

Airfoil Tools

Search 1638 airfoils



Tweet

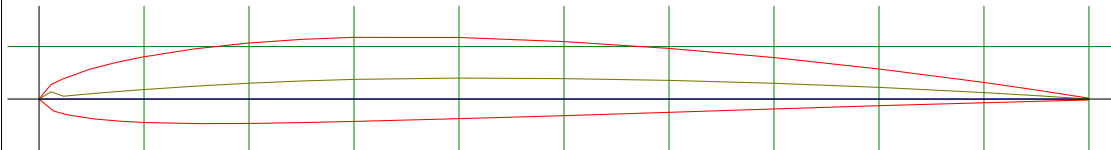
Like 1.3K

You have 0 airfoils loaded.

Your Reynold number range is 50,000 to 1,000,000. ([set](#)) Search

NACA 2408 (naca2408-il)

NACA 2408 - NACA 2408 airfoil



Details

(naca2408-il) NACA 2408
NACA 2408 airfoil
Max thickness 8% at 29.9% chord.
Max camber 2% at 40% chord
Source [UIUC Airfoil Coordinates Database](#)
[Source dat file](#)
The dat file is in Selig format

Dat file

```
NACA 2408
1.00000 0.00084
0.95033 0.00855
0.90054 0.01575
0.80078 0.02858
0.70081 0.03942
0.60068 0.04820
0.50039 0.05473
0.40000 0.05800
```

Parser

No parser warnings

[Send to airfoil plotter](#)
[Add to comparison](#)
[Lednicer format dat file](#)
[Selig format dat file](#)

Similar airfoils

GOE 622 AIRFOIL	Preview	Details
AG36	Preview	Details
GOE 795 AIRFOIL	Preview	Details
GOE 795 smoothed	Preview	Details
RG 14 AIRFOIL	Preview	Details
S6062 8%	Preview	Details
MH 43 8.5%	Preview	Details
GOE 564 AIRFOIL	Preview	Details
AG37	Preview	Details
MH 30 7.84%	Preview	Details

Polars for NACA 2408 (naca2408-il)

Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source
<input checked="" type="checkbox"/>	naca2408-il	50,000	9	37.4 at $\alpha=5.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2408-il	50,000	5	36.7 at $\alpha=5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2408-il	100,000	9	52.6 at $\alpha=5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2408-il	100,000	5	48.9 at $\alpha=4.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2408-il	200,000	9	66.6 at $\alpha=4^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2408-il	200,000	5	58.9 at $\alpha=3.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2408-il	500,000	9	80.6 at $\alpha=3.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2408-il	500,000	5	67.5 at $\alpha=2.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca2408-il	1,000,000	9	86.8 at $\alpha=2.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca2408-il	1,000,000	5	82.1 at $\alpha=7.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details

Update plots

[Reynolds number calculator](#)

Set Reynolds number and Ncrit range

Update Range

Reynolds Number

Ncrit

Low

50,000

7

High

1,000,000

9

