

Airfoil Tools

Search 1638 airfoils

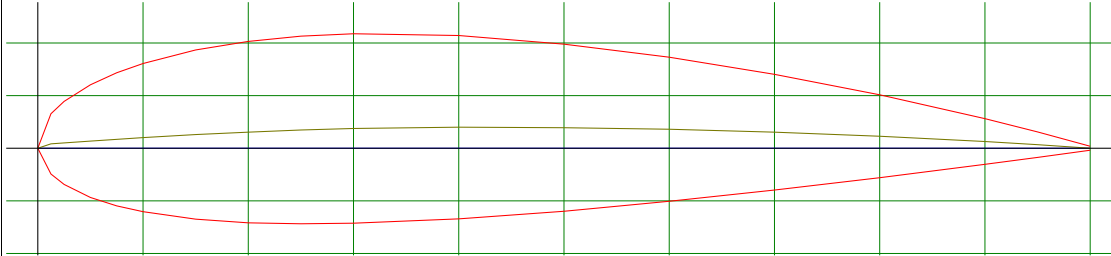


You have 0 airfoils loaded.

Your Reynold number range is 50,000 to 1,000,000. ([set](#))

NACA 2418 (naca2418-il)

NACA 2418 - NACA 2418 airfoil



Details

(naca2418-il) NACA 2418
NACA 2418 airfoil
Max thickness 18% at 30% chord.
Max camber 2% at 40% chord
Source [UIUC Airfoil Coordinates Database](#)
[Source dat file](#)
The dat file is in Selig format

Dat file

```
NACA 2418
1.0000 0.0019
0.9500 0.0155
0.9000 0.0281
0.8000 0.0508
0.7000 0.0702
0.6000 0.0865
0.5000 0.0989
0.4000 0.1000
```

Parser

No parser warnings

[Send to airfoil plotter](#)
[Add to comparison](#)
[Lednicer format dat file](#)
[Selig format dat file](#)

Similar airfoils

NACA 63	Preview	Details
NACA 64(3)-418	Preview	Details
EPPLER 550 AIRFOIL	Preview	Details
MH 102 16.99%	Preview	Details
EPPLER 1213 AIRFOIL	Preview	Details
EPPLER 549 AIRFOIL	Preview	Details
EPPLER 552 AIRFOIL	Preview	Details
NACA 64(3)-218	Preview	Details
EPPLER E1212 AIRFOIL	Preview	Details
CLARK YM-18 AIRFOIL	Preview	Details

Polars for NACA 2418 (naca2418-il)

Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source	
<input checked="" type="checkbox"/>	naca2418-il	50,000	9	8.6 at $\alpha=6^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	naca2418-il	50,000	5	27.3 at $\alpha=7.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	naca2418-il	100,000	9	42.9 at $\alpha=8.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	naca2418-il	100,000	5	44 at $\alpha=6.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	naca2418-il	200,000	9	60.8 at $\alpha=6.75^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	naca2418-il	200,000	5	58.6 at $\alpha=6^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	naca2418-il	500,000	9	83.4 at $\alpha=6.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	naca2418-il	500,000	5	75.3 at $\alpha=7^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	naca2418-il	1,000,000	9	97.1 at $\alpha=7^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	naca2418-il	1,000,000	5	87.5 at $\alpha=7.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details

[Reynolds number calculator](#)

Set Reynolds number and Ncrit range

Reynolds Number

Low

50,000

High

1,000,000

NCrit

7 9

