART	IEND	IINC	JSTART	JEND	JINC	KSTART	KEND	KINC
1	0	1	001	1	01	999	1	1	999	1
2	0	1	001	1	01	999	1	1	999	1
3	0	1	001	1	01	999	1	1	999	1
4	0	1	001	1	01	999	1	1	999	1
5	0	1	001	1	01	999	1	1	999	1
6	0	1	001	1	01	999	1	1	999	1
7	0	1	001	1	01	999	1	1	999	1
8	0	1	001	1	01	999	1	1	999	1
9	0	1	001	1	01	999	1	1	999	1
10	0	1	001	1	01	999	1	1	999	1
11	0	1	001	1	01	999	1	1	999	1
12	0	1	001	1	01	999	1	1	999	1
13	0	1	001	1	01	999	1	1	999	1
14	0	1	001	1	01	999	1	1	999	1

CONTROL SURFACES:

NCS  
7

GRID	ISTA	IEND	JSTA	JEND	KSTA	KEND	IWALL	INORM
7	1	2	999	999	0	0	0	1
9	1	2	999	999	0	0	0	1
11	1	2	999	999	0	0	0	1
13	1	2	999	999	0	0	0	1
1	1	2	1	1	0	0	0	0
3	1	2	1	1	0	0	0	0
5	1	2	1	1	0	0	0	0

MOVING GRID DATA - TRANSLATION

NTRANS  
9

LREF  
1.0

GRID	ITRANS	RFREQ	UTRANS	VTRANS	WTRANS
7	1	0.	0.	0.	-0.0897
8	1	0.	0.	0.	-0.0897
9	1	0.	0.	0.	-0.0897
10	1	0.	0.	0.	-0.0897
11	1	0.	0.	0.	-0.0897
12	1	0.	0.	0.	-0.0897
13	1	0.	0.	0.	-0.0897
14	1	0.	0.	0.	-0.0897
0	1	0.	0.	0.	-0.0897

GRID	DXMAX	DYMAX	DZMAX
7	0.	0.	-24.23514
8	0.	0.	-24.23514
9	0.	0.	-24.23514
10	0.	0.	-24.23514
11	0.	0.	-24.23514
12	0.	0.	-24.23514
13	0.	0.	-24.23514
14	0.	0.	-24.23514
0	0.	0.	-24.23514

MOVING GRID DATA - ROTATION

NROTAT  
0

LREF

GRID	IROTAT	RFREQ	OMEGAX	OMEGAY	OMEGAZ	XORIG	YORIG	ZORIG
7	1	0.	0.	0.	0.	0.	0.	0.
8	1	0.	0.	0.	0.	0.	0.	0.
9	1	0.	0.	0.	0.	0.	0.	0.
10	1	0.	0.	0.	0.	0.	0.	0.
11	1	0.	0.	0.	0.	0.	0.	0.
12	1	0.	0.	0.	0.	0.	0.	0.
13	1	0.	0.	0.	0.	0.	0.	0.
14	1	0.	0.	0.	0.	0.	0.	0.
0	1	0.	0.	0.	0.	0.	0.	0.

GRID THXMAX THYMAX THZMAX

DYNAMIC PATCH INPUT DATA

NINTER

7								
INT	IFIT	LIMIT	ITMAX	MCXIE	MCETA	C-0	IORPH	ITOSS
1	1	1	30	0	0	0	0	1
2	1	1	30	0	0	0	0	1
3	1	1	30	0	0	0	0	1
4	1	1	30	0	0	0	0	1
5	1	1	30	0	0	0	0	1
6	1	1	30	0	0	0	0	1
7	1	1	30	0	0	0	0	1
INT	TO	XIE1	XIE2	ETA1	ETA2	NFB		
1	122	0	0	0	0	6		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	721	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	921	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1121	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1321	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	721	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	24.23514	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	921	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	24.23514	0.	0.	0.		
INT	TO	XIE1	XIE2	ETA1	ETA2	NFB		
2	322	0	0	0	0	6		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	921	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1121	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1321	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	721	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	24.23514	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	921	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	24.23514	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1121	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	24.23514	0.	0.	0.		
INT	TO	XIE1	XIE2	ETA1	ETA2	NFB		
3	522	0	0	0	0	6		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1121	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	
	1321	0	0	0	0	0.	0.	
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ		
	0.	0.	0.	0.	0.	0.		
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK	

	721	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	921	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	1121	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	1321	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	24.23514	0.	0.	0.	
INT	TO	XIE1	XIE2	ETA1	ETA2	NFB	
4	721	0	0	0	0	4	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	122	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	0.	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	522	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	-24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	322	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	-24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	122	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	-24.23514	0.	0.	0.	
INT	TO	XIE1	XIE2	ETA1	ETA2	NFB	
5	921	0	0	0	0	5	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	122	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	0.	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	322	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	0.	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	522	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	-24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	322	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	-24.23514	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	122	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	-24.23514	0.	0.	0.	
INT	TO	XIE1	XIE2	ETA1	ETA2	NFB	
6	1121	0	0	0	0	5	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	322	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	0.	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	522	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	0.	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	122	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	
	0.	0.	0.	0.	0.	0.	
	FROM	XIE1	XIE2	ETA1	ETA2	FACTJ	FACTK
	522	0	0	0	0	0.	0.
	DX	DY	DZ	DTHETX	DTHETY	DTHETZ	

```

0.      0. -24.23514      0.      0.      0.
FROM    XIE1    XIE2    ETA1    ETA2    FACTJ    FACTK
322      0      0      0      0      0.      0.
DX      DY      DZ    DTHETX    DTHETY    DTHETZ
0.      0. -24.23514      0.      0.      0.
INT  TO    XIE1    XIE2    ETA1    ETA2    NFB
7  1321    0      0      0      0      4
FROM    XIE1    XIE2    ETA1    ETA2    FACTJ    FACTK
522      0      0      0      0      0.      0.
DX      DY      DZ    DTHETX    DTHETY    DTHETZ
0.      0.      0.      0.      0.      0.
FROM    XIE1    XIE2    ETA1    ETA2    FACTJ    FACTK
322      0      0      0      0      0.      0.
DX      DY      DZ    DTHETX    DTHETY    DTHETZ
0.      0.      0.      0.      0.      0.
FROM    XIE1    XIE2    ETA1    ETA2    FACTJ    FACTK
122      0      0      0      0      0.      0.
DX      DY      DZ    DTHETX    DTHETY    DTHETZ
0.      0.      0.      0.      0.      0.
FROM    XIE1    XIE2    ETA1    ETA2    FACTJ    FACTK
522      0      0      0      0      0.      0.
DX      DY      DZ    DTHETX    DTHETY    DTHETZ
0.      0. -24.23514      0.      0.      0.

```

The convergence histories for residual, mass flow, and rotor lift coefficient as shown in Figure 9-19 should be obtained.

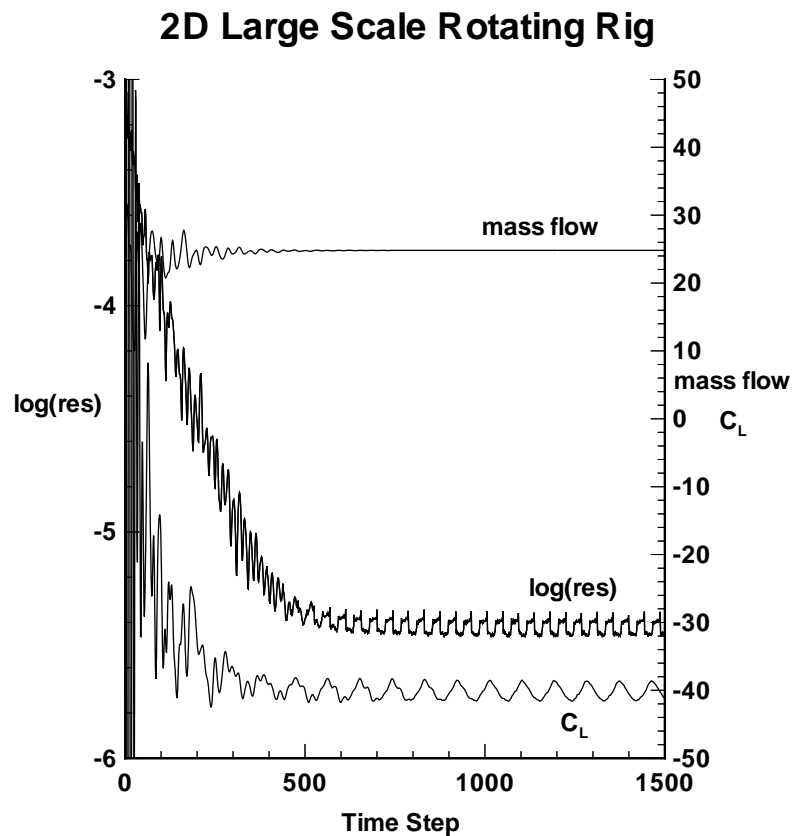


Figure 9-19. Convergence histories for rotor-stator case.

## 9.2 Three-dimensional Test Cases

### 9.2.1 Axisymmetric Bump Flow

This test case solves for the turbulent flow over an axisymmetric bump. The flow is modeled in 3-d using two computational planes (separated by an angle of 1 degree), with periodic boundary conditions; hence **bctype** is 2005 and **dthtx** is -1.0 and 1.0 on the I0 and IDIM boundaries, respectively. The grid consists of a single zone with a total of 36562 points. The memory requirement for this example is 4.9 million words. A typical timing for this case is 1026 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of October 1996). A close-up of the grid near the bump is shown in Figure 9-20.

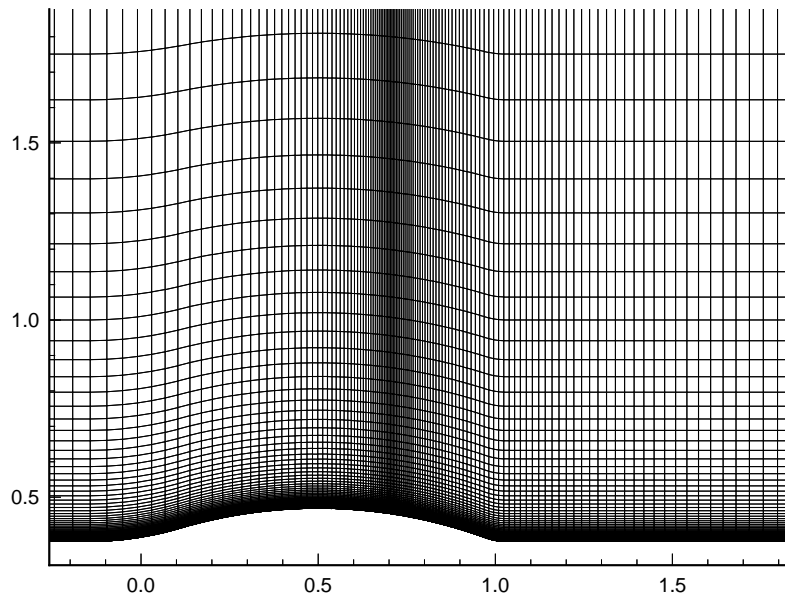


Figure 9-20. Axisymmetric bump grid.

Besides the CFL3D code, the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
bumpv5periodic.inp	input for CFL3D
bump.grd	formatted single plane grid
gridaxi.f	converter for creating 2 grid planes

The steps for running this case on the YMP are as follows:



Step 1

Compile the grid converter code:

```
cft77 gridaxi.f
```

Step 2

Link the grid converter object file:

```
segldr -o gridaxi gridaxi.o
```

Step 3

Run the grid converter program (the binary file bumpgrd.bin will be output):

```
gridaxi
```

In answer to the question, type:

```
bump.grd
```

Step 4

Use the makefile to compile, link, and create the executable for the precfl3d code (be sure precfl.h is in the current directory):

```
make -f makeprecfl3d_cray
```

Step 5

Run the precfl3d code (the cflx.h files will be output):

```
precfl3d < bumpv5periodic.inp
```

Step 6

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

Step 7

Run the CFL3D code:

```
cfl3d < bumpv5periodic.inp
```

The input file for this case is:

```
I/O FILES
bumpgrd.bin
plot3dq.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
```

```

cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
  Axisymmetric bump flow, 3-d, using 2 planes and periodic BCs
    XMACH      ALPHA      BETA  REUE,MIL  TINF,DR  IALPH  IHIST
    0.8750     00.000      0.0   02.660   460.0    0      0
    SREF       CREF       BREF      XMC      YMC      ZMC
    1.00000    1.00000    1.0000  0.00000  0.00    0.00
    DT         IREST      IFLAGTS  FMAX     IUNST     CFLTAU
    -05.000    0          000      5.0000   0        10.
    NGRID      NPLOT3D    NPRINT    NWREST    ICHK      I2D      NTSTEP      ITA
    1          1          1          1200     0         0         1          1
    NCG        IEM        IADVANCE  IFORCE   IVISC(I)  IVISC(J)  IVISC(K)
    2          0          0          001      0         0         07
    IDIM       JDIM       KDIM
    02         181       101
    ILAMLO     ILAMHI     JLAMLO     JLAMHI     KLAMLO     KLAMHI
    0          0          0          0          0          0
    INEWG      IGRIDC     IS         JS         KS         IE         JE         KE
    0          0          0          0          0          0         0         0
    IDIAG(I)   IDIAG(J)   IDIAG(K)  IFLIM(I)  IFLIM(J)  IFLIM(K)
    1          1          1          3          3          3
    IFDS(I)    IFDS(J)    IFDS(K)  RKAP0(I)  RKAP0(J)  RKAP0(K)
    1          1          1      0.3333  0.3333  0.3333
    GRID       NBCI0      NBCIDIM    NBCJ0      NBCJDIM    NBCK0      NBCKDIM    IOVRLP
    1          1          1          1          1          1         1         0
I0:  GRID     SEGMENT    BCTYPE     JSTA      JEND      KSTA      KEND      NDATA
    1          1          2005        0         0         0         0         4
        NBLP      DTHTX      DTHTY      DTHTZ
        1          -1.0       0.        0.
IDIM: GRID     SEGMENT    BCTYPE     JSTA      JEND      KSTA      KEND      NDATA
    1          1          2005        0         0         0         0         4
        NBLP      DTHTX      DTHTY      DTHTZ
        1          +1.0       0.        0.
J0:  GRID     SEGMENT    BCTYPE     ISTA      IEND      KSTA      KEND      NDATA
    1          1          1003        0         0         0         0         0
JDIM: GRID     SEGMENT    BCTYPE     ISTA      IEND      KSTA      KEND      NDATA
    1          1          1003        0         0         0         0         0
K0:  GRID     SEGMENT    BCTYPE     ISTA      IEND      JSTA      JEND      NDATA
    1          1          2004        0         0         0         0         2
        TWTYPE     CQ
        0.          0.
KDIM: GRID     SEGMENT    BCTYPE     ISTA      IEND      JSTA      JEND      NDATA
    1          1          1003        0         0         0         0         0
    MSEQ      MGFLAG    ICONSF      MTT      NGAM
    1          1          0          0          02
    ISSC      EPSSSC(1)  EPSSSC(2)  EPSSSC(3)  ISSR      EPSSSR(1)  EPSSSR(2)  EPSSSR(3)
    0          0.3      0.3      0.3          0          0.3      0.3      0.3
    NCYC      MGLEVG      NEMGL      NITFO
    1100      03          00          000
    MIT1      MIT2      MIT3      MIT4      MIT5      MIT6      MIT7      MIT8
    01        01        01        01        01        1        1        1
  1-1 BLOCKING DATA:
    NBLI
    0
  NUMBER  GRID      :      ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  NUMBER  GRID      :      ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
  PATCH SURFACE DATA:
    NINTER
    0
  PLOT3D OUTPUT:
  BLOCK IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
    1          0      0      0      0      0      0      0      0      0
IMOVIE
  0
  PRINT OUT:
  BLOCK IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
    1          0      1      1      1      0      0      0      1      1
CONTROL SURFACE:

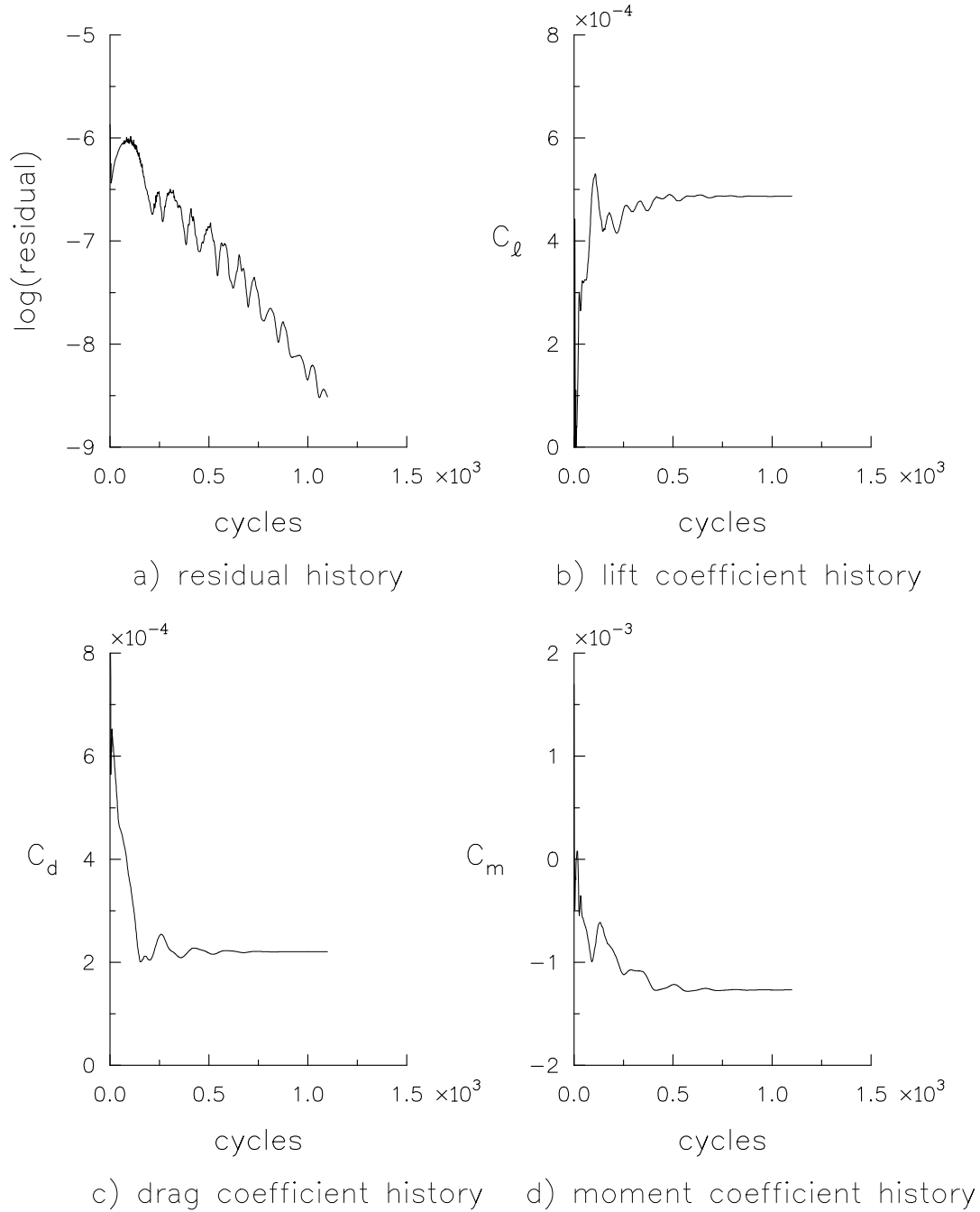
```

```

NCS
0
GRID ISTART IEND JSTART JEND KSTART KEND IWALL INORM

```

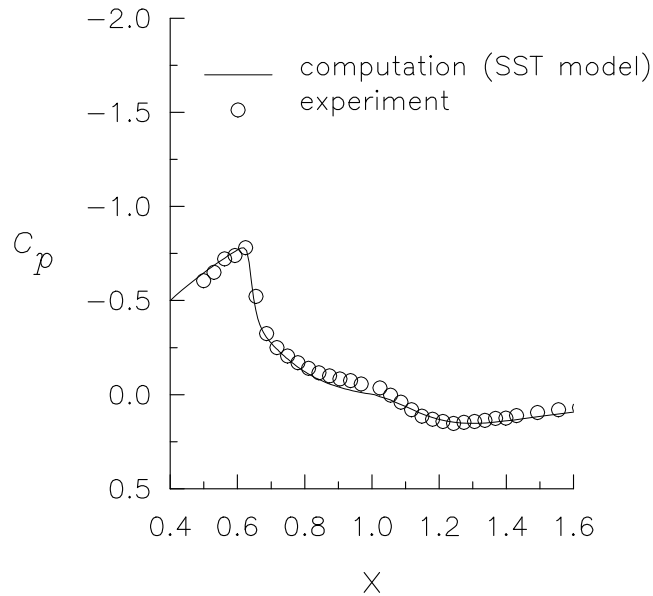
After running this test case, the convergence history plots shown in Figure 9-21 should be duplicated.



**Figure 9-21.** Residual and coefficient histories for axisymmetric bump flow case

$$M_\infty = 0.875, Re_{\tilde{L}_R} = 2.66 \times 10^6.$$

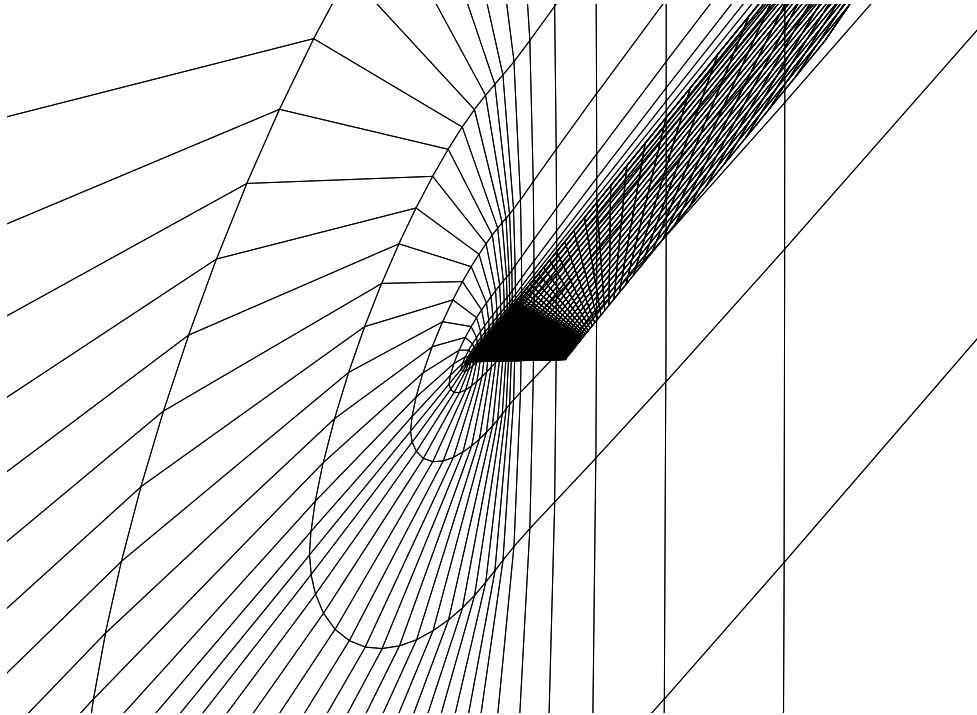
Also, a result such as that shown in Figure 9-22 should be obtained. In the figure, surface pressure coefficients are plotted along with experimental data for this case. The computational surface pressures can be obtained from file `cf13d.prout`. Experimental surface pressure coefficients from Bachalo et. al<sup>8</sup> are included with this test case for comparison purposes. The file is called `bumpcpdata.dat`.



**Figure 9-22.** Pressure coefficients for axisymmetric bump case  
 $M_{\infty} = 0.875$ ,  $Re_{\bar{L}_R} = 2.66 \times 10^6$ .

### 9.2.2 F-5 Wing

The inviscid flow over an F-5 wing is solved in this test case. The grid consists of a single grid zone with a C-H mesh topology and is composed of 210,177 points. The memory requirement for this example is 10.5 million words. A typical timing for this case is 984 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of September 1996). The wing surface grid and wake, as well as the plane of symmetry grid are illustrated in Figure 9-23.



**Figure 9-23.** Single zone F-5 wing surface grid and plane of symmetry grid.

Besides the CFL3D code the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
f5wing.inp	input for CFL3D
f5grid.dat	formatted wing section grid
f5wing_grid.f	grid converter

The steps for running this case on the YMP are as follows:

#### Step 1

Compile the grid converter code:

```
cft77 f5wing_grid.f
```

### Step 2

Link the grid converter object file:

```
segldr -o f5wing_grid f5wing_grid.o
```

### Step 3

Run the grid converter program to generate the 3-d volume grid (the binary file `f5wing.grd` will be output):

```
f5wing_grid
```

### Step 4

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 5

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < f5wing.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the `CFL3D` code:

```
make -f makecfl3d_cray
```

### Step 7

Run the `CFL3D` code:

```
cfl3d < f5wing.inp
```

The input file for this case is:

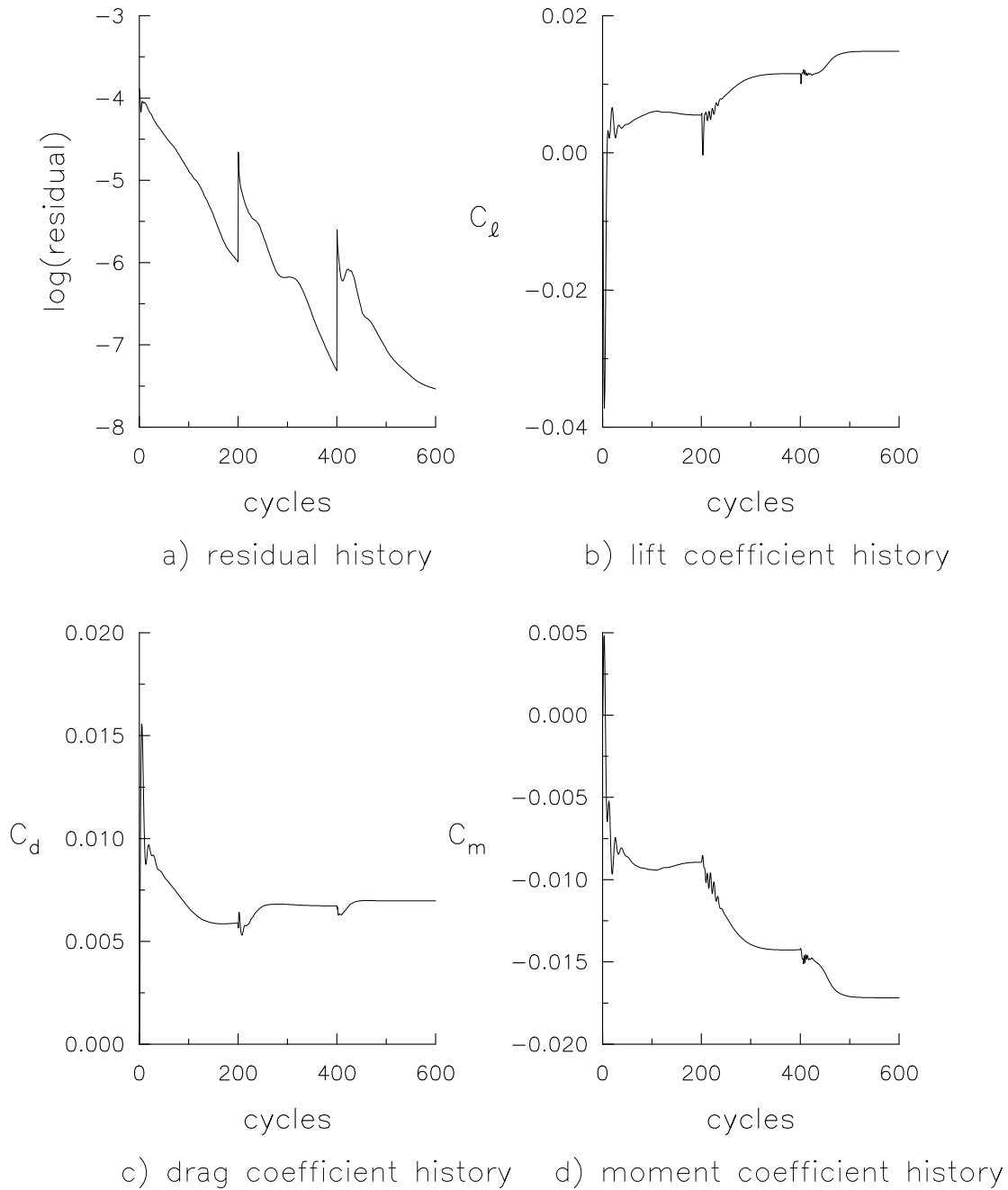
```
I/O FILES:
f5wing.grd
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
      F5 Wing, cfl3d type grid
      XMACH      ALPHA      BETA  REUE,MIL  TINF,DR  IALPH  IHIST
```

```

0.950    00.000    0.0    0.950    460.0    0    0
SREF      CREF      BREF      XMC      YMC      ZMC
1.00000  1.00000  1.0000  0.25000  0.00    0.00
DT        IREST      IFLAGTS      FMAX      IUNST      CFLTAU
-5.000    0    000    1.000    0    10.
NGRID     NPLOT3D     NPRINT     NWREST      ICHK      I2D      NTSTEP      ITA
1    0    0    100    0    0    1    1
NCG      IEM      IADVANCE      IFORCE      IVISC(I)      IVISC(J)      IVISC(K)
2    0    0    1    0    0    0
IDIM      JDIM      KDIM
33    193    33
ILAMLO    ILAMHI      JLAMHI      JLAMHI      KLAMLO      KLAMHI
00    00    000    000    0    0000
INEWG     IGRIDC      IS      JS      KS      IE      JE      KE
0    0    0    0    0    0    0    0
IDIAG(I)  IDIAG(J)  IDIAG(K)  IFLIM(I)  IFLIM(J)  IFLIM(K)
1    1    1    3    3    3
IFDS(I)   IFDS(J)   IFDS(K)   RKAP0(I)  RKAP0(J)  RKAP0(K)
1    1    1    .3333  .3333  .3333
GRID      NBCI0      NBCIDIM      NBCJ0      NBCJDIM      NBCK0      NBCKDIM      IOVRLP
1    1    1    1    1    4    1    0
IO:  GRID  SEGMENT  BCTYPE  JSTA  JEND  KSTA  KEND  NDATA
1    1    1001    0    0    0    0    0
IDIM: GRID  SEGMENT  BCTYPE  JSTA  JEND  KSTA  KEND  NDATA
1    1    1002    0    0    0    0    0
JO:  GRID  SEGMENT  BCTYPE  ISTA  IEND  KSTA  KEND  NDATA
1    1    1003    0    0    0    0    0
JDIM: GRID  SEGMENT  BCTYPE  ISTA  IEND  KSTA  KEND  NDATA
1    1    1003    0    0    0    0    0
K0:  GRID  SEGMENT  BCTYPE  ISTA  IEND  JSTA  JEND  NDATA
1    1    0    1    33    1    41    0
1    2    1005    1    21    41    153    0
1    3    0    21    33    41    153    0
1    4    0    1    33    153    193    0
KDIM: GRID  SEGMENT  BCTYPE  ISTA  IEND  JSTA  JEND  NDATA
1    1    1003    0    0    0    0    0
MSEQ      MGFLAG      ICONSF      MTT      NGAM
3    1    0    0    02
ISSC EPSSSC(1) EPSSSC(2) EPSSSC(3) ISSR EPSSSR(1) EPSSSR(2) EPSSSR(3)
0    0.3    0.3    0.3    0    0.3    0.3    0.3
NCYC      MGLEVG      NEMGL      NITFO
200    01    00    000
200    02    00    000
200    03    00    000
MIT1      MIT2      MIT3      MIT4      MIT5
01    01    01    01    01
01    01    01    01    01
01    01    01    01    01
1-1 BLOCKING DATA:
NBLI
2
NUMBER  GRID  :  ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
1    1    :  1    1    1    33    41    1    1    2
2    1    :  21   41    1    33    97    1    1    2
NUMBER  GRID  :  ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
1    1    :  1    193   1    33    153   1    1    2
2    1    :  21    153   1    33    97    1    1    2
PATCH SURFACE DATA:
NINTER
0
PLOT3D OUTPUT:
GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
MOVIE
0
PRINT OUT:
GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
CONTROL SURFACE:
NCS
0
GRID ISTART IEND JSTART JEND KSTART KEND IWALL INORM

```

After this test case is run, the convergence history, found in file `cfl3d.res`, should look like that plotted in Figure 9-24. The two sharp spikes in the residual history are at the iterations at which the grid levels change in the mesh sequencing procedure.



**Figure 9-24.** Convergence histories for single grid F-5 wing case;  
 $\alpha = 0.0$ ,  $M_\infty = 0.95$ .



### 9.2.3 Onera M-6 Wing

In this case, a turbulent Navier-Stokes computation is performed over the Onera M-6 wing, on a coarse grid, using a grid in PLOT3D-type format is performed. The grid consists of a single grid zone with a C-O mesh topology and is composed of 41,225 points. (Keep in mind that one needs a grid at least double this size in each direction, e.g.  $193 \times 49 \times 33$  or larger, to actually resolve the flow. A coarser grid is used here to shorten the test run.) The wing surface grid and wake, as well as the plane of symmetry grid are illustrated in Figure 9-25. The memory requirement for this example is 3.4 million words. A typical timing for this case is 453 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of September 1996).

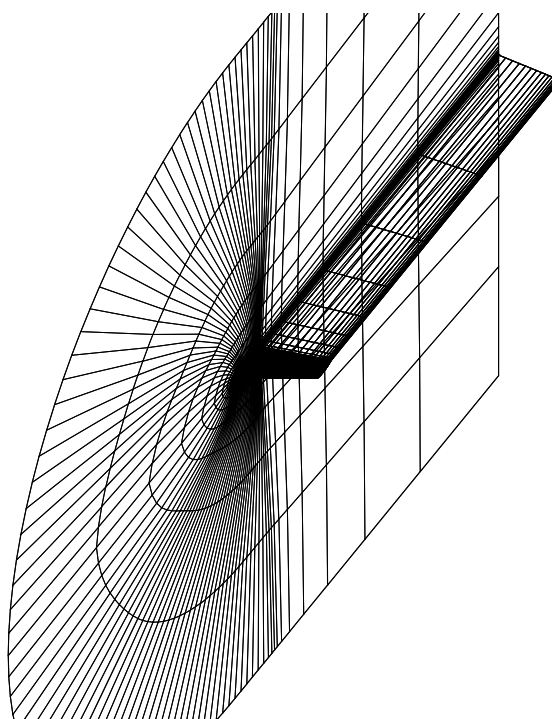


Figure 9-25. Single zone Onera wing surface grid and plane of symmetry grid.

The viscous direction in this PLOT3D-formatted grid is taken as the  $j$  direction rather than the  $k$  direction as generally recommended. (Due to the order in which CFL3D approximately factors the three index directions, the CFL3D code is usually most efficient when the primary viscous direction is taken as the  $k$  direction.) In this case, the convergence is not hurt by the altered directionality. (In some cases, however, it can be!) Note, however, that this case *is* more efficient (CPU timewise) when run on a vector machine with  $k$  as the viscous direction, due to the distribution of individual  $i$ ,  $j$ ,  $k$  index lengths and the way the code is vectorized. For this case, the difference on Sabre is a factor of 17% (with a CFL3D-type  $k$  viscous grid, the code runs in 374 seconds as opposed to 453 sec-

onds). It is possible to duplicate this result by changing the hard-wired parameter `iplot3d` to 0 in the `form2bin.f` file and using the input file `oneram6.inp_cfl3d` instead of `oneram6.inp_p3d`. This exercise will demonstrate the differences between PLOT3D-type and CFL3D-type grids, as well as the corresponding differences in the input files.

Besides the CFL3D code the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
<code>oneram6.inp_p3d</code>	input for CFL3D
<code>m6_i97.fmt_p3d</code>	formatted grid
<code>form2bin.f</code>	grid converter

The steps for running this case on the YMP are as follows:

### Step 1

Compile the grid converter code:

```
cft77 form2bin.f
```

### Step 2

Link the grid converter object file:

```
segldr -o form2bin form2bin.o
```

### Step 3

Run the grid converter program to generate the 3-d volume grid (the binary file `m6_i97.grd_p3d` will be output):

```
form2bin
```

### Step 4

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 5

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < oneram6.inp_p3d
```

### Step 6

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

### Step 7

Run the CFL3D code:

```
cfl3d < oneram6.inp_p3d
```

The input file for this case is:

```
I/O FILES:
m6_i97.grd_p3d
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin

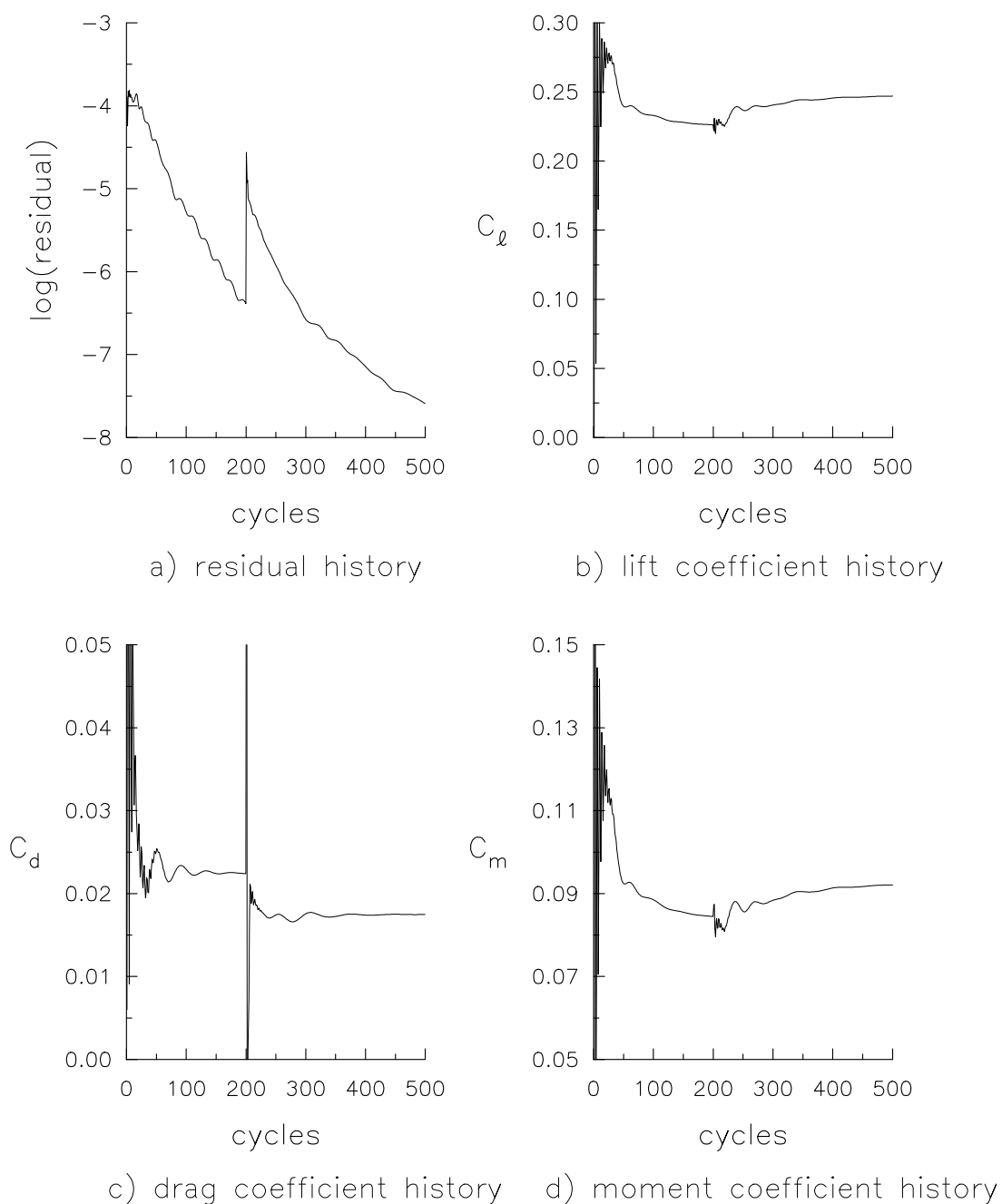
ONERA M6 Wing, plot3d type grid, coarse grid
  XMACH      ALPHA      BETA  REUE,MIL    TINF,DR      IALPH      IHIST
    0.8400    03.060      0.0    21.660     540.0         1         0
  SREF      CREF      BREF      XMC      YMC      ZMC
    0.53080    1.00000    3.9249    0.00000    0.00      0.00
    DT      IREST      IFLAGTS      FMAX      IUNST      CFLTAU
   -5.000      0         000      1.000      0         10.
  NGRID  NPLOT3D  NPRINT  NWREST      ICHK      I2D      NTSTEP      ITA
    -1         1         0         100      0         0         0         1
    NCG      IEM  IADVANCE  IFORCE  IVISC(I)  IVISC(J)  IVISC(K)
      2         0         0         10         0         5         0
    IDIM      JDIM      KDIM
     97         25         17
  ILAMLO  ILAMHI  JLAMLO  JLAMHI  KLAMLO  KLAMHI
    00         00         000      000      0      0000
  INEWG  IGRIDC  IS      JS      KS      IE      JE      KE
    0         0         0         0         0         0         0         0
  IDIAG(I)  IDIAG(J)  IDIAG(K)  IFLIM(I)  IFLIM(J)  IFLIM(K)
    1         1         1         3         3         3
  IFDS(I)  IFDS(J)  IFDS(K)  RKAP0(I)  RKAP0(J)  RKAP0(K)
    1         1         1      .3333      .3333      .3333
  GRID      NBCI0  NBCIDIM  NBCJ0  NBCJDIM  NBCK0  NBCKDIM  IOVRLP
    1         1         1         3         1         1         1         0
I0:  GRID  SEGMENT  BCTYPE  JSTA  JEND  KSTA  KEND  NDATA
    1         1         1003      0         0         0         0         0
IDIM: GRID  SEGMENT  BCTYPE  JSTA  JEND  KSTA  KEND  NDATA
    1         1         1003      0         0         0         0         0
J0:  GRID  SEGMENT  BCTYPE  ISTA  IEND  KSTA  KEND  NDATA
    1         1         0         1         13         0         0         0
    1         2      2004      13         85         1         17         2
      TWTYPE      CQ
        0.         0.
    1         3         0         85         97         0         0         0
JDIM: GRID  SEGMENT  BCTYPE  ISTA  IEND  KSTA  KEND  NDATA
    1         1         1003      0         0         0         0         0
K0:  GRID  SEGMENT  BCTYPE  ISTA  IEND  JSTA  JEND  NDATA
    1         1         1001      0         0         0         0         0
KDIM: GRID  SEGMENT  BCTYPE  ISTA  IEND  JSTA  JEND  NDATA
    1         1         0         0         0         0         0         0
  MSEQ      MGFLAG  ICONSF      MTT      NGAM
    2         1         0         0         02
  ISSC  EPSSSC(1)  EPSSSC(2)  EPSSSC(3)      ISSR  EPSSSR(1)  EPSSSR(2)  EPSSSR(3)
    0         0.3         0.3         0.3         0         0.3         0.3         0.3
  NCYC      MGLEVG      NEMGL      NITFO
```

```

      200      02      00      000
      300      02      00      000
      MIT1    MIT2    MIT3    MIT4    MIT5
        01      01      01      01      01
        01      01      01      01      01
1-1 BLOCKING DATA:
  NBLI
    2
NUMBER  GRID      :   ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
   1      1      :     1    1    1    13    1    17    1    3
   2      1      :     1    1    17   49    25   17    1    2
NUMBER  GRID      :   ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
   1      1      :    97    1    1    85    1    17    1    3
   2      1      :    97    1    17   49    25   17    1    2
PATCH SURFACE DATA:
  NINTER
    0
PLOT3D OUTPUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
   1      0      0      0      0    0      0    0      0      0
MOVIE
  0
PRINT OUT:
  GRID IPTYPE ISTART  IEND  IINC JSTART  JEND  JINC KSTART  KEND  KINC
CONTROL SURFACE:
NCS
  0
  GRID ISTART  IEND  JSTART  JEND  KSTART  KEND  IWALL  INORM

```

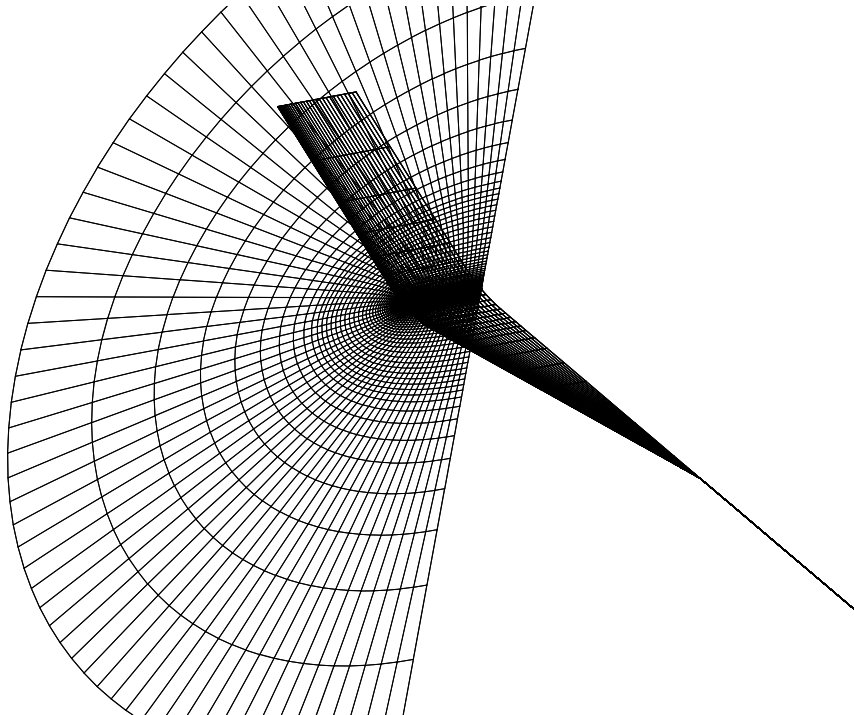
After this test case is run, the convergence histories, found in file `cfl3d.res`, should look like those plotted in Figure 9-26. The sharp spikes in the plots indicate the iteration at which the grid level changes in the mesh sequencing process.



**Figure 9-26.** Convergence histories for single grid Onera wing case;  
 $\alpha = 3.06$ ,  $M_\infty = 0.84$ .

### 9.2.4 Delta Wing

The laminar flow over a  $75^\circ$  swept delta wing is solved in this test case. The grid consists of a single grid zone with 156,325 points. (Note that this grid is coarser than what one would normally use to resolve this flow.) The memory requirement for this example is 8.0 million words. A typical timing for this case is 2236 CPU seconds on a CRAY YMP (NASA LaRC's Sabre as of September 1996). The surface grid ( $k = 1$ ) and a trailing edge grid plane are shown in Figure 9-27.



**Figure 9-27.** Single zone delta wing surface grid and trailing edge plane grid.

Besides the CFL3D code the following files are needed to run this test case:

<u>File</u>	<u>Description</u>
delta.inp	input for CFL3D
delta.fmt	formatted grid
form2bin.f	grid converter

The steps for running this case on the YMP are as follows:

#### Step 1

Compile the grid converter code:

```
cft77 form2bin.f
```

### Step 2

Link the grid converter object file:

```
segldr -o form2bin form2bin.o
```

### Step 3

Run the grid converter program to generate the 3-d volume grid (the binary file `delta.bin` will be output):

```
form2bin
```

### Step 4

Use the makefile to compile, link, and create the executable for the `precfl3d` code (be sure `precfl.h` is in the current directory):

```
make -f makeprecfl3d_cray
```

### Step 5

Run the `precfl3d` code (the `cflx.h` files will be output):

```
precfl3d < delta.inp
```

### Step 6

Use the makefile to compile, link, and create the executable for the CFL3D code:

```
make -f makecfl3d_cray
```

### Step 7

Run the CFL3D code:

```
cfl3d < delta.inp
```

The input file for this case is:

```
I/O FILES
delta.bin
plot3dg.bin
plot3dq.bin
cfl3d.out
cfl3d.res
cfl3d.turres
cfl3d.blomax
cfl3d.out15
cfl3d.prout
cfl3d.out20
ovrlp.bin
patch.bin
restart.bin
75 Degree Swept Delta Wing - 37x65x65 - Laminar
  XMACH      ALPHA      BETA  REUE,MIL  TINF,DR      IALPH      IHIST
```

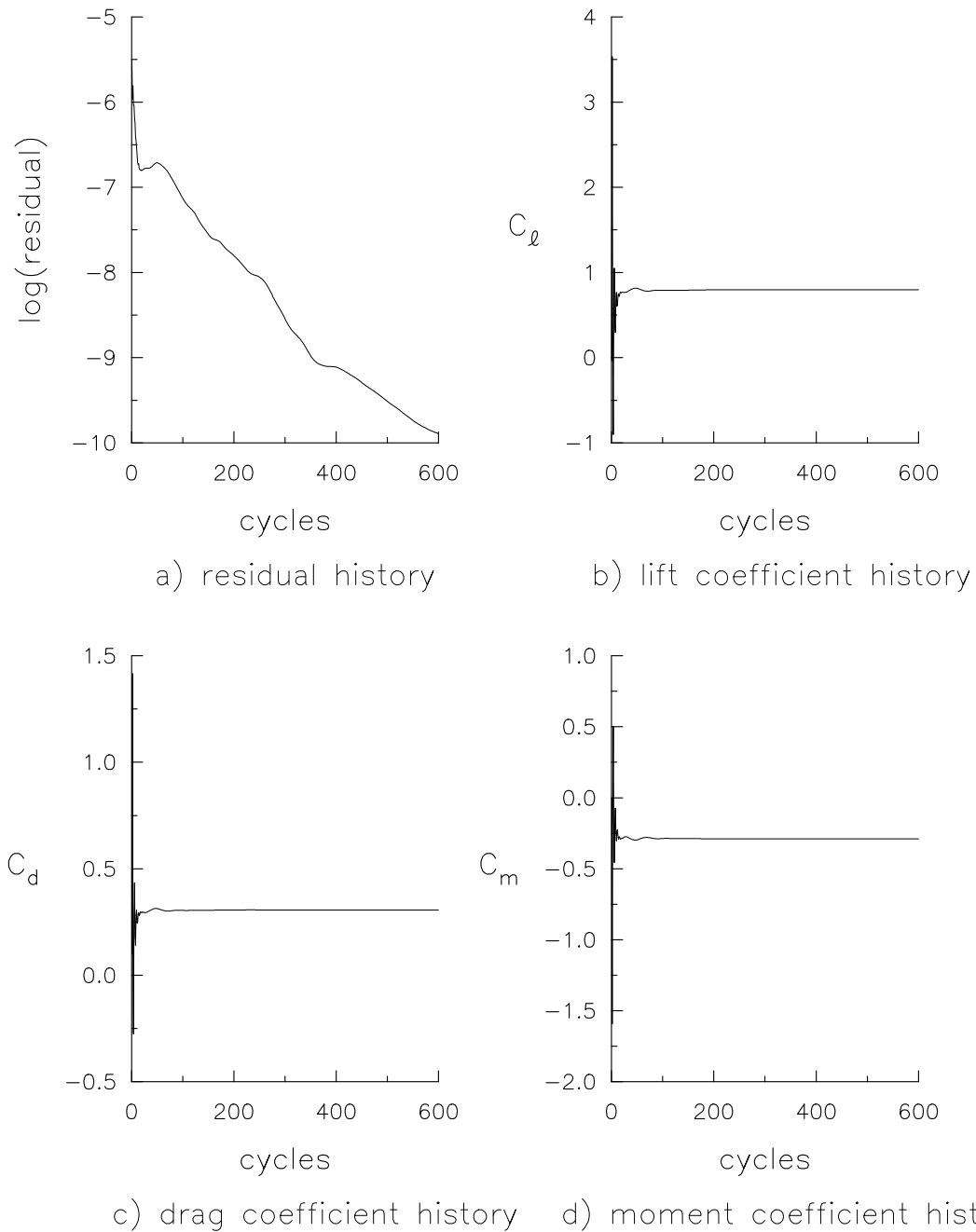
```

0.300    20.500    0.0    0.500    460.0    0    0
SREF      CREF      BREF      XMC      YMC      ZMC
.13398    1.00000    .53590    0.25000    0.
DT      IREST      IFLAGTS      FMAX      IUNST      CFLTAU
-10.0    0    0    1.0    0    10.
NGRID    NPL0T3D    NPRINT    NWREST      ICHK      I2D      NTSTEP      ITA
1    1    0    0100    0    0    1    1
NCG      IEM      IADVANCE      IFORCE      IVISC(I)      IVISC(J)      IVISC(K)
2    0    0    001    0    0    1
IDIM      JDIM      KDIM
37      65      65
ILAMLO    ILAMHI    JLAMLO    JLAMHI    KLAMLO    KLAMHI
00      00      000    000    0    0000
INEWG      IGRIDC      IS      JS      KS      IE      JE      KE
0    0    0    0    0    0    0    0
IDIAG(I)  IDIAG(J)  IDIAG(K)  IFLIM(I)  IFLIM(J)  IFLIM(K)
1    1    1    4    4    4
IFDS(I)   IFDS(J)   IFDS(K)   RKAP0(I)  RKAP0(J)  RKAP0(K)
1    1    1    .3333  .3333  .3333
GRID      NBCI0    NBCIDIM    NBCJ0    NBCJDIM    NBCK0    NBCKDIM    IOVRLP
1    1    1    1    1    3    1    0
I0:  GRID  SEGMENT  BCTYPE  JSTA  JEND  KSTA  KEND  NDATA
1    1    1003  1    65  1    65  0
IDIM: GRID  SEGMENT  BCTYPE  JSTA  JEND  KSTA  KEND  NDATA
1    1    1003  1    65  1    65  0
J0:  GRID  SEGMENT  BCTYPE  ISTA  IEND  KSTA  KEND  NDATA
1    1    1001  1    37  1    65  0
JDIM: GRID  SEGMENT  BCTYPE  ISTA  IEND  KSTA  KEND  NDATA
1    1    1001  1    37  1    65  0
K0:  GRID  SEGMENT  BCTYPE  ISTA  IEND  JSTA  JEND  NDATA
1    1    1011  1    9    1    65  0
1    2    2004  9    25  1    65  2
      TWTYPE      CQ
      -1.      0.
KDIM: GRID  SEGMENT  BCTYPE  ISTA  IEND  JSTA  JEND  NDATA
1    1    1003  1    37  1    65  0
MSEQ      MGFLAG      ICONSF      MTT      NGAM
1    1    1    0    02
ISSC EPSSSC(1) EPSSSC(2) EPSSSC(3) ISSR EPSSSR(1) EPSSSR(2) EPSSSR(3)
1    0.3    0.3    0.3    0    0.3    0.3    0.3
NCYC      MGLEVG      NEMGL      NITFO
600      03      00      0000
MIT1      MIT2      MIT3      MIT4      MIT5      MIT6      MIT7      MIT8
01      01      01      01      01      1    1    1
1-1 BLOCKING DATA:
NBLI
1
NUMBER  GRID  :  ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
1    1    :  25    1    1    37    33    1    1    2
NUMBER  GRID  :  ISTA  JSTA  KSTA  IEND  JEND  KEND  ISVA1  ISVA2
1    1    :  25    65  1    37    33    1    1    2
PATCH SURFACE DATA:
NINTER
0
PLOT3D OUTPUT:
GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
1    0    0    0    0    0    0    0    0    0
MOVIE
0
PRINT OUT:
GRID IPTYPE ISTART IEND IINC JSTART JEND JINC KSTART KEND KINC
CONTROL SURFACE:
NCS
0
GRID ISTART IEND JSTART JEND KSTART KEND IWALL INORM

```

After this test case is run, the convergence histories, found in file `cf13d.res`, should look like those plotted in Figure 9-28.





**Figure 9-28.** Convergence histories for single grid delta wing case;  $\alpha = 20.5$ .

