

Numerical study of the airflow over a high-altitude pseudo-satellite wing

PhD update

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**von KARMAN INSTITUTE
FOR FLUID DYNAMICS**

PhD presentation after 1.5 years

Sections

1 Numerical Methods

- ▶ Numerical Methods
- ▶ Performances
- ▶ Synthetic Eddy Method
- ▶ Laminar Separation Bubble
- ▶ Other activities
- ▶ Element of novelty
- ▶ Next steps

Numerical Method implemented

1 Numerical Methods

The numerical method currently implemented are:

- Variational Multiscale Method (VMS)
- Streamline upwind Petrov–Galerkin (SUPG)

Different solution method are available for all of them

- Non-linear (NLIN)
- Linearized-Coupled (LC-VMS)
- Linearized-Segregated (LS-VMS)

Variational Multiscale Method

1 Numerical Methods

- Evolution of the SUPG
- Implicit LES
- It does not need calibration
- Residual-based stabilization

Galkerkin formulation

1 Numerical Methods

Conservation of mass:

$$\nabla \cdot \vec{u} = 0 \quad (1)$$

Conservation of momentum:

$$\frac{\partial \vec{u}}{\partial t} + (\vec{u} \cdot \nabla) \vec{u} + \nabla p - \nu \Delta \vec{u} - f = 0 \quad (2)$$

Variational Formulation

$$B^G = \int_{\Omega} \frac{\partial \vec{u}}{\partial t} \cdot \vec{v} \, d\Omega + \int_{\Omega} (\vec{u} \cdot \nabla) \vec{u} \cdot \vec{v} \, d\Omega + \int_{\Omega} \nabla(p) \cdot \vec{v} \, d\Omega + \int_{\Omega} \nu \nabla \vec{u} \cdot \nabla \vec{v} \, d\Omega - \int_{\Omega} f \cdot \vec{v} \, d\Omega + \int_{\Omega} q(\nabla \cdot \vec{u}) \, d\Omega = 0 \quad (3)$$

Stabilization equations

1 Numerical Methods

$$B^{SUPG}(t, (\vec{u}, p), (\vec{v}, q)) = \int_{\Omega} (\tau_m (\vec{u} \cdot \nabla \vec{v} + \nabla q) \cdot \vec{R}_m) d\Omega + \int_{\Omega} \tau_c (\nabla \cdot \vec{v}) R_c d\Omega \quad (4)$$

$$B^{VMS1}(t, (\vec{u}, p), (\vec{v}, q)) = \int_{\Omega} (\vec{u} \cdot \nabla \vec{v}') \odot (\tau_m \vec{R}_m) d\Omega \quad (5)$$

$$B^{VMS2}(t, (\vec{u}, p), (\vec{v}, q)) = - \int_{\Omega} (\nabla \vec{v} \odot (\tau_m \vec{R}_m \otimes \tau_m \vec{R}_m)) d\Omega \quad (6)$$

Stabilization parameters

1 Numerical Methods

$$\tau_m = \left(\frac{4}{\Delta t^2} + \vec{u} \cdot G \vec{u} + C_I \nu^2 G : G \right)^{-1/2} \quad (7)$$

$$\tau_c = (\tau_c \vec{g} \cdot \vec{g})^{-1} \quad (8)$$

Where G is the inverse of the gradient of the map cell. For a cubed shaped element, with h the edge length, $G_{ij} = \frac{1}{h^2} \delta_{ij}$, where δ_{ij} is the Kronecker delta.

Linearization

1 Numerical Methods

$$\tilde{u} = 2.1875u^n - 2.1875u^{n-1} + 1.3125u^{n-2} - 0.3125u^{n-3} \quad (9)$$

$$(\vec{u} \cdot \nabla)\vec{u} \Rightarrow (\tilde{\vec{u}} \cdot \nabla)\vec{u} \quad (10)$$

Boundary Layer Initialization

1 Numerical Methods

- Avoid Instabilities (close to the leading edge)
- Avoid velocity ramping
- Higher time-step

Using the function exploited by RANS solvers for the wall distance for computing turbulent parameters but now used to detect the airfoil's contour

$$\begin{cases} \nabla \cdot (|\nabla u_p|^{p-2} \nabla u_p) = -1 & x \in \Omega \\ u_p = 0 & x \in \Omega_D \end{cases} \quad (11)$$

The true-wall distance is provided by solving the equation for $p \rightarrow \infty$ Solved using a method that resembles Picard's

Boundary Layer Initialization

1 Numerical Methods

The velocity in the x direction in the region identified can be with a simple cubic function (12) where $dn = d/\delta_{99}$, d is the minimum distance to the airfoil, u_∞ is the free-stream flow speed.

$$f(dn) = \begin{cases} u_\infty & dn > 1 \\ (-dn^2 + 2 \cdot dn) \cdot u_\infty & dn < 1 \end{cases} \quad (12)$$

The boundary layer function has been obtained by fixing the following boundary conditions:

- Continuity with the external flow, $f(1) = 1$
- Smooth transition between boundary layer and external flow, $f'(1) = 0$
- Non-slip condition at the wall, $f(0) = 0$

Boundary Layer Initialization

1 Numerical Methods

It results in a low-speed zone close to the airfoil, avoiding high speed in really small cells useful for capturing the boundary layer.

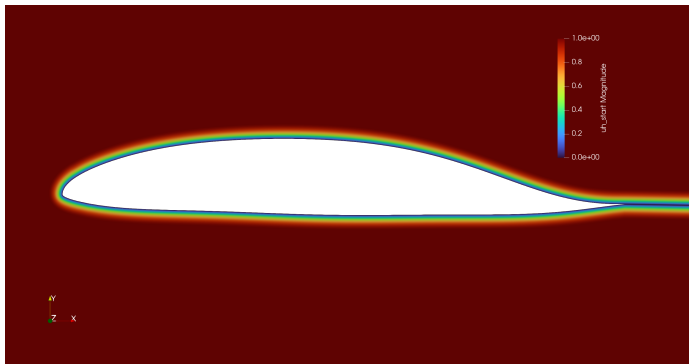


Figure: Boundary Layer Initialization

Sections

2 Performances

- ▶ Numerical Methods
- ▶ **Performances**
- ▶ Synthetic Eddy Method
- ▶ Laminar Separation Bubble
- ▶ Other activities
- ▶ Element of novelty
- ▶ Next steps

Weak Scaling

2 Performances

Weak scalability: the solution time almost does not change with constant problem size per processor

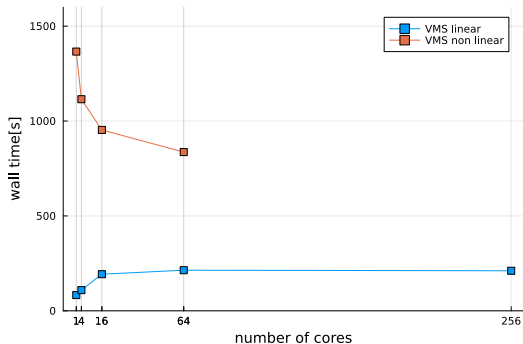


Figure: Taylor Green weak scalability

Strong Scaling

2 Performances

Strong scalability: doubling the number of processors halves the solution time

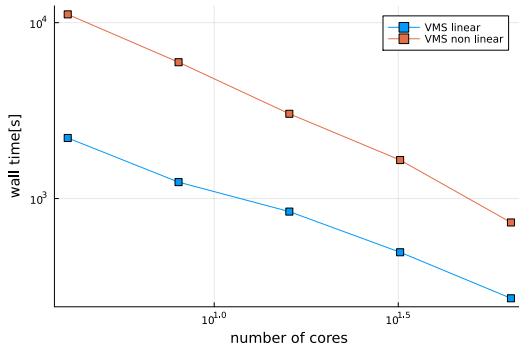


Figure: Taylor Green Strong scalability

Sections

3 Synthetic Eddy Method

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- ▶ Performances
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SyntheticEddyMethod.jl

3 Synthetic Eddy Method

- publication
- presented at *JuliaCon2023* at MIT

Features:

- Create fluctuations that respect the divergence-free condition (DFSEM)
- Create velocity fluctuations for inlet boundary conditions
- Create coherenteddies in 3D domain
- Define custom Reynolds Stress Tensor
- Import from file custom Reynolds Stress Tensor

Synthetic Eddy Method

3 Synthetic Eddy Method

Reynolds decomposition:

$$\vec{u}(\vec{x}, t) = \vec{U}(\vec{x}, t) + \vec{u}'(\vec{x}, t) \quad (13)$$

Compute velocity fluctuations, using a suitable shape function:

$$u_i(\mathbf{x}) = U_i(\mathbf{x}) + \frac{1}{\sqrt{N}} \sum_{k=1}^N a_{ij} \epsilon_j^k f_{\sigma(\mathbf{x})}(\mathbf{x} - \mathbf{x}^k) \quad (14)$$

Sections

4 Laminar Separation Bubble

- ▶ Numerical Methods
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Simulations

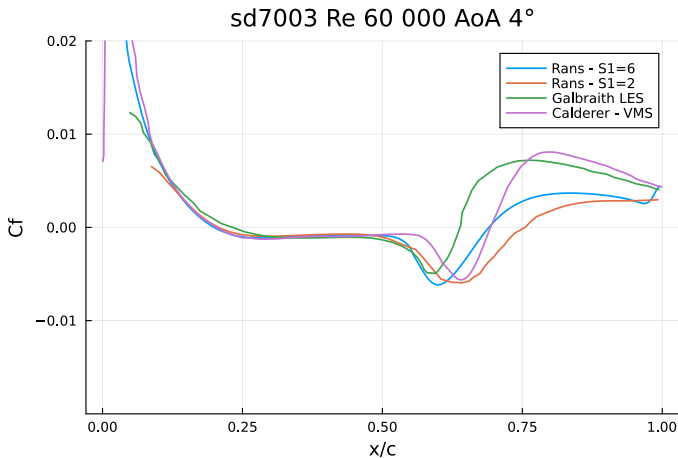
4 Laminar Separation Bubble

- VMS linearized coupled sd7003s - Re 60 000 - AoA 4°
- VMS linearized segregated DU89 - Re 250 000 — 500 000 - AoA 1° — 5°

Models

4 Laminar Separation Bubble

sd7003s - Re 60 000 - AoA 4°



VMS linearized coupled sd7003s

4 Laminar Separation Bubble

Copuled: velocity and pressure are solved at the same time. $Re\ 60\ 000 - AoA\ 4^\circ$.

Initialization with velocity-ramping

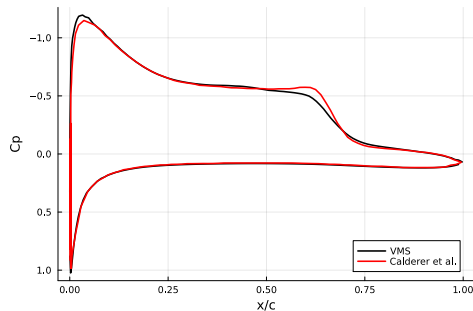
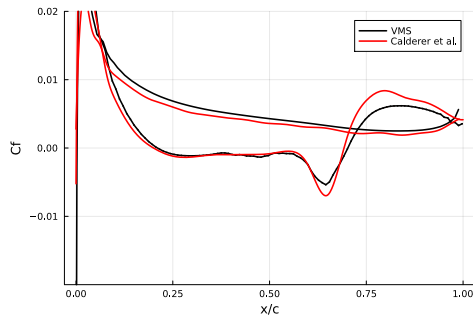


Figure: Comparison with VMS literature results

VMS linearized coupled sd7003s

4 Laminar Separation Bubble

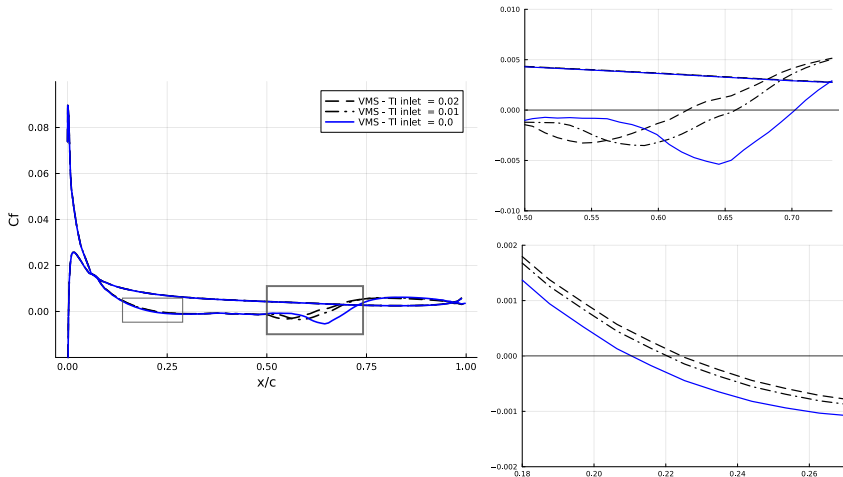


Figure: Bubble position function of freestream turbulence intensity

VMS Linearized-Segregated

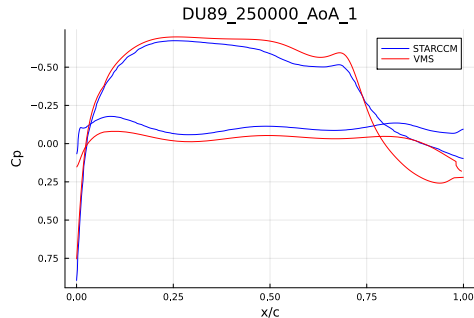
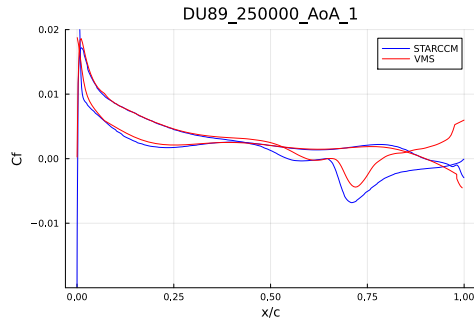
4 Laminar Separation Bubble

Segregated: each time step pressure and velocity system are solved one after the other multiple times. It is possible to re-use the matrices and preconditioner. It is an iterative method.

VMS Linearized-Segregated

4 Laminar Separation Bubble

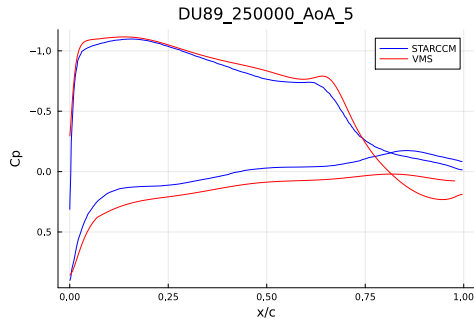
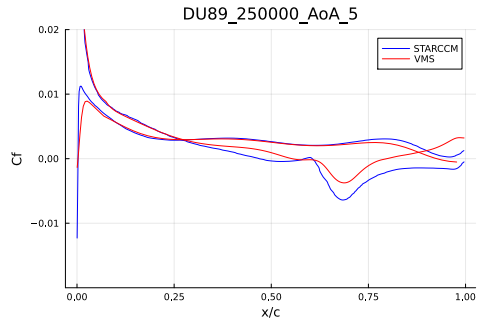
PhD research aims to simulate new a new airfoil. Re 250 000 - AoA 1°



VMS Linearized-Segregated

4 Laminar Separation Bubble

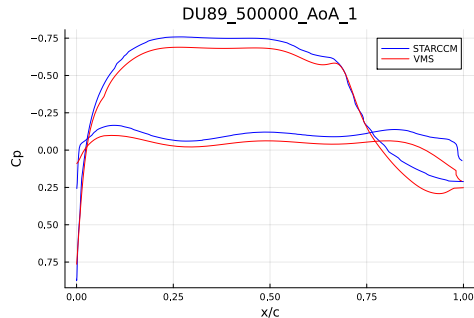
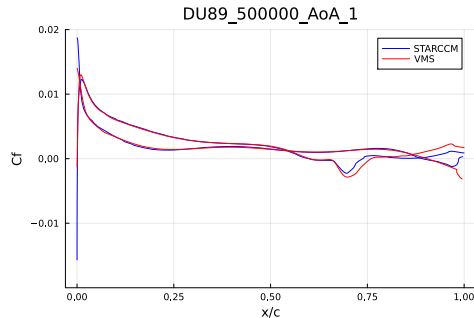
Re 250 000 - AoA 5°



VMS Linearized-Segregated

4 Laminar Separation Bubble

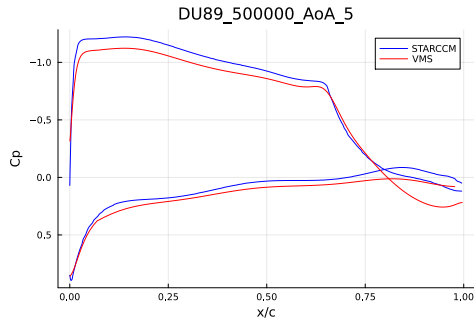
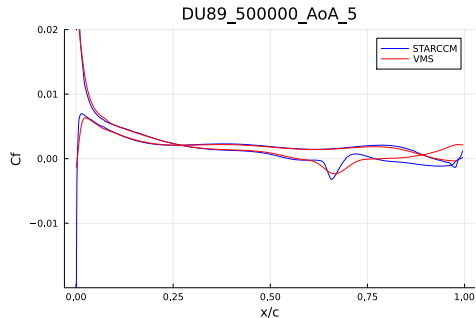
Re 500 000 - AoA 1°



VMS Linearized-Segregated

4 Laminar Separation Bubble

Re 500 000 - AoA 5°



Sections

5 Other activities

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Other activities

5 Other activities

- Synthetic Eddy Method publication
- JuliaCon 2023 conference
- Co-author AIAA 2023 conference paper
- Testing standard passive flow controls (no improvement in aerodynamic efficiency)

Sections

6 Element of novelty

- ▶ Numerical Methods
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Elements of novelty

6 Element of novelty

- Systematic usage of Julia in fluid-dynamics
- Usage of VMS for high Reynolds airfoil
- First LES code in Julia - fully parallelized - working with 3D airfoils up to Re 500 000
- Synthetic Eddy Method coded in Julia and coupled with the VMS

It has been challenging, but it seems we are on the right path!

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7 Next steps

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Next steps

7 Next steps

Expected papers:

- VMS paper (prof. Janssens reading)
- Experimental validation of the LS-VMS - publish paper
- LC-VMS - (publish paper?)

Expected conferences:

- AIAA2024 conferences (LS-VMS, LC-VMS)
- DLES14 (VMS usage, validation test cases)
- ICAS2024 (Passive Flow Controls)

Expected research:

- Uncertainty Quantification using Polynomials Chaos Transformation
- Start coding the adjoint optimization
- Test a passive flow control with the VMS