SLB-GOLDS-DOC-v0.1

GOLDS-UFSC Documentation

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Nomenclature

ACS	*Attitude Control System.*
AEB	*Agência Espacial Brasileira.*
AIT	*Assembly, Integration and Test.*
CG	*Center of Gravity.*
EDC	*Environmental Data Collection.*
EPS	*Electrical Power System.*
FPGA	*Field-Programmable Gate Array.*
GOLDS	*Global Open Collecting Data System*
INPE	*Instituto Nacional de Pesquisas Espaciais.*
LIT	*Laboratório de Integração e Testes.*
LNA	*Low Noise Amplifier.*
MPPT	*Maximum Power Point Tracking.*
OBDH	*On-Board Data Handling.*
PA	*Power Amplifier*
PCB	*Printed Circuit Board.*
PCD	*"Plataforma de Coleta de Dados", or Data Collection Platform*
SBCD	*Sistema Brasileiro de Coleta de Dados.*
SDRAM	*Synchronous Dynamic Random-Access Memory.*
SDR	*Software Defined Radio.*
SNR	*Signal To Noise Ratio*
SoC	*System-On-a-Chip.*
TTC	*Telemetry, Tracking and Command Module.*

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CHAPTER 1

Introduction

The GOLDS-UFSC is a satellite project of a 2U CubeSat (10×10×22,70 cm). This nanosatellite is the sequence project of the FloripaSat-I CubeSat [1], both developed by SpaceLab [2]. This second project is being developed in partneship with INPE (*Instituto Nacional de Pesquisas Espaciais*), who is supplying the main payload of the mission: The EDC board (*Environmental Data Collection*) [3]. The name "GOLDS" stands for "Global Open Collecting Data System", and it is the nomenclature of a collaborative CubeSat constellation for environmental data collection planned as part of the Brazilian space program [4].

This project started just after the launch of FloripaSat-I (first half of 2020) and is planned to be launched in 2022. Most of the embedded electronics is partially or totally based on the FloripaSat-I satellite, with the same and/or improved versions of the modules. In other words, this project has at some level a flight heritage.

1.1 Mission Description

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1.2 Mission Objectives

The main objectives of this mission are enumerated below:

1. To serve as a host platform for the EDC payload.
2. Validate the EDC payload in orbit.
3. Validate EDC functionality in orbit.
4. Validate core-satellite functions in orbit.
5. Evaluate the behavior of the core modules in a 2U mission.
6. Perform experiments on radiation effects in electronic components in orbit.
7. Serve as relay for amateur radio communications, as a contribution to the amateur radio community.

1.3 Project Members

All people involved in the project are students, professors and researchers from Federal University of Santa Catarina (UFSC), the National Institute for Space Research (INPE) and the Brazilian Space Agency (AEB).

A list with the current members directly related to the project (2021/02/08) can be seen in Table 1.1.

Name	**Title**	**Position**	**Institution**
Anderson Wedderhoff Spengler	Ph.D.	Professor	UFSC
Eduardo Augusto Bezerra	Ph.D.	Professor	UFSC
Richard Demo Souza	Ph.D.	Professor	UFSC
Laio Oriel Seman	Ph.D	Researcher	UFSC
Manoel Jozeane Mafra de Carvalho	Ph.D.	Researcher	INPE
José Marcelo Duarte	Ph.D.	Researcher	INPE
Rodrigo Leonardi	Ph.D.	Researcher	AEB
Cezar Antônio Rigo	M.Sc.	Ph.D. Student	UFSC
Edemar Morsch Filho	M.Sc.	Ph.D. Student	UFSC
Gabriel Mariano Marcelino	M.Sc.	Ph.D. Student	UFSC
Thiago Martins	M.Sc.	Ph.D. Student	UFSC
Vinicius Pimenta Bernardo	B.Eng.	Master's Student	UFSC
Amanda Medeiros	-	Undergraduate Student	UFSC
André Martins Pio de Mattos	-	Undergraduate Student	UFSC
Augusto Cezar Boldori Vassoler	-	Undergraduate Student	UFSC
Daniel Baron	-	Undergraduate Student	UFSC
João Cláudio Elsen Barcellos	-	Undergraduate Student	UFSC
Lorenzo Maturano	-	Undergraduate Student	UFSC
Matheus Wagner	-	Undergraduate Student	UFSC
Maurício Sinigaglia	-	Undergraduate Student	UFSC
Tatiane dal Ross	-	Undergraduate Student	UFSC
Victor Noster	-	Undergratuate Student	UFSC
Yan Castro de Azeredo	-	Undergraduate Student	UFSC

Table 1.1: Project members (2021/02/08).

All the used modules and methods used in this project are based in a lot of past works, most of it being the FloripaSat-I and the EDC projects. The list with the indirectly involved people are much bigger.

1.4 Mission Patch

The mission patch of the GOLDS-UFSC can be seen in Figure 1.1, it is inspired by the FloripaSat-I patch [1].

Figure 1.1: GOLDS-UFSC mission patch.

CHAPTER 2

Mission Requirements

1. The power system shall be able to harvest solar energy.
2. The power system shall be able to store energy for use when GOLDS-UFSC is eclipsed.
3. The power system shall supply energy to all other modules.
4. The data handling system shall communicate with the other modules and store their data.
5. The communications system shall send a beacon signal periodically using VHF radio.
6. The communications system shall send the CubeSat telemetry using UHF radio.
7. The communications system shall be able to receive telecommands and respond to them accordingly.
8. The attitude system shall be able to perform a 1-axis stabilization of the CubeSat.
9. GOLDS-UFSC shall have the capability to receive and execute a shutdown telecommand, therefore ceasing all transmissions.
10. The downlink transmissions shall be done once at a time, either telemetry or beacon.
11. The ground station shall operate under the proper radio frequency communication licenses.
12. GOLDS-UFSC shall comply with international and Brazilian radio license agreements and restrictions.
13. The team shall build and operate a ground station for full communication with GOLDS-UFSC.

CHAPTER 3

Mission Schedule

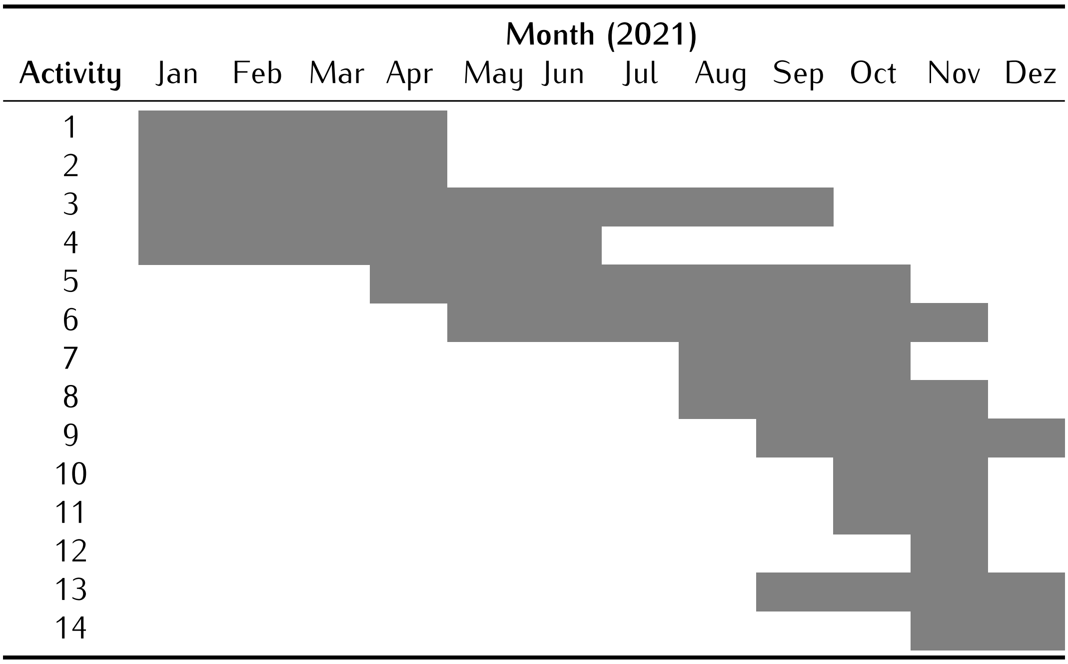

Table 3.1: Mission schedule.

Each activity of Table 3.1 is decribed below:

1. Acquisition and manufacturing of critical elements and components for the solo platform.
2. Acquisition and manufacture of elements and components critical to the payload.
3. Acquisition and manufacturing of critical elements and components for the solo segment.
4. Compatibility tests between platform and payload in SpaceLab UFSC.
5. Integration of the engineering model in SpaceLab UFSC.
6. Preparation and suitability of the ground segment.
7. Verification and validation of the engineering model at SpaceLab UFSC.
8. Verification and validation of the flight model at SpaceLab UFSC.

9. Data collection platforms installation.
10. Verification and validation tests of Engineering Model compatibility with EMMN in the INPE / CRN in Natal.
11. Environmental tests at the Integration and Testing Laboratory (LIT/INPE).
12. Flight model acceptance and ground segment review.
13. Ground segment delivery.
14. Flight model delivery.

CHAPTER 4

Overall Description

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4.1 General Diagrams

The CubeSat's subsystems are positioned in the 2U physical structure as exemplified in Figure 4.1. An exploded 3D view of the satellite is showed in Figure 5.1.

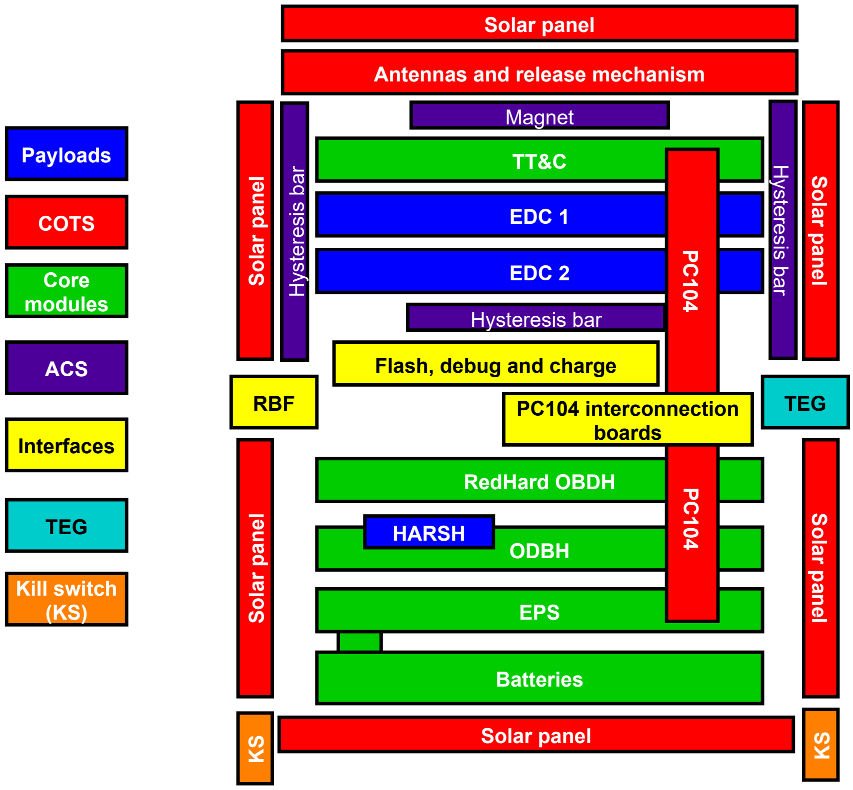

Figure 4.1: Subsystems positioning.

4.2 General Behaviour

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4.3 Orbit Parameters

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4.4 Power Budget

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4.5 Link Budget

The link budget of all radio links of the satellite is available in Table 4.1.

Variable	Beacon	Downlink	Uplink	Uplink (Payload)	Unit
Frequency	145,97	436,9	436,9	401,635	MHz
Modulation	MSK	MSK	MSK	BPSK	-
Protocol	NGHam	NGHam	NGHam	SBCD	-
Transmit power	30	30	47	??	dBm
FSPL	144,8	154,3	154,3	??	dB
Other losses	5	5	7	5	dB
Receive antenna gain	12	15.5	0	0	dBi
Receiver noise temp.					K
Antenna noise temp.					K
System noise temp.					K
Data rate	1200	4800	4800	400	bps
Received SNR	30,87	17,35	31,60	??	dB
SNR required for 10^{-5} BER*	9,6	9,6	9,6	9,6	dB
Link margin	$\leq$ 21,27	$\leq$ 7,75	$\leq$ 22	$\leq$??	dB

Table 4.1: Link budget results.

All equations and steps used to obtain the results of Table 4.1 are available in Appendix A.

4.6 PC-104 Bus

4.7 Telecommunication

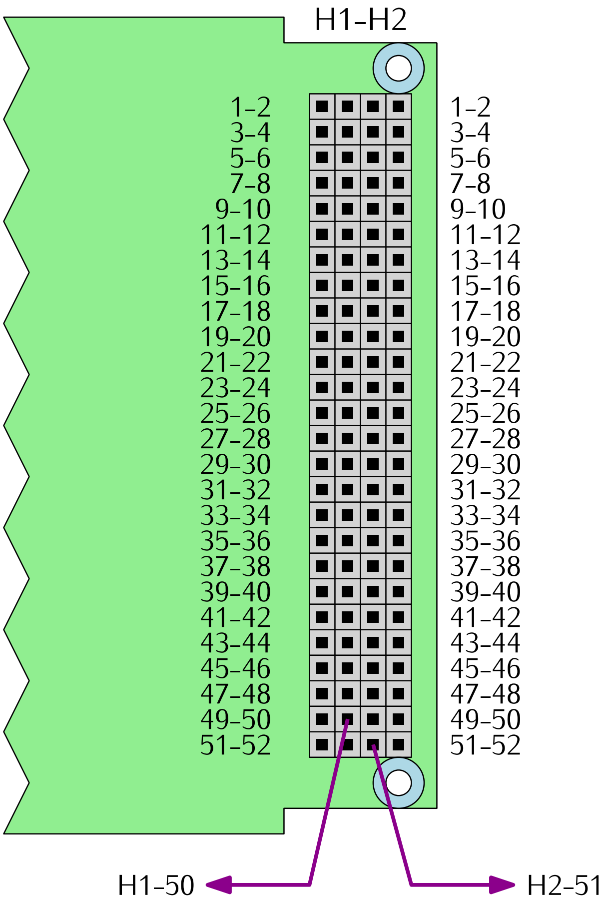

Figure 4.2: Reference diagram of the PC-104 bus.

Pin Row	H1 Odd	H1 Even	H2 Odd	H2 Even
1-2	-	-	-	-
3-4	-	-	EDC_1_EN	EDC_2_EN
5-6	-	-	BE_UART_RX	-
7-8	RA_GPIO_0	RA_GPIO_1	BE_UART_TX	GPIO_0
9-10	RA_GPIO_2	BE_EN	-	-
11-12	RA_RESET	RA_EN	BE_SPI_MOSI	BE_SPI_CLK
13-14	-	-	BE_SPI_CS	BE_SPI_MISO
15-16	-	-	-	-
17-18	EDC_UART_RX/TX	PLX_EN	-	GPIO_1
19-20	EDC_UART_TX/RX	GPIO_2	-	GPIO_3
21-22	-	-	-	GPIO_4
23-24	-	-	-	-
25-26	-	-	PL_VCC	PL_VCC
27-28	-	-	TTC_VCC	TTC_VCC
29-30	GND	GND	GND	GND
31-32	GND	GND	GND	GND
33-34	-	-	-	-
35-36	RA_SPI_CLK	-	ANT_VCC	ANT_VCC
37-38	RA_SPI_MISO	-	-	-
39-40	RA_SPI_MOSI	RA_SPI_CS	-	-
41-42	PL_I2C_SDA	-	-	GPIO_5
43-44	PL_I2C_SCL	-	-	-
45-46	OBDH_VCC	OBDH_VCC	BAT_VCC	BAT_VCC
47-48	PL_VCC	PL_VCC	-	-
49-50	RA_VCC	RA_VCC	EPS_I2C_SDA	-
51-52	BE_VCC	BE_VCC	EPS_I2C_SCL	-

Table 4.2: PC-104 bus pinout.

Signal	Pin(s)	Used By	Description
GND	H1-29/30/31/32, H2-29/30/31/32	All	Ground reference
BAT_VCC	H2-45, H2-46	EPS	Battery terminals (+)
ANT_VCC	H2-35, H2-36	EPS, ANT	Antenna power supply (3.3 V)
OBDH_VCC	H1-45, H1-46	EPS, OBDH	OBDH power supply (3.3 V)
TTC_VCC	H2-27, H2-28	EPS, TTC	TTC power supply (3.3 V)
PL_VCC	H1-47/48, H2-25/26	EPS, EDC 1/2, Payload X	Payloads power supply (5 V)
RA_VCC	H1-49, H1-50	EPS, TTC	Main radio power supply (5 V)
BE_VCC	H1-51, H1-52	EPS, TTC	Beacon power supply (6 V)
RA_SPI_CLK	H1-35	OBDH, TTC	CLK signal of the main radio SPI bus
RA_SPI_MISO	H1-37	OBDH, TTC	MISO signal of the main radio SPI bus
RA_SPI_MOSI	H1-39	OBDH, TTC	MOS signal of the main radio SPI bus
RA_SPI_CS	H1-40	OBDH, TTC	CS signal of the main radio SPI bus
EPS_I2C_SDA	H2-49	OBDH, EPS	SDA signal of the EPS I2C bus
EPS_I2C_SCL	H2-51	OBDH, EPS	SCL signal of the EPS I2C bus
BE_UART_RX	H2-5	EPS, TTC	EPS TX, Beacon RX (UART bus)
BE_UART_TX	H2-7	EPS, TTC	EPS RX, Beacon TX (UART bus)
EDC_UART_TX/RX	H1-25	OBDH, EDC 1/2	OBDH TX, EDCs RX (UART bus)
EDC_UART_RX/TX	H1-27	OBDH, EDC 1/2	OBDH RX, EDCs TX (UART bus)
BE_EN	H1-10	EPS, TTC	Beacon radio power enable
RA_EN	H1-12	EPS, OBDH	Main radio power enable
EDC_1_EN	H2-3	OBDH, EDC 1	EDC 1 enable signal
EDC_2_EN	H2-4	OBDH, EDC 2	EDC 2 enable signal
PLX_EN	H1-18	OBDH, Payload X	Payload X enable (GPIO)
PL_I2C_SDA	H1-41	OBDH, Payload X	SDA signal of the payload I2C bus
PL_I2C_SCL	H1-43	OBDH, Payload X	SCL signal of the payload I2C bus
GPIO_N	H1-20, H2-8/18/20/22/42	OBDH	GPIO pin (not used)

Table 4.3: PC-104 bus signal description.

Link	*Packet Name*	*Payload*				*Access*
		ID	*Source Callsign*	*Data (up to 220 bytes)*	*Size (bytes)*	
Beacon	EPS data	00h	"0" + "PY0EGU"	EPS data	58	Public
	TTC Data	01h		TTC data	18	Public
Downlink	Telemetry	20h	"0" + "PY0EGU"	Flags + OBDH/EPS data	220	Public
	Ping answer	21h		Requester callsign	15	Public
	Data request answer	22h		Req. callsign + data	15 to 155	Public
	Message broadcast	23h		Req. + dst. callsign + message	22 to 60	Public
	Hibernation feedback	24h		Req. callsign + hibernation in hours	17	Public
	EDC info	25h		PTT decoder + HK info + system state	79	Public
	EDC samples	26h		Timestamp + pkt. counter + samples	219	Public
	TC feedback	27h		Req. callsign + TC packet ID + timestamp	13	Public
Uplink	Ping Request	40h	Any Callsign	None	8	Public
	Data Request	41h		Data flags + count + origin + offset	16	Public
	Broadcast Message	42h		Dst. callsign + message	15 to 46	Public
	Enter hibernation	43h		Req. callsign + hibernation in hours + key	29	Private
	Leave hibernation	44h		Command key	16	Private
	Activate beacon	45h		Command key	16	Private
	Deactivate beacon	46h		Command key	16	Private
	Activate downlink	47h		Command key	16	Private
	Deactivate downlink	48h		Command key	16	Private
	Activate EDC	49h		Command key	16	Private
	Deactivate EDC	4Ah		Command key	16	Private
	Activate Payload X	4Bh		Command key	16	Private
	Deactivate Payload X	4Ch		Command key	16	Private
	Get EDC info	4Dh		Command key	16	Private

Table 4.4: Telecommunication packets and their content.

CHAPTER 5

Subsystems

This chapter presents a description of all subsystems of the space segment of the mission, which can be seen in the exploded view of the satellite, available in Figure 5.1.

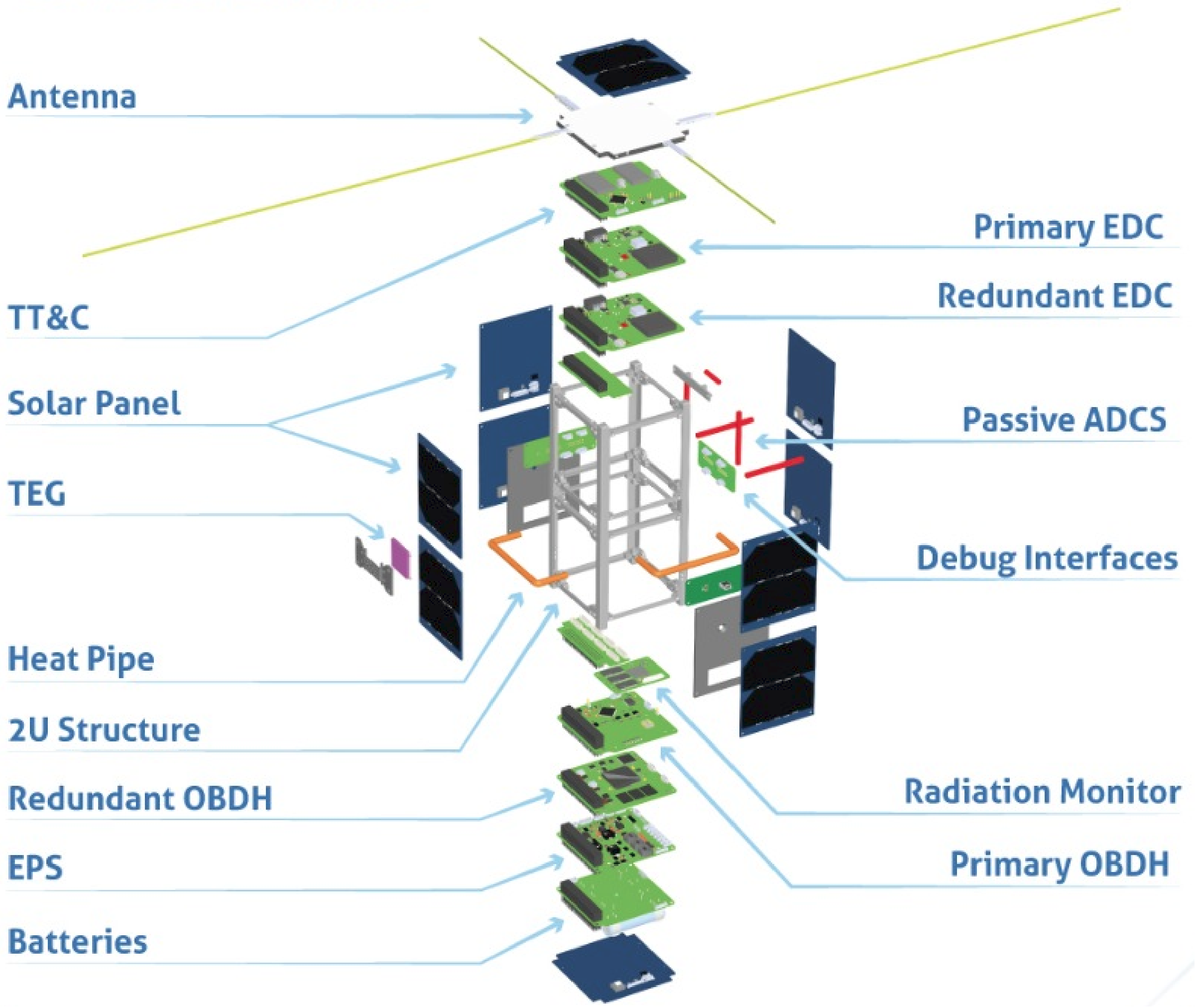

Figure 5.1: Exploded view of the GOLDS-UFSC satellite.

Most of the subsystems presented here have their own documentation, with a deeper technical description of it. When available, there is a reference to the respective document. This chapter is intended to show an overview of each subsystem, in a macro context of the mission.

5.1 On-Board Data Handling

The OBDH 2.0 is an On-Board Computer (OBC) module designed for nanosatellites. The module is responsible for synchronizing actions and the data flow between other modules (i.e., power module, communication module, payloads) and the Earth segment. It packs the generated data into data frames and transmit back to Earth through a communication module, or stores it on a non-volatile memory for later retrieval. Commands sent from Earth segment to the CubeSat are received by radio transceivers located in the communication module and redirected to the OBDH, which takes the appropriate action or forward the commands to the target module.

The module is a direct upgrade from the OBDH of FloripaSat-1 [1], which grants a flight heritage rating. The improvements focus on providing a cleaner and more generic implementation in comparison with the previous version, more reliability in software and hardware implementations, and adaptations for the new mission requirements. The board of the module can be seen in Figure 5.2.

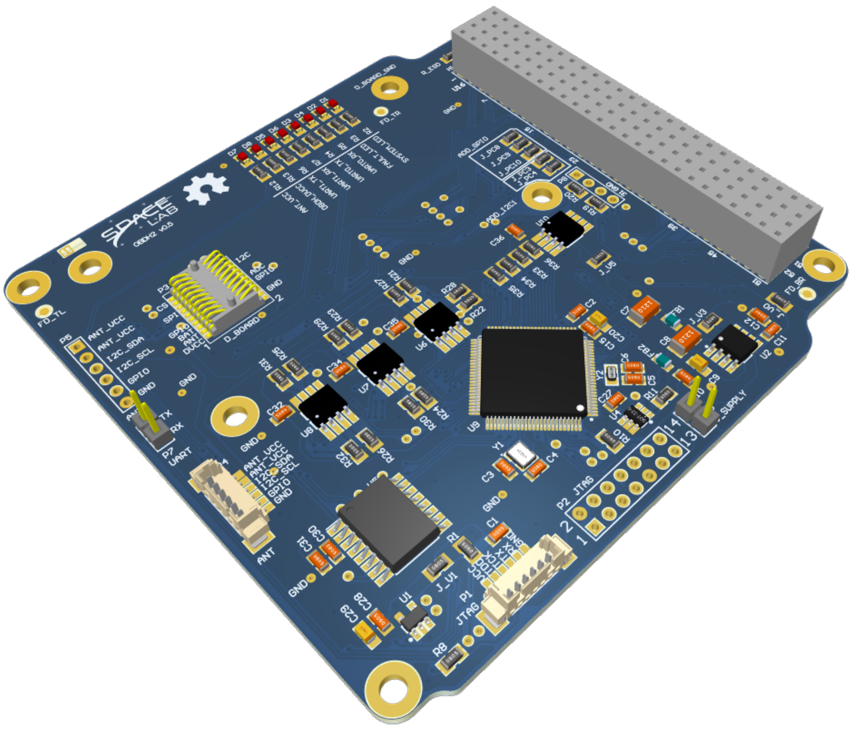

Figure 5.2: OBDH module.

More information about this module can be found in [5].

5.2 Telemetry, Tracking and Command Module

The TTC (or TT&C) is responsible to make the communication between the earth (a ground station) and the satellite, and is divided in two sub-modules: Beacon and downlink/uplink. The beacon is a independent sub-module who transmits a periodic signal containing an identification data (ID) of the satellite and some basic telemetry data. The downlink/uplink sub-module is the main communication device. It has a bidirectional data link to receive telecommands from the earth and transmit all available data back to Earth. The board of the module can be seen in Figure 5.3.

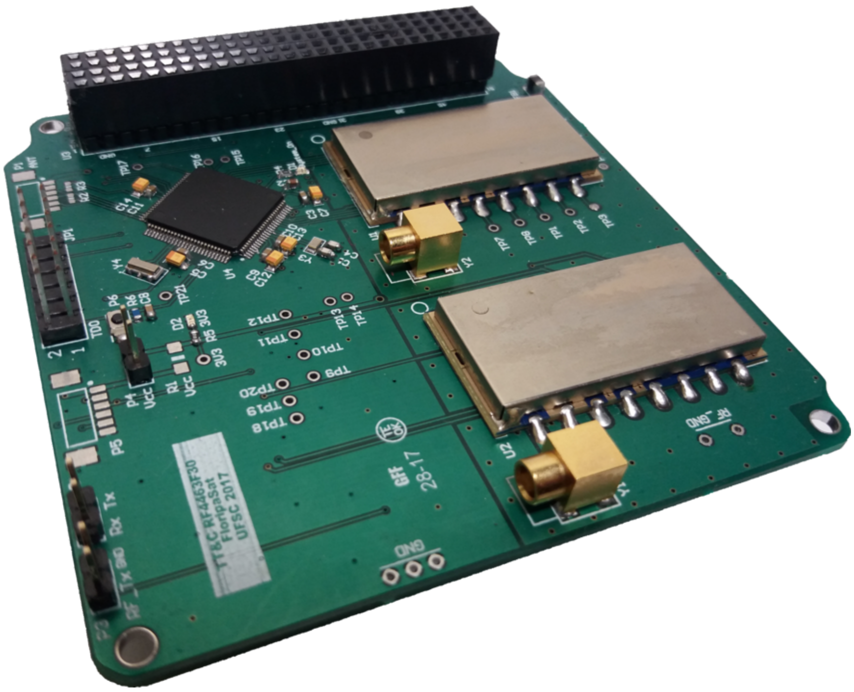

Figure 5.3: TTC module.

More information about this module can be found in [6].

5.2.1 Antenna Module

The used antenna module is the CubeSat deployable VHF and UHF antenna from ISISpace [7]. It is a four monopole antenna built with tape strings (up to 55 cm) and compliant with the CubeSat standard (dipole or turnstile options are also available). The deployment method is the burning wire and it can be controlled digitally through a I^2C interface. To allow redundancy, there are two independent deployment controllers that can be activated separately. Also, the construction of this module allows the installation of a solar panel at the top side. The RF gain is about 0 dBi.

A picture of the antenna module (with all antennas released) can be seen in Figure 5.4.

The chosen configuration for this mission can be seen below (using Figure 5.5 as reference):

- Configuration: 4 monopoles (1x VHF + 3x UHF)
  - Antenna 1: VHF - 145,97 MHz (beacon)
  - Antenna 2: UHF - 401,635 MHz (EDC)
  - Antenna 3: UHF - 436,9 MHz (downlink/uplink)
  - Antenna 4: UHF - 401,635 MHz (redundant EDC)
- Tuning structure size: 2U
- Mounting position: Top

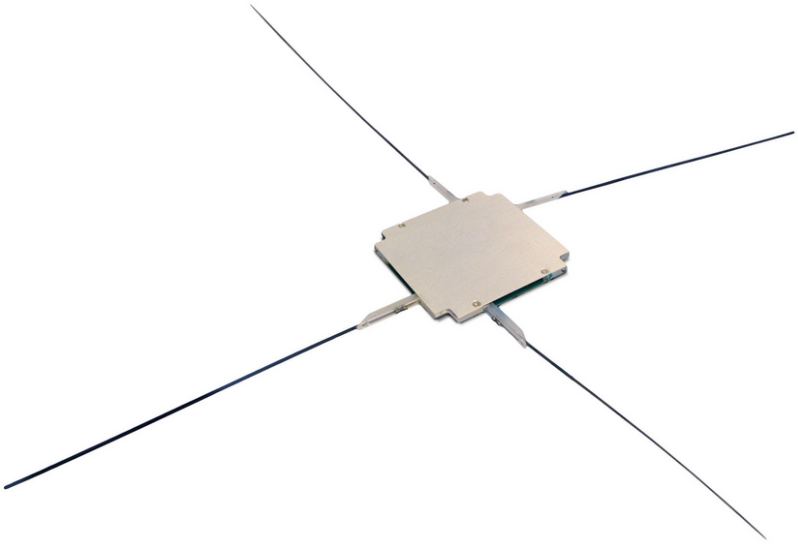

Figure 5.4: Antenna module from ISISpace.

- Supply voltage: 3,3 V
- I^2C control type: Dual bus
  - Primary I^2C address: 31h (7-bit address)
  - Redundant I^2C address: 32h (7-bit address)
- I^2C watchdog: Enabled with a time out of 60 seconds.

In the digital interface, a temperature sensor and the state of four deployment switches (1 per monopole) are also available. These switches indicate if a monopole is released or not, and can be used as feedback of the deployment process.

5.3 Electrical Power System

The EPS is the module designed to harvest, store and distribute energy for the satellite. The energy harvesting system is based on solar energy conversion through te solar panels attached to the CubeSat structure. The EPS is designed to operate the solar panels at their maximum power point (MPPT). The board also measures the solar panels current, voltage and the temperature of the batteries. The harvested solar energy is stored in a battery module connected to the EPS. The energy distribution is done by several integrated buck DC-DC converters. The full EPS system is composed of the solar panels, the EPS PCB and the battery module. A general view of the EPS board can be seen in Figure 5.6.

The module is a direct upgrade from the EPS of FloripaSat-1 [1], which grants a flight heritage rating. The improvements focus on providing a cleaner and more generic implementation in comparison with the previous version, more reliability in software, and adaptations for the new mission requirements.

More information about this module can be found in [8].

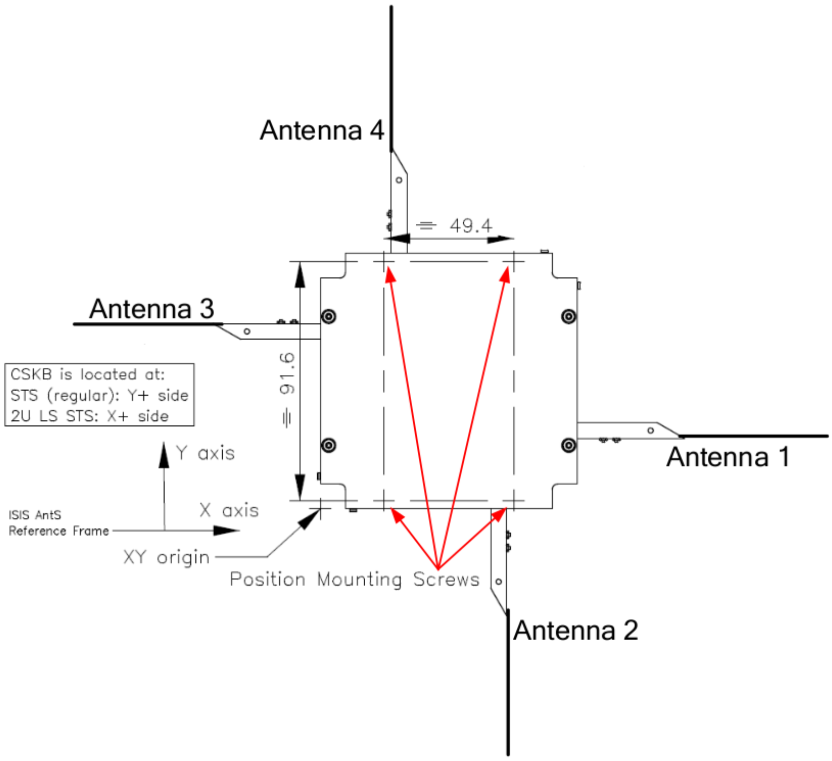

Figure 5.5: Configuration reference of the antenna module.

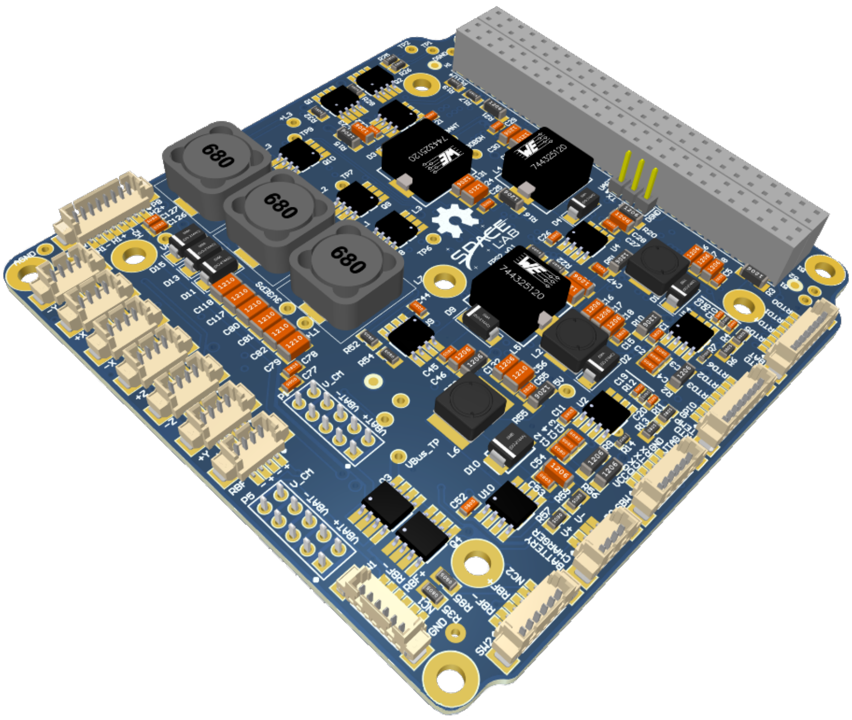

Figure 5.6: EPS module.

5.3.1 Battery Module

The used battery module is the "*Battery Module 4C*", that is a separate battery module from the EPS board and composed by four lithium-ion 18650 cells. Besides the cells, the board has connectors for interfacing signals and power lines with the EPS module, 2 power resistors to operate as heaters to maintain the cell's temperature during eclipse periods, and 4 temperature sensors. The batteries used are the ICR18650-30B lithium-ion cells from Samsung [9], which are connected in series and parallel (two sets of two parallel cells in series) to supply the required voltage and current. Each cell is fixed with 18650 metal holders and between the pairs there is the power resistor attached with a thermal element in the middle. A mechanical mount is placed over the batteries and screwed to the board, providing better stress resistance. Also, there are PC-104 through hole pads present on the board for a connector that could be used for making mechanical integration with the EPS, or with future improvements a interface for power, data or control signals. The board is a direct improvement from the first battery board used in the FloripaSat-1 mission [1].

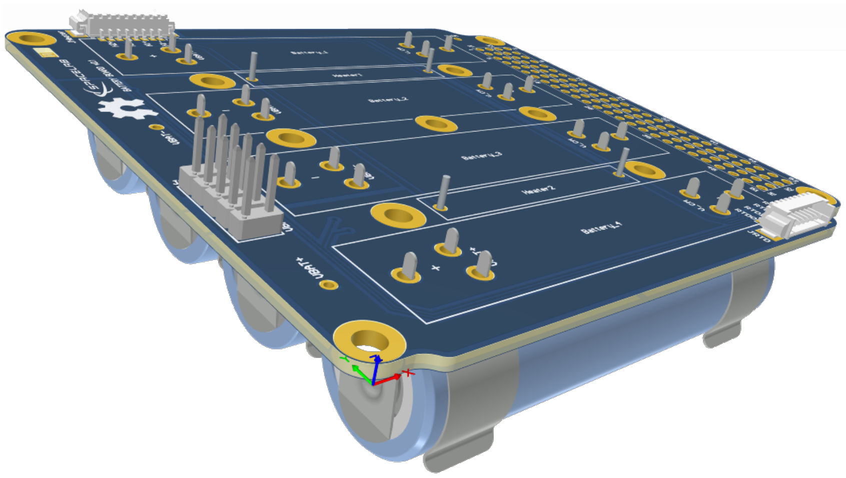

Figure 5.7: Battery module board.

More information about the battery module can be found in [10].

5.3.2 Solar Panels

The solar panels are a set of 5 custom made panels manufactured by ORBITAL, a Brazilian company, and a single panel from ISISpace. The panels features protection diodes and high-efficiency solar cells, which are the CESI's CTJ-30 [11] with dimensions 6,9 $\times$ 3,9 cm (area 26,5 cm^2). This cell is qualified for space use by ESA with an efficiency of 29,5 % (AM0, BOL). The panels do not include magnetorquers, sensors and others devices. The top solar panel is a model from ISISpace to ensure mechanical compatibility with the antenna module (also from ISISpace). These two types of solar panels can be seen in Figures 5.8 and 5.9.

5.3.3 Kill-Switches and RBF

Two electronic switches have been implemented into the design as to allow for the (redundant) deployment detection of the CubeSat when it is deployed from the POD. This electronic microswitch can be used to prevent the satellite from starting up during launch as is required for all CubeSat launches and hence acts as a Kill-Switch. The Kill-Switch is the Panasonic AV4 microswitch (AV402461), as can be seen in Figure 5.10.

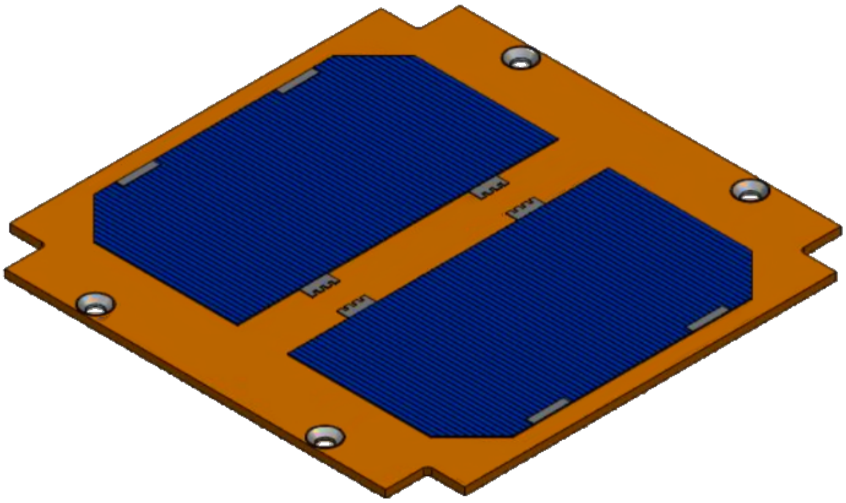

Figure 5.8: Conceptual solar panel from ORBITAL.

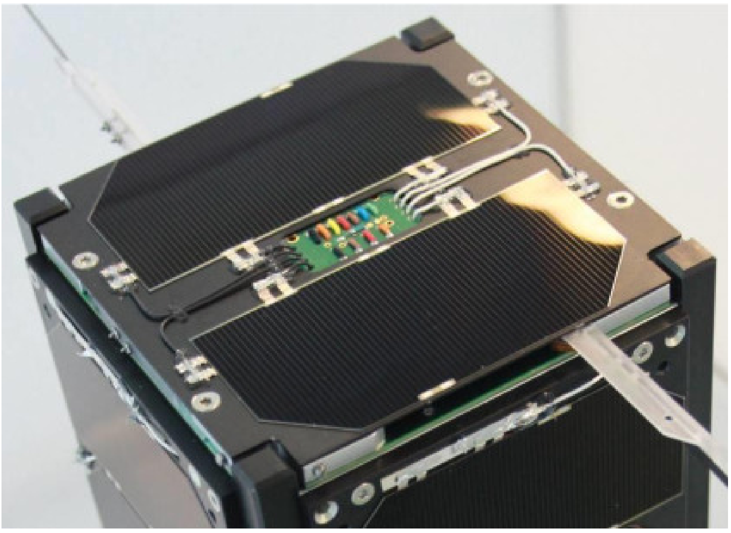

Figure 5.9: Top solar panel from ISISpace.

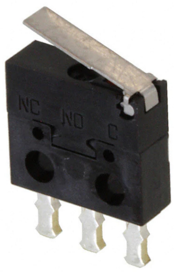

Figure 5.10: Panasonic AV402461 Microswitch.

The Kill-Switch mechanism in the mechanical structure has combined the function of providing deployment and detection (Figure 5.11). The travel of the actual switch of the Kill-Switch itself is so short that the Kill-Switch could “detect deployment” of the

CubeSat from the launch adapter simply due to launch vibrations. To overcome this issue the Kill-Switch has been rotated so that there is a positive obstruction in front of the switch which needs 8 mm of deployment before deployment can be detected with the Kill-Switch. In Figure 5.11 the Kill-Switch parts are highlighted and the stowed and deployed configuration is shown.

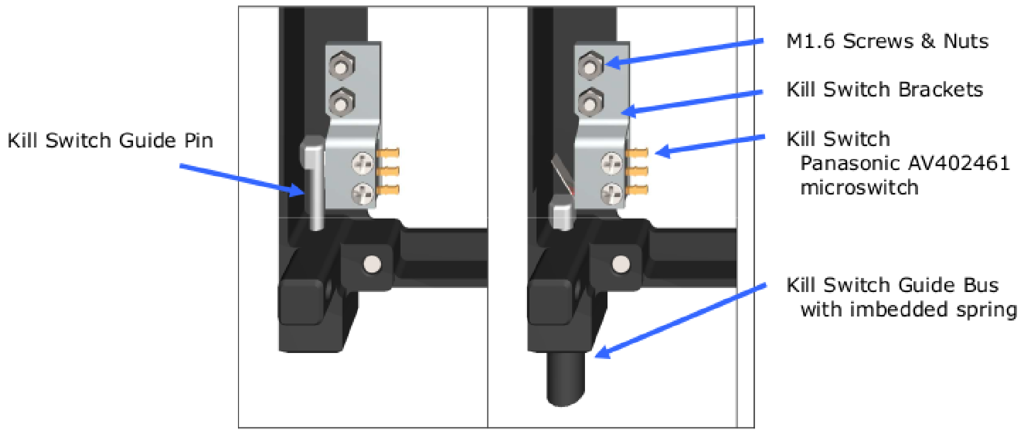

Figure 5.11: Kill-Switches installed in the mechanical structure.

The contact arrangement of the microswitch and the current rating are detailed in Figure 5.12 and Table 5.1.

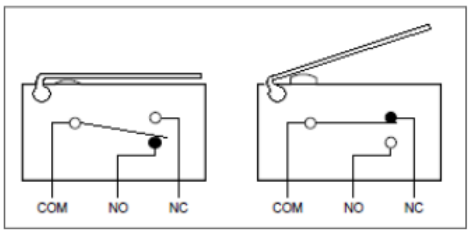

Figure 5.12: The contact arrangement of the microswitch.

Characteristic	Minimum	Typical	Maximum	Unit
Switch Current	2	50	100	mA
DC Voltage across switch contacts	n/a	n/a	30	V
Contact resistance microswitch	n/a	n/a	200	mΩ

Table 5.1: Kill-Switch current rating and voltage range.

5.4 Attitude Control System

The Attitude Control System (ACS) is a passive attitude control system, which depends on the Earth's magnetic field to rotate and stabilize the satellite [12, 13]. The system is composed of one permanent magnet to create a force to align the magnet with the Earth's magnetic field and four hysteresis bars to damp the cube oscillations and achieve stabilization.

When equilibrium is achieved, the permanent magnet aligns itself to the Earth's field lines. The hysteresis bars convert oscillation and rotation energy into heat, maintaining the alignment through magnetic moment. The components are placed in positions as to minimize the magnet's interaction with the hysteresis bars, which limits the magnetic moment of the magnet [14]. Figure 5.13 shows the mounting of the hysteresis bars (green) and the permanent magnet (red) on the mechanical structure. The whole passive ACS was implemented according to [14].

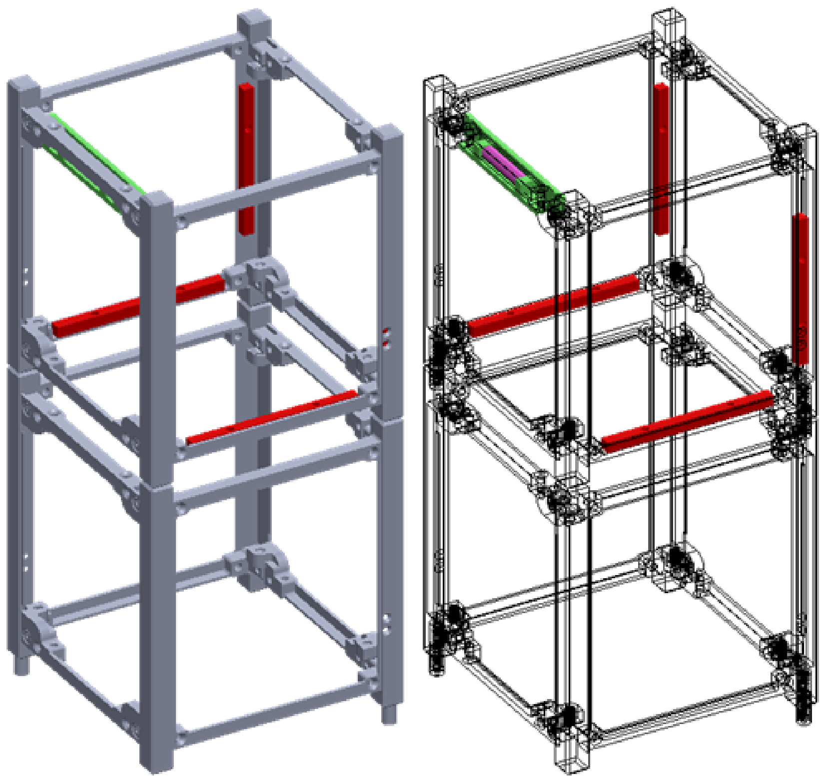

Figure 5.13: ACS subsytem. Rare earth magnet (pink) and hysteresis bars (red) installed in the structure.

As a passive magnetic attitude control system is used, it is possible to stabilize only one axis, and so, the CubeSat will still slowly (due to hysteresis bars) rotate around this axis, even after stabilized. A N45 neodymium magnet and 4 hysteresis bars of Permanorm 5000 H2 are used (courtesy of Vacuumschmelze GmbH & Co. KG). The material of the hysteresis bar is shaped in order to maximize the stabilization, which is the most important part of the attitude control.

Many conditions impact on the detumbling time, which is the time required for the satellite to stabilize. Magnetic passive attitude stabilization systems such as the one developed for this mission achieve the equilibrium state within a few weeks of operation [12].

The GOLDS-UFSC satellite does not feature an orbit control subsystem.

5.5 Mechanical Structure

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5.6 Interconnection Modules

5.6.1 PC-104 Interconnection Boards

The PC-104 interconnection boards are intended to be used as an interconnection of the two PC-104 bus segments of the 2U structure (top and bottom units). This interconnection is made with a set of PicoBlade cables between the top and bottom boards. The set of two boards can be seen in Figure 5.14.

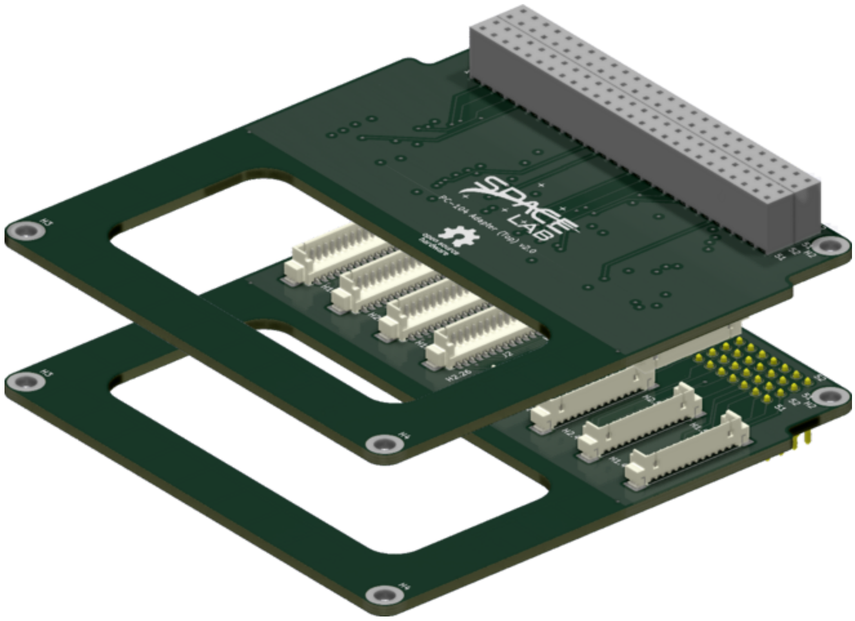

Figure 5.14: PC-104 adapter boards (top and bottom).

More information about these boards can be found in [15].

5.6.2 External Connection Boards

The Interstage Interface Panels (IIP) are three vertical internally mounted PCBs designed to give external access up to four modules inside of a 2U CubeSat during final assembly, integration and testing (AIT) before launch. The complete set of the boards allow the nanosatellite to be charged, programed and debugged. The usage of this hardware platform is taking into account the use of a MSP-FET: MSP430 Flash Emulation Tool from Texas Instruments for JTAG programing and debugging, UART debugging through a mini USB type B port interfacing the FT4232H USB bridge IC from FTDI, a JST XH header for charging internal batteries and a Remove Before Flight (RBF) pin header. The boards can seen in Figure 5.15.

More information about these boards can be found in [16].

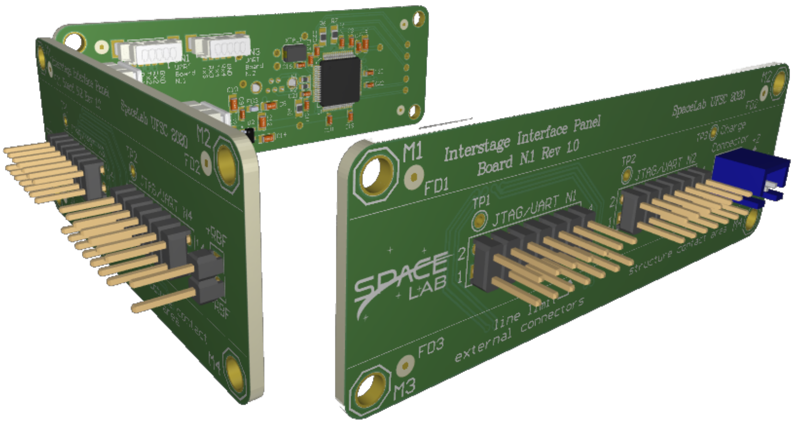

Figure 5.15: Set of external connection boards.

5.7 Payloads

The GOLDS-UFSC satellite is planned to carry three different payloads on-board: "*EDC*", "*Payload-X*" and the "*Harsh Payload*". Each one of these payloads are presented next.

5.7.1 Environmental Data Collection

The Environmental Data Collector (EDC) is a CubeSat-compatible payload that decodes signals from Platform Transmitter Terminals (PTTs) belonging to the Brazilian Environmental Data Collection System (SBCD) and the Argos-2 System. It is the main payload of the GOLDS-UFSC mission.

The main featues of this payload are listed below, a 3D model of the EDC board can be seen in Figure 5.16.

- Reception/decoding of SBCD and Argos-2 signals on the 401.635 MHz $\pm$ 30 kHz frequency range.
- Can decode up to 12 PTT signals simultaneously.
- Attaches a header to decoded messages with frequency, time, and signal strength information.
- Full speed I^2C interface (400 kbit/s) for the OBC communication.
- Full-duplex RS-485 interface with fail-safe for the OBC communication.
- 5 V power supply.
- Memory capable of storing up to 64 decoded user messages.
- Generates housekeeping information including current supply, board temperature, digitized signal RMS level, front-end PLL synchronism state and overcurrent events.
- Can capture a 2048 samples sequence (16 ms window) from the received signal upon request.

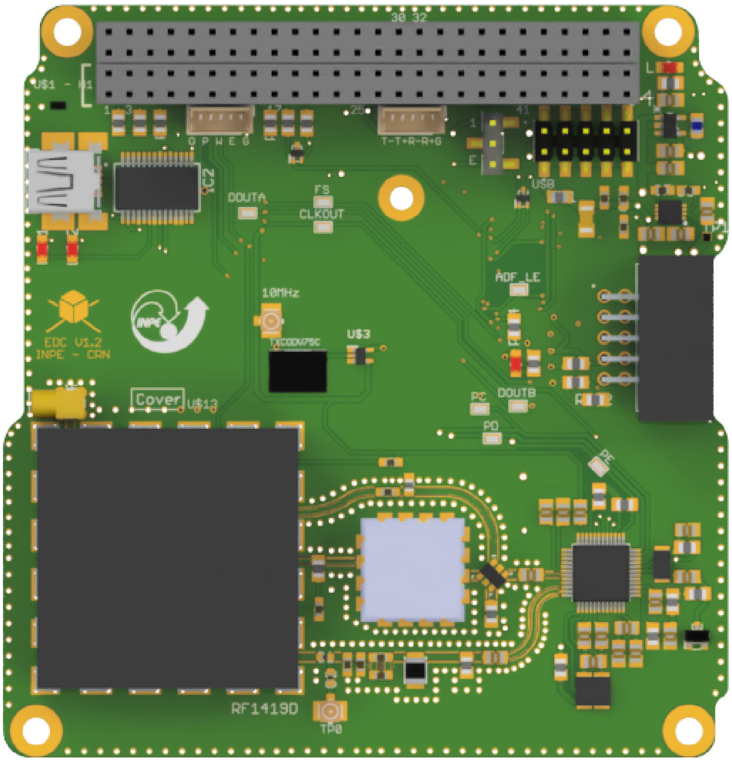

Figure 5.16: EDC board.

As can be seen in Figure 5.1, for this mission, two identical EDC boards will be used, in a cold redundancy configuration. More information about this payload can be found in [3].

5.7.2 Redundant OBDH (Payload-X)

The Payload-X is a radiation-hardened reconfigurable hardware platform designed for a radioactive environment, having as a main feature the possibility to change the hardware configuration of the FPGA through remote uplink of its bitstream.

More information about this payload can be found in [17].

5.7.3 Radiation Monitor (Harsh Payload)

The Radiation Monirot (or Harsh Payload) is a payload capable of evaluate the radiation effects on three SDRAM memories with different manufacturing nodes. This payload will test this chips in the real harsh space environment by flying aboard of GOLDS-UFSC CubeSat mission. These particular SDRAM memories were previous characterized on laboratory experiments, then by exposing them to the real environments and executing the same tests routines will not only generate more results for analysis, but also provide an opportunity to assess the test methodologies themselves. Also, after collecting sufficient data to be analysed, this payload could be used to provide a meaningful health status, concerning the radiation doses which the satellite were exposed, to the entire satellite subsystems and further missions. A picture of the harsh payload board is available in Figure 5.18.

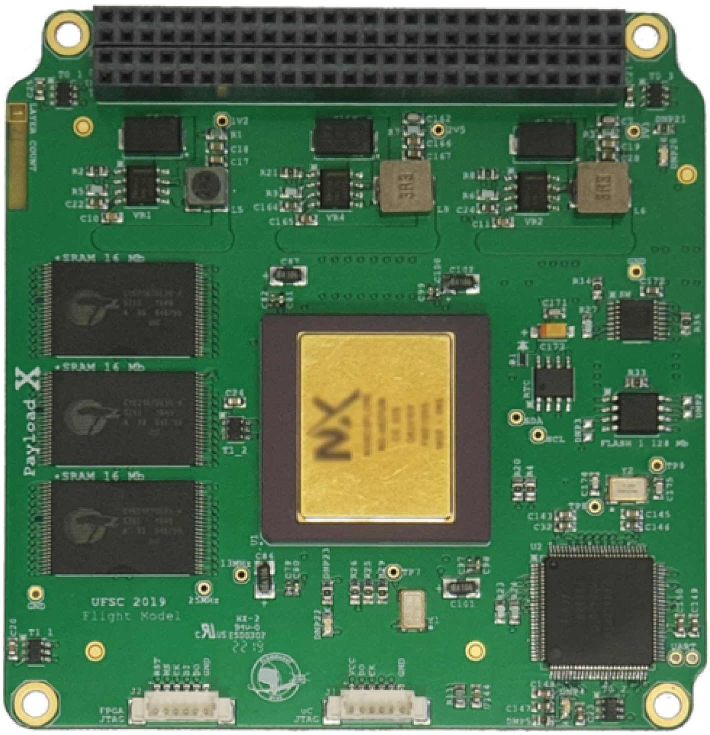

Figure 5.17: Payload-X board.

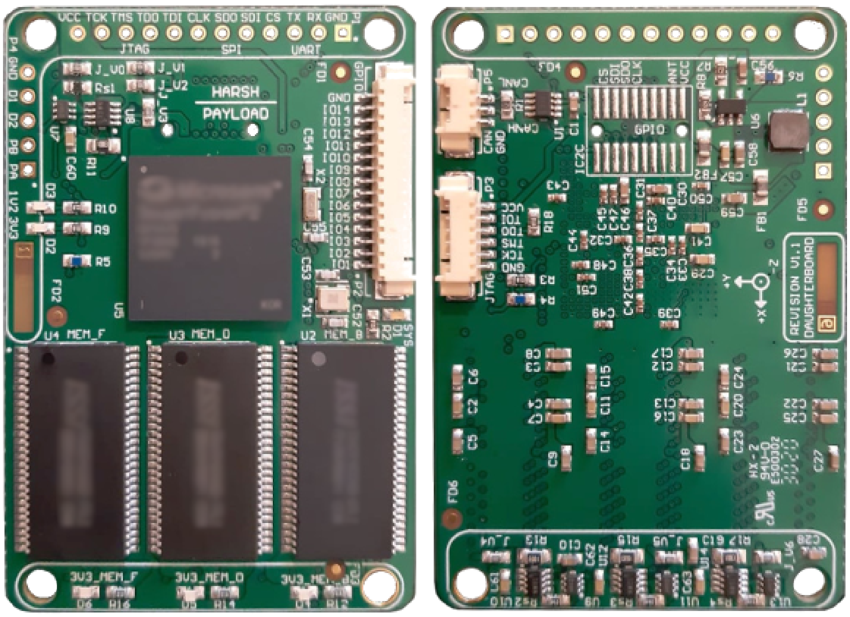

Figure 5.18: Radiation monitor board. Top (left) and bottom (right) sides.

In order to accomplish this objectives, the payload is designed to follow the OBDH DaughterBoard standard of SpaceLab, which defines the connectors, shape and size of the board. This standard allows the utilization of the module throughout future SpaceLab core missions in reason of its low space occupation inside the CubeSat, being considered further as an expansion module instead of a payload experiment. A picture of the exploded view of the harsh payload and the OBDH can be seen in Figure 5.19.

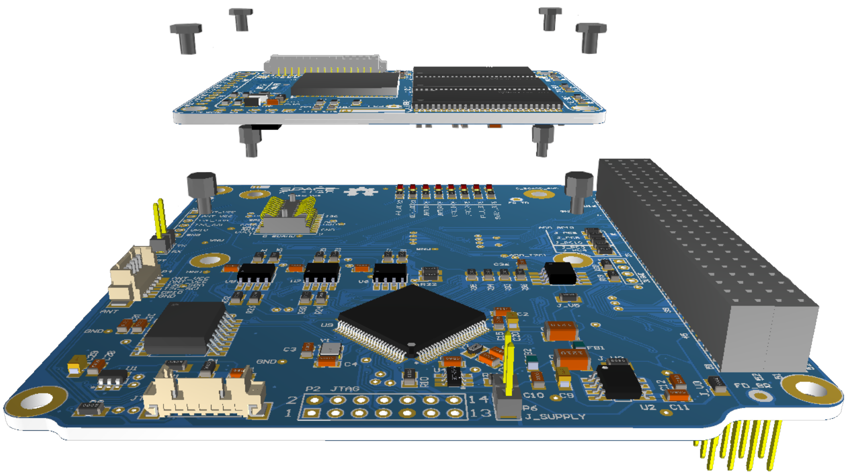

Figure 5.19: Integration of the radiation monitor payload in the OBDH.

Also, due to the mission limited power budget, the developed board should consider reduce power consumption and define clever power management strategies. In addition, methods for anti latch-up, a type of short circuit which can occur inside an IC, are considered in the design. Therefore, combining all these requirements, the payload architecture consists of the following modules: a control and management subsystem, operated by a System-On-a-Chip (SoC) solution with an integrated FPGA, power converters for properly voltage level supply, anti latch-up circuitry, communication and interface buses, debug module and the SDRAM memory chips.

CHAPTER 6

Ground Segment

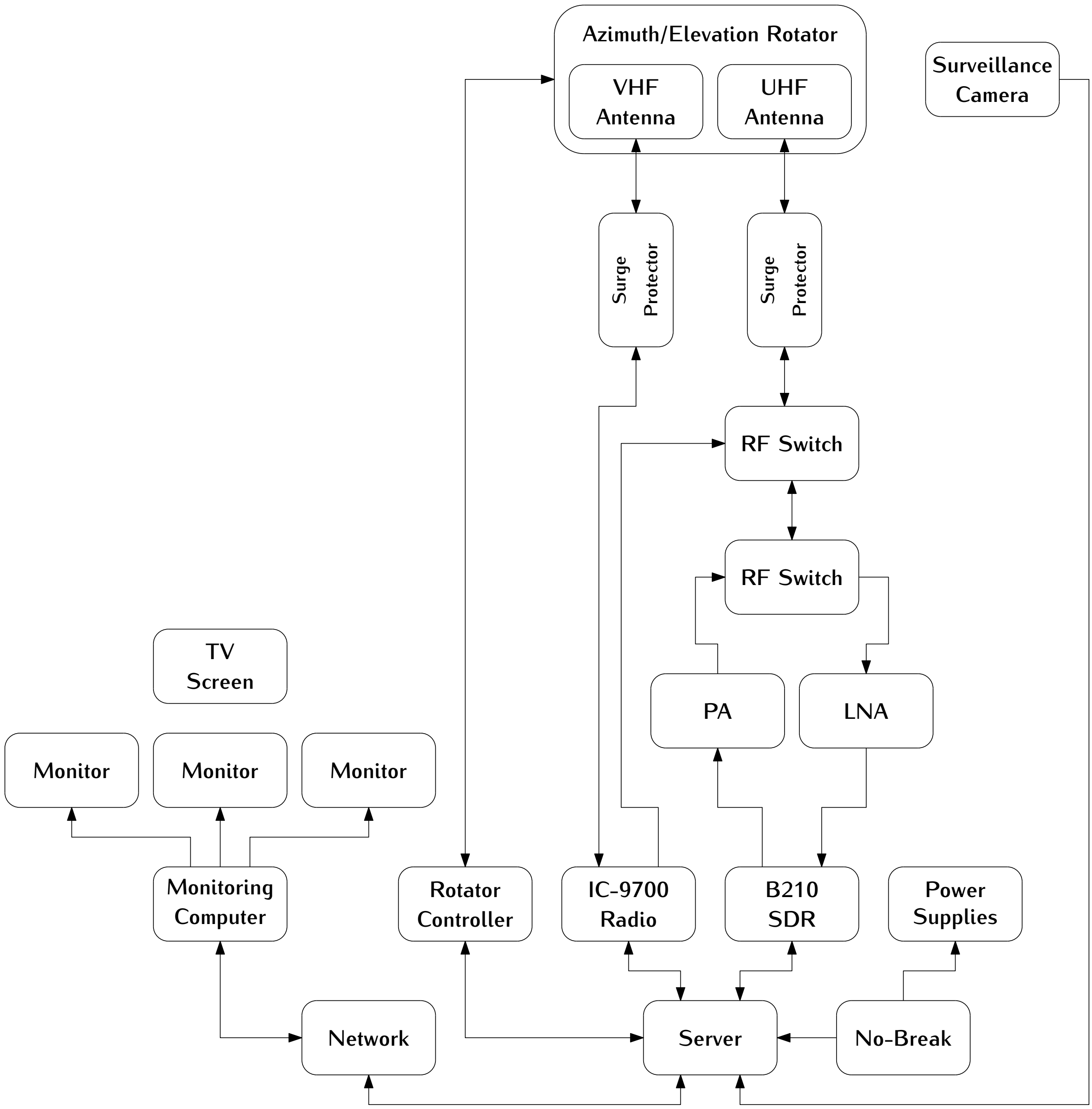

Figure 6.1: Block diagram of the ground segment.

6.1 Hardware

6.1.1 Antennas

There are two antennas in the ground station: One for VHF and one for the UHF band. The main characteristics of these antennas can be seen in Table 6.1

Characteristic	VHF Antenna	UHF Antenna	Unit
Brand	M^2	Cushcraft	-
Model	2MCP14	A719B	-
Type	Yagi	Yagi	-
Number of elements	14	19	-
Frequency range	143-148	430-450	MHz
Gain	12,34	15,5	dBi
Power rating	1500	2000	W
Boom length	3,2	4,1	m
Longest element	1,02	0,34	m
Weight	2,72	2,55	kg

Table 6.1: Main characteristics of the ground segment antennas.

More information about the VHF and UHF antennas can be found in [18] and [19] respectively.

Surge Protector

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6.1.2 Rotators

Both antennas (VHF and UHF) track the satellite through a two axis rotator (azimuth and elevation). The used model is the Yaesu G-5500, which provides 450° azimuth and 180° elevation control of medium and large size unidirectional satellite antenna arrays under remote control from station operation position.

A picture of the G-5500 rotator (and controller) can be seen in Figure 6.2, the main characteristics can be found in Table 6.2.

More information about the ground station rotator can be found in [20].

6.1.3 Amplifiers

Power Amplifier

PA...

A picture of the power amplifier can be seen in Figure 6.3, the main characteristics are available in Table 6.3.

Low Noise Amplifiers

LNA...

Figure 6.2: Yaesu G-5500 rotator and controller.

Figure 6.3: Mini-Circuits ZHL-50W-52-S+ power amplifier.

6.1.4 Radios

The Icom IC-9700 [21] is an RF direct sampling receiver for 2 m and 70 cm. The IF receiver consists of a single, down conversion for 23 cm that is between 311 and 371 MHz. The PA provides 100 W on 2 m, 75 W on 70 cm, and 10 W on 23 cm.

In addition to band specific memory channels, the IC-9700 allows band specific receiver and transmitter settings. For transmit, users can make adjustments to RF power, TX power Limit, Limit Power, and TX Delay by band. Basic receiver settings, like the Noise Blanker, Noise Reduction, and others can be tweaked by band with a dynamic Notch and Filter setup by band/mode.

A picture of the IC-9700 radio can be seen in Figure 6.4.

Characteristic	Value	Unit
Brand	Yaesu	-
Model	G-5500	-
Voltage requirement	110-120 or 200-240	V_{AC}
Motor voltage	24	V_{AC}
Rotation time (elevation, 180°)	67	s
Rotation time (azimuth, 360°)	58	s
Maximum continuous operation	5	min
Rotation torque (elevation)	14	kg-m
Rotation torque (azimuth)	6	kg-m
Braking torque (elevation and azimuth)	40	kg-m
Vertical load	200	kg
Pointing accuracy	± 4	%
Wind surface area	1	m^2
Weight (rotator)	9	kg
Weight (controller)	3	kg

Table 6.2: Main characteristics of antennas' rotators.

Characteristic	Value	Unit
Brand	Mini-Circuits	-
Model	ZHL-50W-52-S+	-
Frequency range	50-500	MHz
Gain	47-52	dB
Noise figure	4,5-7,0	dB
DC supply voltage	24-25	V
Max. supply current	9,3	A

Table 6.3: Main characteristics of the ZHL-50W-52-S+ power amplifier.

Figure 6.4: Icom IC-9700 radio transceiver.

Software Defined Radio

As presented in Figure 6.1, the ground segment also has an SDR (Software Defined Radio) as transceiver. The used model is the USRP B210, from Ettus Research [22], which is a fully integrated, single-board SDR with continuous frequency coverage from 70 MHz to 6 GHz. It combines the AD9361 RFIC direct-conversion transceiver providing up to 56 MHz of real-time bandwidth, an open and reprogrammable Spartan6 FPGA, and USB 3.0 connectivity. Also, a full support for the USRP Hardware Driver (UHD) software allows the use with the GNURadio framework.

A picture of the USRP B210 SDR (with enclosure) can be seen in Figure 6.5.

Figure 6.5: Ettus USRP B210 SDR.

6.1.5 Processing and Control

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6.2 Satellite Tracking

To track the satellite and for orbit prediction, the GPredict software [23] will be used. Gpredict is a real-time satellite tracking and orbit prediction application. It can track a large number of satellites and display their position and other data in lists, tables, maps, and polar plots (radar view). Gpredict can also predict the time of future passes for a satellite, and provide you with detailed information about each pass. Gpredict is free software licensed under the GNU General Public License. A picture of the main window of GPredict can be seen in Figure 6.6.

6.3 Packet Decoding

[24]

6.4 PCDs

PCD...

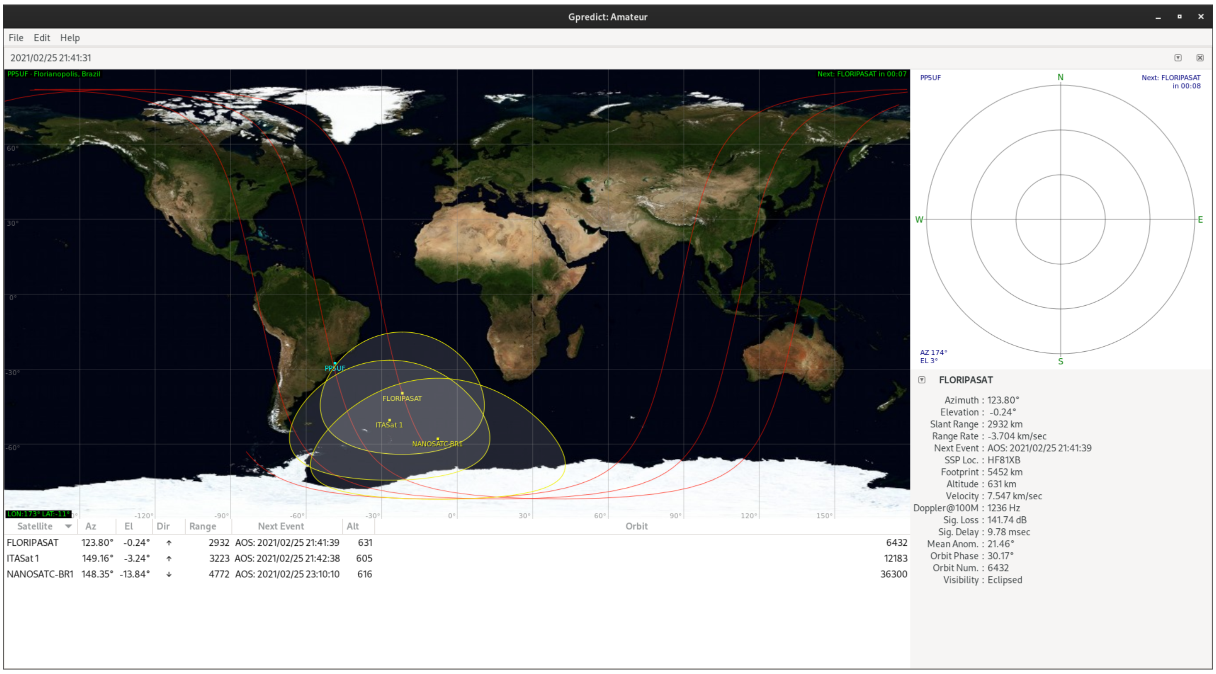

Figure 6.6: Main window of GPredict.

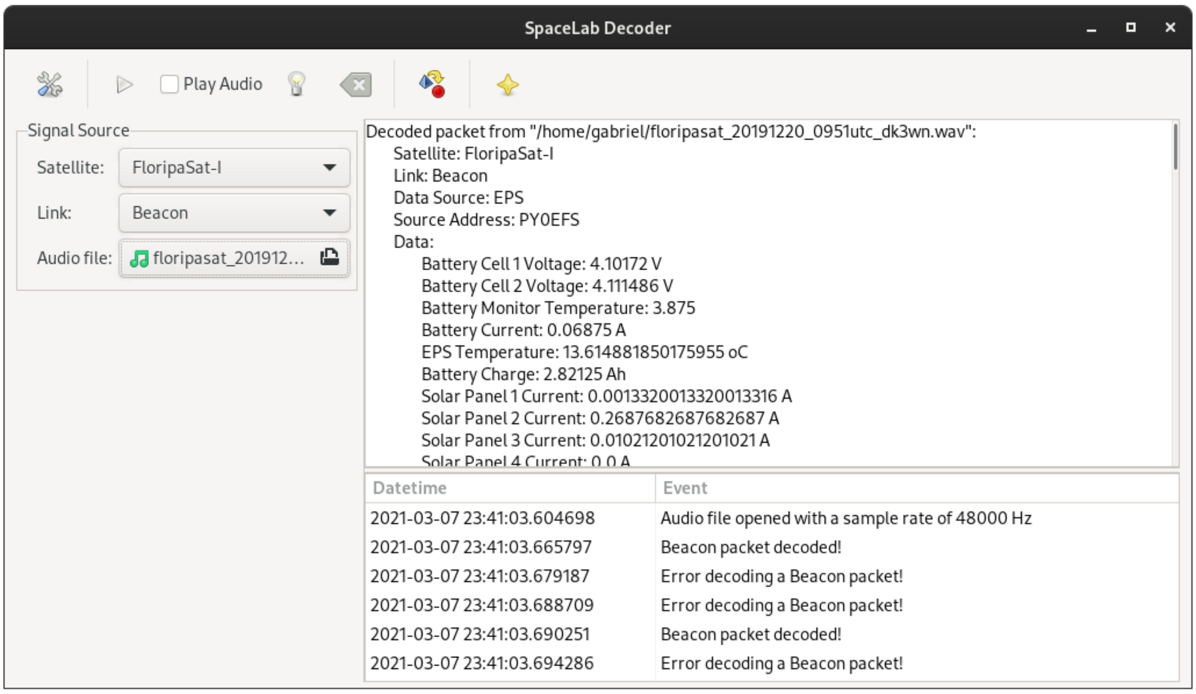

Figure 6.7: Main window of the SpaceLab Decoder application.

CHAPTER 7

Test Plan and Results

7.1 Test Procedure

7.1.1 Hardware

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7.1.2 Software

Unit Tests

- Hardware checks (might require mock circuitry).
- Driver operation checks: not extensive, might use loopback and fake sensor data schemes for hardware checks.
- Device operation checks: one test file for each device implemented, more extensive than driver checks, but should avoid development overhead.
- Standalone application checks: evaluate the application logic (masking or faking operating system calls, such as waiting for queue or a delay). It should be implement without the operating system, in other words, evaluate inputs/outputs in dedicated main file.

Integration Tests

- Operating system initialization: assert memory allocation (RAM, stack, heap), hooks and etc;
- Boot sequence (as similar to the actual procedure as possible).
- Operating system task/queue/interrupts priority, constraints, size, depth and delay checks: use dummy task/queue/interrupts (same config as actual system).
- Short-term system check: after 1 hour, exit without error logs.
- Mid-term system check: after 1 day, exit without error logs.
- Long-term system check (used in flatsat): after 1 week, exit without flatsat/integration error logs.

Workflow

- Always it is a build->flash->test, change main and repeat.
- It must have a test folder containing subfolders (hardware, drivers, devices, app, integration) and a json file (with name, path and type).
- Inside the workflow is called a python script that read this json and setup variables to allow running multiple main file swaps for each test type.
- There are 5 different workflows, one for each test type: hardware, drivers, devices, app, integration;
- The workflow, tests and scripts must be reviewed before each release.
- Idea: for short/mid/long-term tests, the workflow should evaluate the log messages offline instead of real time, in which a job is scheduled to run just after this period and "a script" will read the log file and search for the test criteria, giving the actual CI result.
- Idea: Inside the code, using the log message approach, we might create our ultra lightweight framework that consists of only log types (colors) and log messages (specific strings). This way we do not modify our current workflow and we can add a simple scheme to access the flight code.
- Unit Tests = Tests performed per firmware unit.
- Integration Tests = Tests performed per firmware component (several units abstracted).

7.1.3 Flatsat

To test all modules during the development of the projet, a flatsat platform was developed. The FlatSat Platform is a testbed for CubeSat PCB modules. FlatSats enable easier, faster and a secure method for testing subsystens independently while been integrated in a flat design before going to integration on a CubeSat form factor. The PCB can support up to 7 modules, all PC-104 pins are interligated to flexibilize its use, only the particularity connection between modules need to be be taken into account. One PC-104 has inverted pinout, the board also makes it possible to have two seperate power supplies, a UART to USB converter for 4 modules, kill-switches activation though SPDTs, Remove Before Flight (RBF) pin header, connector for charging batteries and SMA connectors for antennas. A picture of the flatsat board can be seen in Figure 7.1.

More information about the Flatsat Platform can be found in [25].

7.1.4 Environmental Tests

LIT

[26]

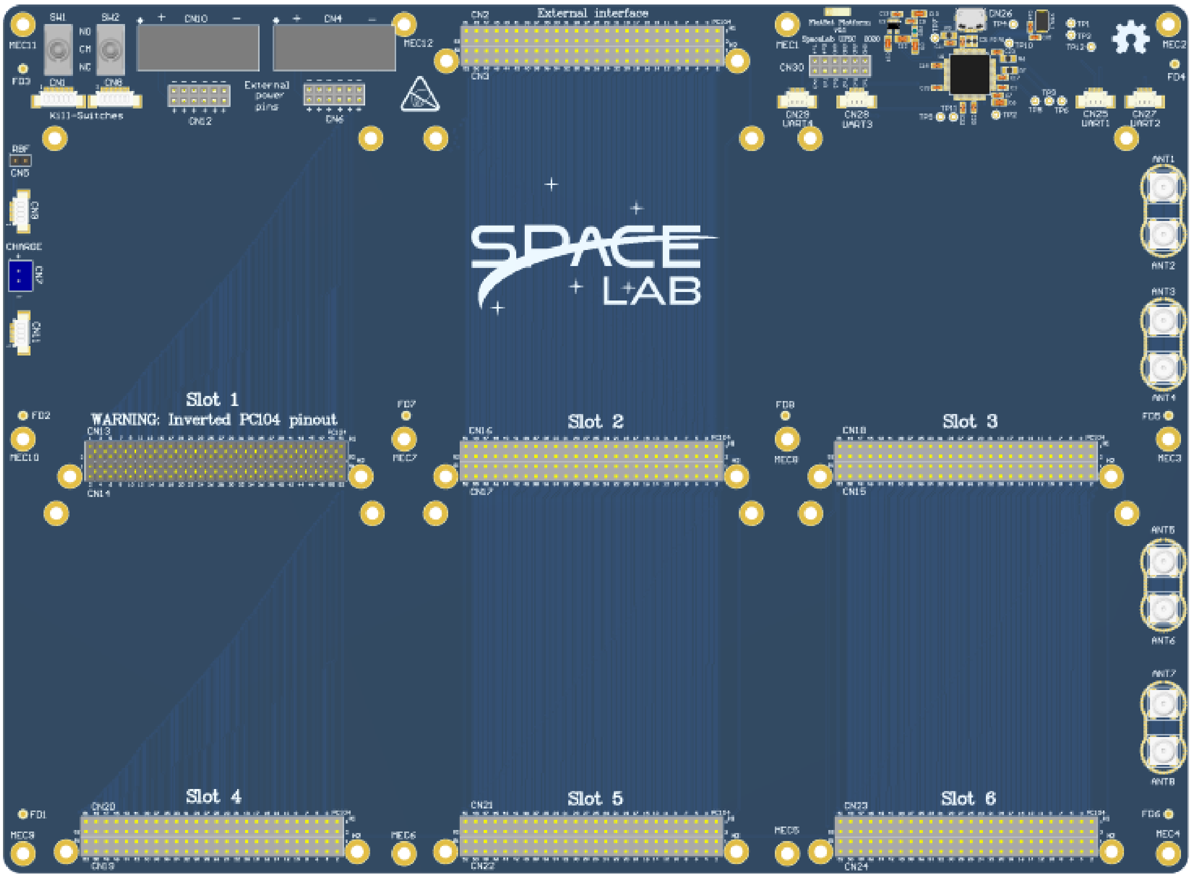


Figure 7.1: Top view of the flatsat board.

Mass Verification

This test checks the total mass of the satellite (without RBF tag), which must be less than 2,66 kg [27]. The verification is made with a precision balance. Figure 7.2 examplifies this process with FloripaSat-I total mass.

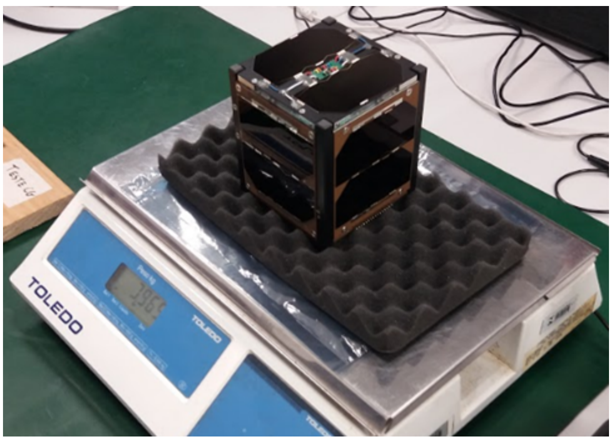
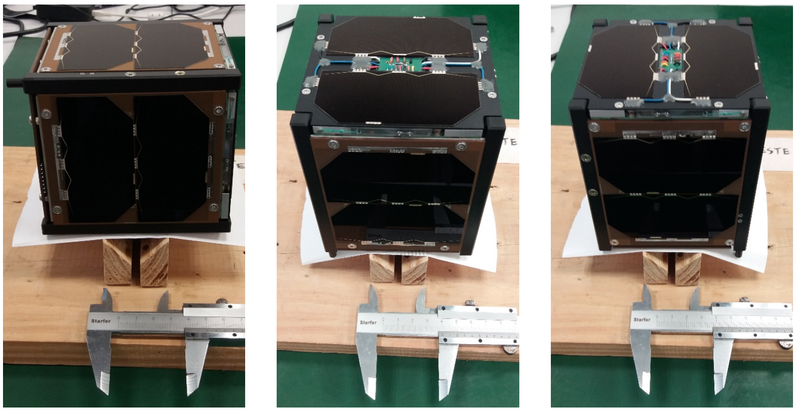


Figure 7.2: Mass verificatiton of FloripaSat-I.

Center of Gravity

This test checks the center of gravity (CG) of the satellite, which must be less than 2 cm from the geometric center (see Figure 7.3) [27]. To perform this test, a simple test-bench based on two parallel bars fixed on a plate (4 cm from each other) can be used. The geometric center of the satellite is put in the middle of the bars and, if the satellite does not fall, the CG is within the radius of 2 cm. This strategy does not measure the location of CG, however, it does prove if the satellite follows the requirement.

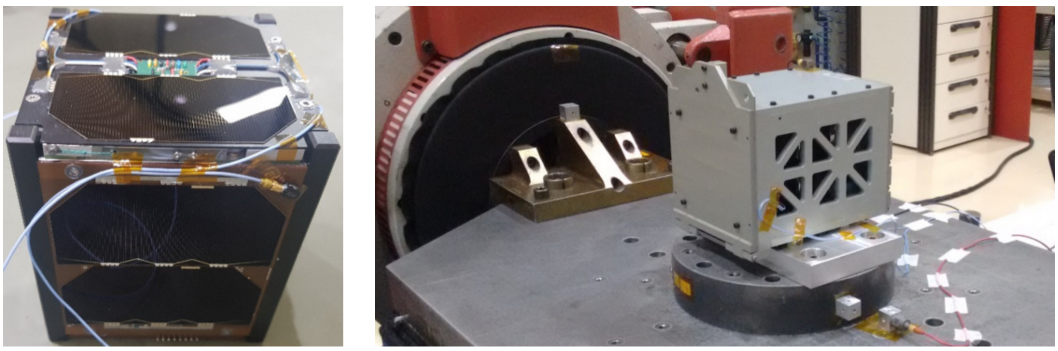


(a) X axis. (b) Y axis. (c) Z axis.

Figure 7.3: Center of gravity of FloripaSat-I within 2 cm from geometric center.

Vibration Test

To measure and control the acceleration profile during the dynamic tests, accelerometers should be positioned on three external surfaces of the satellite, one on each axis, over areas without solar cells. The satellite should be fixed on a shaker. Figure 7.4(a) shows some of the accelerometers and Figure 7.4(b) shows the satellite during a vibration test.



(a) Position of the accelerometers. (b) Shaker.

Figure 7.4: Vibration test.

The CubeSat should be tested entirely off, with RBF pin removed but with the Kill-Switches pressed, in a 2U Test POD, simulating the normal launching condition. The set of vibration tests follows Figure 7.5.

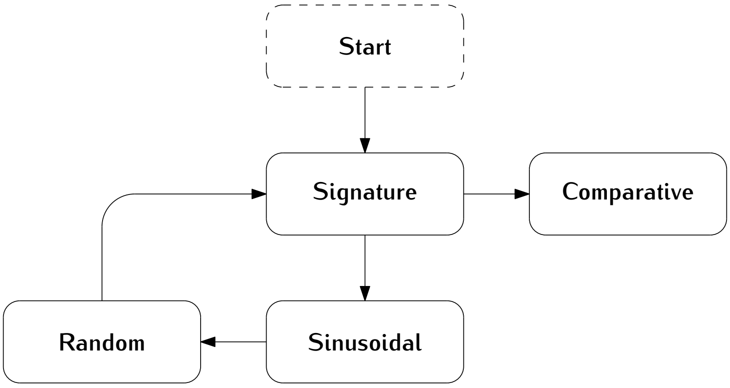


Figure 7.5: Sequence of dynamic tests.

A signature testing should be conducted before and after the tests (sinusoidal and random vibration), in order to identify the presence of significant variations in the dynamic response, a condition that may represent mechanical failures. For the signature task, Table 7.1 presents the specifications.

Name	**Parameter**
Test method	Sinusoidal sweep testing
Frequency range	5 - 2000 Hz
Vibration level	0,25 g
Sweep rate	2 octaves per minute
Number of sweeps	1 (5 - 2000 Hz)
Test axes	3 (X, Y, Z)

Table 7.1: Resonance survey test (signature).

Thermal Test

For the thermal tests, thermocouples should be attached on different points on the surface of the satellite, including over the solar panels and structure. As an example, **??** shows FloripaSat-I ready for thermal tests. The parameters of the tests are indicated in Table 7.2.

7.2 Preliminary Results

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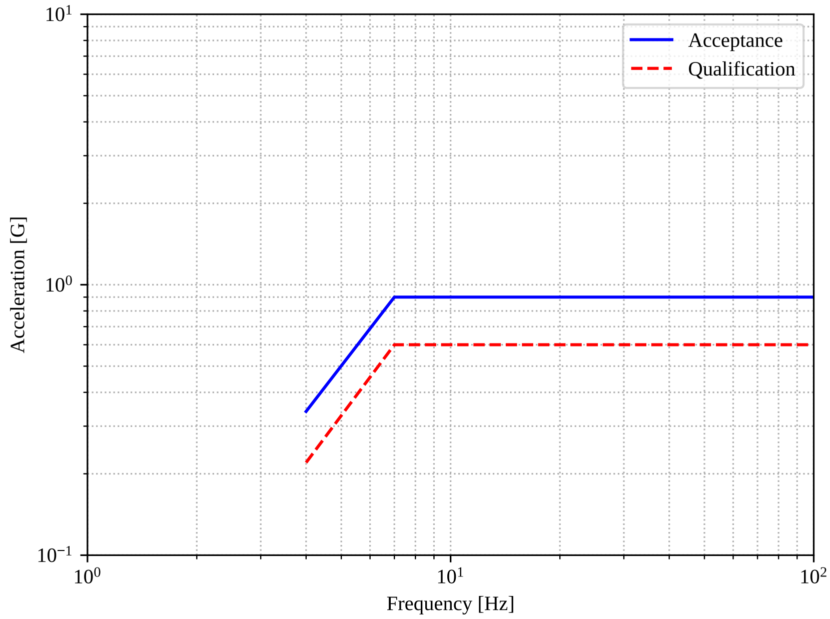

Figure 7.6: Sinusoidal sweeping vibration curve.

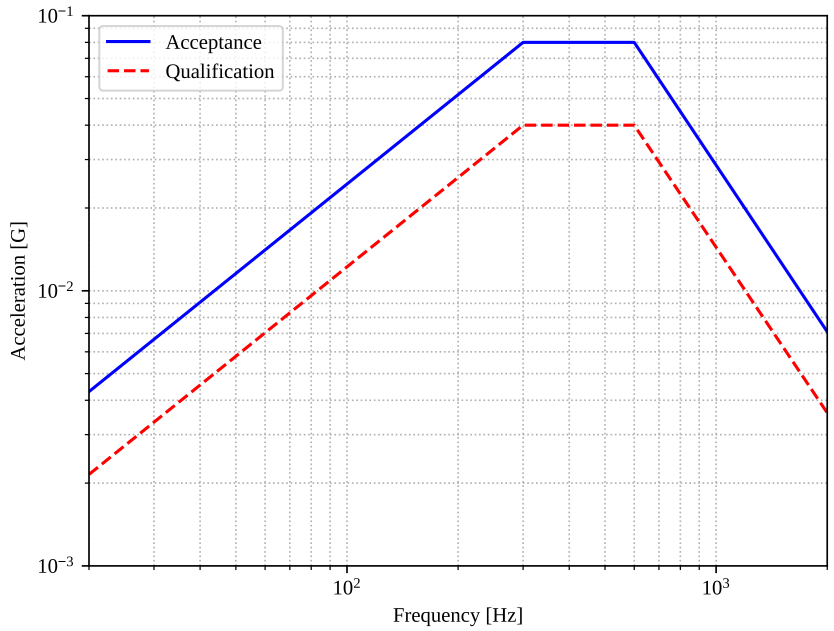

Figure 7.7: Random vibration curve.

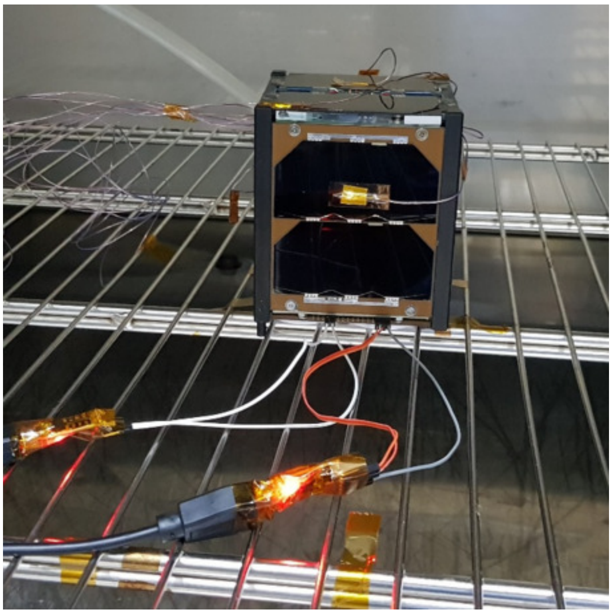

Figure 7.8: FloripaSat-I during the thermal cycling (with thermocouples).

Thermal cycle		Bake out	
Parameter	**Value**	**Parameter**	**Value**
Number of cycles	2	Part 1	
Min. temp. (T_{min})	-15 °C	Pressure	$<1\times10^{-4}$ mbar
Max. temp. (T_{max})	+50 °C	Temperature	23 °C
Duration in T_{min}	30 min	Duration	12 hours
Duration in T_{max}	60 min	Part 2	
Heating rate	5.5 °C/min	Pressure	$<1\times10^{-4}$ mbar
Cooling rate	3.5 °C/min	Temperature	60 °C
Stabilization criteria	1 °C/10 min	Duration	6 hours

Table 7.2: Parameters for the bake out and thermal cycling.

CHAPTER 8

Assembly, Integration and Test

AIT...

8.1 Assembly Instructions

8.1.1 Preparation and Required Material

- .
- .

8.1.2 Assembly Steps

1. .
2. .

8.2 Environmental Testing

- Mass verification
- Dimensions verification (fit check)
- Center of gravity (CG) verification
- Vibration test
- Thermal test
- Bake out test

8.2.1 Mass, Center of Gravity and Fit Check

1. .
2. .

8.2.2 Vibration Test

1. .
2. .

8.2.3 Thermal Cycling

1. .
2. .

8.2.4 Bake Out

1. .
2. .

8.3 Pre-launch Preparation

1. .
2. .

8.3.1 Keys of the Telecommands

1. .
2. .

8.3.2 Firmware Upload

1. .
2. .

8.3.3 Memory Reset

1. .
2. .

8.4 Transport to Launch

8.4.1 Packing the Satellite

1. .
2. .

8.4.2 Unpacking the Satellite

1. .

2. .

CHAPTER 9

Operation Planning

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APPENDIX A

Link Budget Calculation

This appendix shows the link budget calculation of all the satellite links (including the radio links of the payloads). The used method was taken from [28] (section 13.3).

A.1 Distance to Satellite at Horizon

The distance to satellite at horizon (the maximum theoretical distance between the satellite and a ground station) can be calculated using Equation A.1.

$$d = \sqrt{2 \cdot R_e \cdot h + h^2} \tag{A.1}$$

Where:

- R_e = Earth radius = 6378 km
- h = Satellite altitude = 600 km
- d = Distance to sattellite at horizon

So, the distance to satellite at horizon is:

$$d = \sqrt{2 \cdot 6378 \cdot 600 + 600^2} = \mathbf{2830\ km} \tag{A.2}$$

A.2 Free-Space Path Loss

The free-space path loss ($FSPL$) can be calculated using Equation A.3.

$$FSPL = \left(\frac{4\pi df}{c}\right)^2 \tag{A.3}$$

Where:

- d = Distance between the satellite and the ground station
- f = Radio frequency
- c = Speed of light

The FSPL value in decibels can be calculated with Equation A.4.

$$\begin{aligned} FSPL^{dB} &= 20\log\left(\frac{4\pi}{c}\right) + 20\log(d) + 20\log(f) \\ &= 32,45 + 20\log\left(\frac{d}{1\ km}\right) + 20\log\left(\frac{f}{1\ MHz}\right) \end{aligned} \tag{A.4}$$

The minimum distance between the satellite and a ground station is the satellite altitude, in this case: 600 km. The maximum distance is the distance at horizon, defined by Equation A.2.

A.2.1 Beacon

Considering the frequency of the beacon as 145,97 MHz, the minimum and maximum FSBL is:

$$FSPL^{dB}_{max} = 32,45 + 20\log\left(\frac{2830}{1\ km}\right) + 20\log\left(\frac{145,97}{1\ MHz}\right) = \mathbf{144,8\ dB} \tag{A.5}$$

$$FSPL^{dB}_{min} = 32,45 + 20\log\left(\frac{600}{1\ km}\right) + 20\log\left(\frac{145,97}{1\ MHz}\right) = \mathbf{131,3\ dB} \tag{A.6}$$

$$\mathbf{131,3 \leq FSPL^{dB} \leq 144,8\ dB} \tag{A.7}$$

A.2.2 Downlink/Uplink

Considering the frequency of the downlink/uplink as 436,9 MHz, the minimum and maximum FSBL is:

$$FSPL_{max} = 32,45 + 20\log\left(\frac{2830}{1\ km}\right) + 20\log\left(\frac{436,9}{1\ MHz}\right) = \mathbf{154,3\ dB} \tag{A.8}$$

$$FSPL_{min} = 32,45 + 20\log\left(\frac{600}{1\ km}\right) + 20\log\left(\frac{436,9}{1\ MHz}\right) = \mathbf{140,8\ dB} \tag{A.9}$$

$$\mathbf{140,8 \leq FSPL^{dB} \leq 154,3\ dB} \tag{A.10}$$

A.2.3 Uplink (Payload)

Considering the frequency of the payload's uplink is 401,635 MHz, the minimum and maximum FSBL is:

$$FSPL_{max} = 32,45 + 20\log\left(\frac{2830}{1\ km}\right) + 20\log\left(\frac{401,635}{1\ MHz}\right) = \mathbf{153,6\ dB} \tag{A.11}$$

$$FSPL_{min} = 32,45 + 20\log\left(\frac{600}{1\ km}\right) + 20\log\left(\frac{401,635}{1\ MHz}\right) = \mathbf{140,1\ dB} \tag{A.12}$$

$$\mathbf{140,1 \leq FSPL^{dB} \leq 153,6\ dB} \tag{A.13}$$

A.3 Signal-to-Noise-Ratio

The Signal-to-Noise-Ratio (SNR) of a transmitted signal at the receiver can be expressed using Equation A.14:

$$SNR = \frac{E_b}{N_0} = \frac{P_t G_t G_r}{k T_s R L_p} \tag{A.14}$$

Where:

- P_t = Transmitter power
- G_t = Transmitter gain
- G_r = Receiver gain
- k = Boltzmann's constant ($\cong 1,3806 \times 10^{-23}\ J/K$)
- T_s = System noise temperature
- R = Data rate in bits per seconds (bps)
- L_p = Free-Space Path Loss (FSPL)

The system noise temperature (T_s) can be defined using Equation A.15.

$$T_s = T_{ant} + T_r \tag{A.15}$$

with:

$$T_r = \frac{T_0}{L_r}(F - L_r) \tag{A.16}$$

and:

$$F = 1 + \frac{T_r}{T_0} = 1 + \frac{323}{290} = 2,114 \tag{A.17}$$

The SNR value in decibels can be calculated using the Equation A.18:

$$\begin{aligned} SNR^{dB} &= 10\log_{10}\left(\frac{E_b}{N_0}\right) = 10\log_{10}\left(\frac{P_t G_t G_r}{k T_s R L_p}\right) \\ &= P_t^{dBm} - 30 + G_t^{dBi} + G_r^{dBi} - L_p^{dB} - 10\log k - 10\log T_s - 10\log R \end{aligned} \tag{A.18}$$

Considering other losses in the system (L_s) (cable and connection losses as example), the Equation A.18 can be corrected as presented in Equation A.19.

$$SNR^{dB} = P_t^{dBm} - 30 + G_t^{dBi} + G_r^{dBi} - L_p^{dB} - L_s^{dB} - 10\log k - 10\log T_s - 10\log R \tag{A.19}$$

A.3.1 Beacon

Using Equations A.19 and A.15, with:

- $P_t = 30\ dBm$
- $G_t = 0\ dBi$
- $G_r = 12\ dBi$
- $L_p = 144,8\ dB$
- $L_s = 5\ dB$
- $R = 1200\ bps$
- $T_0 = 290\ K$
- $T_r = 323\ K$
- $T_{ant} = \ K$

$$F = 1 + \frac{T_r}{T_0} = 1 + \frac{323}{290} = 2,114 \tag{A.20}$$

$$T_r = \frac{T_0}{L_r}(F - L_r) = \frac{290}{L_r}(2,114 - L_r) = \ K \tag{A.21}$$

$$T_s = T_{ant} + T_r == \ K \tag{A.22}$$

$$SNR^{dB} = 30 - 30 + 0 + 12 - 144,8 - 5 + 228,6 - 30,64 - 30,79 = 30,37\ dB \tag{A.23}$$

$$\mathbf{SNR^{dB} \geq 30,87\ dB} \tag{A.24}$$

A.3.2 Downlink

Using Equations A.19 and A.15, with:

- $P_t = 30\ dBm$
- $G_t = 0\ dBi$
- $G_r = 15\ dBi$
- $L_p = 154,3\ dB$
- $L_s = 5\ dB$
- $R = 4800\ bps$
- $T_0 = 290\ K$

- $T_r = \ K$
- $T_{ant} = \ K$

$$SNR^{dB} = 30 - 30 + 0 + 15.5 - 154,3 - 5 + 228,6 - 30,64 - 36,81 = 17,35\ dB \quad \text{(A.25)}$$

$$\mathbf{SNR^{dB} \geq 17,35\ dB} \quad \text{(A.26)}$$

A.3.3 Uplink

Using Equations A.19 and A.15, with:

- $P_t = 30\ dBm$
- $G_t = 15\ dBi$
- $G_r = 0\ dBi$
- $L_p = 154,3\ dB$
- $L_s = 7\ dB$
- $R = 4800\ bps$
- $T_0 = 290\ K$
- $T_r = \ K$
- $T_{ant} = \ K$

$$SNR^{dB} = 47 - 30 + 15.5 + 0 - 154,3 - 7 + 228,6 - 31,39 - 36,81 = 31,60\ dB \quad \text{(A.27)}$$

$$\mathbf{SNR^{dB} \geq 31,60\ dB} \quad \text{(A.28)}$$

A.3.4 Uplink (Payload)

TBD

A.4 Link Margin

From [28], the minimum SNR value at the received considering a 10^{-5} bit error rate is:

- Beacon: $SNR^{dB} \geq 9,6\ dB$
- Downlink/Uplink: $SNR^{dB} \geq 9,6\ dB$
- Uplink (payload): $SNR^{dB} \geq XX\ dB$

And considering the link margin as the SNR of the link minus the SNR threshold for a given bit error, the link margin of the radio links of the satellite are:

- Beacon: $30,87 - 9,6 = \mathbf{21,27\ dB}$
- Downlink: $17,35 - 9,6 = \mathbf{7,75\ dB}$
- Uplink: $31,60 - 9,6 = \mathbf{22\ dB}$
- Uplink (payload): $X - X = \mathbf{XX\ dB}$