

# APPLICATION OF ATTOSECOND TECHNIQUES TO CONDENSED MATTER SYSTEMS

DISSERTATION

Presented in Partial Fulfillment of the Requirements for the Degree Doctor of  
Philosophy in the Graduate School of The Ohio State University

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2020

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# Chapter 1

## INTRODUCTION

### 1.1 Ultrafast Dynamics in Condensed Matter Systems

timescales and processes in solids

### 1.2 Attosecond Transient Absorption Spectroscopy (ATAS)

**why are you doing it with HHG?**

include figure of pulse duration vs photon energy, showing different light sources (synchrotrons, HHG sources, XFEL, etc.) tie this into the timescales necessary to probe condensed matter physics.

#### 1.2.1 overview of the technique

references [25]

**induced dipole interpretation**

**population transfer and probing interpretation**

**comparison of absorptive and reflective measurements**

In a transient absorption experiment, we measure the transmission  $T$  of a sample in response to excitation by an external field. Generally speaking,  $T$  depends on both parts of the complex refractive index:  $\tilde{n} = n + ik$ . However, in a normal transmission geometry it turns out that the contribution of  $\text{Im}(\tilde{n})$  dominates the measured signal, and to a good approximation the role of  $\text{Re}(\tilde{n})$  can be ignored. Note that in a non-normal reflection geometry, both parts of  $\tilde{n}$  make significant contributions to the measured signal. In the following discussion we will analyze the Fresnel equations to see why this is the case. This section will draw from arguments made in reference [22].

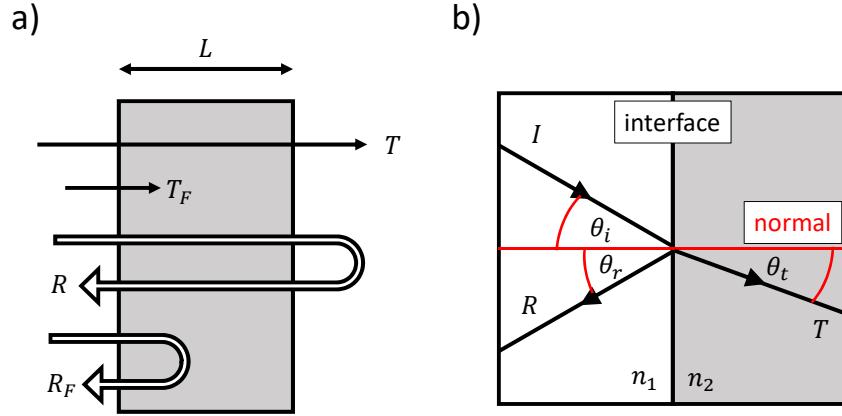


Figure 1.1: Normal and non-normal incident geometries. **a)** Normal incidence geometry showing Fresnel coefficients  $R_F$ ,  $T_F$  for interfaces and total transmission  $T$  and reflectance  $R$  for a slab of thickness  $L$ . Figure recreated from [22]. **b)** Non-normal geometry showing definitions of angles  $\theta_i$ ,  $\theta_r$  and  $\theta_t$  with respect to each interface.

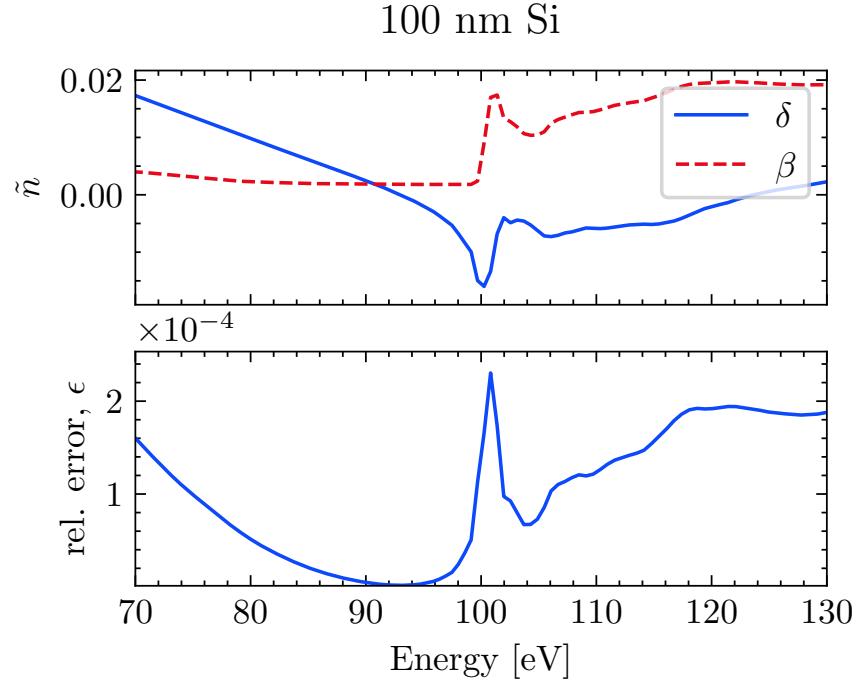


Figure 1.2: Consequences of ignoring the real part of  $\tilde{n}$  when calculating the transmission  $T$  of a thin sample. Top panel: complex refractive index of silicon using the notation from Eq. (1.1). The Si  $L$ -edge absorption feature is visible near 100 eV. Data from [12]. Bottom panel: relative error in  $T$ , as defined in Eq. (1.8), introduced by ignoring the contribution of  $\text{Re}(\tilde{n})$ . An infinite number of bounces (e.g., Eq. (1.6)) is assumed.

First, we consider the normal geometry shown in the left panel of Fig. 1.1. We write the complex index of refraction in the following form:

$$\begin{aligned}\tilde{n} &= n - ik \\ &= (1 - \delta) - i\beta\end{aligned}\tag{1.1}$$

The Fresnel coefficients  $R_F$  and  $T_F$  describe the interface reflectance and transmittance and depend on both parts of the complex index  $\tilde{n}$ . For normal incidence, they are:

$$\begin{aligned}R_F &= \left| \frac{n - ik - 1}{n - ik + 1} \right|^2 \\ T_F &= \frac{4n}{|n - ik + 1|^2}\end{aligned}\tag{1.2}$$

Absorption in the bulk is described via the absorption length  $\alpha$ :

$$\alpha = 4\pi k/\lambda\tag{1.3}$$

Ignoring interface effects, the transmisison through the bulk is:

$$T_{\text{bulk}} = \exp(-\alpha L)\tag{1.4}$$

Note that  $\alpha$  and  $T_{\text{bulk}}$  only depend on  $k$ .

The total reflectance  $R$  and transmission  $T$  are the result of interface effects plus bulk effects. We must consider the case where the detected light is the result of multiple reflections within the sample. Neglecting interference, we consider the case of  $2N$  bounces where the laser's coherence length is less than the thickness of the bulk. In this case, the sum is incoherent with the expressions for  $T$  and  $R$  given by:

$$\begin{aligned}R &= R_F + R_F T_F^2 T_{\text{bulk}}^2 \sum_{m=0}^N [R_F T_{\text{bulk}}]^{2m} \\ T &= T_F^2 T_{\text{bulk}} \sum_{m=0}^N [R_F T_{\text{bulk}}]^{2m}\end{aligned}\tag{1.5}$$

For the case of an infinite number of bounces, Eq. (1.5) simplifies to:

$$\begin{aligned}R &= R_F + \frac{R_F T_F^2 T_{\text{bulk}}^2}{1 - R_F^2 T_{\text{bulk}}^2} \\ T &= \frac{T_F^2 T_{\text{bulk}}}{1 - R_F^2 T_{\text{bulk}}^2},\end{aligned}\tag{1.6}$$

whereas if only a single bounce occurs, Eq. (1.5) reduces to:

$$\begin{aligned} R &= R_F + R_F T_F^2 T_{\text{bulk}}^2 \\ T &= T_F^2 T_{\text{bulk}} \end{aligned} \quad (1.7)$$

We now consider the fractional error introduced by ignoring the interface effects described by  $T_F$  and  $R_F$ . That is, what would happen if we assume that the interfaces have no effect on the transmitted intensity? We introduce the relative error  $\epsilon$  made by ignoring the Fresnel coefficients of Eq. (1.6):

$$\epsilon \equiv \frac{T_{\text{bulk}}}{T} - 1 \quad (1.8)$$

As an example, consider a 100 nm thick Si sample measured in transmission near the Si  $L$ -edge (about 100 eV), as shown in Fig. 1.2. The relative error is in the range of one part in  $10^4$  to  $10^5$ , well below our experimental detection limit. Silicon was chosen due to its data availability above and below the absorption edge, but this behavior should hold for all materials in normal transmission.

The real part of the complex index becomes important when the sample isn't normal to the beam, as shown in the right panel of Fig. 1.1. In this case, the Fresnel equations are a bit messier:

$$\begin{aligned} R_s &= \left| \frac{\tilde{n}_1 \cos \theta_i - \tilde{n}_2 \cos \theta_t}{\tilde{n}_1 \cos \theta_i + \tilde{n}_2 \cos \theta_t} \right|^2 \\ R_p &= \left| \frac{\tilde{n}_1 \cos \theta_t - \tilde{n}_2 \cos \theta_i}{\tilde{n}_1 \cos \theta_t + \tilde{n}_2 \cos \theta_i} \right|^2 \end{aligned} \quad (1.9)$$

$$T_s = 1 - R_s$$

$$T_p = 1 - R_p$$

Here, the subscripts  $s$  and  $p$  denote the polarization relative to the surface normal. For a sample in vacuum,  $\tilde{n}_1 = 1$  and  $\tilde{n}_2$  is the index of the sample. We can extract the relevant physics without any additional manipulation of Eq. (1.9). Right away, we can see that unlike Eq. (1.2), Eq. (1.9) is symmetric in the real and imaginary parts of the sample's complex index,  $\tilde{n}_2$ . In the limit of a thick slab, ( $L \gg \alpha$ ), the light is attenuated before it can reflect off the back surface and we have  $T \rightarrow 0$  and  $R \rightarrow R_{s,p}$ . That is, the only contributions to the reflected intensity are from the interface and possibly the sample volume within  $z \approx 1/\alpha$  of the interface. As a result, both parts of  $\tilde{n}_2$  will make significant contributions to the reflected intensity. This geometry is common in transient reflection-absorption experiments [5, 17].

complex refractive index

sample requirements and preparation

pointing stability (in reflection, sample is an XUV optic)

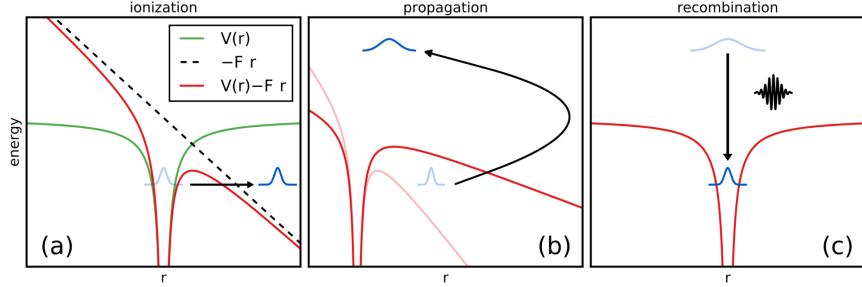


Figure 1.3: 3 step model from S. Schoun's thesis[26].

### 1.2.2 previous work

what is state of the art?

previous work in condensed matter (Si, Ge, Si-Ge, etc)

motivation for long-wavelength studies in condensed matter

### 1.2.3 physical observables in ATAS

limited k-space information (requires single crystal)

transmission geometry measures imaginary and not the real part of n

### 1.2.4 interpretation of experimental data

## 1.3 High Harmonic Generation (HHG)

### 1.3.1 three-step model

single atom response

cycle-averaged quiver energy

$$U_p = \frac{q_e^2 F_0^2}{4m_e \omega^2} \propto I_0 \lambda^2 \quad (1.10)$$

cutoff energy:

$$\omega_{cutoff} = I_p + 3.17 U_p \quad (1.11)$$

semi-classical three step model

### 1.3.2 Phase Matching

macroscopic response

things i need to talk about:

$$\Delta k = \Delta k_{atomic} + \Delta k_{plasma} + \Delta k_{Gouy} + \Delta k_{dipole} \quad (1.12)$$

### 1D phase matching model

for the  $q^{th}$  harmonic, the number  $N_{out}$  of photons emitted on axis per unit time and of area is proportional to [6]:

$$\rho^2 A_q^2 \frac{4L_{abs}^2}{1 + 4\pi^2(L_{abs}^2/L_{coh}^2)} \left[ 1 + \exp\left(-\frac{L_{med}}{L_{abs}}\right) - 2 \exp\left(\frac{\pi L_{med}}{L_{coh}}\right) \exp\left(-\frac{L_{med}}{2L_{abs}}\right) \right] \quad (1.13)$$

where  $L_{coh} = \pi/\Delta k$  is the coherence length ( $\Delta k = k_q - qk_0$ )

- critical phase matching pressure (kazamias?)

# Chapter 2

## EXPERIMENTAL APPARATUS

### 2.1 Introduction

what will be discussed in this chapter?

- the need for high vacuum
- all about XUV optics
- recombination mirror (hole mirror)
- modular design for beamline
- HHG gas sources (nozzle, LPC, HPC, pulsed valve)
- description of optical path
- overview of TABLE, including main parts of the beamline. refer the reader to Steve's dissertation [13] for details on the target chamber and photon spectrometer

### 2.2 Laser System

#### 2.2.1 Spitfire and TOPAS

We use a commercial mid-IR laser system (Spectra Physics Spitfire ACE), which delivers 12 mJ of 800 nm light at a variable 100 – 1,000 Hz repetition rate with a 60 fs FMHM pulse duration. This system utilizes the chirped pulse amplification (CPA) technique to amplify the pulse energy from a weak seed pulse. In this scheme, a low energy femtosecond seed pulse is stretched in time, amplified and compressed [30]. As such, the Spitfire consists of an oscillator, a grating stretcher, a regenerative amplifier, a single-pass amplifier and a grating compressor.

The laboratory layout is shown in Fig. 2.2. All of the lasers in the DiMauro research group are located in a centralized laser bay, where walking traffic is kept to a minimum and

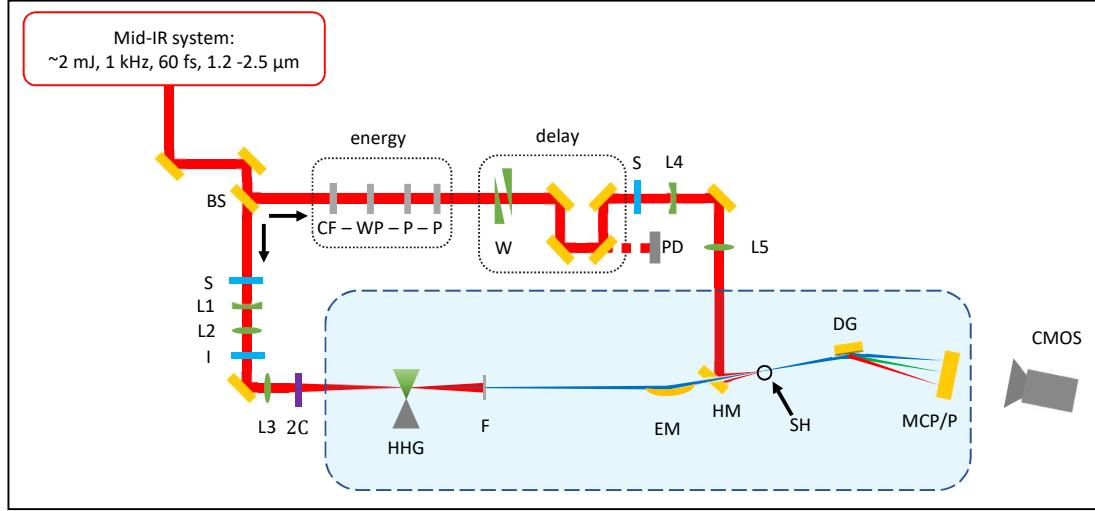


Figure 2.1: Schematic of the Transient Absorption BeamLine (TABLE). Blue shaded region represents vacuum. BS: beam splitter, S: computer-controlled shutter, L: lens, I: iris, 2C: optics for two-color generation, HHG: high harmonic generation, F: metallic filter, EM: ellipsoidal mirror, HM: hole mirror, SH: sample holder, CF: long-pass color filter, WP:  $\lambda/2$  waveplate, P: wire-grid polarizer, W: delay wedges, PD: photodiode and associated optics, DG: dispersive grating, MCP/P: micro-channel plate and phosphor.

air quality is nominally higher than the surrounding laboratory areas. This minimizes air disturbances around the laser systems and reduces the accumulation of dirt and debris on their optics. Experiments are performed in the adjacent target rooms which contain the vacuum systems and other experimental equipment. The Spitfire shares the laser bay with two home-built ultrafast laser systems (the “2 micron system” and the “4 micron system”, not shown in Fig. 2.2), as well as some laser development. The Spitfire is positioned so that its light can be directed to either the East, South or Southeast Target Rooms, depending on the needs of the researchers. To reduce air currents, welding curtains surround each optical table in the laser bay. When propagating the beam to target rooms, the beam path is enclosed in PVC tubing to reduce air curtains and to increase user safety. A  $\text{CaF}_2$  window is used to block air currents between the laser bay and the target rooms.

Referring to Fig. 2.2, the transient absorption beamline (TABLE) is located in the southeast target room. Laser light from the amplifier must be propagated uncompressed to the target room to avoid nonlinear propagation effects. To understand why we can compute the  $B$  integral, which provides a measure of the nonlinear phase accumulated during propagation:

$$B = \frac{2\pi}{\lambda} \int n_2 I(z) dz \quad (2.1)$$

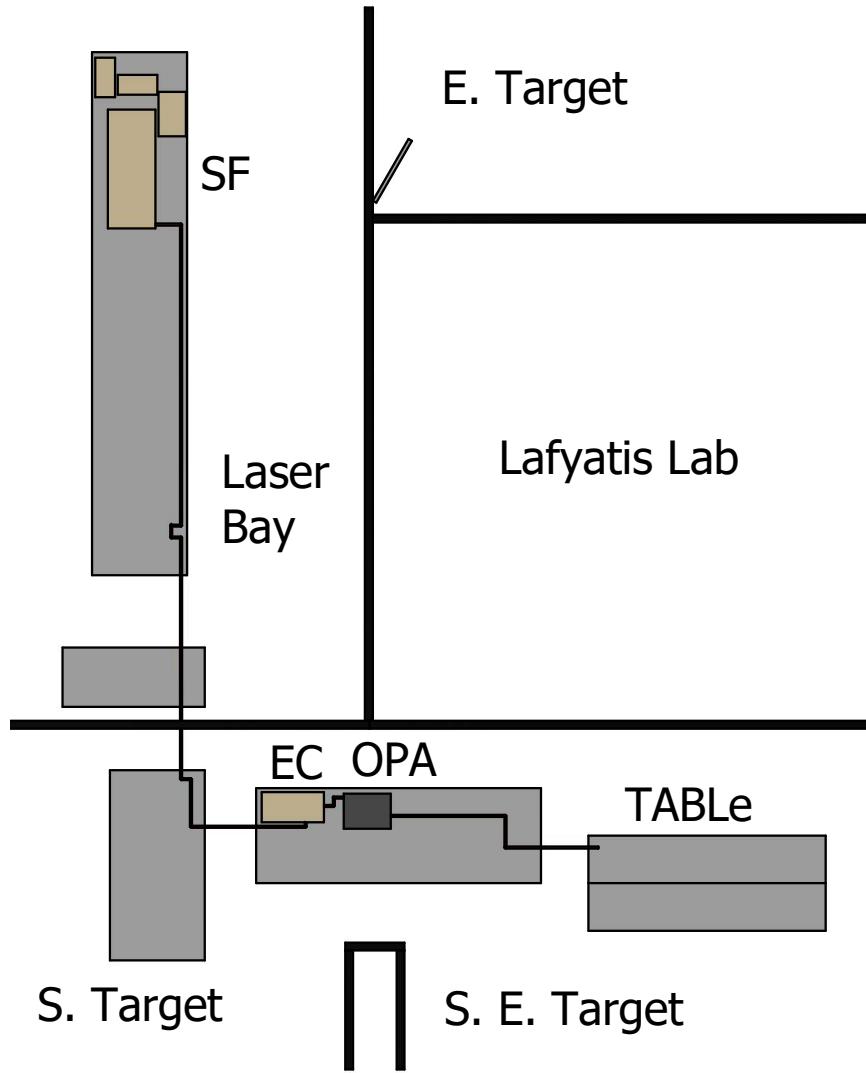


Figure 2.2: Block diagram of part of the DiMauro lab showing the laser path from the laser bay to the southeast target room. Other experiments and laser systems are omitted for visual clarity. Optical tables are represented as gray boxes. SF: Spitfire laser system consisting of a MaiTai oscillator, two Empower pump lasers, interal stretcher, amplifier & internal compressor (bypassed for this experiment); EC: Spitfire external compressor; OPA: Light Conversion HE TOPAS Prime; TABLE: 4' × 10' optical table for the transient absorption beamline.

The distance between the amplifier and the south target room is approximately 13 meters. For a propagation distance of 13 meters, a beam radius of 0.8 cm, a pulse energy of 12 mJ and a FWHM pulse duration of 60 fs,  $B = 1.53$ , which indicates that nonlinear propagation effects are significant [33]. On the other hand, the uncompressed pulse has slightly higher pulse energy (15 mJ, owing to the 20% transmission losses of the compressor), but a significantly longer pulse duration ( $\sim 10^3$  longer), resulting in a negligible  $B$  value. For this reason, we use an external compressor centrally located between the south and south east target rooms, as shown in Fig. 2.2. This positioning allows the Spitfire to be used for either the TABLE in the southeast target room or the RABBITT apparatus [4, 11, 18] in the south target room (not shown in Fig. 2.2). The external compressor has an efficiency of 80%, giving us 12 mJ of 800 nm light with a FWHM pulse duration of 60 fs at the entrance of the OPA.

The output of the external compressor is sent into a commercial optical parametric amplifier (Light Conversion HE TOPAS Prime), which converts the 800 nm light to longer wavelengths ranging from 1.2 to 2.2  $\mu\text{m}$  while roughly maintaining pulse duration. To minimize nonlinear propagation effects, the TOPAS is located immediately after the external compressor with only two steering mirrors between the external compressor and the TOPAS. Details of the TOPAS operation, alignment and optimization can be found in the user manual. Briefly, it utilizes a nonlinear process called optical parametric amplification (OPA), where the 800 nm pump ( $p$ ) is converted into two longer wavelength photons (the signal  $s$  and the idler  $i$ ) that obey the following energy conservation relation:

$$\frac{1}{\lambda_p} = \frac{1}{\lambda_s} + \frac{1}{\lambda_i} \quad (2.2)$$

Inside the TOPAS, a white light generation process creates a broadband seed pulse, followed by three stages of amplification in BBO crystals. The signal  $\lambda_s$  and idler  $\lambda_i$  wavelengths are determined by phase matching conditions inside the nonlinear crystals, which is controlled by the crystal angle relative to the incident laser light. The BBO crystals are mounted on encoded motorized stages, and the entire system is computer controlled and calibrated so the crystal angles change when the user specifies the desired wavelength. The conversion efficiency of the TOPAS ranges from 40 to 50 % (combined signal + idler pulse energy of 5 - 6 mJ), depending on the degree of optical alignment into the TOPAS and the desired wavelength. During the amplification process, all three beams are collinear. After the final amplification stage, a dichroic mirror inside the TOPAS separates the depleted 800 nm pump from the signal + idler, and a wavelength separator immediately outside the TOPAS splits the signal from the idler.

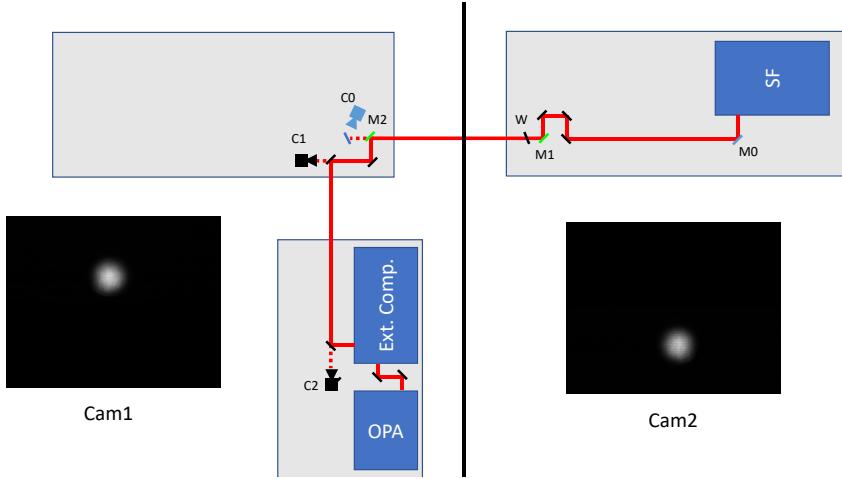


Figure 2.3: Implementation of active stabilization systems between the amplifier in the laser bay and external compressor in the target room (not to scale). M0 & C0 are the motorized mirror and digital camera used for the single-point correction scheme. M1, M2, C1 & C2 are the motorized mirrors and cameras used for the two-point correction scheme. W is an uncoated CaF<sub>2</sub> window used to reduce air currents between the laser bay and target rooms. PVC tubes are omitted for visual clarity. Inset images show the attenuated beam as imaged by C1 & C2.

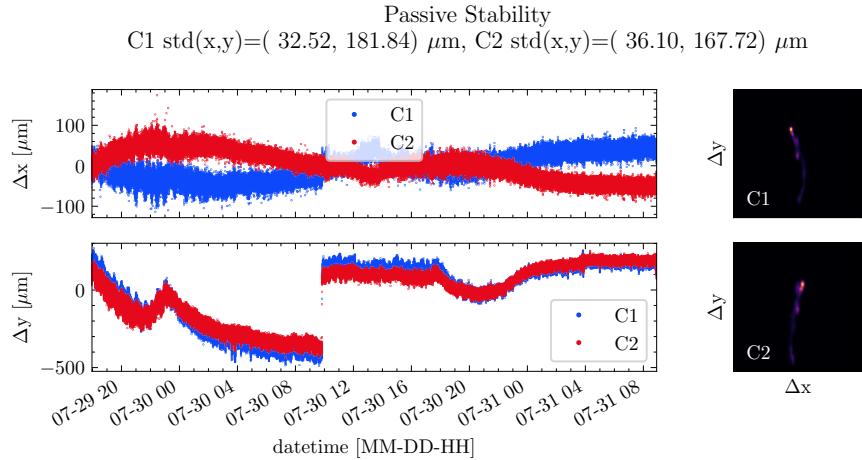


Figure 2.4: Typical passive pointing stability of the Spitfire's amplifier as measured on cameras C1 & C2. Left panels:  $x$  and  $y$  coordinates of the centroid vs. time; right panels: 2D histogram of beam centroid positions for the same time period. The 2-point correction scheme was activated between 9 am and 5 pm on 07-30. Long term thermal drift is apparent at all other times.

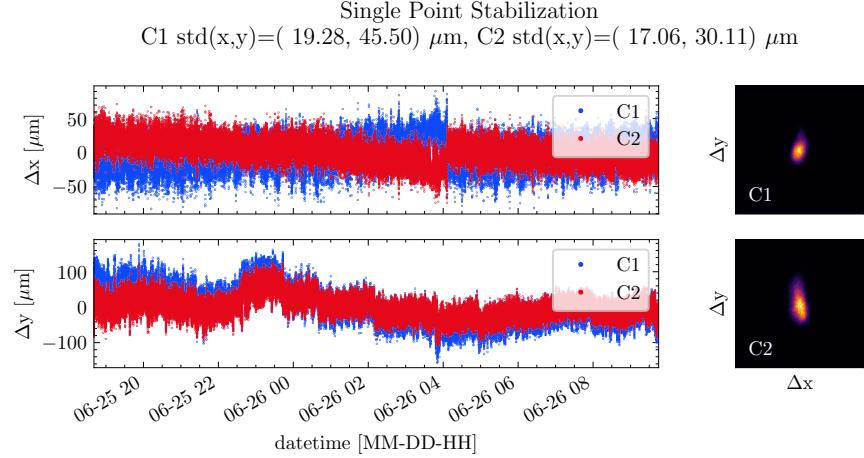


Figure 2.5: Single-point stabilization of the Spitfire’s amplifier as measured on cameras C1 & C2. This dataset contains about 10 correction events, which are visible as abrupt jumps in the time series. The skew of the centroid distributions highlights the limitations of the single-point correction scheme.

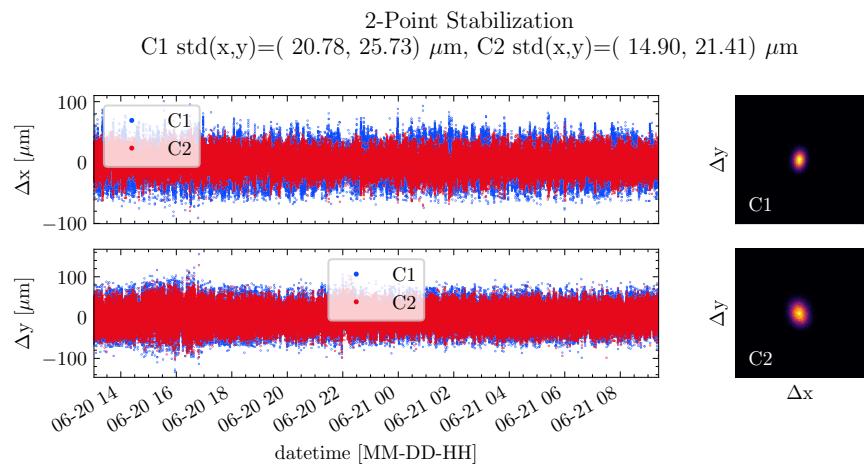


Figure 2.6: Two-point stabilization of the Spitfire’s amplifier as measured on cameras C1 & C2.

### 2.2.2 Active Pointing Correction Systems

As a nonlinear device, the performance of the TOPAS is extremely sensitive to input pointing, laser pulse parameters and laboratory environmental conditions. The large optical path length ( $\approx 15.5$  m) between the amplifier and the TOPAS puts stringent requirements on the angular tolerances of the amplifier's output pointing. According to the specification sheet, the rms beam pointing stability of the amplifier at constant temperature is  $< 5 \mu\text{rad}$  ( $\approx 75 \mu\text{m}$  at 15.5 m) at constant temperature, which is sufficient for our purposes. Unfortunately, the temperature in the laser bay varies significantly throughout the day – sometimes by several degrees – as a function of the occupancy of the physics research building, building-wide energy conservation measures, and activity within the laser bay. Under these conditions, the amplifier's pointing changes by up to  $20 \mu\text{rad}/^\circ\text{C}$  ( $= 310 \mu\text{m}/^\circ\text{C}$  at 15.5 m).

To combat this slow pointing drift, we actively stabilize the beam pointing between the amplifier and the external compressor as shown in Fig. 2.3. Note that there is not enough space to implement a pointing solution between the external compressor and the TOPAS. Over the years, we have used both a home-built single-point correction scheme and a commercial two-point correction scheme.

Fig. 2.4 shows the passive pointing stability of the amplifier as measured by cameras C1 & C2 over the course of 28 hours. The left panel shows the  $x$  and  $y$  coordinates of the centroid (expressed as a deviation from the average position) sampled at 2 Hz. Note that the sign of  $\Delta x$  is reversed between C1 and C2; this is an artifact of the camera geometry; otherwise the data from the two cameras is self-consistent. The thermal drift is most apparent in the vertical direction; from midnight to 9 am the beam drifts vertically nearly  $500 \mu\text{m}$ . Between 9 am and 5 pm we activated the two-point stabilization system (described below), which accounts for the good performance during this time period. At 5 pm, the stabilization system was turned off and the slow drift resumes. The right panels show 2D histograms of the centroid position, calculated from the time series data. From these plots, it is apparent that the laser drifts in a large arc pattern, with the primary deviation occurring in the vertical direction.

The single-point correction system was programmed and implemented by Dietrich Kiesewetter, who was a graduate student at the time [18]. In this scheme, a digital camera (C0 in Fig. 2.3) located approximately 7.8 m after the amplifier monitors the transmitted light of a high reflective mirror incident on a card. The position of the centroid is calculated on a rolling average basis and compared to a saved set point. When the centroid position deviates from the set point by more than a minimum correction size, a correction signal is sent to a motorized mirror (M0, located 33 cm after the amplifier). The minimum correction size is set by the user to avoid frequent small corrections to the beam, which results in high-frequency pointing jitter. The correction time interval is also

user-adjustable, with typical values of 1 - 60 seconds. If the correction signal requires too large of a step, or if the integrated intensity of the beam falls below a set value, then the locking algorithm assumes that something is wrong and breaks the correction loop without taking corrective action. This prevents the system from taking corrective action in the event the beam is partially blocked by a third party.

The performance of the single-point stabilization system as reported by cameras C1 & C2 is shown in Fig. 2.5. Here, we can see the limitations of a single-point correction scheme. The system is very good at maintaining the position of the beam at C0, but it has no control over the pointing of the beam. As a result, the beam centroid continues to drift on downstream optics, albeit with reduced magnitude compared to the uncorrected case. This effect is apparent in the skewed 2D histograms in Fig. 2.5. The single-point stabilization system works well over short periods of time, but its geometry necessitates weekly realignment of the external compressor and all other downstream optics. Given the technical demands of our experiments and the sensitivity of the TOPAS to input pointing, this can be a prohibitively time consuming process. Unlike a single-point correction system, a two-point system only needs to be set once.

For the two-point stabilization system, a commercial system was chosen over a home-built solution to reduce the development and implementation time. We use a Newport GuideStar II, which utilizes two cameras (C1 & C2 in Fig. 2.3) and two motorized mirrors (M1 & M2) to monitor the beam centroid and make corrections. A patented correction algorithm running on purpose-built computational hardware is applied to the output of the cameras to solve for the necessary correction signals up to 3 times per second [9]. Beam pointing is usually recovered with a single corrective action, and even especially large pointing drifts are corrected well within 1 second. For numerical stability and increased sensitivity, the distance between M1 and M2 is made as large as possible (3.1 meters); the distance between C1 and C2 (1.75 meters) is also maximized and made similar to the M1-M2 distance; the distance between C1 and M2 is made as small as possible (75 cm).

The passive and actively-controlled pointing stability of the amplifier is shown in Fig. 2.6. Owing to the high correction frequency, individual corrections are small and not visible in the time series. The two-point system maintains both the centroid position and propagation direction, so there is minimal correlation between the reported centroid positions on C1 & C2. As a result, the pointing into the TOPAS rarely needs to be optimized under normal operation, saving valuable time and making experiments more repeatable.

The GuideStar II performs extremely well, but its software lacks the safety features of our home-built system. Specifically, there is no maximum allowable correction size, intensity or beam mode quality monitoring to prevent run-away corrections. For example, it is common to insert a paper card into the beam path to inspect the beam mode just before the external compressor. As a result, camera C2 will briefly see a partially clipped beam

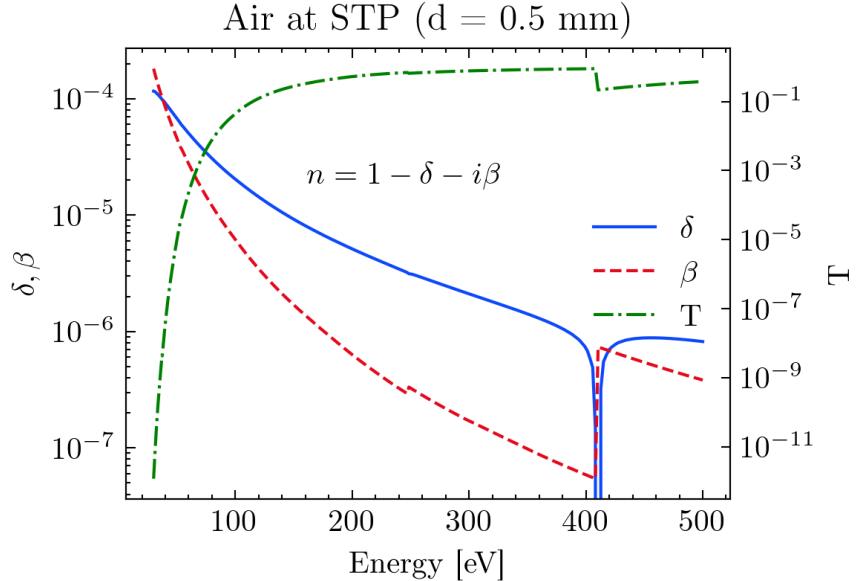


Figure 2.7: Estimation of XUV propagation losses through 0.5 mm of air at STP. Atomic scattering data obtained from [12, 15]. Calculation follows Eqs. (2.3) to (2.5).

with a centroid displaced by approximately the beam radius, and the GuideStar will take *immediate corrective actions* to adjust the pointing. These actions may result in a 15 mJ beam pointing in an unsafe direction. **Users are cautioned to keep clear of the beam path between the amplifier and the GuideStar cameras when the GuideStar locking algorithm is enabled.** To mitigate this issue, the GuideStar cameras are kept in a plastic enclosure and workers are made aware of the limitations of the GuideStar system.

**say something about the 2photon photodiode in the external compressor**

### 2.2.3 Beam routing into the TABLe interferometer

The output of the TOPAS is sent approximately 3 meters towards the transient absorption beamline.

## 2.3 Vacuum System

### 2.3.1 The Need for High Vacuum

The light generated by the high harmonic process is absorbed strongly by air, as most gases have at least one electronic transition in the XUV regime. The magnitude of absorption can be estimated using the atomic scattering factors  $f = f_1 + i f_2$ , which were taken from [15]. The photoabsorption cross section  $\mu_a$ , the transmission ratio  $T$ , and the complex index of

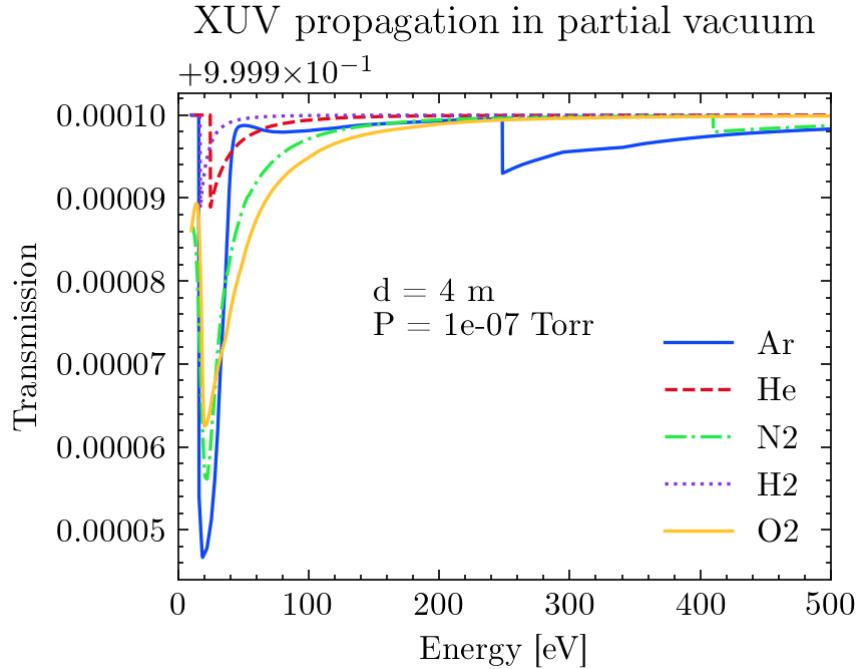


Figure 2.8: Estimation of XUV propagation losses through a vacuum level of  $10^{-7}$  Torr and a distance of 4 meters. Transmission is well over 99.99% for this pressure-length product. Atomic scattering data obtained from [12, 15]. Calculation follows Eqs. (2.3) to (2.5).

refraction  $\hat{n}$  of a gas can be calculated from these factors:

$$\mu_a = 2r_0\lambda f_2 \quad (2.3)$$

$$T = \exp(-N\mu_a d) \quad (2.4)$$

$$\hat{n} = 1 - \frac{1}{2\pi}Nr_0\lambda^2(f_1 + if_2) \quad (2.5)$$

In the above equations,  $\lambda$  is the photon wavelength,  $N$  is the number of atoms per unit volume,  $d$  is the optical path length and  $r_0 = 2.8179403227(19) \times 10^{-6}$  nm is the classical electron radius. The results for air at standard temperature and pressure are shown in Fig. 2.9. From this figure it is apparent that any XUV light we generate will be effectively attenuated to zero in less than 1 millimeter if the beam is propagated in air.

For reference, the XUV portion of the transient absorption beamline is about 4 meters long. Fig. 2.8 shows the expected XUV transmission losses for XUV propagation through a partial vacuum of common gases: argon and helium are often used for generation, while nitrogen, hydrogen and oxygen are common UHV system contaminants. From this figure,

we can see that the XUV transmission exceeds 99.99% for an average pressure of  $10^{-7}$  Torr. Note that this calculation does not include reflection losses from the ellipsoidal mirror, which are on the order of 10%. This simple analysis tells us that the XUV portion of the beamline must be kept under relatively high vacuum to avoid needlessly reducing the XUV flux.

The microchannel plate (MCP) assembly in the photon spectrometer puts additional constraints on the vacuum level. Contaminants in the spectrometer chamber (originating from a finite chamber pressure) lower the effective electrical resistance between the highly charged plates, resulting in a somewhat periodic current surge between the plates. This effect manifests itself in the data as a bright point source at a random location on the detector. In addition to reducing the fidelity of the data, each current surge counts towards the lifetime limit of the MCP assembly, reducing its lifetime [20].

The low pressure condition required to minimize XUV absorption and instrumentation malfunction is in direct conflict with the requirements for high harmonic generation (HHG) and gas-phase attosecond transient absorption spectroscopy (ATAS) experiments. HHG requires a gas source to be placed near the IR focus in the generation chamber, and a gas-phase ATAS experiment requires a similar gas source to be placed near the XUV/IR focus in the target chamber. The gas from these sources will diffuse into neighboring chambers, raising the pressure of the entire beamline. The vacuum system was designed to localize the gas density at the interaction regions while allowing a range of optical configurations to be used.

### 2.3.2 Vacuum System Introduction

The DiMauro group has extensive experience working with home-built vacuum hardware and attosecond interferometry. At the beginning of this project, we drew heavily on the expertise of group members when designing our vacuum system. Because the technical requirements of our vacuum system are very similar to that of the RABBITT apparatus [4], we will frequently make comparisons between the two devices.

The vacuum system was designed to be as modular as possible. In this sense, the TABLE apparatus can be thought of a permanently installed XUV light source & interferometer that has the ability to accept modular end stations. This design principle has already allowed the study of electron rescattering in strong infrared fields by another graduate student [18]; going forward, new target chambers can be designed to meet the needs of future experiments while maintaining the integrity of the XUV light source and interferometer.

We employ magnetically levitated turbomolecular pumps which were not commercially available at the time of the RABBITT apparatus' construction. These pumps are designed so that the vacuum side of the blade assembly does not make mechanical contact with the drive shaft and housing, which dramatically reduces vibrations during operation. Mag-

CAD drawing  
showing an  
overhead view  
of the TABLe  
system with  
labels and  
optical table

Figure 2.9: this is an overhead view of the TABLe's vacuum chambers. each chamber is labelled and the optical table is visible. this figure will span an entire page so the reader can actually see details.

lev turbopumps have a noise power spectrum an order of magnitude smaller than that of traditional turbopumps. This ultra-quiet operation allows us to mount the vacuum hardware directly on the interferometer's optical table. Likewise, we keep our rough vacuum system in an adjacent pump room to minimize acoustics, vibrations and oil contamination in the target rooms.

Large vacuum chambers have the benefit of being able to accommodate a seemingly limitless amount of optics and internal hardware, but they are very uncomfortable and difficult to work around. For this reason, we tried to keep the physical size of the chambers as small as possible while still allowing sufficient internal space for our current and reasonable future equipment needs (*in vacuo* motorized stages, optics, gas and electric feedthroughs, etc.). Additionally, *in vacuo* optics are difficult to optimize. Vacuum compatible optic mounts exist, but they are expensive, cumbersome to work with and generally considered specialty items that need to be custom ordered. Adjusting a non-motorized optic requires a venting / pumping cycle, which can take several hours. Early on in the design process, we made the decision to keep as many optics as possible outside of the vacuum system, greatly reducing the overall size of the system. Excluding the modular endstations, our vacuum chambers have a footprint that is approximately one third that of the RABBITT apparatus, which leaves more than half of our 4' × 10' optical table unoccupied.

### 2.3.3 Vacuum System Details

In this section, we will provide a brief overview each vacuum chamber in the TABLe vacuum system.

#### Generation & Differential Pumping Chambers

The generation chamber is designed to house the HHG gas nozzle or cell. It is here that the XUV light is created for the ATAS experiments. As such, it must be large enough to house an XYZ translation stage, as well as the necessary electric and gas line feedthroughs. The turbo pump must be large enough to maintain milliTorr or lower pressures while being subjected to large gas throughputs for the duration of an experiment. The geometry of the chamber must accommodate a range of focal lengths and optical layouts. Viewports were added to help with optical alignment to the HHG interaction region.

We use a differential pumping scheme to reduce the flow of gas from the generation chamber to the other parts of the beamline.

physical dimensions of the chamber. 10 inch 8-way cross. distance between the center of the chamber (HHG position) and the entrance and exit walls.

The chamber is a standard 6-way 10" ConFlat (CF) cross that has been modified to have four additional 2.5" CF ports. These additional ports are positioned at the corners of

the top flange and are directed towards the center of the chamber.

generation turbo: Oerlikon Leybold Turbovac Mag W 1300 iP. differential pump turbo: Oerlikon Leybold Turbovac Mag W 400 iP. mirror chamber: Oerlikon Leybold Turbovac Mag W 700 iP. target chamber: Oerlikon Leybold Turbovac Mag W 1300 iP. target chamber differential: Oerlikon Leybold Turbovac T450i (hybrid bearing). spectrometer: Oerlikon Leybold Turbovac Mag W 400 iP.

A 10 mm diameter vacuum aperture is placed at the end of the generation chamber. This size was a compromise between minimizing gas flow and maximizing light transmission. This aperture was designed to be interchangeable for other sizes should experimental needs change in the future.

talk about the expected gas flow here. turbo pump specifications and model. gas throughput calculations.

optics: CaF<sub>2</sub> window. focusing geometry: supported focal lengths, telescope, 2-color generation. XYZ translation stage and modular gas nozzle holder.

vacuum aperture

rough vacuum feedthrough for HPC

### **differential pump chamber**

### **IR diagnostic port and metallic filter**

### **mirror chamber**

CaF<sub>2</sub> window, EM motorization, HM location (before or after EM)

### **target chamber and 2nd differential pump chamber**

designed to accommodate an electron spectrometer (not implemented)

sample holder (gas and condensed phase)

vacuum aperture and differential pumping

### **photon spectrometer**

VLS gratings and motorization

MCP-P

movable detector plane (cage and crank)

IR diagnostic mirrors and port

### **OMRON safety system and rough vacuum details**

generation stack: blower = Leybold WSU 501 roots blower, rough pump = Leybold Trivac D65B

mirror-target-spec stack: blower = Ruvac WA251 roots blower, rough pump = Leybold Trivac D65B

pressure gauges at mouth of pumps = Agilent Type 0531 thermocouple vacuum gauge  
pressure gauge controller (electronic, ethernet enabled remote pressure monitoring) =  
Kurt J. Lesker KJL615TC-E

## 2.4 Interferometer Design

### 2.4.1 Design Goals

compare to the old attosecond beamline, cite razvan's thesis [4]. main differences between the new and old beamlines:

- in-air pump design allows for complex optics (THz, OPA, etc.) in pump arm
- smaller vacuum chambers, easier to work inside
- not as well isolated from external environment (bellows feet not implemented)
- photon spectrometer, can accomodate simultaneous electron & photon measurements
- modular endstation design

talk about the split optical table design here, since you didn't talk about it in the vacuum section.

### 2.4.2 XUV Optics

#### Fresnel Reflection from a Rough Surface

need a figure showing the interface, defining angles, indices, etc.

We use the familiar Fresnel equations to model reflection from the surface of a conductive surface [34]. In the equations that follow, the vacuum is denoted by  $j = 1$  and the conductive material is  $j = 2$ . The incident electric field is  $E_I$  and the reflected component is  $E_R$ . Following standard convention, the  $s$ -polarized light refers to light that has a polarization normal to the surface;  $p$ -polarized light is polarized parallel to the surface. The complex reflection amplitudes  $r_{s,p}$  are written in terms of the complex impedance  $\hat{Z}_j = \mu_j c / \hat{n}_j$  and the angle measured from the normal in each medium  $\theta_j$ :

$$\hat{r}_s \equiv \left[ \frac{E_R}{E_I} \right]_s = \frac{\hat{Z}_2 \cos \theta_1 - \hat{Z}_1 \cos \theta_2}{\hat{Z}_2 \cos \theta_1 + \hat{Z}_1 \cos \theta_2} \quad (2.6)$$

$$\hat{r}_p \equiv \left[ \frac{E_R}{E_I} \right]_p = \frac{\hat{Z}_1 \cos \theta_1 - \hat{Z}_2 \cos \theta_2}{\hat{Z}_1 \cos \theta_1 + \hat{Z}_2 \cos \theta_2} \quad (2.7)$$

Next, we assume non-magnetic media ( $\mu_1 = \mu_2 = \mu_0$ ) and write the amplitudes in terms of the complex indices of refraction  $\hat{n}_j = n_j + ik_j$ :

$$\hat{r}_s = \frac{\hat{n}_1 \cos \theta_1 - \hat{n}_2 \cos \theta_2}{\hat{n}_1 \cos \theta_1 + \hat{n}_2 \cos \theta_2} \quad (2.8)$$

$$\hat{r}_p = \frac{\hat{n}_1 \cos \theta_2 - \hat{n}_2 \cos \theta_1}{\hat{n}_1 \cos \theta_2 + \hat{n}_2 \cos \theta_1} \quad (2.9)$$

The reflectance is the modulus squared of the reflection amplitudes:

$$\hat{R}_s = \left| \frac{\hat{n}_1 \cos \theta_i - \hat{n}_2 \cos \theta_t}{\hat{n}_1 \cos \theta_i + \hat{n}_2 \cos \theta_t} \right|^2 \quad (2.10)$$

$$\hat{R}_p = \left| \frac{\hat{n}_1 \cos \theta_t - \hat{n}_2 \cos \theta_i}{\hat{n}_1 \cos \theta_t + \hat{n}_2 \cos \theta_i} \right|^2 \quad (2.11)$$

Finally, we assume that the first medium is vacuum ( $\hat{n}_1 = 1$ ) and apply Snell's law to write the transmitted angle  $\theta_2$  in terms of the incident angle  $\theta_1$ :

$$\hat{R}_s = \left| \frac{\cos \theta_1 - \hat{n}_2 \sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2}}{\cos \theta_1 + \hat{n}_2 \sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2}} \right|^2 \quad (2.12)$$

$$\hat{R}_p = \left| \frac{\sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2} - \hat{n}_2 \cos \theta_1}{\sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2} + \hat{n}_2 \cos \theta_1} \right|^2 \quad (2.13)$$

The above analysis is valid for a perfectly smooth interface, but real optics have finite roughness. Loosely speaking, the surface roughness becomes important when it is on the order of the photon wavelength. As a point of comparison, a high quality off-the-shelf optic will have a surface roughness of  $\lambda/10 = 63.2$  nm, which is 5 times larger than the wavelength of a 100 eV photon. Clearly, we need specialized optics if we want to work in the XUV.

We will summarize the results of the analytic treatment of surface roughness in the literature [10, 28]. The surface roughness can be thought of as a local modulation  $\delta z(\vec{r})$  of the ideal interface height  $z(\vec{r})$  with a variance  $\langle z^2 \rangle$ , and a characteristic in-plane length scale  $\delta r$  over which the modulation occurs. There are two limiting cases to consider, based on the relative magnitudes of  $\delta r$  and the extinction length  $1/k_2$ . For reference, the extinction length is on the order of 1  $\mu m$  for hard x-rays, and about 10 nm for XUV light.

First, consider the case where the characteristic length scale of the roughness is much larger than the extinction length ( $\delta r \gg 1/k_2$ ). In this case, the local electric field can be

expressed in the following form:

$$E_j(x, y) = \left( A_j^+ \exp(i k_{j,z} z) + A_j^- \exp(-i k_{j,z} z) \right) \exp(i \omega t - k_i \ln ||r_{||}) \quad (2.14)$$

In Eq. (2.14),  $A_j^+(A_j^-)$  denotes the amplitude of the incoming (outgoing) wave and  $k_i \ln ||r_{||} = k_j \cos \theta_j$  denotes ???.

Then, the incoming field does not

$$r^{rough} = \frac{\langle A_1^+ \rangle_x}{A_1^-} = r \langle \exp(2i k_{1,z} z) \rangle = r \exp(-2k_{1,z} \langle z^2 \rangle) \quad (2.15)$$

Eq. (2.15) is the Debye-Waller factor.

Then, the modulation is slow enough that there is a phase relationship between the incident and reflected fields over a large area. This is the Debye-Waller limit, and the reflectance is modified by a factor of  $\exp(-4k_1^2 \langle z^2 \rangle)$ .

If the modulation length scale is much less than the extinction length ( $\delta r \ll 1/k_2$ ) then there is no short-scale correlations between the electric field and the roughness. This is the Croce-Nérot limit, and the reflectance is modified by a factor of  $\exp(-4k_1 k_2 \langle z^2 \rangle)$ .

For hard x-rays, the Nevot-Croce Surface roughness has the effect of reducing the reflectance by an exponential factor.

The finite surface roughness of a real mirror reduces the reflectance by a factor of  $\exp(-4k_1^2 \sigma^2)$ , called the *Debye-Waller factor* [10]. Here,  $\sigma$  is the root-mean-square (rms) of the vertical roughness, and  $k_1$  is the normal component of the wave-vector in vacuum. The final result is below:

Combining Eqs. (2.12) and (2.13) with Eq. (2.15) yields the reflectance of a rough metallic surface:

$$\hat{R}_s = \left| \frac{\cos \theta_1 - \hat{n}_2 \sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2}}{\cos \theta_1 + \hat{n}_2 \sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2}} \right|^2 \left( \frac{2\pi\sigma \cos \theta_1}{\lambda} \right)^2 \quad (2.16)$$

$$\hat{R}_p = \left| \frac{\sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2} - \hat{n}_2 \cos \theta_1}{\sqrt{1 - \left( \frac{1}{n_2} \sin \theta_1 \right)^2} + \hat{n}_2 \cos \theta_1} \right|^2 \left( \frac{2\pi\sigma \cos \theta_1}{\lambda} \right)^2 \quad (2.17)$$

The mid spatial frequency roughness (MSFR) of the ellipsoidal mirror's surface is  $\leq 0.3$  nm when sampled at a spatial frequency of 1-200  $\mu m$ .

The extinction length in gold is on the order of 10 nm for photon energies below 500 eV. This length scale is much smaller than the spatial variation of the height of our ellipsoidal

mirror ( $1 - 200 \mu m$ ), so we treat the using the Debye-Waller factor [10, 29]:

$$r^{rough} = r^{smooth} \exp(-2k_{z,0}k_{z,1}\sigma^2) \quad (2.18)$$

- the extinction length for photon range 30 - 500 eV in gold is on the order of 0.07 - 0.2 nm.
- the 0.3 nm rms roughness is based on a measurement frequency of 1 - 200 microns. this corresponds to a relatively long-range / slowly varying roughness.
- therefore we should use the debye-waller factor.

Nevot-Croce model assumes random vertical roughness with a Gaussian distribution. for the low spatial frequencies of the roughness spectrum, the Fresnel reflection coefficient is usually multiplied with a Debye-Waller factor  $\exp(-16\pi^2 k_1^2 \sigma^2)$ , while for the high spatial frequencies, the correction coefficient is given by the Nevot-Croce correction factor  $\exp(-16\pi^2 k_1 k_2 \sigma^2)$ , where  $\sigma$  is the root-mean-square (rms) of the vertical roughness, and  $k_1$  and  $k_2$  are the normal components of the wave-vectors in the two media [29].

FRED calculations: toroid demagnification vs ellipsoid demagnification

why an ellipsoid doesn't have aberrations

aligning the ellipsoidal mirror - what the spot looks like when misaligned

## Material Choice

Fig. 2.10 shows the Fresnel reflectance from various materials with an ideal interface at fixed angle. While other materials provide have better reflectance, gold was chosen due to its chemical inertness.

We compare two common materials.

Gold was chosen due to its chemical inertness and generally good reflectivity.

thickness of gold layer – penetration depth (in general and at 5 degrees)

40 nm of gold with a sticking layer ... explain your reasoning

## Angle Choice

why did we go with 5 degrees?

footprint of beam on the optic / size of the optic [10]:

$$F = \frac{t_1}{\sin \alpha} t_2 \quad (2.19)$$

in the above equation,  $t_1, t_2$  are the rectangular dimensions of the input beam and  $\alpha$  is the grazing angle of the light. then  $F$  is the footprint of the beam on the angled mirror.

5 degrees was a good compromise between reflectivity, footprint on optic, and footprint of optic

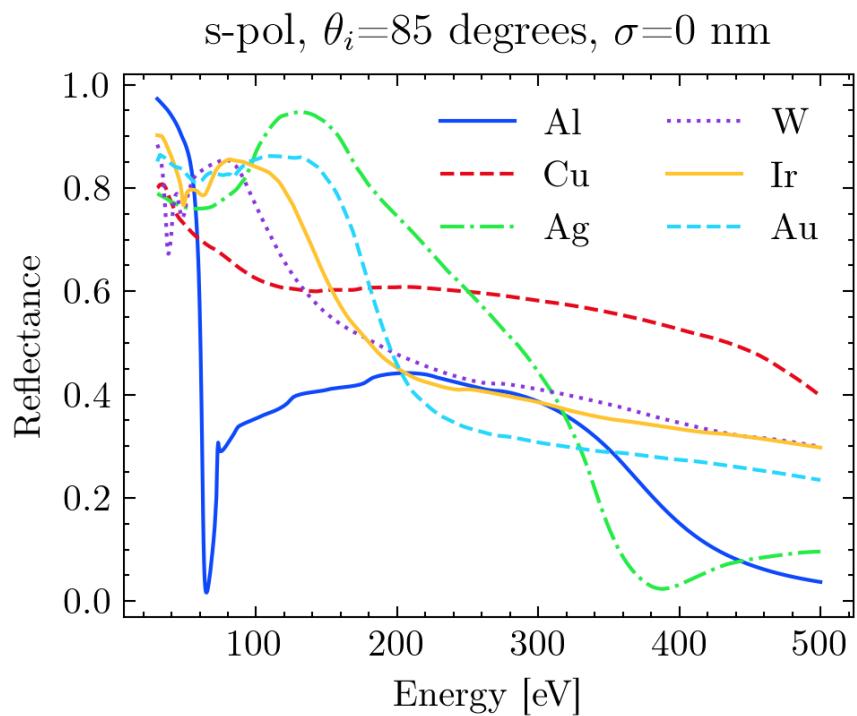


Figure 2.10: Fresnel reflectance for s-polarized light on smooth metal mirrors at a grazing angle of 5 degrees. Refractive index data obtained from [12, 15]. Calculation follows Eqs. (2.12) and (2.13).

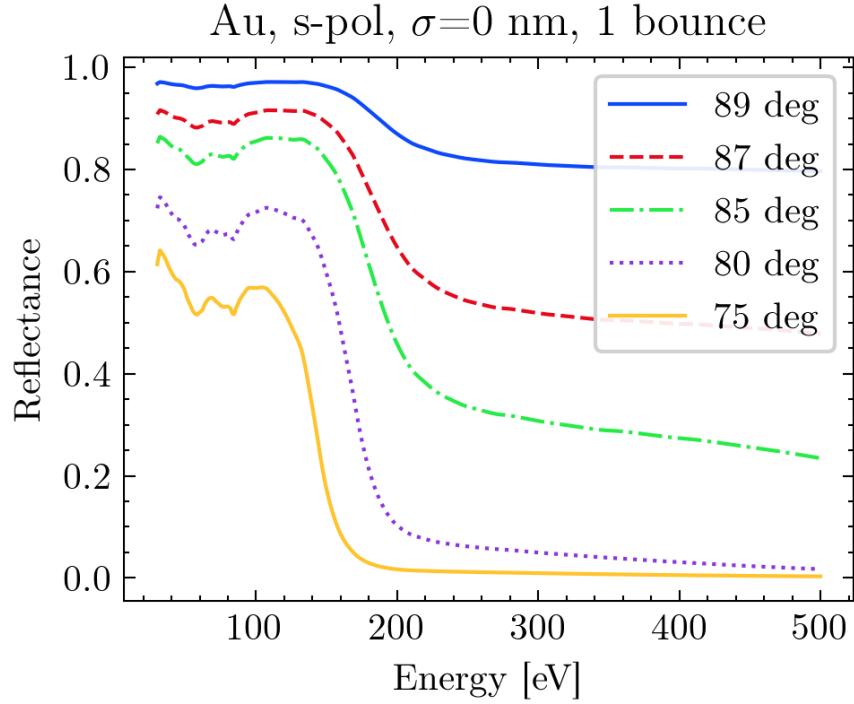


Figure 2.11: Fresnel reflectance for *s*-polarized light from a single smooth gold mirror as a function of incident angle. Calculation follows Eqs. (2.12) and (2.13).

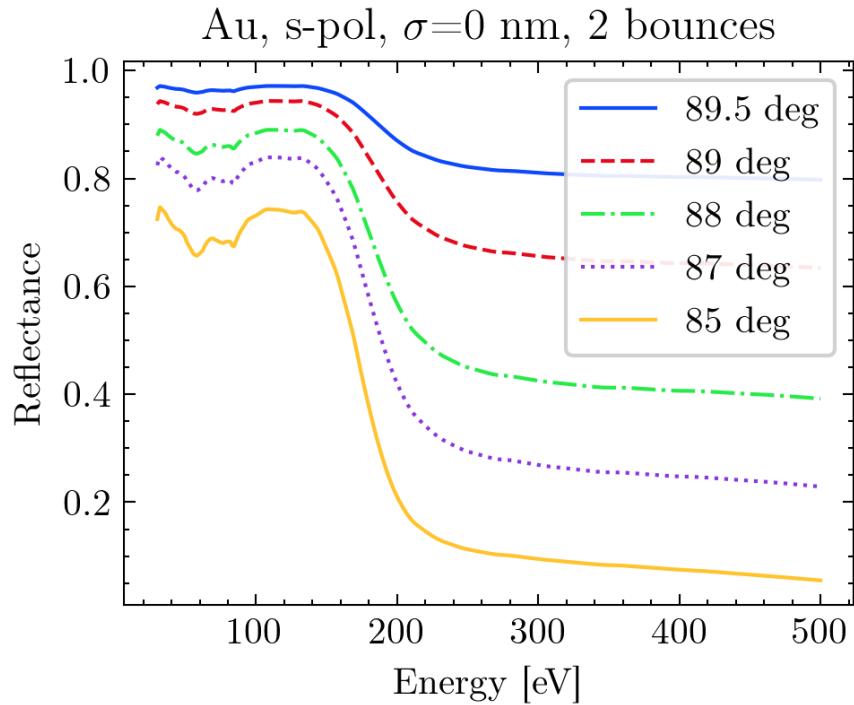


Figure 2.12: Fresnel reflectance for *s*-polarized light from two smooth gold mirrors as a function of incident angle. Calculation follows Eqs. (2.12) and (2.13).

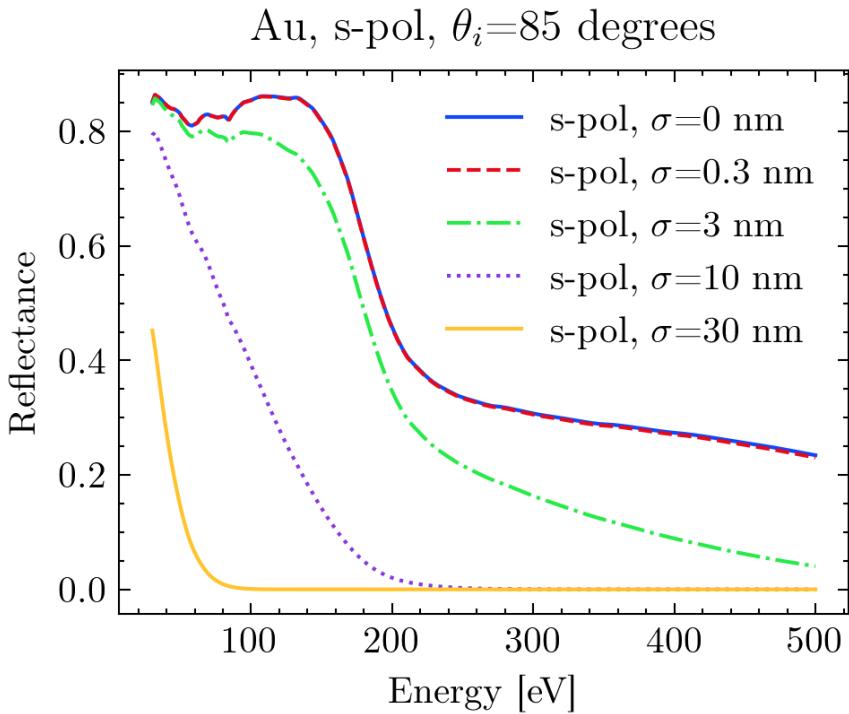


Figure 2.13: Effect of surface roughness on reflectance. Calculation follows Eq. (2.16).

### Roughness Choice

why did we go with 0.3 nm rms? because it was state of the art, and very close to the performance of a ideal/smooth surface

#### 2.4.3 Types of Reflective Optics

compare toroid, ellipsoid and multilayer mirrors

our ellipsoidal mirror specifications

FRED simulations of the ellipsoidal mirror focus

#### 2.4.4 Focusing the IR Pump Arm

hole mirror

the need for collinear IR-XUV geometry (spatial overlap, temporal smearing)

HM placement: before or after EM?

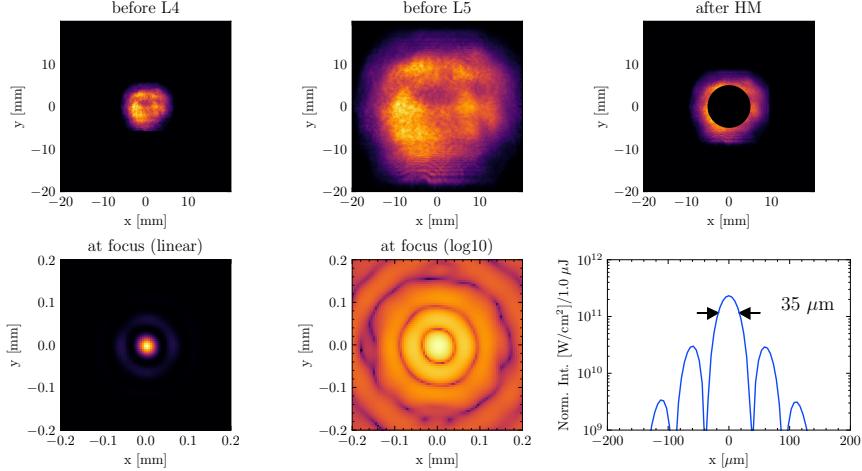


Figure 2.14: Numerical propagation of the IR ( $\lambda=1500$  nm) beam through the pump arm. Beam path layout follows Fig. 2.1. Each panel shows the intensity of the beam as the beam propagates towards the focus. The first panel shows the measured intensity (Electrophysics PV320 thermal camera), all other panels are calculations. The arrows on the lineout indicate the FWHM. All calculations are for vacuum ( $n = 1$ ). See text for details.

## description of optics used

### Calculation of IR intensity at Focus

The profile of the TOPAS output at  $\lambda=1500$  nm was measured immediately before the pump arm's diverging lens (L4 in Fig. 2.1) using an Electrophysics PV320 thermal camera. The beam was propagated numerically using the Python package *Lighthpipes for Python* [32] through the remainder of the pump arm to the focus using a grid size of  $2^{13} \times 2^{13}$ . The result of this calculation is shown in Fig. 2.14. Optical element parameters are as follows: L5 is a  $f = +500$  mm Thorlabs LA1380-C, located 68.5 cm after L4 ( $f = -300$  mm, Thorlabs LF1015-C). The hole mirror HM is located 53 cm after L5 and has an inner radius of 5 mm. Clear apertures of 22.86 mm (L4), 45.72 mm (L5) and 50.8 mm (HM) were used. A 65 fs Gaussian temporal profile containing 1  $\mu\text{J}$  of energy was assumed.

Reflection losses from 2 Ag mirrors, 2 AR-coated lenses and uncoated CaF<sub>2</sub> vacuum window are responsible for a 26.2% reduction in transmitted power. Additionally, the geometry of the hole mirror causes only 56% of incident power to be incident on a reflective surface (the rest is lost to the central aperture). In total, the pump arm transmits 41.3% of the power from before L4 to the focus.

The IR intensity makes an Airy-like diffraction pattern at the focal plane. There is an intense bright spot surrounded by a series of rings, with the intensity of each ring as the distance from the center increases. The rings exhibit a periodic modulation in intensity with

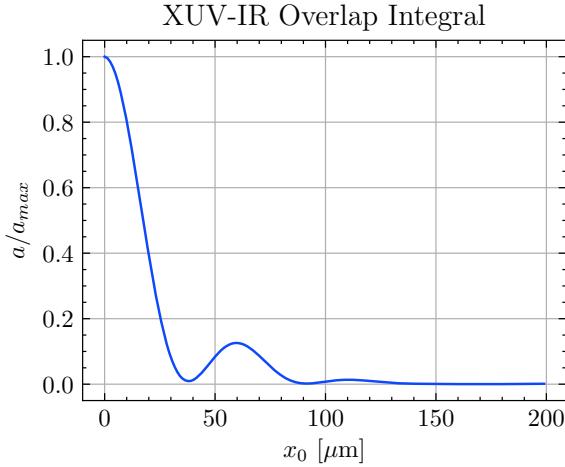


Figure 2.15: XUV-IR overlap function, as defined in Eq. (2.21), calculated using the numerical simulation results of Fig. 2.14 and a Gaussian XUV beam with a 6  $\mu\text{m}$  waist. The result has been normalized to perfect overlap,  $a_{max}$ .

respect to angle  $\phi$ . This four-fold symmetry is due to the square-like spatial profile of the TOPAS output, whereas the clipping from the hole mirror's aperture is responsible for the central peak and ring structure. The central lobe has a peak intensity of  $\sim 2.3 \times 10^{11} \text{ W/cm}^2$  per 1  $\mu\text{J}$  input pulse energy and a FWHM of 35  $\mu\text{m}$ . The first ring has a radius of 59  $\mu\text{m}$  and a peak intensity  $\sim 2.9 \times 10^{10} \text{ W/cm}^2$  per input  $\mu\text{J}$  pulse energy. Thus the central lobe's peak intensity is about an order of magnitude larger than the ring's intensity. However, the central spot only contains about 49% of the total power, as the rings cover a much larger area.

Note that the above calculations assume perfect alignment into the hole mirror (i.e., the IR beam is centered on the central aperture of the hole mirror). If the IR beam is misaligned to the hole mirror, then the transmission to the focus will increase as the most intense part of the beam is no longer clipped by the central aperture. Therefore, a drift in the laser's pointing during an experiment can effect the sample's interaction intensity.

#### 2.4.5 Pulse energy control, calibration & monitoring

#### 2.4.6 Delay control & calibration

#### 2.4.7 XUV-IR spatial overlap

**need to re-write this intro paragraph. excitation fraction hasn't been computed yet**

The excitation fraction can be computed for each spatial coordinate on the sample using the above method and the predicted intensity distribution from the numerical beam prop-

agation calculations. Because the electrons are being excited via a single-photon process, the excited carrier density will be proportional to the fluence, and thus proportional to the intensity shown in Fig. 2.14. Because the intensity of the XUV is very weak, the absorption of the XUV by the sample is also linear. Thus, we should expect the ATAS signal to be proportional to the XUV-IR overlap integral:

$$a = \frac{\int dV I_{\text{IR}} I_{\text{XUV}}}{\int dV I_{\text{IR}} \int dV I_{\text{XUV}}} \quad (2.20)$$

Here, the integration volume is over the entire sample. If we assume that the intensity distribution does not appreciably change over the thickness of the sample, we can simplify the above equation. This is a reasonable assumption because the sample ( $L = 100$  nm) is much thinner than the Rayleigh range ( $z_R \sim 1$  mm), and the absorption is low ( $\sim 4\%$ ). So we assume the sample is a  $\delta$ -function in thickness and only evaluate the intensities at the focal plane. With this assumption, the overlap integral becomes:

$$a = L \frac{\int dA I_{\text{IR}} I_{\text{XUV}}}{\int dA I_{\text{IR}} \int dA I_{\text{XUV}}} \quad (2.21)$$

Knife edge measurements have been performed on the XUV light, showing that it has a Gaussian spatial profile with a beam waist of  $6 \mu\text{m}$ . We write down the spatial profile of the XUV light at the focus:

$$I^{\text{XUV}} = I_0^{\text{XUV}} \exp(-2((x - x_0)^2 + (y - y_0)^2)/w_{\text{XUV}}^2) \quad (2.22)$$

Here,  $I_0^{\text{XUV}}$  is the peak intensity and  $w_{\text{XUV}}$  is the beam waist (radius), defined as the point where the intensity falls to  $e^{-2} = 13.5\%$  of its maximum. The lateral shift from the center of the IR focal spot in the horizontal and vertical directions is  $x_0$  and  $y_0$ , respectively. With this formulation, and using the simulation results for the IR spot, the XUV-IR overlap integral is calculated as a function of XUV-IR misalignment ( $x_0, y_0$ ). This result is shown in Fig. 2.15. Here, XUV beam is translated relative to the IR beam in the horizontal direction ( $x_0$  with  $y_0 = 0$ ) and the overlap is computed from Eq. (2.21).

Fig. 2.15 shows the sensitivity of a condensed matter ATAS experiment to relative alignment.<sup>1</sup> A spatial overlap deviation of  $10 \mu\text{m}$  will cause the XUV-IR overlap - and thus the measured signal - to drop by 20%. Note that a  $10 \mu\text{m}$  displacement of the IR at the sample corresponds to a  $15 \mu\text{rad}$  tilting of the hole mirror (HM). There are two ways misalignment can affect experimental results. If the relative positions of the XUV and IR focal spots changes as an experiment is performed, then the recorded ATAS signal would be a function of both the laser-induced dynamics and the XUV-IR spatial misalignment.

<sup>1</sup>Note that the relevant parameter in Eq. (2.21) is the relative positions of the two focal spots. We have yet to calculate the sensitivity of spatial overlap to deviations in the input laser pointing.

On the other hand, if the entire experiment is performed using a constant misalignment, we would be exciting the sample to some peak excitation fraction  $f$ , but our probe would be measuring a lower excitation fraction ( $\approx fa/a_{max}$ ). Consequently, the measured ATAS signal would be lower than otherwise expected, and any attempts to boost the signal by increasing the interaction intensity could result in permanent laser-induced sample damage.

A condensed matter ATAS experiment has much tighter alignment tolerances than a gas phase experiment. This discrepancy is a simple consequence of sample geometry and density. In either experiment, the measured signal comes from the region of space where the sample density, XUV intensity and IR intensity overlap. The transmission of XUV through the sample is, to first order,  $T = \exp(-n\mu_a d)$ , where  $n$  is the number density,  $d$  is the sample thickness and  $\mu_a$  is the photoabsorption cross section. As discussed above, for technical reasons the experiment should be designed with  $T \approx 1/2$ . Therefore, the product  $n\mu_a d$  will be approximately constant for any transient absorption experiment.

The number density of a condensed phase sample is determined by the chemistry of the compound and is on the order of  $4 \times 10^{22}$  atoms/cm<sup>3</sup>. The experimentalist is free to engineer clever sample geometries, heterostructures and/or nanopatterns, but the high atomic density (and thus absorption coefficient) dictates a total sample thickness on the order of 100 nm. On the other hand, the spatial profile and density of a gas phase sample is determined by the gas nozzle design and its backing pressure, respectively. A typical nozzle used in our lab produces a gas plume with lateral dimensions on the order of 200 - 500  $\mu$ m. This effectively creates a sample that is three orders of magnitude thicker than a condensed phase sample, which relaxes the alignment constraints significantly. This has important consequences for the alignment of the sample.

If the XUV and IR are perfectly collinear, then the beam overlap region is effectively infinite in the propagation direction. In this case, the XUV-IR overlap integral will be positive regardless of any displacement of the sample plane from the focal plane, and maximal when the sample lies in the focal plane. However, if there is a small angle  $\delta\theta$  between their  $k$ -vectors, then the beams will only spatially overlap within a finite region. In this case, the position of the sample plane relative to the beam crossing plane becomes a critical experimental parameter. For an infinitely thick sample (i.e., a chamber effusively filled with gas), it wouldn't matter where the beams crossed as long as they overlapped somewhere within the chamber. Then, the overlap integral would decrease as a function of  $\delta\theta$ , but it would never go to zero. For a thin sample, the bounds of Eq. (2.20) must enclose the beam overlap region, or else the integral will be zero. Thus, the signal strength of a condensed phase ATAS experiment is roughly 3 orders of magnitude more sensitive to the  $z$ -position of the sample relative to the focal plane than a gas phase ATAS experiment.

## 2.5 XUV Photon Spectrometer

### 2.5.1 basic description

### 2.5.2 2D CMOS sensor

### 2.5.3 Spectral Calibration

## 2.6 old stuff

All elements in the periodic table have at least one electronic transition between 30 and 500 eV. Each transition has a characteristic photoabsorption cross section line shape, resulting in poor (good) transmission above (below) the transition. In the terminology of x-ray spectroscopy, this spectroscopic feature is called an *x-ray absorption edge*. Typical attenuation lengths<sup>2</sup> for most elements near an edge at energies of interest to our lab are measured in the tens of nanometers.

This basic feature of XUV light-matter interaction is both a blessing and a curse. On one hand, we can exploit the strong absorption to perform element-specific transient absorption spectroscopy on a myriad of elements and compounds (materials science permitting). On the other hand, the strong absorption puts severe experimental limits on any apparatus that uses XUV.

- modular endstation design
- in-air convenience
- split level table
- magnetic turbo pumps

## 2.7 XUV Optics

The XUV light is divergent after its generation and needs to be focused onto a sample for the experiment. The short relatively wavelength of the XUV puts strong requirements on the surface quality of the focusing optic, raising the cost and manufacturing time significantly. Alignment of the optic is frustrated by the need for in-vacuum propagation and the invisible nature of XUV light. These factors lead us to pursue a "one size fits all" broadband optic rather than a series of interchangeable narrow bandwidth optics tailored for individual experiments.

The presence of strong absorption edges over the bandwidth of the XUV pulse precludes the use of transmissive optics. Narrow bandwidth transmissive optics have been designed to exploit the dispersion near an absorption edge [8]. However, these techniques cannot be extended to support the entire bandwidth of our XUV pulses. Reflective dielectric coatings

<sup>2</sup>The attenuation length is defined as the depth into the material measured along the surface normal where the intensity of x-rays falls to  $1/e$  of its value at the surface.

have been designed for the XUV (**citation needed - Dietrich, Krauss?**), but their reflectance is only good for 10-20 eV (**need a source. this range might be incorrect.**). These considerations leave reflective optics as the only good choice for broadband XUV light.

The IR intensity profile at the focus is an important parameter for every experiment. In a transient absorption experiment, the experimental signal originates from the spatial overlap of the sample, the XUV and the IR light in the interaction region. Ideally, neither pulse would have any spatial structure at the interaction plane. In the real world, we design the optics to minimize the intensity variation of the IR over the spatial extent of the XUV spot. For finite beam sizes, this is accomplished by focusing the XUV tighter than the IR while minimizing aberrations. We chose to demagnify the XUV spot size by a factor of three, which reduces the XUV spot area by a factor of nine. This allows us to more strongly focus the IR pump pulse, resulting in a nine-fold increase of interaction intensity without changing the relative spot sizes of the XUV and the IR. While not required for a condensed matter experiment, the increased IR intensity was essential to doubly ionize helium gas for  $e - 2e$  experiments [18].

In the following sections, we outline the decision-making process for choosing an XUV optic. We start with a review of the physics of reflection from a rough metallic surface. Using these results we determine the optimal material, thickness, surface roughness and angle of the optic. We then compare the most common mirror configurations and discuss their suitability for a 3x demagnification. We conclude by describing

### 2.7.1 XUV mirror choices

As discussed above, we are restricted to smooth grazing incident angle metallic mirrors, but the shape this mirror has not yet been discussed. We considered several focusing geometries for our beamline before ultimately choosing an ellipsoidal mirror. Spherical mirrors introduce significant aberration when operated away from normal incidence. Toroidal mirrors, when operated in a  $2f - 2f$  configuration, can re-image the source point without introducing any aberrations at the focus [4]. Additionally, the toroid's constant curvature allows it to be manufactured to a higher degree of accuracy than other shapes [16]. However, toroids introduce significant coma when the entrance and exit arms are unequal (i.e., when  $M \neq 1$ ). These aberrations can be minimized by using a two-toroid configuration, but this introduces other problems. The second bounce greatly reduces the transmission of the optical system as a whole, as shown in Fig. 2.12. The reflection losses of the second optic could be offset by reducing the grazing angle of the pair, but at the cost of increasing the footprint of each optic. For example, an optic pair must operate at 88 degrees to achieve the reflectivity of a single optic at 85 degrees. Applying Eq. (2.19), we can see that for a fixed input beam size, this 3 degree change increases the required dimensions of each mirror by 250%. Even

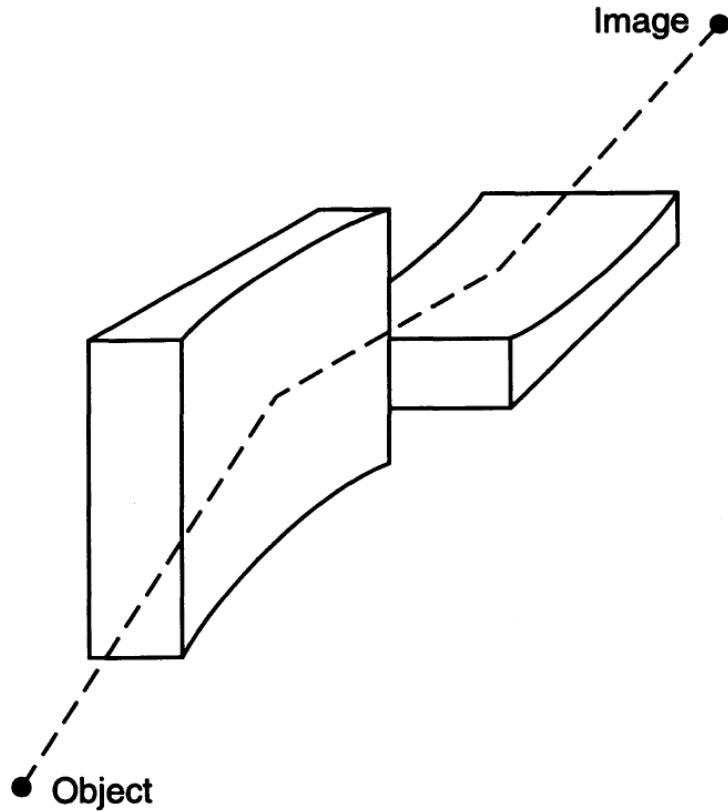


Figure 2.16: placeholder text.

if the spacing between the two mirrors was negligible, a two-optic solution would increase the overall footprint of the refocusing system by a factor of 5. Furthermore, the presence of an additional reflective optic within the interferometer reduces the interferometric stability and makes the optical alignment of the system more difficult. The above arguments can also be applied to Kirkpatrick-Baez (KB) pairs [19] or other two-optic configurations. clearly a single optic solution was required.

connection between aberrations and temporal smearing [1]

### KB pairs

original citation [19]

geometric description

avoided due to size of optics

disadvantage: multi-mirror configuration. difficult to align, extra bounce causes more reflection losses.

# 3D line picture showing bicycle tire geometry of toroidal mirror

Figure 2.17: placeholder text.

## toroidal mirrors

geometric description of toroidal mirror, including equations / parameters

$$\frac{1}{r} + \frac{1}{r'} = \frac{2}{R \cos \alpha} \quad (2.23)$$

$$\frac{1}{r} + \frac{1}{r'} = \frac{2 \cos \alpha}{\rho} \quad (2.24)$$

$$f_t = \frac{1}{2} R \cos \alpha \quad (2.25)$$

$$f_s = \frac{1}{2} \frac{\rho}{\cos \alpha} \quad (2.26)$$

setting  $f_t = f_s$ , we obtain  $r/R = \cos^2 \alpha$ .

$$\Delta y'_{lc} = \frac{1}{2} \frac{l^2 \sin \alpha}{\rho} (M - 1) \quad (2.27)$$

where  $M = r/r'$  is the magnification.

figure showing geometry

equal path length equations - choosing the optimal source, focal positions  
coma and aberration when magnification is not unity

# FRED output showing B-2 bomber coma of toroidal mirror

Figure 2.18: placeholder text.

multi-mirror configurations (avoided due to complexity, size)

## **ellipsoidal mirrors**

geometric description, including equations. what parameters describe this optic?

figure showing geometry

coma and aberration are not an issue - why not?

advantage: single-mirror configuration

disadvantage: difficulty in manufacturing a non-spherical / non-cylindrical surface. solution: active

## **2.7.2 entrance and exit arm length choice**

2250 mm entrance, 750 mm exit arm.

due to the modular nature of the beamline, we were limited on how short the exit arm could be. given the chamber geometries and the placement of the recombination optic, the exit arm had to be at least 750 mm.

## **2.7.3 mirror physical dimensions choice**

this had to do with the assumed divergence of the XUV light and the entrance arm length.

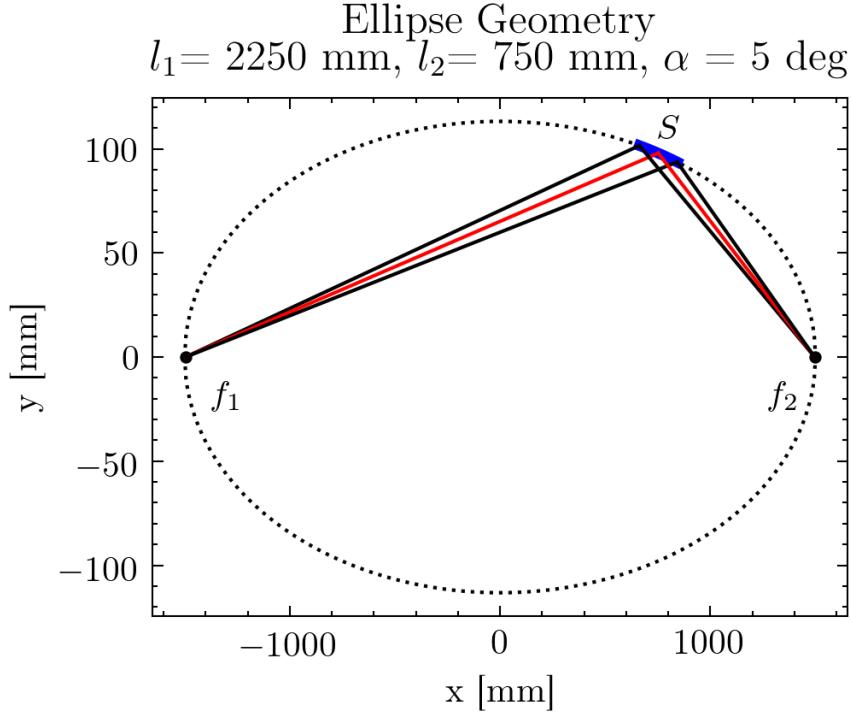


Figure 2.19: Overhead view of the ellipsoidal mirror geometry. Mirror surface  $S$  is shown in blue; foci  $f_{1,2}$  are represented by black dots. Rays that strike the center (red) and the edges (black) of the mirror are depicted as lines. Vertical scale is enlarged to show detail.

#### 2.7.4 ellipsoid mirror math

The surface of an ellipsoidal mirror is a section of an ellipsoid of revolution. In a Cartesian coordinate system, an ellipsoid of revolution is described by parameters  $a$  and  $b$ :

$$\frac{x^2}{a^2} + \frac{y^2 + z^2}{b^2} = 1 \quad (2.28)$$

where the  $xy$  plane is the optical plane and  $z$  is the vertical direction. Due to the symmetry between  $y$  and  $z$ , much of the analysis can be done on a two-dimensional ellipse:

$$\frac{x^2}{a^2} + \frac{y^2}{b^2} = 1 \quad (2.29)$$

In the  $xy$  plane, the surface of the mirror is a segment of the curve formed by Eq. (2.29). The center of the mirror is centered at position  $(x, y) = (x_M, y_M)$ . The ellipse has eccentricity  $\epsilon$ ,

$$\epsilon = \sqrt{1 - \frac{b^2}{a^2}}, \quad (2.30)$$

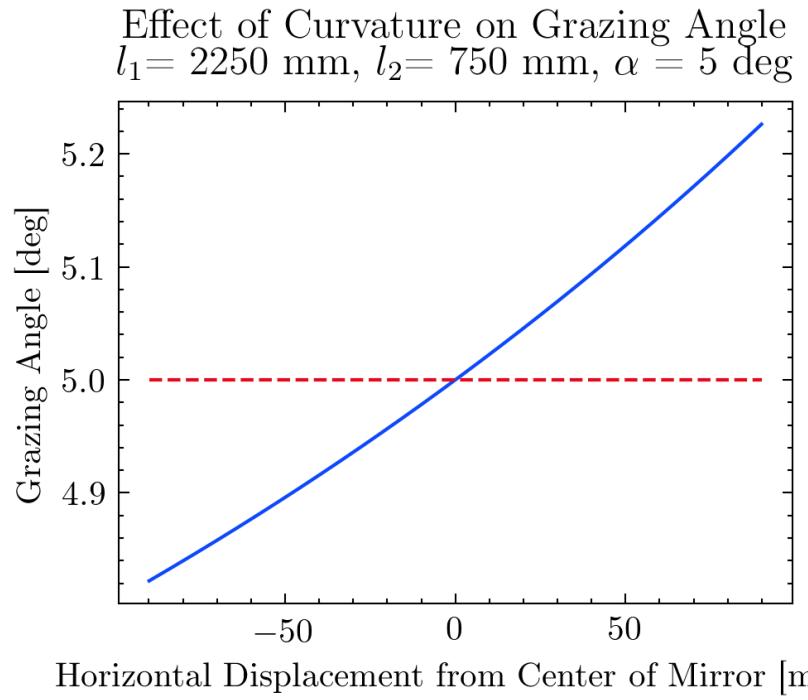


Figure 2.20: The effect of curvature on the local grazing angle. Light that strikes the edges of the mirror will experience a slightly different grazing angle than the design angle.

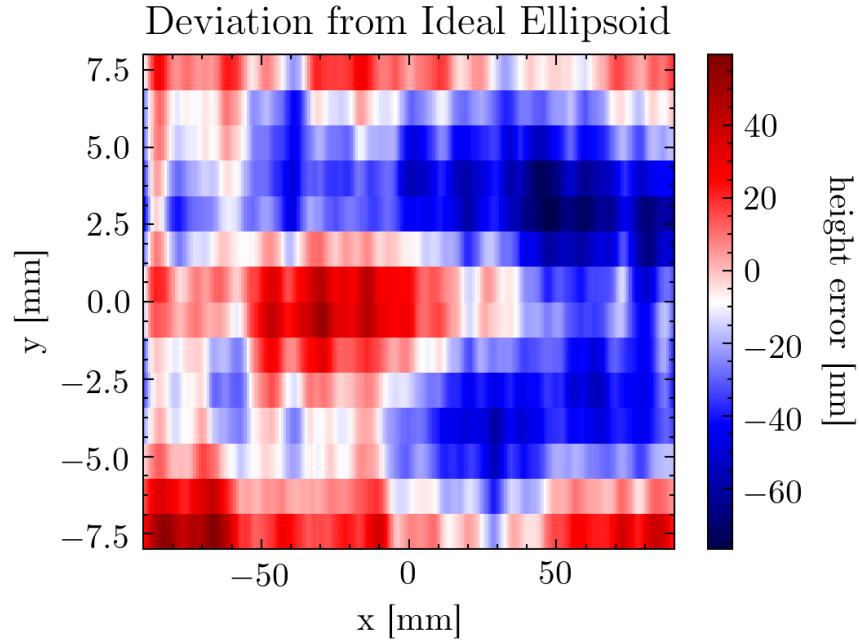


Figure 2.21: Data provided by Carl Zeiss Laser Optics GmbH.

and focal points at positions  $(x, y) = (\pm a\epsilon, 0)$ . Consider a light source at the first focal point,  $f_1 = (-a\epsilon, 0)$ : rays emanating from  $f_1$  will strike surface of the ellipse at some point  $S = (x_0, y_0)$  and focus to  $f_2 = (a\epsilon, 0)$ . This geometry is shown in Fig. 2.19. This optical system has an entrance arm  $l_1 = f_1 S$ , exit arm  $l_2 = S f_2$ , and demagnification  $M$  given by:

$$l_1(x_0, y_0) = \sqrt{(x_0 + a\epsilon)^2 + y_0^2} \quad (2.31)$$

$$l_2(x_0, y_0) = \sqrt{(x_0 - a\epsilon)^2 + y_0^2} \quad (2.32)$$

$$M = \frac{l_1}{l_2} \quad (2.33)$$

Once the  $f_1$ ,  $f_2$  and  $S$  are defined, the grazing angle  $\alpha$  of reflection can be found by applying the Law of Cosines:

$$\alpha = \frac{1}{2} \arccos \left( \frac{(2a\epsilon)^2 - l_1^2 - l_2^2}{2l_1 l_2} \right) \quad (2.34)$$

The finite spatial extent of a physical mirror will result in a range of grazing angles across the surface of the mirror. The grazing angle for our mirror is shown in Fig. 2.20. The curvature of the ellipsoidal mirror is characterized by three parameters: the semi-major axis  $a$ , semi-minor axis  $b$ , and off-axis position  $x_M$ , or equivalently by the entrance arm  $l_1$ , exit arm  $l_2$  and grazing angle  $\alpha$ . Additionally, the spatial extent (clear aperture) of the mirror must be specified by the tangential length  $L$  and the out-of-plane width  $W$ .

The problem with an adhesion layer is that over time it diffuses to the surface, so eventually you may observe Cr lines in your spectrum. However, without the adhesion layer the Au film is quite fragile.

The actual roughness-values will be somewhere between 0.3 and 5nm RMS, degrading to the edge.

## 2.8 HHG Gas Source

Transient absorption experiments require a high XUV photon flux for a variety of reasons. First, the sample thickness is usually chosen such that the XUV transmission is roughly 50% near the spectral feature of interest. This optical density represents a compromise between the incompatible goals of having a strong ground state absorption (which allows you to easily detect small changes in optical density) while simultaneously avoiding the noise floor of the detector (which is required for good statistics). As a result, half of the XUV photons will never reach the detector, so you better be making a lot of them. Second, a high XUV flux will reduce the number of laser shots required for a given data point, which in turn reduces the chances of inadvertently damaging your sample with the infrared laser. Finally, a high flux reduces the overall time required to complete an experiment. Besides the obvious benefit of happier graduate students, the ability to quickly perform an

experiment increases data fidelity by reducing the effects of unavoidable experimental noise sources such as long-term laser drift (either pointing or energy) or environmental changes caused by the building's HVAC system.

Depending on the energy of the spectral feature, obtaining a high photon flux can range from trivial to challenging. There are many (usually interdependent) experimental parameters (gas type, interaction pressure and length, wavelength, intensity, confocal parameter, focal position relative to gas source, etc.) that can be tuned to optimize photon flux. Physically, these parameters can change the microscopic single atom response, the macroscopic coherent addition of dipole emitters (via phase matching), or both. Each experiment will usually require a unique combination of experimental settings to achieve a usable light source. For example, optimizing the harmonic yield at 100 eV for a Si L-edge measurement will usually come at the expense of harmonics yield in the 30-50 eV range, which are used to measure the transition metal M-edges.

In general, an experimentalist has neither perfect knowledge nor control over all the variables that contribute towards phase matching. Setting aside the complicated topic of phase matching, the one dimensional on-axis phase matching model[6] shows that the photon flux is proportional to the square of the pressure-length product of the interaction gas. That is, so long as we can remain phase matched and below the critical phase matching pressure[24], we can universally increase the harmonic flux of our experiments by increasing the pressure-length product.

Unfortunately, one cannot ignore phase matching. Oftentimes, the spectral feature of interest lies beyond the harmonic cutoff when using the more convenient shorter wavelengths. In this case, the fundamental wavelength is increased to extend the cutoff (which scales as  $\lambda^2$ ). However, the critical phase matching pressure also scales as  $\lambda^2$  [24], and the single atom response scales as  $\lambda^{-(5-6)}$  [31]. These two combined effects result in a dramatically decreased photon flux if intensity and pressure are kept constant with increasing wavelength, often to the point that the resulting flux is insufficient for a transient absorption experiment, even though your cutoff has been extended to the proper energy. While some of the flux can be recovered by increasing the backing pressure of the continuous free expansion nozzle, the generation chamber's finite pumping speed limits the efficacy of pressure tuning at the longer wavelengths. Even at 800 nm, the maximum backing pressure of the continuous free expansion nozzle results in an interaction pressure below the critical phase matching pressure. Practically speaking, the continuous free expansion gas nozzle is not suitable for transient absorption experiments using the signal wavelengths ( $\lambda > 1.6 \mu\text{m}$ ) or with spectral features greater than the aluminum edge at 72 eV.

Providing the lab with a brighter harmonic source was the ultimate goal of the high pressure cell, and for the most part this goal was achieved. Below, we will review the basic design considerations, drawbacks and advantages of the three main types of gas sources

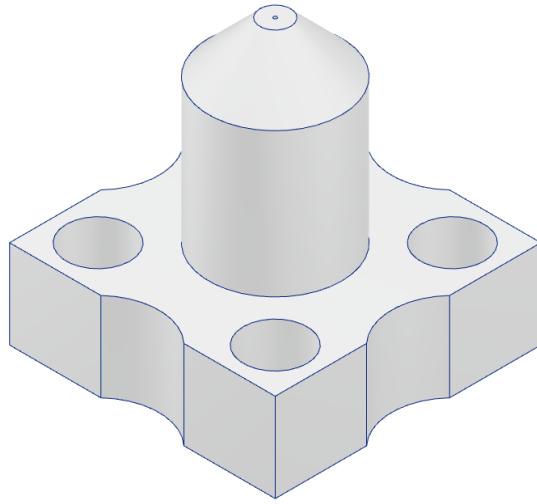


Figure 2.22: The continuous free expansion nozzle. Gas flows from the base of the nozzle and out of the  $200\ \mu\text{m}$  aperture. The large through holes on the base of the nozzle are for mounting to the gas delivery system; the sidewall cuts are for clearance for other mounting hardware. The top surface is beveled to reduce the minimum allowable distance between the laser axis and the nozzle.

used in this thesis: the free expansion nozzle, the low pressure cell and the high pressure cell. A primer on how to install and use the high pressure cell can be found in Appendix ??.

### 2.8.1 Continuous Free Expansion Nozzle

When gas flows from a high pressure region ( $P_0$ ) to a low pressure region ( $P_b$ ) through a small aperture, a plume will form in the low pressure region. If the pressure ratio  $P_0/P_b$  exceeds a critical value  $G \equiv ((\gamma+1)/2)^{\gamma/(\gamma-1)}$ , then the gas flow may exceed the local speed of sound. This critical value is at most 2.1 for all gases, so we easily exceed it in all of our experiments.<sup>3</sup> It is therefore necessary to understand the basic properties of supersonic gas plumes.

The continuous free expansion nozzle consists of a small diameter hole drilled in a block of aluminum, shaped to be convenient for gas delivery and assembly.<sup>4</sup> The basic design is shown in Fig. 2.22. The structure of the resulting supersonic plume is shown in Fig. 2.23. The physics of supersonic gas flow has been extensively studied in the literature and will

<sup>3</sup>The highest chamber pressures in our experiments are on the order of  $P_b \approx 10\ \text{mTorr}$ . Therefore, any nozzle that is backed by more than  $P_0 \approx 21\ \text{mTorr}$  will result in a supersonic gas flow. Typical backing pressures for harmonic generation are on the order of 250 Torr, putting us well within the supersonic regime.

<sup>4</sup>To reduce the gas load on the pumps, we used  $200\ \mu\text{m}$  diameter nozzles. This was the smallest size hole the machine shop could readily drill into aluminum.

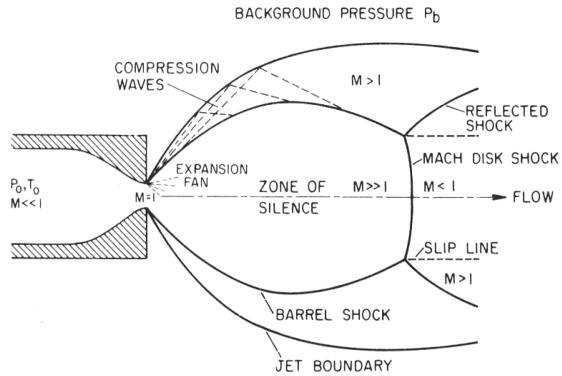


Figure 2.23: The structure of the supersonic gas plume after leaving a gas nozzle. This figure was taken from Ref [21].

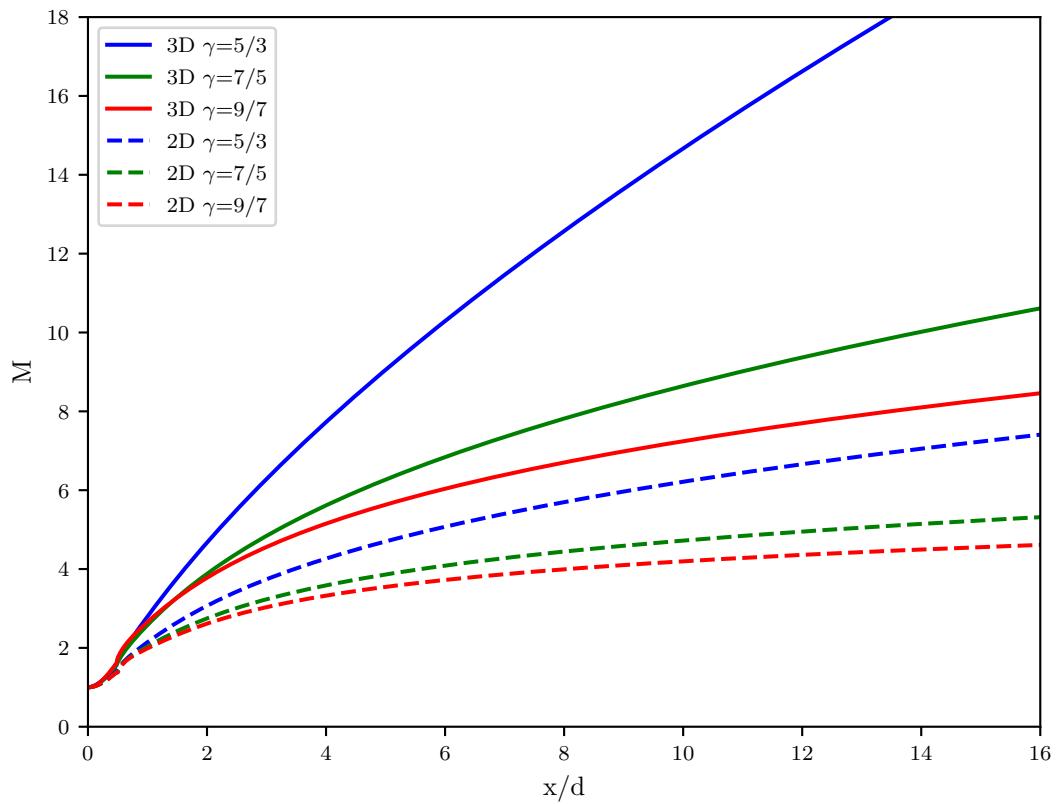


Figure 2.24: Centerline Mach number versus distance in nozzle diameters for 2D (planar) and 3D (axisymmetric) geometries, calculated using Eq. (2.40).

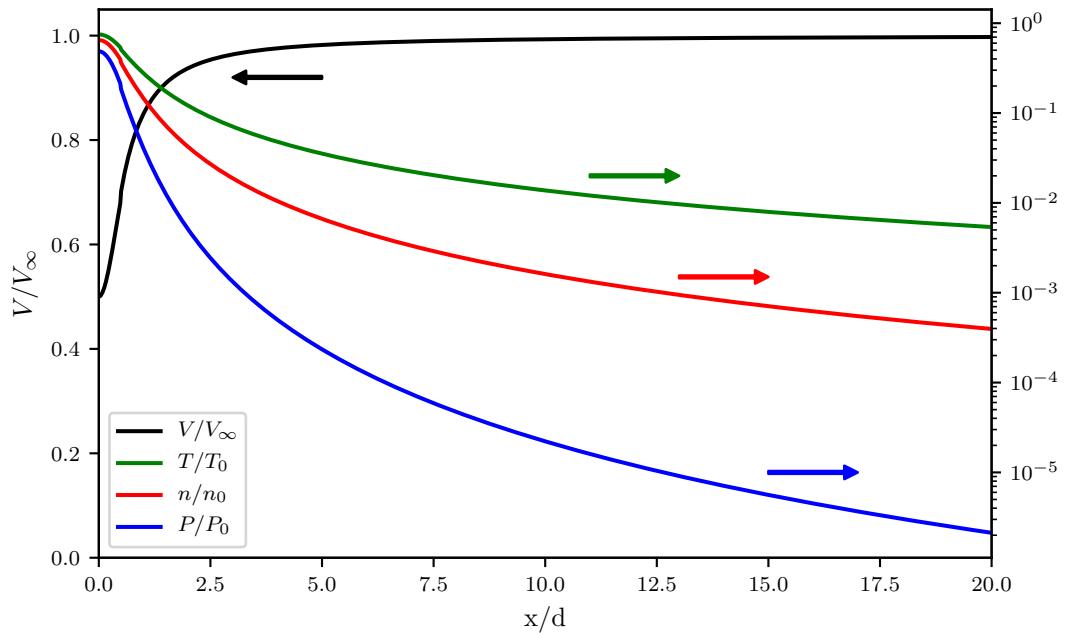


Figure 2.25: Free jet centerline properties versus distance in nozzle diameters for helium gas ( $\gamma=5/3$ ,  $W=4$ ). Mach number is calculated using Eq. (2.40), and the centerline properties are calculated using Eq. (2.38). Velocity  $V$  is scaled by terminal velocity  $V_\infty$ ; temperature  $T$ , number density  $n$  and pressure  $P$  are normalized by source stagnation values  $T_0$ ,  $n_0$ ,  $P_0$ .

not be discussed at length here. Below is a brief overview of the relevant physics required to understand the gas nozzles used for HHG in our lab. For a more detailed review of the field, see Ref [21].

energy equation.  $V$  is velocity,  $h$  is enthalpy per unit mass.

$$h + V^2/2 = h_0 \quad (2.35)$$

for ideal gases,  $dh = \hat{C}_p dt$ , and we have

$$V^2 = 2(h_0 - h) = 2 \int_T^{T_0} \hat{C}_p dT \quad (2.36)$$

For an ideal gas,  $\hat{C}_p = \gamma/(\gamma - 1)(R/W)$ , where  $\gamma = C_p/C_V$  is the ratio of the specific heats,  $R$  is the gas constant,  $W$  is the molecular weight. if the gas is cooled substantially in the expansion ( $T \ll T_0$ ), then we have:

$$V_\infty = \sqrt{\frac{2R}{W} \left( \frac{\gamma}{\gamma - 1} \right) T_0} \quad (2.37)$$

For an ideal gas, the speed of sound is  $a = \sqrt{\gamma RT/W}$  and the Mach number is  $M = V/a$ . Assuming  $\hat{C}_p$  is constant, we can recast Eq. (2.36) in terms of  $\gamma$  and  $M$ . Using these assumptions, one can obtain the following relationships for the temperature  $T$ , velocity  $V$ , pressure  $P$ , mass density  $\rho$  and number density  $n$  in the gas jet scaled to those parameters at the stagnation point ( $T_0, P_0, \rho_0, n_0$ ):

$$(T/T_0) = \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-1} \quad (2.38a)$$

$$V = M \sqrt{\frac{\gamma R T_0}{W}} \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-1/2} \quad (2.38b)$$

$$(P/P_0) = (T/T_0)^{\gamma/(\gamma-1)} = \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-\gamma/(\gamma-1)} \quad (2.38c)$$

$$(\rho/\rho_0) = (n/n_0) = (T/T_0)^{1/(\gamma-1)} = \left( 1 + \frac{\gamma - 1}{2} M^2 \right)^{-1/(\gamma-1)} \quad (2.38d)$$

Therefore, once we know the Mach number  $M$ , we can calculate the above properties for the gas jet. The Mach number is found by solving the fluid mechanics equations dealing with the conversation of mass, momentum and energy:

| Source | $j$ | $\gamma$ | $C_1$ | $C_2$   | $C_3$  | $C_4$   | $A$   | $B$    |
|--------|-----|----------|-------|---------|--------|---------|-------|--------|
| 3D     | 1   | 5/3      | 3.232 | -0.7563 | 0.3937 | -0.0729 | 3.337 | -1.541 |
| 3D     | 1   | 7/5      | 3.606 | -1.742  | 0.9226 | -0.2069 | 3.190 | -1.610 |
| 3D     | 1   | 9/7      | 3.971 | -2.327  | 1.326  | -0.311  | 3.609 | -1.950 |
| 2D     | 2   | 5/3      | 3.038 | -1.629  | 0.9587 | -0.2229 | 2.339 | -1.194 |
| 2D     | 2   | 7/5      | 3.185 | -2.195  | 1.391  | -0.3436 | 2.261 | -1.224 |
| 2D     | 2   | 9/7      | 3.252 | -2.473  | 1.616  | -0.4068 | 2.219 | -1.231 |

Table 2.1: Gas parameters used in free expansion calculations, with Eq. (2.40). Table recreated from Ref [21].

$$\text{mass: } \nabla \cdot (\rho \mathbf{V}) = 0 \quad (2.39\text{a})$$

$$\text{momentum: } \rho \mathbf{V} \cdot \nabla \mathbf{V} = -\nabla P \quad (2.39\text{b})$$

$$\text{energy: } \mathbf{V} \cdot \nabla h_0 = 0 \text{ or } h_0 = \text{constant along streamlines} \quad (2.39\text{c})$$

$$\text{equation of state: } P = \rho \frac{R}{W} T \quad (2.39\text{d})$$

$$\text{thermal equation of state: } dh = \hat{C}_P dT \quad (2.39\text{e})$$

The above equations are valid for an isentropic, compressible flow of a single component ideal gas molecular weight  $W$  and constant specific heat ratio  $\gamma$ . A steady state is assumed and viscosity and heat conduction are neglected. These equations have been numerically solved in the literature for two source geometries: a “slit” nozzle (2D, planar) and a circular aperture (3D, axisymmetric). The numerical solutions to each geometry scale with the nozzle diameter  $d$ , and have been fit to the following analytical functions:

$$\frac{x}{d} > 0.5 : \quad M = \left( \frac{x}{d} \right)^{(\gamma-1)/j} \left[ C_1 + \frac{C_2}{(\frac{x}{d})} + \frac{C_3}{(\frac{x}{d})^2} + \frac{C_4}{(\frac{x}{d})^3} \right] \quad (2.40\text{a})$$

$$0 < \frac{x}{d} < 1.0 : \quad M = 1.0 + A \left( \frac{x}{d} \right)^2 + B \left( \frac{x}{d} \right)^3 \quad (2.40\text{b})$$

**question: why does  $M$  increase without bound with increasing  $x$ , while  $\mathbf{V}$  is limited to a finite value? scoles has a discussion, you should address it here.**

The fitting coefficients for Eq. (2.40) are listed in Table 2.1. A plot of the results for different source geometries and gases are shown in Fig. 2.24.

Table 2.3 shows the centerline Mach numbers used in the following equations:

$$M = A \left( \frac{x - x_0}{d} \right)^{\gamma-1} - \frac{\frac{1}{2} \left( \frac{\gamma+1}{\gamma-1} \right)}{A \left( \frac{x-x_0}{d} \right)^{\gamma-1}} \quad (2.41a)$$

$$\frac{\rho(y, x)}{\rho(0, x)} = \cos^2(\theta) \cos^2 \left( \frac{\pi\theta}{2\phi} \right) \quad (2.41b)$$

$$\frac{\rho(R, \theta)}{\rho(R, 0)} = \cos^2 \left( \frac{\pi\theta}{2\phi} \right) \quad (2.41c)$$

$$\left( \frac{x}{d} \right) > \left( \frac{x}{d} \right)_{\min} \quad (2.41d)$$

The gas nozzle throughput  $\hat{T}$  is calculated from:

$$\hat{T} \text{ (torr} \cdot \text{l/s}) = \hat{S} \cdot P_b = C \left( \frac{T_C}{T_0} \right) \sqrt{\frac{300}{T_0}} (P_0 d) d \quad (2.42)$$

where  $C$  is the gas constant from ??,  $P_0$  is the nozzle's backing pressure in Torr,  $T_C$  and  $T_0$  are the vacuum chamber and backing temperatures, respectively, in Kelvin,  $P_0$  is the backing pressure in Torr, and  $d$  is the nozzle's diameter in cm.

S = pumping speed? Pb = chamber pressure

(how was this equation derived?)

Note that the gas nozzle throughput is proportional to both backing pressure and diameter of the nozzle. For our vacuum system, the generation chamber has a pumping speed of approximately ???;

Relevant gas parameters are listed in ??.

condition for supersonic flow: the pressure ratio  $P_0/P_b$  exceeds a critical value  $G \equiv ((\gamma + 1)/2)^{\gamma/(\gamma-1)}$ , which is less than 2.1 for all gases. Since the vacuum chamber pressure is at most 10 mTorr, just about any backing pressure will result in supersonic flow out of the gas nozzle.

Mach disk location:  $x_M/d = 0.67(P_0/P_b)^{1/2}$ . for example, for a chamber pressure of 10 mTorr and a backing pressure of -5 psig ( $\sim 450$  Torr), the Mach disk is located about 45 nozzle diameters away from the orifice. for a 200 micron diameter nozzle, that's about 9 cm.

basic design of free expansion nozzle

throughput calculations (Scoles)

harmonic yield results

note: i did not design this cell

advantages: easy to align, cheap to produce, free-expansion cooling (for alignment experiments)

disadvantages: small pressure length product. very low interaction pressure. impossible to phase match longer wavelengths. overall low yield.

| Gas             | $\epsilon/k$ (K) | $\sigma$ (angstrom) | $C_6/k$ ( $10^{-43}$ K · cm $^6$ ) | $Z_r$ | C (l/cm $^2$ /s);<br>Eq. (2.42) |
|-----------------|------------------|---------------------|------------------------------------|-------|---------------------------------|
| He              | 10.9             | 2.66                | 0.154                              | -     | 45                              |
| Ne              | 43.8             | 2.75                | 0.758                              | -     | 20                              |
| Ar              | 144.4            | 3.33                | 7.88                               | -     | 14                              |
| Kr              | 190              | 3.59                | 16.3                               | -     | 9.8                             |
| Xe              | 163              | 4.3                 | 41.2                               | -     | 7.9                             |
| H <sub>2</sub>  | 39.6             | 2.76                | 0.7                                | ~300  | 60-63                           |
| D <sub>2</sub>  | 35.2             | 2.95                | 0.93                               | ~200  | 42                              |
| N <sub>2</sub>  | 47.6             | 3.85                | 6.2                                | ~2.5  | 16                              |
| CO              | 32.8             | 3.92                | 4.76                               | ~4.5  | 16                              |
| CO <sub>2</sub> | 190              | 4.0                 | 31.1                               | ~2.5  | 12-13                           |
| CH <sub>4</sub> | 148              | 3.81                | 18.1                               | ~15   | 21                              |
| O <sub>2</sub>  | 115              | 3.49                | 8.31                               | ~2    | 15                              |
| F <sub>2</sub>  | 121              | 3.6                 | 10.5                               | ~3.5  | 14                              |
| I <sub>2</sub>  | 550              | 4.98                | 336                                | ~1    | 5.2                             |

Table 2.2: Gas parameters used in free expansion calculations. Table recreated from Ref [21].

| $\gamma$ | $x_0/d$ | $A$  | $\phi$ | $(x/d)_{\min}$ |
|----------|---------|------|--------|----------------|
| 1.67     | 0.075   | 3.26 | 1.365  | 2.5            |
| 1.40     | 0.4     | 3.65 | 1.662  | 6              |
| 1.2857   | 0.85    | 3.96 | 1.888  | 4              |
| 1.20     | 1.00    | 4.29 | -      | -              |
| 1.10     | 1.60    | 5.25 | -      | -              |
| 1.05     | 1.80    | 6.44 | -      | -              |

Table 2.3: Centerline Mach Number and Off-Axis Density Correlations for Axisymmetric Flow. Table recreated from Ref [21].

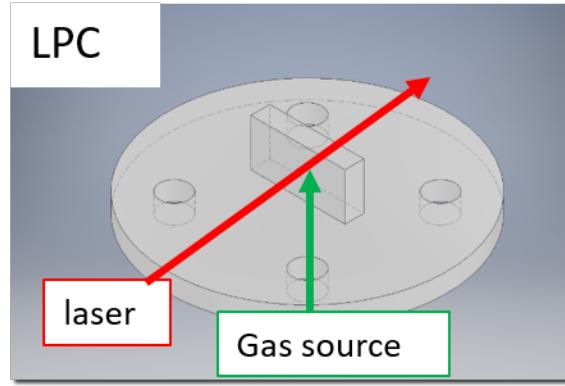


Figure 2.26: Detail of the LPC interaction region.

### 2.8.2 low pressure cell

basic design of low pressure cell – gas load, Rayleigh range, spot size, laser drift

gas load calculations (simple model)

harmonic yield results

note that i did not design this cell. design is from (now Dr.) Zhou Wang.

advantages: increased interaction length - brighter! easy to align.

disadvantages: relative to the free expansion nozzle, you don't get any cooling.

### 2.8.3 high pressure cell

- why didn't you go with semi-infinite gas cell, or fiber-cell?

- basic design of high pressure cell

- limited pump speed → differential pumping is required

- gas load calculations (simple model)

- harmonic yield results

- advantages: much brighter due to pressure-length product. future application: can operate in low-pressure mode and reduce downstream generation gas contamination of target chamber.

- disadvantages: difficult to align and initially install (once it's installed, alignment is easy). messed up mode. HHG instability at higher pressures.

### 2.8.4 pulsed valve

expensive

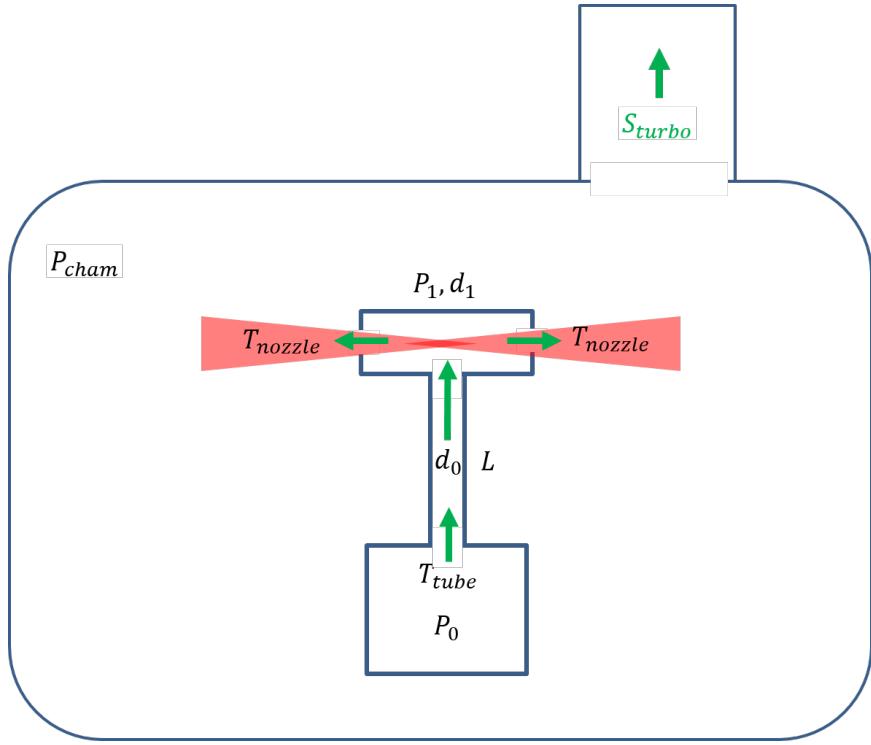


Figure 2.27: Gas flow schematic of the LPC. The green arrows indicate the direction of gas flow, and the red shaded region indicates the laser path. An infinite reservoir of gas with backing pressure  $P_0$  supplies the laser interaction region with gas via a thin capillary of diameter  $d_0$ , length  $L$  and throughput  $T_{tube}$ . The interaction region has pressure  $P_1$  and diameter  $d_1$ . The interaction region acts as a pressure source for two diametrically opposed supersonic gas jets, each with throughput  $T_{nozzle}$ . The generation chamber has a turbopump with pumping speed  $S_{turbo}$  and an equilibrium pressure  $P_{cham}$ .

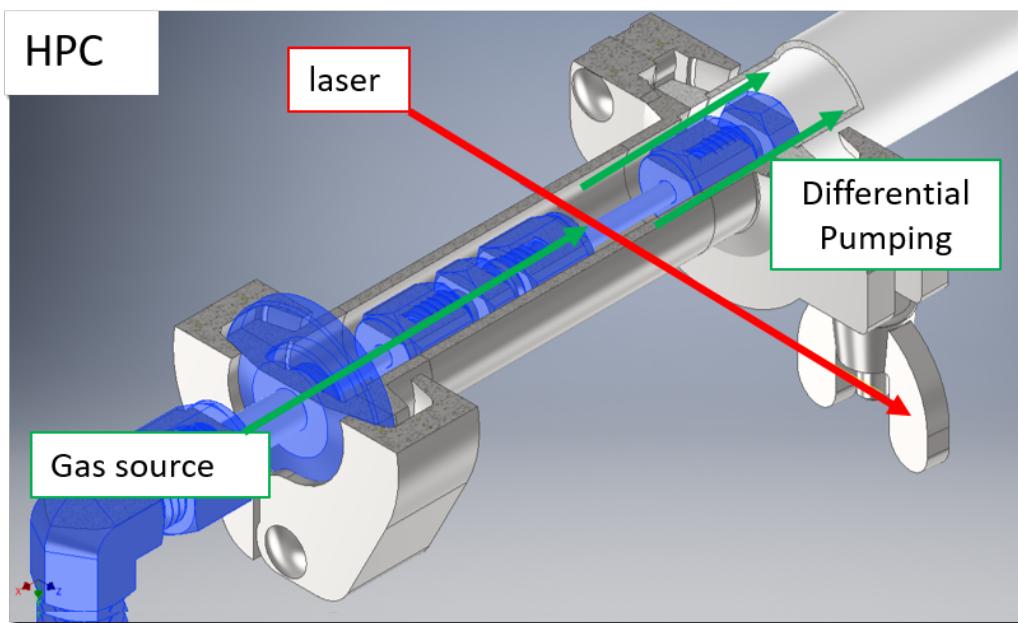


Figure 2.28: Detail of the HPC interaction region. From bottom left to top right: welded gas feedthrough, concentric inner & outer pipes, edge-welded bellows. The high pressure region is shaded blue. The green lines indicate the gas flow direction; the red line indicates the laser propagation direction.

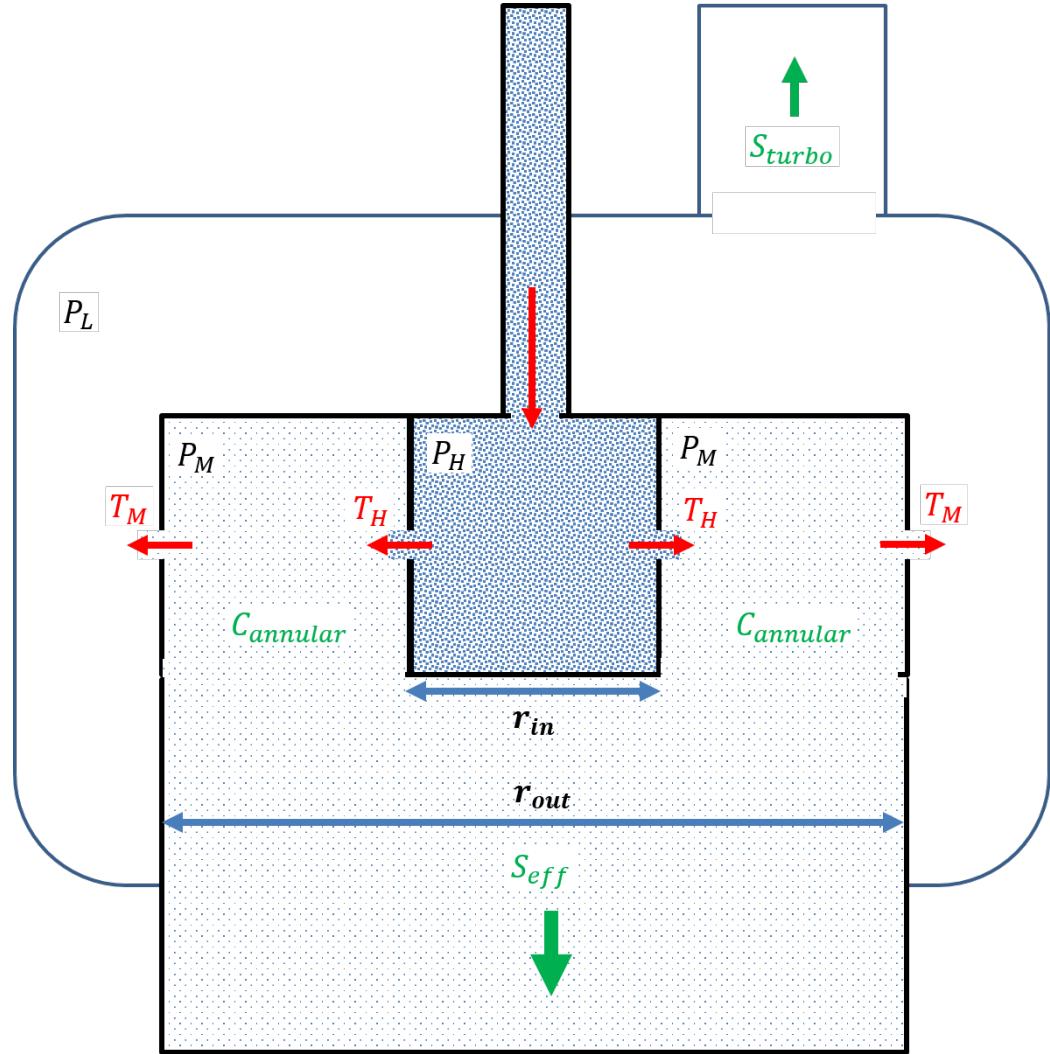


Figure 2.29: Schematic used to calculate the pressures inside the HPC and generation chamber. The dark blue region represents the inner pipe, the light blue region represents the outer pipe. Red arrows and text indicate gas sources, green arrows and text indicate flow towards the vacuum pumps; blue arrows and text indicate physical dimensions.  $P_H$ ,  $P_M$ , and  $P_L$  are the pressures of the inner pipe, outer pipe, and generation chamber, respectively;  $S_{turbo}$ ,  $S_{eff}$  and  $C_{annular}$  are the turbo pumping speed, effective rough pump speed and annular conductance, respectively;  $T_H$  ( $T_M$ ) is the gas throughput from the high (medium) pressure region into the medium (low) pressure region.

# Chapter 3

## XUV LIGHT SOURCE DESIGN AND APPARATUS PERFORMANCE

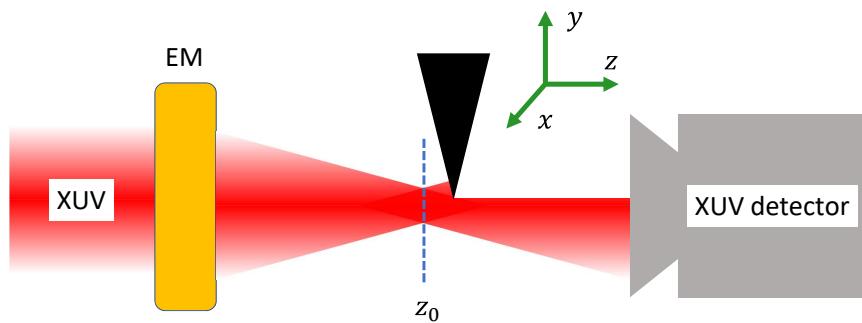


Figure 3.1: Schematic of XUV knife edge measurement. EM: ellipsoidal mirror,  $z_0$ : XUV focal plane.

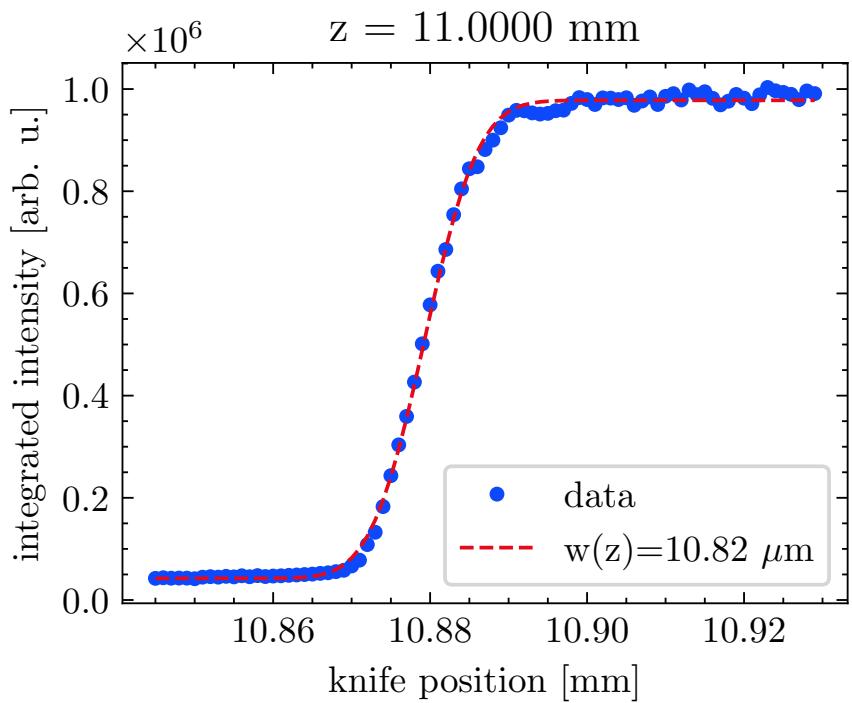


Figure 3.2: A typical XUV knife edge measurement near the focal plane. The sample motor position is  $k = 11.0000 \text{ mm}$ . A fit to equation Eq. (3.3) yields a beam waist of  $10.82 \mu\text{m}$  at this position.

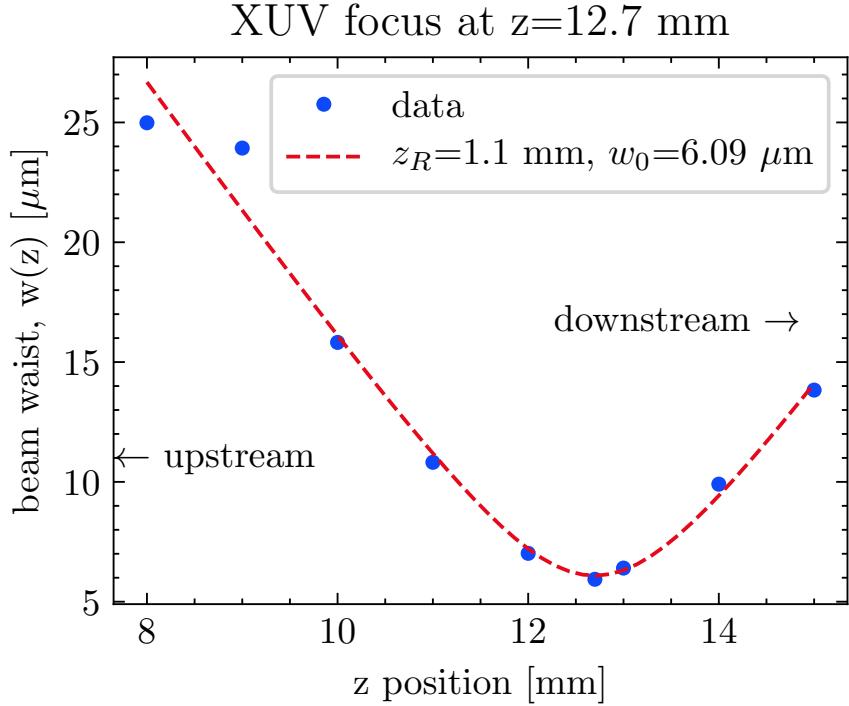


Figure 3.3: Evolution of XUV beam waist as a function of propagation direction,  $z$ . The Rayleigh range  $z_R$  and beam waist  $w_0$  are extracted from the fit to Eq. (3.2).

### 3.1 The need for high XUV flux in ATAS experiments

### 3.2 Harmonic Gas Sources

#### 3.2.1 free expansion gas jet nozzle

#### 3.2.2 low pressure cell

#### 3.2.3 high pressure cell

#### 3.2.4 amsterdam pulsed piezoelectric valve

### 3.3 characterization of XUV source

#### 3.3.1 knife edge measurements at XUV focus

We characterize the XUV focus in the target chamber by performing knife edge measurements at different  $k$ -positions, as depicted in Fig. 3.1. We use the interior angled edge of

the Si frame on a broken sample heterostructure as a knife edge (see ??). This frame makes an excellent knife edge as it has a very well-defined geometry and fits in the sample holder. Recalling Gaussian optics, the assumed profile of the XUV beam is:

$$I(x, y, z) = I_0 \left( \frac{w_0}{w(z)} \right)^2 \exp \left( -2((x - x_0)^2 + (y - y_0)^2)/w(z)^2 \right), \quad (3.1)$$

using the coordinate system defined in Fig. 3.1. The XUV focus is at position  $(x_0, y_0, z_0)$ . The beam waist  $w(z)$  will evolve as:

$$w(z) = w_0 \sqrt{1 + \left( \frac{z - z_0}{z_R} \right)^2}, \quad (3.2)$$

where  $z_R$  is the Rayleigh range. If we use the knife edge to block the transmission as depicted in Fig. 3.1, then the transmitted power will be:

$$P(x, z) = P_0 + \frac{P_{max}}{2} \left( 1 - \text{erf} \left( \frac{\sqrt{2}(x - x_0)}{w(z)} \right) \right), \quad (3.3)$$

where  $x$  is the insertion of the knife in the beam,  $z$  represents the location of the knife plane in the propagation direction, and erf is the error function.

A typical knife edge measurement is shown Fig. 3.2. In this measurement, the knife edge is translated across the XUV spot in  $1 \mu\text{m}$  steps until the XUV light is completely blocked. A 2D spectrum is saved at each knife edge position. Each image is background subtracted, normalized and summed (integrating over all divergences and wavelengths), which yields the XUV flux as a function of knife position. The resulting curve is fit to Eq. (3.3) and the beam waist  $w(z)$  is extracted for this  $z$ -position.

The knife edge measurement is repeated at different  $z$ -positions until enough data has been acquired to determine the focal plane. The evolution of the XUV beam waist is shown in Fig. 3.3. In this figure, the beam waist has been fit to Eq. (3.2) to determine the focal plane  $z_0$ , the Rayleigh range  $z_R$  and the beam waist  $w_0$ . In both figures, a reasonably good fit is obtained, indicating that the XUV light has a Gaussian spatial profile near the focus.

- 3.3.2 harmonic yield stability**
- 3.3.3 XYV spectra optimized for various HHG conditions**
- 3.3.4 Measured Transmission of Metallic Filters**
- 3.3.5 Ground State Measurements of Condensed Matter Samples**
- 3.4 characterization of interferometric stability**
- 3.5 MCP response**

scaling of yield and noise with respect to MCP voltage

# Chapter 4

## ATAS EXPERIMENTS IN GERMANIUM

### 4.1 Introduction

### 4.2 Experimental Considerations

#### 4.2.1 sample requirements

There are several sample requirements for a successful condensed matter transient absorption experiment. First and foremost, the sample needs to have an absorption edge within the bandwidth of the XUV source. Second, the material must be the correct thickness for a transmission measurement, given the capabilities of the XUV light source and detector. If the material is too thick, the ground state will absorb most of the XUV flux and the recorded spectrum will be too close to the noise floor of the apparatus. If it is too thin, the laser-induced change of the ground state (on the order of 1 – 10%) will be lost in the noise. As a general guideline, a sample that absorbs 50% at the spectral feature of interest provides a good compromise between these conflicting requirements. Fig. 4.1 shows the expected transmission of several materials, calculated from the atomic scattering factors [12]. We can see that a typical sample will be on the order of 10 - 200 nm thick, depending on the material.

Another upper bound for sample thickness comes from material dispersion. In any material, the XUV light ( $n_{\text{IR}} \sim 1$ ) will outpace the IR light ( $n_{\text{IR}} > 1$ ). This effect can be significant even for ultrathin films. In order to keep the phase slippage between the XUV and IR light below half an IR period, the sample thickness  $L$  must obey the following relationship:

$$L \leq \frac{1}{2} \frac{\lambda_{\text{IR}}}{n_{\text{IR}} - n_{\text{XUV}}} \quad (4.1)$$

For germanium excited with  $\lambda_{\text{IR}} = 1430$  nm and probed with 30 eV XUV at the  $M_{4,5}$  edge,  $n_{\text{IR}} = 4.2481$  [23] and  $n_{\text{XUV}} = 0.992536$  [12], which gives a maximum thickness of 220 nm.

Next, the sample needs to be excitable using laser sources present in our lab (i.e., ultrafast pulses with wavelengths between 800 nm and a couple of microns). To minimize

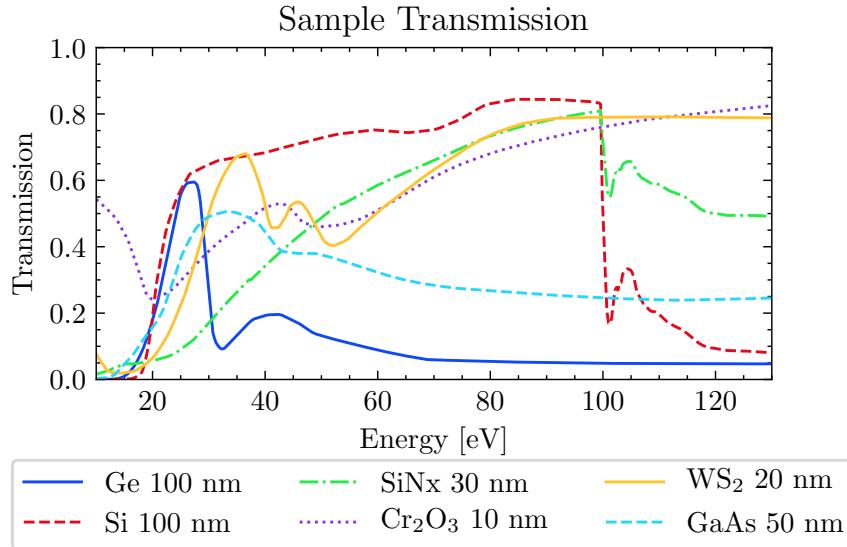


Figure 4.1: Calculated XUV transmission of various materials. Data from [12].

the slow build up of heat (on the order of seconds) and laser-induced damage, the sample needs to be rastered through the laser focus as the experiment is performed. This rastering method necessitates both a large clear aperture ( $\sim 1 \text{ mm}^2 - 1 \text{ cm}^2$ ) and good sample uniformity. Samples that meet the above thickness and clear aperture requirements are extremely delicate, with thicknesses between 5,000 and 100,000 times smaller than their freestanding lateral dimensions. As such, one should expect most samples to break before, during and after measurements, so a successful experiment will have a materials pipeline that is capable of producing multiple, consistent samples in a short time frame.

#### 4.2.2 rastering of sample through focus to avoid heating, charge build-up

#### 4.2.3 XUV maps of samples

#### 4.2.4 IR propagation in thin films (TMM starting with LightPipes output)

#### 4.2.5 orbital-resolved excitation probability vs wavelength (band structure calculations)

#### 4.2.6 laser damage

#### 4.2.7 estimation of excited carrier density

need to include orbital-resolved excitation probability results in this section

We are concerned with two quantities: the peak excitation fraction in the sample and

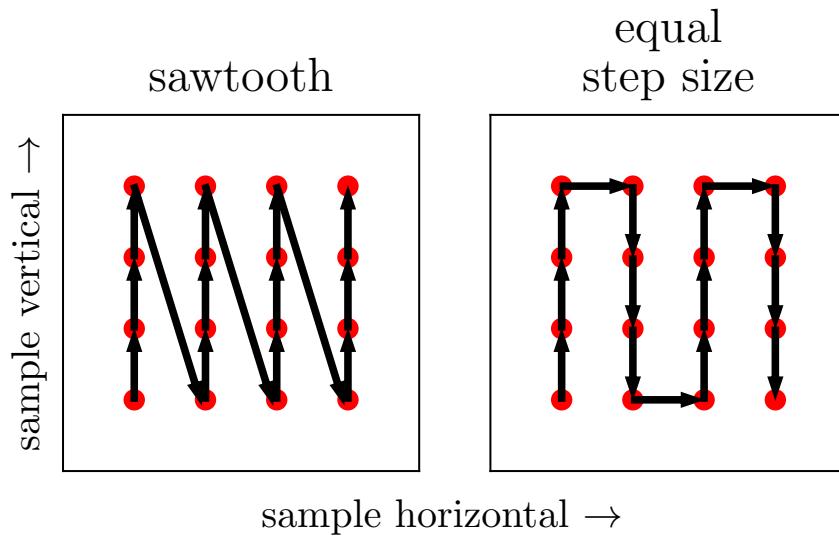


Figure 4.2: Schematic of competing raster methods, shown in the sample's reference frame. The clear aperture of the sample is represented by the interior of the black square. The laser propagation direction is out of the page. The laser focal spots are shown as red circles, and the movement of the sample holder relative to the laser focus is indicated by arrows. A  $200 \mu\text{m}$  border exists between the raster array and the perimeter of the sample's clear aperture. This diagram is to scale for a  $1 \times 1 \text{ mm}^2$  clear aperture sample, a  $60 \mu\text{m}$  diameter IR focal spot and a  $200 \mu\text{m}$  step size.

# TMM calculation showing abs. profile & intensity in sample vs thickness

Figure 4.3: thin film calculation made using the TMM package [2, 3].

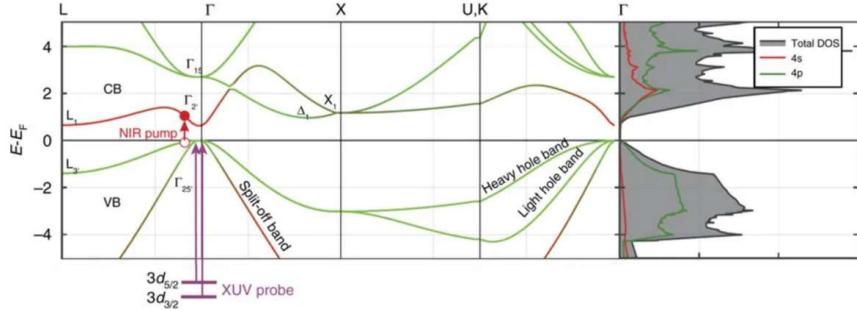


Figure 4.4: Band structure and orbital character of germanium. Purple arrows indicate XUV-induced transitions from the  $3d$  core levels to the valence bands. Red arrow indicates IR-induced transition across the direct band gap. Figure adapted from [35].

the average excitation fraction at the location of the XUV focus. The former quantity is relevant when considering sample damage, whereas the latter will be proportional to the measured signal. If the XUV and IR spots are perfectly overlapped at the sample plane, then these two quantities are approximately equal. We first calculate the peak excitation fraction, then we consider how a misaligned beam will affect the measured signal.

The laser propagation calculations in Fig. 2.14 were done for vacuum, but we are concerned with the field in our sample. The electric field inside a dielectric  $E_{\text{int}}$  is related to the external electric field by the following equation [27]:

$$E_{\text{int}} = \frac{2}{1 + \sqrt{\epsilon}} E_{\text{ext}} \quad (4.2)$$

where  $E_{\text{int}}$  is the electric field inside the sample,  $E_{\text{ext}}$  is the electric field outside the sample,  $\epsilon$  is the dielectric constant and its square root is the refractive index  $n_{\text{IR}}$ . The internal intensity  $I_{\text{int}}$  is the square of the internal electric field. For germanium at  $\lambda = 1430$  nm,  $n_{\text{IR}} = 4.2481$ , and we have the following relations:

$$\begin{aligned} E_{\text{int}} &= 0.381 \times E_{\text{ext}} \\ I_{\text{int}} &= 0.145 \times I_{\text{ext}} \end{aligned} \quad (4.3)$$

Given our laser parameters, we can estimate the highest carrier density within the sample. First, we estimate the absorbed laser fluence,  $F_{\text{abs}}$  [14]:

$$F_{\text{abs}} = F_{\text{inc}} (1 - R) (1 - \exp(-\alpha L)) (1 + R \exp(-\alpha L)), \quad (4.4)$$

where  $F_{\text{inc}}$  is the incident fluence,  $R$  is the reflectivity equal to the square of the Fresnel coefficient,  $\alpha$  is the absorption coefficient and  $L$  is the sample thickness. The bracketed terms in Eq. (4.4) are the fraction of fluence transmitted by the first surface, the fraction

absorbed by a single pass through the sample, and the additional absorption due to a back reflection off the rear face of the sample. Note that the back-propagating beam will arrive (on average) at a delay of  $n_{\text{IR}}L/(2c) \approx 0.7$  fs later than the forward-propagating beam. This time scale is nearly two orders of magnitude less than the IR pulse duration, so we should expect any electron dynamics initiated by the back reflection to contribute to the measured signal.

If each absorbed photon corresponds to an excited electron, then the excited carrier density  $\Delta N$  is given by the following expression [7]:

$$\Delta N = \frac{F_{\text{abs}}}{\hbar\omega} \frac{1}{L}, \quad (4.5)$$

where  $\hbar\omega$  is the IR photon energy. In Eq. (4.5), the quantity  $F_{\text{abs}}/(\hbar\omega)$  represents the number of absorbed photons per unit area; dividing this quantity by the sample thickness gives the number of absorbed photons per unit volume. This assumes that the skin depth of the material is greater than membrane thickness, which is true for germanium at these wavelengths.

Finally, we convert the excited carrier density to a fractional excitation. Germanium has  $N_{\text{u.c.}} = 2$  valence electrons per unit cell, and each unit cell has a volume  $V_{\text{u.c.}} = 4.527 \times 10^{-23} \text{ cm}^3$ . Therefore the fractional carrier excitation is

$$f = \Delta N \frac{V_{\text{u.c.}}}{N_{\text{u.c.}}} \quad (4.6)$$

We can use literature values for 100 nm of germanium pumped at  $\lambda = 1430$  nm light. From the literature [23],  $R = 0.38315$ ,  $\alpha = 5803.4 \text{ cm}^{-1}$ , and so  $F_{\text{abs}} = 0.0413 \times F_{\text{inc}}$ . Therefore, only about 4.13% of the incident fluence is absorbed by the sample.

According to the calculations in Fig. 2.14, for each 1  $\mu\text{J}$  energy input pulse (measured at L4), 0.413  $\mu\text{J}$  makes it to the focal plane. 49% of that energy is within the main lobe, which contains 0.202  $\mu\text{J}$  of energy. Approximating the central lobe as a Gaussian beam with a FWHM of 35  $\mu\text{m}$  and a pulse energy of 0.202  $\mu\text{J}$ , the peak fluence is calculated by dividing the total energy of the Gaussian by  $\pi w^2/2$ . The Gaussian beam waist  $w$  is related to the FWHM via  $w^2 = \text{FWHM}^2/(2 \ln 2)$ . Thus, for each 1  $\mu\text{J}$  input energy, the peak fluence in the central lobe is 14.6 mJ/cm<sup>2</sup> and the absorbed peak fluence is 0.60 mJ/cm<sup>2</sup>. This corresponds to an peak excited carrier density of  $4.3 \times 10^{20} \text{ cm}^{-3}$  and an excitation fraction of 0.98% (per 1  $\mu\text{J}$  of input energy).

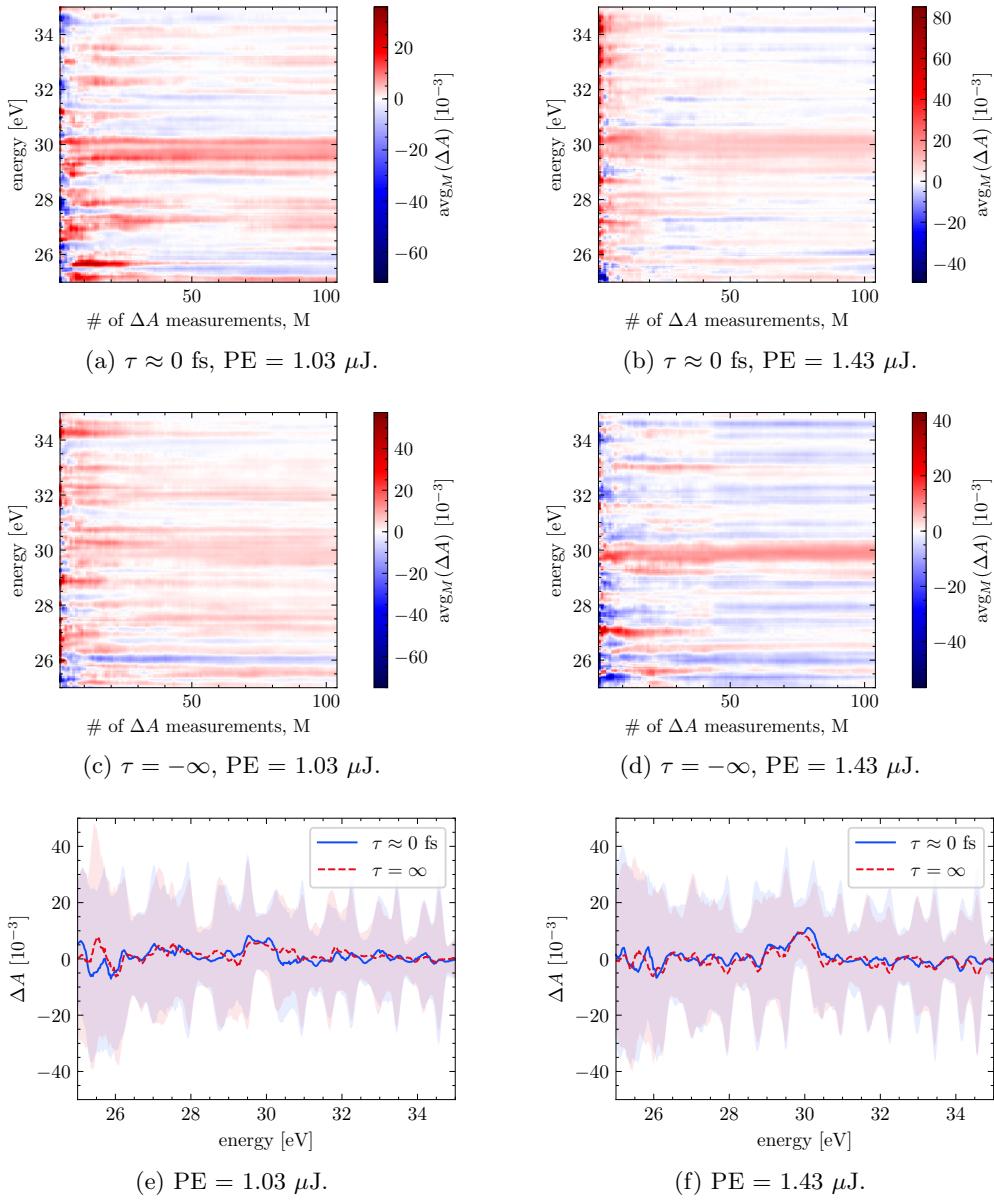


Figure 4.5: 1 kHz fixed-delay ATAS measurements on 100 nm Ge using a  $\lambda = 1450$  nm excitation pulse. See text for details.

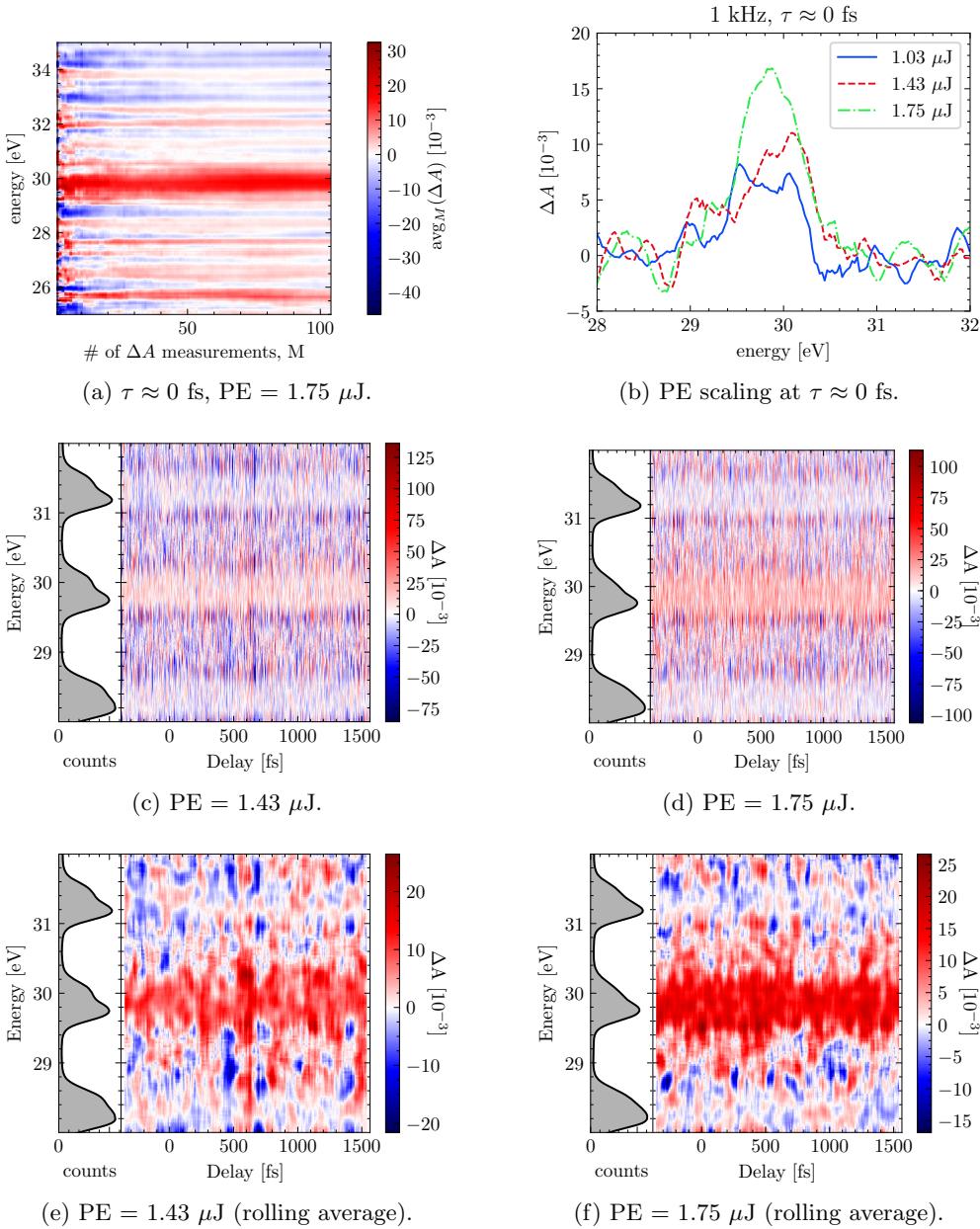


Figure 4.6: 1 kHz ATAS measurements in Ge using a  $\lambda = 1450$  nm excitation pulse. Fig. 4.6a: fixed-delay ATAS measurements with a pulse energy of 1.75  $\mu\text{J}$ . Fig. 4.6b: Pulse energy scaling at overlap of 1 kHz measurements. Figs. 4.6c to 4.6f: delay scans at 1 kHz. Figs. 4.6c and 4.6d: raw delay scan data. Figs. 4.6e and 4.6f: rolling average of the raw data with a 65 fs window (20 delay points). The left panel on each spectrogram shows the ground state spectrum  $S_{gs}(E)$ . See text for details.

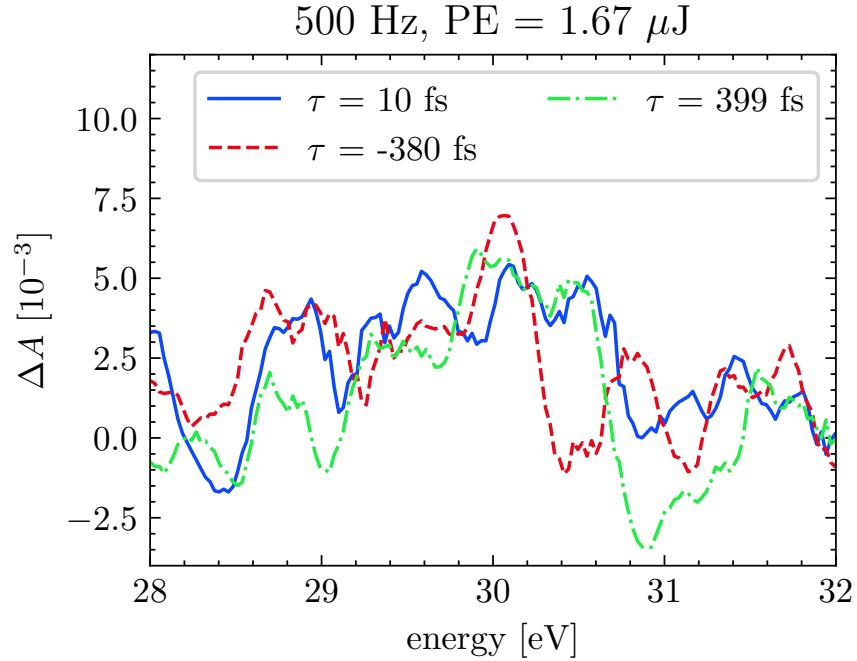


Figure 4.7: 500 Hz ATAS measurements in Ge using a  $\lambda = 1450$  nm,  $1.67 \mu\text{J}$  excitation pulse. Each delay curve is an average of 104 identical measurements. The sample shows no delay dependance within the uncertainty of the measurement.

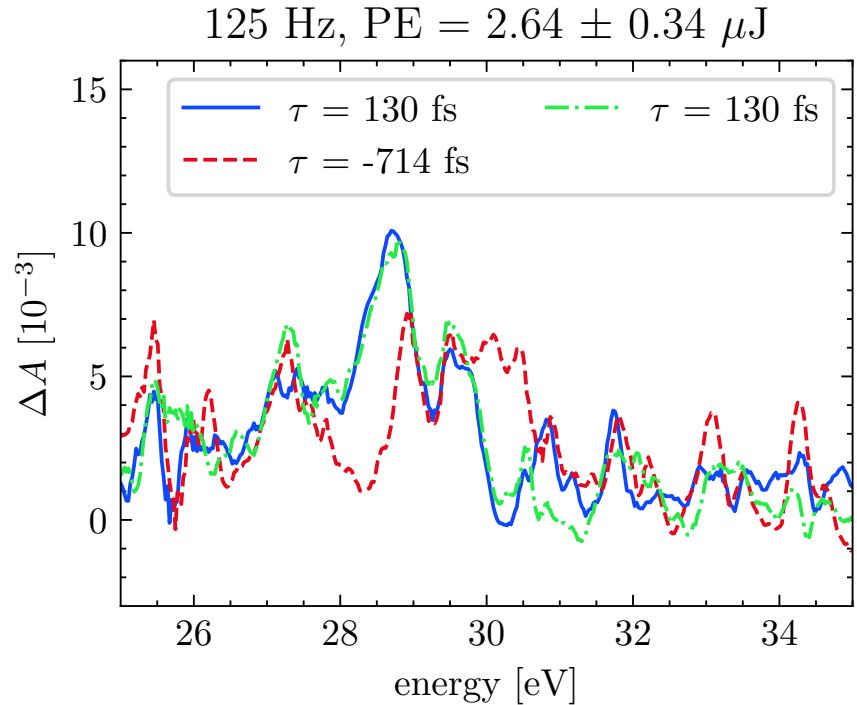


Figure 4.8: 125 Hz ATAS measurements in Ge using a  $\lambda = 1450$  nm,  $2.64 \mu\text{J}$  excitation pulse. Each lineout represents the average of 394 measurements. See text for details.

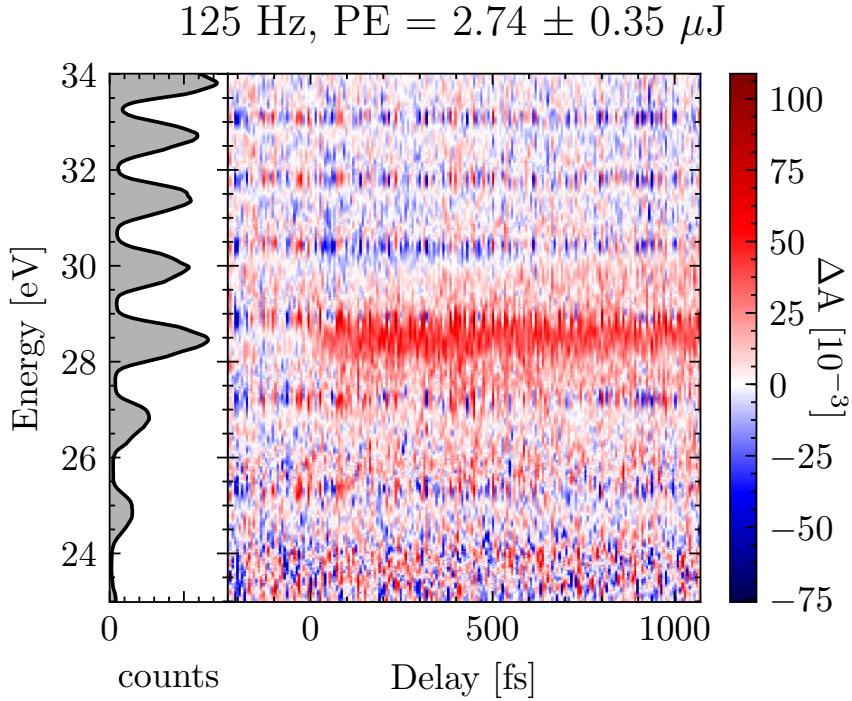


Figure 4.9: 125 Hz delay scan in Ge using a  $\lambda = 1450$  nm,  $2.74 \pm 0.35 \mu\text{J}$  excitation pulse. This is an average of 3 repeated measurements. See text for details.

### 4.3 Optimizing experimental ATAS parameters for Germanium thin films

#### 4.3.1 rep rate (avoiding ms-scale excitation)

We performed exploratory experiments at 1 kHz to determine the optimal excitation pulse energy. For this set of measurements, we generated harmonics in Argon using  $\lambda = 1450$  nm, a 200 nm Al filter and the 2C optics removed (see Fig. 2.1). Two delay points were recorded: with the XUV and IR pulses overlapped ( $\tau = 0$ ) and with the XUV pulse arriving about 300 fs before the IR ( $\tau = -\infty$ ). For each delay point, we used three pulse energies: 1.03, 1.43 and 1.75  $\mu\text{J}$ . To increase the signal-to-noise, we performed each experiment 104 times and averaged the datasets. The exposure time was 0.5 seconds, and the sample was rastered so that each measurement was recorded at a new position on the sample. The results are shown Figs. 4.5, 4.6a and 4.6b.

Figs. 4.5a to 4.5d and 4.6a show the average  $\Delta A$  as a function of the number of averaged measurements,  $M$ . Spectral features are apparent after averaging about 10 datasets together, but the fidelity of the signal does not appreciably improve after the first  $M = 50$  datasets. Figs. 4.5e and 4.5f show the average signal (lines) and the standard deviation

(shaded area), as calculated from the entire ensemble of measurements. The data is noisy, but we can see a spectral feature near 30 eV which scales with pulse energy (or perhaps average power). This behavior is evident in Fig. 4.6b. However, this feature is delay-independent:  $\Delta A$  has nearly the same value regardless of whether the XUV arrives before or after the IR pulse.

To confirm the delay-independence of this feature, we performed delay scans at two different pulse energies (1.43 and 1.75  $\mu\text{J}$ ). These measurements are shown in Fig. 4.6. Delay was controlled by inserting a fused silica wedge into the interferometer (W in Fig. 2.1). Each delay step corresponds to approximately 3.25 fs (25  $\mu\text{m}$  of wedge insertion).

The raw data is shown in Figs. 4.6c and 4.6d. As this experiment was only performed once ( $M = 1$ ), the contrast is poor and the 30 eV feature is barely visible. The spectral feature becomes more prominent after performing a rolling average over 20 delay points (65 fs), which is shown in Figs. 4.6e and 4.6f. These delay measurements confirm our suspicion that the absorption feature is delay-independent at 1 kHz.

One possible origin of a delay-independent signal can be a very long-lived excited state with lifetime  $1/\Gamma$ . Each laser shot initiates an assortment of electron and phonon dynamics, each with their own time scales. If any of these excited states have time scales that approach the inverse rep. rate of the laser ( $1/RR$ ), then the dynamics from the previous shot will still be evolving by time the next shot arrives. Since each exposure integrates over hundreds or thousands of laser shots, measurements at a nominal delay  $\tau$  will contain information from several delays  $\tau_i$ , each separated by the time between laser shots:  $\tau_i = \tau, \tau - \frac{1}{RR}, \tau - \frac{2}{RR}, \dots$ , with the amplitude of each contribution weighed by an exponential decay factor  $\exp(+\tau_i\Gamma)$ . If this is the case, then the magnitude of the delay-independent signal should decrease as time between laser shots is increased. As the time between laser shots is increased past the lifetime of the state, the excited state population from the previous laser shot will be small enough to observe the dynamics of the shorter-lived states. Experimentally, we can accomplish this by adjusting the rep. rate divider on the Spitfire amplifier (which reduces the rep. rate), and/or using an optical chopper.

The rep. rate was halved to 500 Hz using the Spitfire's rep. rate divider ( $m = 2$ ) and the exposure time was doubled to 1 second so that the number of laser shots per exposure was held constant. A series of fixed-delay measurements were recorded using a 1.67  $\mu\text{J}$  pulse energy, shown in Fig. 4.7. The spectral feature at 30 eV is still present, but it does not exhibit any delay dependence within the uncertainty of the measurement. It is notable that, for a similar pulse energy, the magnitude of the feature at 500 Hz is half that of the 1 kHz measurement. This is consistent with the hypothesis of a long-lived state contributing to the signal.

The rep. rate was lowered to 125 Hz using a combination of the rep. rate divider ( $m = 4$ ) and an optical chopper ( $T = 50\%$ ) placed after the TOPAS. The exposure time was

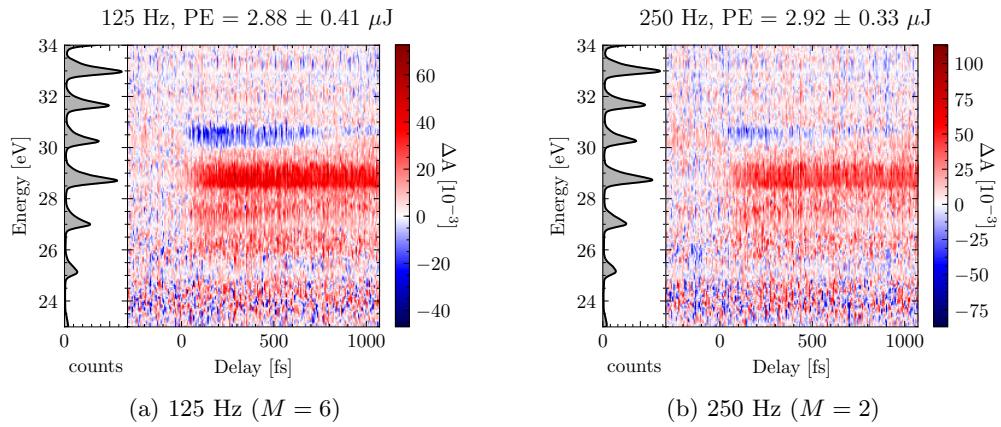


Figure 4.10: 125 vs. 250 Hz measurements at  $\lambda = 1430$  nm.

increased to 4 seconds to maintain sufficient counts on the detector. An ATAS spectrum was recorded about 130 fs after temporal overlap and several hundred fs before overlap using a  $2.64 \pm 0.34 \mu\text{J}$  excitation pulse ( $\lambda = 1450 \text{ nm}$ ), as shown in Fig. 4.8. A second measurement after overlap was recorded to ensure repeatability. At negative delays (red curve), we see the familiar 30 eV feature, albeit weaker at  $6 \times 10^{-3}$ . Near overlap, we see a new feature emerge at 28.7 eV, with a magnitude of  $\approx 10 \times 10^{-3}$ . This observation is consistent with long-lived excited state at 30 eV. A delay scan at 125 Hz, shown in Fig. 4.9, reveals that the 28.7 eV feature is indeed time-dependent with a ps-scale lifetime.

### 4.3.2 IR pulse energy

### 4.3.3 harmonic spectrum ( $\lambda$ , 2-color generation)

The fundamental wavelength was decreased by 20 nm to 1430 nm, where the absorption length in Ge is about 5% shorter. For a fixed pulse energy, this increases the excitation fraction and thus the strength of the  $\Delta A$  signal. Changing the wavelength also changes which initial and final states near the Fermi level are populated, according to the band structure calculations in Fig. 4.4. Using this shorter wavelength we can observe a more robust sample response, as shown in Fig. 4.10.

At 1430 nm, we can see a sample response from 25.7 to 31 eV. From 25.7 to 30 eV, there is a broad increase in absorption, with the largest increase occurring between 28.4 and 29.5 eV. A decrease in absorption occurs between 30 and 31 eV. These features are present in both the 125 and 250 Hz data. The 30 eV negative delay feature persists in the 250 Hz dataset, but at  $12 \times 10^{-3}$  it does not overwhelm the rest of the sample response. Further measurements were performed at either 125 Hz to suppress the static feature, or at 250 Hz to minimize data collection time.

# cartoon showing 2D image -> lineout -> deltaA calculations & normalization

Figure 4.11: this cartoon shows the data pipeline. it is an overview of all the processing steps i do on the data.

**4.3.4 optimized ATAS Ge experimental results**

**4.3.5 post-experiment analysis: verify we didn't permanently damage sample**

## **4.4 Data Analysis**

**4.4.1 description of data pipeline**

**going from 2D image to 1D spectra**

background subtraction, selecting a divergence window, normalization by exposure time & divergence window, integration over divergence window

# energy calibration: Ar fano features vs pixel

Figure 4.12: this figure shows the argon fano features vs pixel for the purpose of calibrating the spectrometer.

**energy calibration**

$A$ ,  $\Delta A$  calculation

4.4.2 systematic noise sources in our experiment

4.4.3 methods to numerically correct for harmonic noise and drift

4.4.4 frequency filtering to remove  $\omega$ ,  $2\omega$  oscillations

4.5 Physical Interpretation of spectra

4.5.1 decomposition of spectral response

4.5.2 description of observed dynamics

# Chapter 5

## CONCLUSIONS

say something here ...

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