

Airfoil Tools

Search 1638 airfoils



Tweet

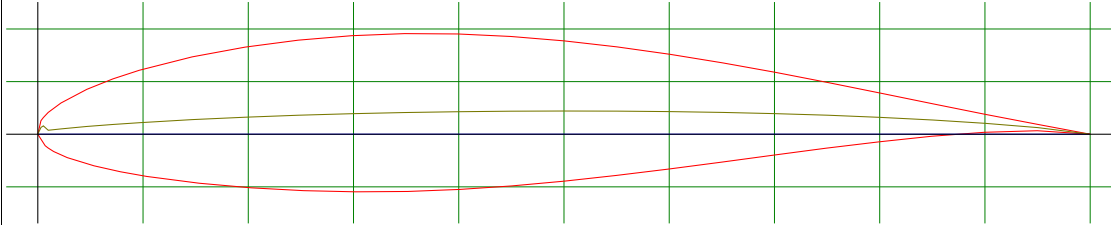
Like 1.3K

You have 0 airfoils loaded.

Your Reynold number range is 50,000 to 1,000,000. ([set](#)) Search

NACA 63-415 AIRFOIL (n63415-il)

NACA 63-415 AIRFOIL - NACA 63(2)-415 airfoil



Details

(n63415-il) NACA 63-415 AIRFOIL
NACA 63(2)-415 airfoil
Max thickness 15% at 34.9% chord.
Max camber 2.2% at 50% chord
Source [UIUC Airfoil Coordinates Database](#)
[Source dat file](#)
The dat file is in Lednicer format

Dat file

```
NACA 63-415 AIRFOIL
26.      26.
0.000000 0.000000
0.003000 0.012870
0.005250 0.015850
0.009910 0.020740
0.021980 0.029640
0.022500 0.029640
```

Parser

No parser warnings

[Send to airfoil plotter](#)
[Add to comparison](#)
[Lednicer format dat file](#)
[Selig format dat file](#)

Similar airfoils

NACA 64(2)-415	Preview	Details
NACA 65(2)-415	Preview	Details
HQ 3.0/15 AIRFOIL	Preview	Details
EPPLER 715 AIRFOIL	Preview	Details
MRC-20	Preview	Details
NACA 63-215 AIRFOIL	Preview	Details
NACA 64-215 AIRFOIL	Preview	Details
NACA 64(2)-215	Preview	Details
NACA 63(2)-615	Preview	Details
NACA 65(2)-415 a=0.5	Preview	Details

Polars for NACA 63-415 AIRFOIL (n63415-il)

Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source	
<input checked="" type="checkbox"/>	n63415-il	50,000	9	30 at $\alpha=9.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	n63415-il	50,000	5	27.8 at $\alpha=8.5^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	n63415-il	100,000	9	53.1 at $\alpha=8.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	n63415-il	100,000	5	52.3 at $\alpha=7.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	n63415-il	200,000	9	75.6 at $\alpha=7.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	n63415-il	200,000	5	70 at $\alpha=6^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	n63415-il	500,000	9	100.9 at $\alpha=6^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	n63415-il	500,000	5	90.7 at $\alpha=4.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details
<input checked="" type="checkbox"/>	n63415-il	1,000,000	9	118.7 at $\alpha=5.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction	Details
<input type="checkbox"/>	n63415-il	1,000,000	5	103 at $\alpha=4.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction	Details

Update plots

[Reynolds number calculator](#)

Set Reynolds number and Ncrit range

Update Range

Reynolds Number

Low

50,000

High

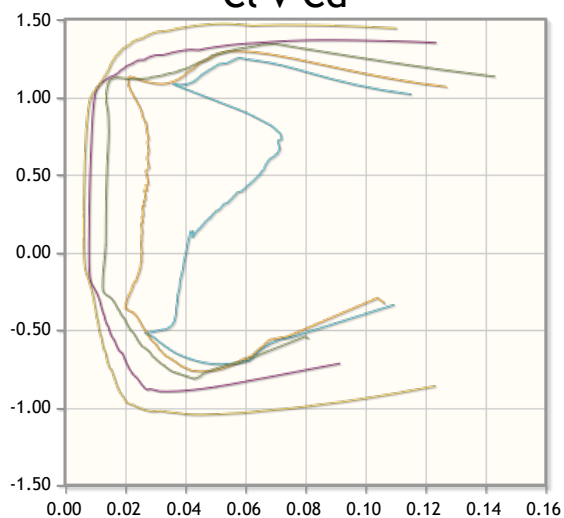
1,000,000

NCrit

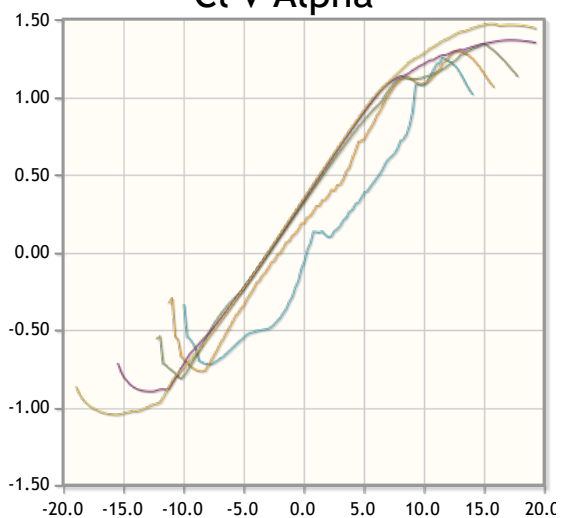
7

9

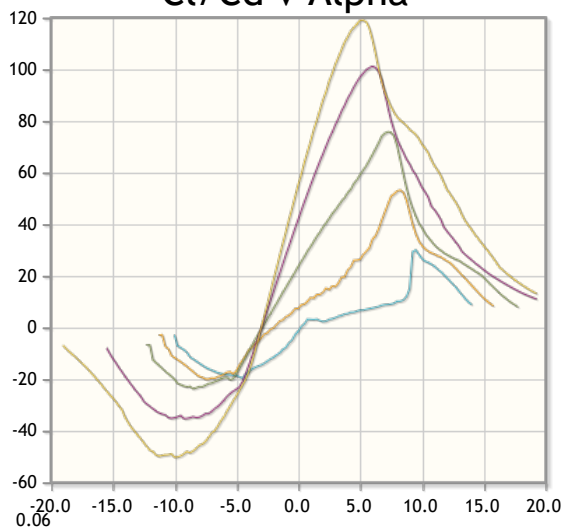
Cl v Cd



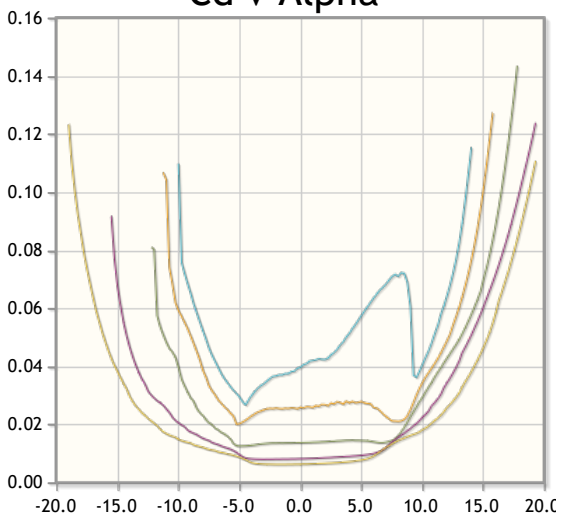
Cl v Alpha



Cl/Cd v Alpha



Cd v Alpha



Cm v Alpha

